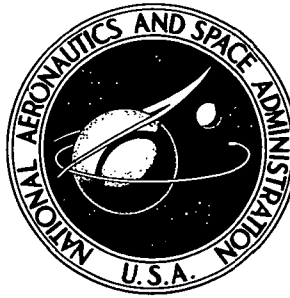


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**TABULATED PRESSURE MEASUREMENTS ON
A LARGE SUBSONIC TRANSPORT MODEL
AIRPLANE WITH HIGH-BYPASS-RATIO,
POWERED, FAN-JET ENGINES**

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Langley Research Center

Hampton, Va. 23365

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16. Abstract <p>An experimental wind-tunnel investigation to determine the aerodynamic interference and the jet-wake interference associated with the wing, pylon, and high-bypass-ratio, powered, fan-jet model engines has been conducted on a typical high-wing logistics transport airplane configuration. Pressures were measured on the wing and pylons and on the surfaces of the engine fan cowl, turbine cowl, and plug. Combinations of wing, pylons, engines, and flow-through nacelles were tested, and the pressure coefficients are presented in tabular form. Tests were conducted at Mach numbers from 0.700 to 0.825 and angles of attack from -2° to 4°. NASA TN D-6067 contains an analysis of the interference drag data for the configurations presented herein.</p>			
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TABULATED PRESSURE MEASUREMENTS ON A LARGE SUBSONIC
TRANSPORT MODEL AIRPLANE WITH HIGH-BYPASS-RATIO,
POWERED, FAN-JET ENGINES

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SUMMARY

An experimental wind-tunnel investigation has been conducted by the National Aeronautics and Space Administration to determine the aerodynamic interference associated with the wing, pylon, and high-bypass-ratio fan-jet engines installed on a typical high-wing logistics transport airplane configuration and the interference due to the jet wake produced by powered model fan-jet engines. These tests were conducted in the Langley 8-foot transonic pressure tunnel at Mach numbers from 0.700 to 0.825 and angles of attack from -2° to 4° . Pressures were measured on the wing and pylons and on the surfaces of the engine fan cowl, turbine cowl, and plug. Combinations of wing, pylons, engines, and flow-through nacelles were tested, and the pressure coefficients are presented in tabular form.

INTRODUCTION

The present investigation is a continuation of a wing-pylon-engine aerodynamic interference investigation being conducted by the Langley Research Center 8-Foot Tunnels Branch of the National Aeronautics and Space Administration. The present test has been performed to determine the jet-wake effect of a high-bypass-ratio engine on the wing-nacelle interference drag of a large, four-engine, subsonic logistics transport airplane of the C-5A type.

A previous investigation (ref. 1) conducted on a two-engine, large, subsonic transport airplane configuration has shown that the jet-wake effects on interference drag of an underwing, pylon-mounted engine may be favorable if the engine is properly located with respect to the wing. The present investigation was conducted on a semispan model of a four-engine, heavy, logistics-type airplane. This wind-tunnel investigation was performed to determine the interference effects associated with the variation in longitudinal and vertical positioning of the engine nacelles, the engine-nacelle incidence, the pylon leading-edge sweep, and the pylon thickness.

A more detailed model description and an analysis of the interference drag data are presented in reference 2. Contained herein are the wing, pylon, and engine surface pressures in coefficient form measured at Mach numbers from 0.700 to 0.825, angles of attack from -2° to 4° , a free-stream stagnation pressure of 96 kN/m^2 , and a free-stream stagnation temperature of 311° K .

SYMBOLS

C_p	pressure coefficient, $\frac{p_l - p_{\infty}}{q_{\infty}}$
c	local chord, meters
\bar{c}	mean geometric wing chord, 53.741 centimeters
i	incidence angle between engine or nacelle and model reference, degrees
M	free-stream Mach number
p_l	local static pressure, newtons/meter ²
p_{∞}	free-stream static pressure, newtons/meter ²
$p_{t,e}/p_{t,\infty}$	fan-exit total-pressure ratio
q_{∞}	free-stream dynamic pressure, newtons/meter ²
x	longitudinal distance, meters
α	angle of attack, degrees

APPARATUS AND EXPERIMENTAL METHODS

Test Facility

The investigation was conducted in the Langley 8-foot transonic pressure tunnel. This facility has a single-return, rectangular test section with a slotted floor and ceiling and is capable of continuous operation through the transonic speed range with negligible effects of choking and blockage. (See ref. 3.) The model has a ratio of wing semispan to tunnel width of 0.82 and a ratio of frontal area to tunnel test-section area of 0.049.

Model Configurations

A drawing of the model is shown in figure 1, and a photograph of the model installed in the wind tunnel is shown in figure 2. Coordinates for all the model components are presented in table 1. Investigations were made on combinations of fuselage, wing, pylons, and engines as shown in table 2.

Fuselage.- The fuselage is similar in shape to the C-5A transport airplane.

Wing.- The model wing has a quarter-chord sweep of $24^{\circ}05'$ inboard of the 46-percent-semispan station (trailing-edge break) and 25° outboard. The wing has an aspect ratio of 7.75, a taper ratio of 0.372, a quarter-chord anhedral of 3.5° , and a twist distribution of 0° at the root, -1.5° at the trailing-edge break station, and -3.5° at the tip.

Engine.- The two-stage model fan is connected directly to a three-stage turbine driven by compressed nitrogen gas. (The gas driving system is explained in refs. 1 and 2.) The model engine is designed to simulate the output of an actual engine having a bypass ratio of 8 and a fan-exit total-pressure ratio of 1.5.

Supplementing the model engine configurations are two flow-through nacelle configurations. One simulates the powered engine by having a short forward fan cowl. The other simulates a long fan cowl engine.

Pylon.- The model engine pylons are made with a hollow spar within an aluminum shell of appropriate shape. The short-cowl flow-through nacelle uses the model engine pylons, and the long-cowl flow-through nacelle uses constant-chord solid aluminum pylons.

A specially designed elongated pylon was used to obtain data for the model engine alone. Figure 3 shows the model engine on this elongated pylon installed in the wind tunnel.

Figure 4 shows the engine- pylon-wing arrangements in each configuration.

Surface Pressure Measurements

Figure 5 shows the location of the instrumentation rows (A to L). Table 3 presents pertinent model geometry data.

Wing.- Pressures were measured along the wing chord directly above each engine-pylon juncture (row A) and on the lower surface 3.81 cm on each side of the pylon center line (rows B and C). In addition, for six configurations (denoted by footnote c in table 2), pressures were measured on both surfaces along the wing chord approximately midway between the two engine stations (center station, rows A and B).

Pylon.- Pressures were measured along the first 15 percent of the local pylon chord at a station approximately one-eighth the distance from the wing to the engine (rows D and F) and along the whole chord at a station approximately midway between the wing and the engine (rows E and G). Pressures were measured on both sides of the pylons.

Engine.- Longitudinal rows of pressure orifices were located at the 30° , 90° , 180° , 270° , and 330° (rows H, I, J, K, and L, respectively) circumferential positions on both the fan and turbine cowls and at the 90° and 270° positions on the plug.

It should be noted that no attempt was made to remove the data obtained from any leaking or plugged orifice. The data from those few orifices that may have been plugged or leaking are generally quite evident when the data are plotted.

Test Conditions

The engines were usually operated at maximum design thrust (45 000 rpm), with a nominal fan-exit total-pressure ratio of 1.5. Additional data were taken at nominal fan-exit total-pressure ratios of 1.3 and 1.0 and with the fan windmilling. These additional data were obtained only for configurations 1 and 12.

The stagnation pressure and temperature in the tunnel were maintained at a nearly constant 96 kN/m^2 and 311° K , respectively.

Boundary-layer transition strips were installed on all configurations. The strips were approximately 3 mm wide and consisted of No. 120 carborundum grains set in a plastic adhesive. They were installed at 10 percent of the local chord on both surfaces of the wing and pylons and approximately 13 mm from the inlet leading edge (inside and outside) of the fan cowl and the flow-through nacelles.

The investigation was conducted at Mach numbers from 0.700 to 0.825 and angles of attack from -2° to 4° .

SUMMARY OF DATA PRESENTED

The surface pressure coefficients are presented in the tables as indicated:

Configuration	Table number for C_p measured on -		
	Wing	Pylon	Engine
1	4	19	31
1a	5	20	32
2	6	21	33
3	7	22	34
4	8	23	35
5	9	24	36
5a	10	25	37
5b	11	26	38
6	12	27	39
7	13	28	40
8	14	29	41
9	15	30	
10	16		
10a	17		
11	18		
12			42
13			43

Langley Research Center,
National Aeronautics and Space Administration,
Hampton, Va., February 15, 1972.

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3. Wright, Ray H.; Ritchie, Virgil S.; and Pearson, Albin O.: Characteristics of the Langley 8-Foot Transonic Tunnel With Slotted Test Section. NACA Rep. 1389, 1958. (Supersedes NACA RM L51H10 by Wright and Ritchie and RM L51K14 by Ritchie and Pearson.)

TABLE 1.- MODEL COORDINATES

(a) Coordinates for wing

[Stations and ordinates in percent wing mean geometric chord]

0-m station; 0° incidence			0.695-m station			0.82-m station; -1.50° incidence			1.081-m station			1.904-m station; -3.50° incidence		
Upper surface		Station	Lower surface		Station	Upper surface		Station	Lower surface		Station	Upper surface		Station
Ordinate	Station		Ordinate	Station		Ordinate	Station		Ordinate	Station		Ordinate	Station	
0	0	0	0	0	0	0	0	0	0	0	0	0	0	0
-0.03	.83	.40	-0.73	.58	.28	-0.50	.53	.25	-0.46	.49	.24	-0.08	.37	.18
.07	1.20	.66	-1.00	.83	.45	-0.69	.04	.76	.41	.63	.38	-0.04	.53	.29
.21	1.49	.90	-1.19	1.03	.61	-0.82	.13	.92	.56	.75	.51	-0.00	.64	.38
.36	1.75	1.11	-1.35	1.20	.76	-0.93	.22	1.10	.70	.85	.64	.04	.75	.46
.52	1.97	1.32	-1.49	1.35	.90	-1.02	.32	1.23	.83	.93	.76	.09	.83	.53
.69	2.18	1.52	-1.61	1.48	1.04	-1.10	.42	1.36	.95	1.01	.87	.14	.91	.61
1.04	2.54	1.91	-1.82	1.73	1.30	-1.23	.63	1.58	1.19	1.13	1.09	.25	1.06	.75
1.41	2.86	2.27	-2.01	1.94	1.56	-1.35	.85	1.77	1.43	1.23	1.29	.36	1.18	.87
2.38	3.54	3.14	-2.40	2.39	2.16	-1.59	1.43	2.18	1.99	1.44	1.79	.68	1.43	1.18
3.39	4.08	3.98	-2.75	2.24	2.75	-1.79	2.03	2.51	2.53	1.62	2.30	1.01	1.62	1.47
7.40	5.48	7.34	-3.92	4.88	3.71	-2.41	4.43	3.39	4.70	2.14	3.08	2.34	2.11	2.61
11.1	6.35	11.0	-4.88	7.38	4.32	-2.87	6.72	3.96	6.98	2.52	5.96	3.59	2.39	3.85
14.8	7.04	14.7	-5.65	9.88	4.81	-3.22	9.00	4.41	9.26	2.79	7.26	4.83	2.61	5.09
22.2	8.09	22.1	-6.84	14.9	5.54	-3.71	13.6	5.09	13.8	3.16	12.1	7.33	2.95	7.54
29.5	8.86	29.4	-7.73	19.9	6.08	-4.05	18.1	5.58	18.4	3.39	16.1	9.84	3.21	9.98
36.9	9.43	36.8	-8.37	24.9	6.48	-4.28	22.7	5.96	22.9	3.55	20.2	12.4	3.40	12.4
44.2	9.84	44.2	-8.82	29.9	6.77	-4.42	27.3	6.23	27.5	3.64	24.3	14.9	3.53	14.8
51.6	10.1	51.6	-9.08	34.9	6.96	-4.49	31.9	6.40	32.0	3.67	28.4	17.4	3.57	17.3
58.9	10.2	59.0	-9.14	39.9	7.05	-4.48	36.5	6.49	36.6	3.65	32.5	19.9	3.55	17.8
66.2	10.2	66.4	-9.01	44.9	7.05	-4.40	41.1	6.50	41.1	3.58	36.5	22.4	3.47	22.3
73.6	9.98	73.8	-8.69	49.9	6.96	-4.24	45.6	6.42	45.7	3.45	40.6	24.8	3.33	24.7
80.9	9.68	81.2	-8.20	54.9	6.77	-4.01	50.2	6.25	50.2	3.26	44.7	27.3	3.13	27.2
88.3	9.25	88.6	-7.55	59.9	6.49	-3.71	54.8	6.00	54.8	3.03	48.8	29.8	2.89	29.7
95.7	8.72	95.9	-6.74	64.9	6.13	-3.34	59.4	5.66	59.3	2.73	52.8	32.3	2.60	32.2
103.1	8.05	103.2	-5.83	69.9	5.67	-2.90	64.0	5.25	63.9	2.38	56.9	34.8	2.27	34.6
110.5	7.24	110.6	-4.84	74.9	5.10	-2.43	68.6	4.72	68.4	2.01	61.0	37.2	1.93	37.1
117.9	6.27	117.9	-3.79	79.9	4.40	-1.95	73.2	4.06	72.9	1.63	65.1	39.7	1.58	39.6
125.3	5.14	125.2	-2.72	85.0	3.55	-1.47	77.7	3.27	77.5	1.25	69.1	42.1	1.23	42.1
132.7	3.78	132.5	-1.72	89.9	2.56	-0.99	82.3	2.34	82.1	0.87	73.2	44.6	0.87	44.6
140.1	2.15	139.9	-0.85	94.9	1.42	-0.54	86.8	1.29	86.7	0.48	77.2	47.1	0.48	47.1
147.4	.19	147.3	-0.19	99.8	.11	-0.11	91.3	.10	91.3	0.10	81.2	49.6	0.05	49.6

TABLE 1.- MODEL COORDINATES - Continued

(b) Coordinates for pylons^a

[Stations and ordinates in percent wing mean geometric chord]

Station							Ordinate
At wing	At engine						
All pylons (b)	Pylon 1	Pylon 2	Pylon 3	Pylon 4	Pylon 5	Pylon 6	
0	0	0	0	0	0	0	0
.10	.10	.10	.10	.10	.10	.10	.40
.21	.21	.22	.21	.21	.21	.21	.57
.31	.31	.36	.31	.31	.31	.31	.71
.41	.41	.52	.41	.41	.41	.41	.85
.52	.52	.62	.52	.52	.52	.52	.93
.62	.62	.80	.62	.62	.62	.62	1.03
.83	.83	1.08	.83	.83	.83	.83	1.18
1.04	1.04	1.36	1.04	1.04	1.04	1.04	1.31
1.56	1.56	2.07	1.56	1.56	1.56	1.56	1.58
2.07	2.07	2.56	2.07	2.07	2.07	2.07	1.81
4.15	4.15	5.90	4.15	4.15	4.15	4.15	2.34
6.22	6.22	8.58	6.22	6.22	6.22	6.22	2.68
8.30	8.30	12.1	8.30	8.30	8.30	8.30	2.93
12.5	12.5	17.8	12.5	12.5	12.5	12.5	3.25
16.6	16.6	22.4	16.6	16.6	16.6	16.6	3.39
20.7	20.7	29.0	20.7	20.7	20.7	20.7	3.51
24.9	24.9	33.2	24.9	24.9	24.9	24.9	3.53
33.2	61.7	61.7	48.6	57.0	52.4	39.5	3.53
37.3	65.8	65.8	52.7	61.1	56.6	43.6	3.51
41.5	70.0	70.0	56.9	65.3	60.7	47.8	3.44
45.6	74.1	74.1	61.0	69.4	64.9	51.9	3.34
49.8	78.3	78.3	65.2	73.6	69.0	56.1	3.18
53.9	82.4	82.4	69.3	77.7	73.2	60.2	2.98
58.1	86.6	86.6	73.4	81.9	77.3	64.4	2.73
62.2	90.7	90.7	77.6	86.0	81.5	68.5	2.43
66.4	94.9	94.9	81.7	90.2	85.6	72.7	2.07
70.5	99.0	99.0	85.9	94.3	89.8	76.8	1.66
74.7	103.2	103.2	90.0	98.5	93.9	81.0	1.19
78.8	107.3	107.3	94.2	102.6	98.1	85.1	.66
83.0	111.5	111.5	98.3	106.8	102.2	89.3	.07

^aSee table 2 to match pylon with configuration.^bAlso at the nacelle for the pylon for the long-duct flow-through nacelle (pylon 7).

TABLE 1.- MODEL COORDINATES - Continued

(c) Coordinates for fan-jet engine

[Stations and radii in percent wing mean geometric chord]

Fan cowl		Turbine cowl		Plug	
Station	Radius	Station	Radius	Station	Radius
0	24.2	34.3	18.2	70.8	8.18
.33	24.9	38.7	17.9	73.5	7.75
1.37	25.6	44.2	17.3	76.8	6.85
3.07	26.3	49.3	16.7	79.6	5.77
5.48	27.0	53.7	16.0	82.1	4.49
10.6	27.7	58.1	15.4	85.2	2.60
17.2	27.7	61.7	14.7	86.5	0
26.1	27.4	65.3	13.9		
30.2	26.8	69.0	13.9		
32.6	26.2	70.8	12.7		
34.3	25.9				

(d) Coordinates for short-duct flow-through nacelle

[Stations and radii in percent wing mean geometric chord]

Station	Inside contour	Outside contour	Station	Inside contour	Outside contour
	Radius	Radius		Radius	Radius
Fan cowl			Turbine cowl		
0	12.1	12.1	28.9	7.09	7.09
.12	11.7	12.2	29.0	6.67	7.47
.24	11.6	12.4	29.2	6.57	7.64
.47	11.5	12.5	29.4	6.43	7.85
.95	11.4	12.7	29.9	6.26	8.11
1.42	11.3	12.9	30.3		8.42
1.89	11.2	13.0	30.8		8.63
2.36	11.2	13.0	31.3		8.75
4.25	11.2	13.4	32.7		9.03
6.14	11.4	13.6	34.1		9.12
8.03	11.5	13.8	36.5		9.01
9.93	11.6	13.9	41.2		8.84
11.8	11.7	13.9	45.9		8.49
16.5	12.0	13.9	50.7		8.23
21.3	12.3	13.9	55.4		7.85
26.0	12.6	13.7	60.1		7.47
28.4	12.7	13.6	64.9		7.02
30.7	12.9	13.4	69.6		6.45
31.7	13.0	13.3	70.0		6.38
32.6	13.0	13.2	70.8	6.26	6.35
33.6	12.9	13.0			
34.3	12.8	12.9			

TABLE 1.- MODEL COORDINATES - Concluded

(e) Coordinates for long-duct flow-through nacelle

[Stations and radii in percent wing mean geometric chord]

Inside contour		Outside contour	
Station	Radius	Station	Radius
0	12.5	0	12.5
.24	12.4	.24	12.8
.47	12.3	.47	13.0
.71	12.2	.71	13.0
1.42	11.9	1.42	13.2
2.13	11.7	2.13	13.3
2.84	11.6	2.84	13.4
3.09	11.5	3.09	13.5
62.3	11.5	3.78	13.5
		4.73	13.7
		5.67	13.7
		6.62	13.7
		7.56	13.8
		8.34	13.8
		20.7	13.8
		20.8	13.7
		22.7	13.7
		24.6	13.7
		26.5	13.5
		28.4	13.4
		30.3	13.3
		31.2	13.1
		32.4	12.9
		34.0	13.0
		35.5	12.9
		40.2	12.7
		44.9	12.7
		49.6	12.5
		53.9	12.2
		58.1	12.0
		62.3	11.6

TABLE 2.- CONFIGURATION DESCRIPTION

[Components marked with X indicate that component is part of the configuration]

Configuration	Fuselage (a)	Wing	Pylon	Engine		Flow-through nacelle ^a				Engine or nacelle incidence angle, deg	Notation of figure 5 in reference 2
				Inboard station	Outboard station	Inboard station	Outboard station	Inboard station	Outboard station		
b1	X	X	1	X	X					0	Pylon 1
1a	X	X	1	X	X					2	
2	X	X	2	X	X					0	Pylon 2
3	X	X	3	X	X					0	Pylon 3
4	X	X	4	X	X					0	Pylon 4
5	X	X	5	X	X					0	Pylon 5
5a	X	X	5	X	X					-3	
5b	X	X	5	X	X					2	
6	X	X	6	X	X					0	Pylon 6
7	X	CX	1	X						0	
8	X	CX	1		X					0	
9	X	CX	1			X	X			0	Short duct
10	X	CX	a7					X	X	0	Long duct
10a	X	CX	a7					X	X	2	
11	X	CX	None								
b12			Special ^a	X						0	
13			Special ^a		X					0	

^aComponent not instrumented to measure surface pressures.^bAdditional data obtained at $P_{t,e}/P_{t,\infty} = 1.3$ and 1.0 and with the fan windmilling.^cPressures at wing center station were also measured.

TABLE 3.- MODEL GEOMETRY DATA

Row (fig. 5)	Configuration	Engine station (fig. 5)		
		Inboard	Center	Outboard
Spanwise station, cm				
A	1 to 11	69.48	85.69	108.14
B	1 to 11	65.67	85.69	104.33
C	1 to 11	73.29		111.95
Wing chord, cm				
A	1 to 11	53.62	48.29	43.65
B	1 to 11	55.04	48.29	44.43
C	1 to 11	52.24		42.86
Distance below pylon-wing juncture, cm				
D, F	1 to 9	1.40		1.30
E, G	1 to 3, 5	4.86		4.32
	6 to 9	4.86		4.32
	4	4.06		3.53
	5a	3.94		3.40
	5b	3.66		3.12
Pylon chord, cm				
D, F ^a	1, 1a, 2, 7, 8, 9	46.4		46.3
	3	45.5		45.5
	4	46.3		46.2
	5, 5a, 5b	45.7		45.7
	6	45.0		45.0
E, G	1, 1a, 2, 7, 8, 9	50.82		50.13
	3	47.91		47.54
	4	49.61		48.96
	5	48.80		48.33
	5a	48.00		47.54
	5b	47.76		47.29
	6	45.97		45.82

^aApproximate.

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(c) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.280	.236	-.160	-.616	.564	.113	-.746	.988	.254	-.884	.770	.362
.025	-.521	.022	-.286	-.823	.375	-.047	-.976	.481	.083	-.1758	.583	.179
.050	-.034	-.336	-.498	-.365	.605	-.197	-.796	.173	-.088	-.677	.319	.028
.075	-.622			-.856			-.1065			-.1159		-.1359
.100	-.538	-.902	-.442	-.884	-.449	-.315	-.1065	-.320	-.137	-.1203	-.067	-.028
.150	-.366	-.907	-.528	-.834	-.449	-.270	-.1065	-.292	-.132	-.1203	-.184	-.067
.200	-.694	-.504	-.459	-.717	-.382	-.264	-.1065	-.275	-.158	-.1203	-.184	-.072
.250	-.650	-.459	-.358	-.722	-.326	-.214	-.740	-.242	-.130	-.1181	-.173	-.061
.300	-.655	-.398	-.319	-.717	-.276	-.186	-.763	-.202	-.135	-.1013	-.151	-.061
.350	-.616		-.772	-.772			-.825			-.834		-.1427
.400	-.622	-.314	-.291	-.800	-.214	-.192	-.841	-.152	-.141	-.868	-.112	-.084
.450												
.500	-.622	-.274	-.246	-.722	-.181	-.175	-.836	-.135	-.124	-.912	-.089	-.084
.550												
.600	-.554	-.150	-.162	-.521	-.115	-.108	-.499	-.085	-.074	-.481	-.056	-.039
.650												
.700	-.442	-.112	-.095	-.437	-.069	-.058	-.421	-.040	-.040	-.396	-.016	.000
.750												
.800	-.263	-.017	-.011	-.236	.009	.009	-.219	.022	.027	-.201	.045	.051
.900	-.011	.073	.073	.009	.087	.087	.022	.305	.100	.034	.107	.124
.975	.140			.121			.128			.124		.114
1.000	.157	.168	.162	.149	.154	.149	.145	.151	.139	.151	.146	.151
Outboard station												
.010	-.283	-.235	-.235	-.526	.262	.100	-.650	.461	.251	-.756	.604	.340
.025	-.576	-.351	-.458	-.846	.026	-.138	-.783	.156	.021	-.1054	.382	.067
.050	-.491	-.503	-.475	-.908	-.149	-.200	-.124	.021	-.087	-.1206	.230	.022
.075	-.520			-.700			-.1062			-.1178		-.046
.100	-.576	-.689	-.497	-.742	-.539	-.234	-.1028	-.301	-.132	-.1144	.125	-.046
.150	-.478	-.801	-.407	-.958	-.458	-.200	-.1073	-.290	.038	-.1218	-.184	-.024
.200	-.525	-.813	-.328	-.801	-.356	-.177	-.1063	-.244	.078	-.1161	-.184	-.040
.250	-.598	-.860	-.322	-.784	-.256	-.172	-.1028	-.182	.042	-.1212	-.125	-.035
.300	-.598	-.491	-.283	-.806	-.428	-.149	-.1039	-.171	-.154	-.1201	-.125	-.040
.350	-.649			-.823			-.1034			-.1184		-.0826
.400	-.655	-.221	-.238	-.846	-.172	-.144	-.938	-.120	-.098	-.1212	-.074	-.046
.450												
.500	-.615	-.153	-.204	-.660	-.104	-.132	-.741	-.075	-.098	-.790	-.035	-.057
.550												
.600	-.525	-.103	-.103	-.543	-.059	-.071	-.594	-.041	-.047	-.474	-.112	-.018
.650												
.700	-.390	-.035	-.013	-.374	-.020	-.003	-.363	.021	.009	-.322	.038	.044
.750												
.800	-.232	.027	.049	-.183	.053	.070	-.160	.054	.077	-.159	.095	.117
.900	.016	.066	.016	.030	.126	.126	.032	.150	.133	.033	.157	.162
.975	.128			.126			.128			.145		.157
1.000	.151	.094	.140	.137	.070	.149	.130	.086	.150	.162	.130	.157
.493												.765
.175												.599
.136												.435
.068												.102
.088												.017
.028												.023
.023												.023
.011												.040
.023												.034
.023												.017
.057												.017
.057												.017
.034												.006
.000												.006
.000												.023
.000												.198
.170												.175

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(d) $M = 0.800$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.016	-.262	.301	-.177	-.542	.574	.116	-.650	.675	.252	-.797	.772	.326
.025	-.499	.072	-.294	-.768	.327	-.070	-.877	.475	.066	-.969	.563	.166
.050	.116	-.283	-.482	-.204	.016	-.220	-.677	.179	-.085	-.536	.385	-.001
.075	.552	-.822	-.482	-.827	-.456	.231	-.984	-.236	.117	-.1066	.130	-.055
.100	-.482	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.150	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.200	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.250	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.300	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.350	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.400	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.450	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.500	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.550	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.600	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.650	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.700	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.750	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.800	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.850	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.900	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
.950	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
1.000	-.531	-.843	-.531	-.848	-.489	.285	-.011	-.327	.193	-.1.119	-.222	-.098
Outboard station												
.016	-.216	-.150	-.229	-.466	.210	.060	-.569	.421	.193	-.641	.553	.300
.025	-.536	-.314	-.465	-.774	.021	-.148	-.867	.162	-.644	-.849	.316	.067
.050	-.465	-.449	-.476	-.904	-.153	-.202	-.1.024	.032	-.092	-.1.106	.131	-.019
.075	-.498	-.498	-.704	-.704	-.558	-.266	-.932	-.304	.141	-.1.074	.198	-.095
.100	-.574	-.725	-.563	-.709	-.521	-.294	-.1.019	-.331	.119	-.1.117	.241	-.073
.150	-.563	-.747	-.633	-.893	-.521	-.294	-.1.019	-.331	.119	-.1.117	.241	-.073
.200	-.530	-.785	-.657	-.844	-.565	-.385	-.1.191	-.497	.109	-.1.168	.259	-.171
.250	-.639	-.720	-.541	-.860	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.300	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.350	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.400	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.450	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.500	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.550	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.600	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.650	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.700	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.750	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.800	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.850	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.900	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
.950	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073
1.000	-.644	-.644	-.303	-.882	-.626	-.486	-.1.303	-.603	.098	-.1.117	.134	-.073

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(e) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.5$

X/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.254	.333	-.179	-.500	.572	.090	-.602	.670	.197	-.710	.752	.292
.025	-.457	.129	-.301	-.692	.343	-.066	-.794	.471	.051	-.871	.019	-.892
.050	-.280	-.218	-.457	-.643	.033	-.210	-.571	.186	-.141	-.404	.260	-.519
.075	-.540	-.763	-.457	-.764	-.454	-.242	-.893	-.270	-.177	-.585	.036	-.524
.100	-.612	-.854	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.150	-.540	-.854	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.200	-.566	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.250	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.300	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.350	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.400	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.450	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.500	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.550	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.600	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.650	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.700	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.750	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.800	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.850	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.900	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
.950	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
1.000	-.571	-.857	-.508	-.790	-.521	-.335	-.918	-.415	-.192	-.688	-.145	-.688
Outboard station												
.010	-.195	-.151	-.276	-.356	.199	.012	-.498	.398	.160	-.575	.499	.236
.025	-.508	-.279	-.451	-.658	.026	-.188	-.790	.144	-.070	-.857	.259	-.002
.050	-.451	-.414	-.497	-.745	-.156	-.255	-.926	.098	-.138	-.803	.087	-.096
.075	-.477	-.477	-.576	-.620	-.557	-.281	-.863	-.415	-.165	-.687	-.373	-.148
.100	-.503	-.727	-.576	-.656	-.797	-.245	-.842	-.474	-.148	-.651	-.325	-.122
.150	-.639	-.665	-.586	-.833	-.625	-.214	-.874	-.331	-.148	-.587	-.283	-.127
.200	-.508	-.717	-.425	-.797	-.425	-.208	-.831	-.237	-.133	-.534	-.226	-.137
.250	-.592	-.766	-.420	-.844	-.292	-.188	-.957	-.221	-.133	-.504	-.215	-.132
.300	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.350	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.400	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.450	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.500	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.550	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.600	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.650	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.700	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.750	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.800	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.850	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.900	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
.950	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132
1.000	-.639	-.690	-.373	-.854	-.250	-.188	-.973	-.221	-.133	-.504	-.215	-.132

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(f) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.3$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.315	.154	-.204	-.704	.525	.144	-.954	.683	.244	-1.144	.755	.389
.025	-.593	-.072	-.277	-.909	.315	.014	-1.146	.491	.142	-1.337	.226	.576
.050	-.647	-.498	-.417	-.998	.027	-.135	-1.018	.199	-.031	-1.177	.591	.406
.075	-.564	-.853	-.411	-.857	-.313	-.172	-.954	-.150	-.083	-1.394	.335	.227
.100	-.532	-.533	-.424	-.736	.319	-.204	-.896	-.224	-.115	-1.106	.037	.157
.150	-.513	-.456	-.379	-.716	.281	-.213	-.781	-.204	-.121	-.908	-.133	.099
.200	-.519	-.392	-.328	-.652	.242	-.165	-.826	-.185	-.102	-.888	-.043	.080
.250	-.526	-.341	-.283	-.614	.217	-.145	-.762	-.172	-.095	-.805	-.043	.080
.300	-.513	-.264	-.228	-.627	.159	-.143	-.743	-.127	-.095	-.758	-.126	.081
.350	-.533	-.213	-.194	-.614	.127	-.121	-.615	-.102	-.095	-.792	-.094	.023
.400	-.532	-.149	-.120	-.575	.082	-.069	-.486	-.053	-.057	-.780	-.062	.010
.450	-.526	-.091	-.072	-.460	.037	-.031	-.429	-.025	-.019	-.517	-.037	.029
.500	-.507	-.008	-.002	-.325	.023	.043	-.256	.033	.014	-.427	-.094	.042
.550	-.411	-.075	.075	-.212	.104	.104	-.206	.067	.110	-.252	.053	.087
.600	-.277	.177	.173	-.142	.174	.168	-.129	.161	.154	-.131	.098	.164
.650	-.043	.138	.133	-.068	.174	.168	.129	.161	.154	-.003	.119	.087
.700	.138	.177	.173	.168	.174	.168	.129	.161	.154	.131	.157	.157
.750	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
.800	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
.850	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
.900	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
.950	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
1.000	.173	.177	.173	.168	.174	.168	.129	.161	.154	.144	.157	.157
Outboard station												
.010	-.292	-.323	-.277	-.882	.246	.137	-.870	.509	.323	-1.029	.633	.416
.025	-.806	-.433	-.401	-.898	.014	-.057	-1.212	.213	.007	-1.351	.401	.143
.050	-.523	-.581	-.414	-.782	-.128	-.141	-1.237	.091	-.025	-1.532	.207	.065
.075	-.529	-.690	-.427	-.805	.379	-.186	-.844	-.219	-.116	-1.454	.063	.031
.100	-.561	-.774	-.355	-.856	.315	-.154	-.773	-.219	-.083	-1.332	-.121	-.031
.150	-.478	-.569	-.285	-.650	.263	-.135	-.780	-.193	-.077	-.804	-.115	-.024
.200	-.484	-.569	-.285	-.650	.263	-.135	-.780	-.193	-.077	-.804	-.115	-.024
.250	-.542	-.389	-.265	-.582	.235	-.128	-.793	-.143	-.077	-.855	-.089	-.038
.300	-.581	-.304	-.247	-.669	.193	-.122	-.760	-.141	-.083	-.910	-.085	-.037
.350	-.555	-.234	-.221	-.644	.135	-.122	-.722	-.116	-.096	-.778	-.079	-.037
.400	-.561	-.169	-.169	-.579	.089	-.115	-.619	-.058	-.090	-.746	-.070	-.037
.450	-.536	-.113	-.095	-.508	.044	-.051	-.528	-.032	-.045	-.636	-.037	-.050
.500	-.491	-.041	-.015	-.386	.001	-.006	-.419	.013	.007	-.566	-.012	-.024
.550	-.382	-.022	-.015	-.219	.032	.032	-.212	.065	.071	-.398	.014	.014
.600	-.234	-.041	-.015	-.089	.113	.359	-.037	.116	.097	-.205	.072	.014
.650	-.094	-.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.700	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.750	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.800	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.850	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.900	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
.950	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117
1.000	-.100	.017	.123	.031	.078	.130	-.025	.104	.110	-.083	.104	.117

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(g) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.3$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	-.303	-.187	-.186	-.715	-.533	-.293	-.815	-.664	-.257	-.966	-.762	-.349
.025	-.484	-.000	-.287	-.825	-.307	-.214	-.1748	-.435	-.027	-.158	-.582	-.191
.050	-.531	-.367	-.432	-.921	-.004	-.248	-.782	-.252	-.055	-.913	-.337	-.087
.075	-.653			-.878			-.130			-.1269		
.100	-.665	-.843	-.425	-.884	-.382	-.232	-.1107	-.189	-.096	-.1287	-.655	-.014
.150	-.571	-.717	-.427	-.756	-.411	-.259	-.891	-.250	-.143	-.1264	-.166	-.072
.200	-.577	-.484	-.455	-.773	-.417	-.242	-.774	-.236	-.131	-.1246	-.168	-.071
.250	-.676	-.414	-.355	-.738	-.446	-.213	-.923	-.207	-.131	-.1246	-.168	-.071
.300	-.612	-.362	-.367	-.773	-.259	-.195	-.785	-.263	-.137	-.130	-.132	-.063
.350	-.641			-.744			-.793			-.866		
.400	-.671	-.285	-.257	-.756	-.207	-.183	-.733	-.137	-.113	-.861	-.107	-.072
.450												
.500	-.641	-.227	-.216	-.592	-.172	-.163	-.692	-.113	-.108	-.720	-.090	-.072
.550	-.461	-.153	-.134	-.516	-.113	-.132	-.545	-.073	-.067	-.574	-.049	-.031
.600												
.650	-.426	-.087	-.076	-.446	-.061	-.055	-.429	-.032	-.032	-.423	-.008	-.008
.700												
.750	-.262	-.006	-.006	-.259	-.004	-.015	-.233	-.027	-.038	-.230	-.051	-.051
.800	-.018	-.082	-.070	-.070	-.080	-.085	-.069	-.051	-.103	-.016	-.103	-.121
.850												
.900	-.146			-.123			-.126			-.127		
.950												
1.000	-.159	-.181	-.175	-.138	-.156	-.144	-.138	-.138	-.144	-.144	-.156	-.153

Outboard station												
.010	-.259	-.268	-.265	-.591	-.266	-.133	-.744	-.456	-.259	-.837	-.612	-.379
.025	-.399	-.359	-.441	-.826	-.044	-.127	-.1073	-.228	-.008	-.143	-.408	-.123
.050	-.500	-.506	-.447	-.822	-.156	-.158	-.1226	-.061	-.062	-.1325	-.167	-.009
.075	-.523			-.767			-.138			-.1290		
.100	-.582	-.593	-.459	-.802	-.456	-.227	-.1044	-.239	-.127	-.1331	-.095	-.056
.150	-.482	-.787	-.382	-.996	-.415	-.137	-.095	-.239	-.086	-.150	-.038	-.038
.200	-.523	-.752	-.303	-.873	-.321	-.168	-.0932	-.227	-.086	-.1213	-.144	-.044
.250	-.576	-.523	-.283	-.750	-.238	-.162	-.768	-.162	-.086	-.1243	-.103	-.044
.300	-.562	-.388	-.271	-.750	-.215	-.162	-.768	-.150	-.092	-.1208	-.103	-.044
.350	-.593			-.761			-.803			-.984		
.400	-.599	-.206	-.224	-.755	-.156	-.150	-.815	-.109	-.098	-.726	-.079	-.053
.450												
.500	-.564	-.148	-.177	-.532	-.109	-.133	-.644	-.069	-.092	-.626	-.050	-.055
.550												
.600	-.506	-.101	-.121	-.544	-.074	-.119	-.538	-.039	-.056	-.532	-.015	-.027
.650												
.700	-.388	-.025	-.025	-.357	-.027	-.003	-.386	-.038	-.008	-.379	-.032	-.032
.750												
.800	-.218	-.023	-.023	-.238	-.039	-.067	-.185	-.067	-.073	-.220	-.085	-.097
.850	-.025	-.091	-.092	-.020	-.044	-.103	-.026	-.114	-.020	-.021	-.126	-.132
.900												
.950	-.132			-.108			-.102			-.100		
1.000	-.134	-.011	-.046	-.126	-.133	-.132	-.125	-.044	-.061	-.135	-.073	-.138

$\alpha = 4^\circ$												
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
.010	-.1177	-.854	-.521	-.1177	-.854	-.521	-.1177	-.854	-.521	-.1177	-.854	-.521
.025	-.1357	-.747	-.374	-.1357	-.747	-.374	-.1357	-.747	-.374	-.1357	-.747	-.374
.050	-.1572	-.554	-.187	-.1572	-.554	-.187	-.1572	-.554	-.187	-.1572	-.554	-.187
.075	-.1468			-.1468			-.1468			-.1468		
.100	-.1491	-.187	-.135	-.1491	-.187	-.135	-.1491	-.187	-.135	-.1491	-.187	-.135
.150	-.1515	-.030	-.071	-.1515	-.030	-.071	-.1515	-.030	-.071	-.1515	-.030	-.071
.200	-.1515	-.017	-.041	-.1515	-.017	-.041	-.1515	-.017	-.041	-.1515	-.017	-.041
.250	-.1509	-.017	-.047	-.1509	-.017	-.047	-.1509	-.017	-.047	-.1509	-.017	-.047
.300	-.1480	-.023	-.036	-.1480	-.023	-.036	-.1480	-.023	-.036	-.1480	-.023	-.036
.350	-.1515			-.1515			-.1515			-.1515		
.400	-.1037	-.023	-.006	-.1037	-.023	-.006	-.1037	-.023	-.006	-.1037	-.023	-.006
.450												
.500	-.693	-.017	-.011	-.693	-.017	-.011	-.693	-.017	-.011	-.693	-.017	-.011
.550												
.600	-.454	-.006	-.018	-.454	-.006	-.018	-.454	-.006	-.018	-.454	-.006	-.018
.650												
.700	-.396	-.030	-.036	-.396	-.030	-.036	-.396	-.030	-.036	-.396	-.030	-.036
.750												
.800	-.238	-.071	-.088	-.238	-.071	-.088	-.238	-.071	-.088	-.238	-.071	-.088
.850	-.011	-.135	-.152	-.011	-.135	-.152	-.011	-.135	-.152	-.011	-.135	-.152
.900												
.950	-.164			-.164			-.164			-.164		
1.000	-.205	-.199	-.216	-.205	-.199	-.216	-.205	-.199	-.216	-.205	-.199	-.216

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(1) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.3$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Inboard station											
.010	-.239	.283	-.224	-.580	.536	.089	-.672	.663	.203	-.775	.747	.319
.025	-.545	-.084	-.287	-.731	.325	-.047	-.865	.459	.077	-.947	.561	.147
.050	-.336	-.368	-.475	-.343	.002	-.203	-.435	.174	-.390	-.597	.363	-.020
.075	-.540			-.801			-.974			-.1071		
.100	-.481	-.847	-.525	-.850	-.510	-.246	-.990	-.236	-.123	-.1109	-.085	-.053
.150	-.540	-.852	-.524	-.850	-.521	-.295	-.1012	-.333	-.204	-.1125	-.209	-.107
.200	-.551	-.864	-.524	-.823	-.424	-.273	-.993	-.295	-.171	-.1125	-.193	-.112
.250	-.578	-.459	-.373	-.747	-.343	-.252	-.993	-.263	-.171	-.1125	-.203	-.095
.300	-.578	-.406	-.336	-.709	-.295	-.213	-.802	-.214	-.139	-.1114	-.161	-.112
.350	-.642			-.763			-.796			-.1093		
.400	-.675	-.325	-.309	-.790	-.235	-.262	-.845	-.247	-.177	-.888	-.123	-.123
.450							-.893			-.942		
.500	-.691	-.282	-.260	-.855	-.203	-.241	-.893	-.150	-.139	-.942	-.107	-.117
.550				-.769			-.802		-.080	-.678	-.074	-.069
.600	-.551	-.196	-.207		-.139	-.122	-.802	-.056	-.080	-.678	-.056	-.056
.650				-.381			-.360		-.031	-.328		-.023
.700	-.486	-.110	-.099		-.101	-.074	-.360	-.053	-.031	-.328	-.026	-.023
.750				-.203		.007	-.177		.017	-.161	.039	.044
.800	-.233	-.013	-.013	-.018	-.004	.007	-.028	.053	.028	-.028	.098	.109
.900	.014	.079	.068	.115	.077	.082	.120		.130	.130		
.975	.138			.131		.136	.141	.147	.136	.147	.152	.141
1.000	.159	.170	.159		.147	.136	.141	.147	.136	.147	.152	.141
x/c	Outboard station											
	Outboard station											
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Outboard station											
.010	-.209	-.208	-.241	-.477	.207	.031	-.570	.408	.193	-.651	.565	.265
.025	-.518	-.291	-.437	-.764	.027	-.152	-.873	.173	.022	-.933	.227	.065
.050	-.464	-.464	-.464	-.900	-.152	-.222	-.1030	.010	-.109	-.101	.172	-.028
.075	-.507			-.639			-.982			-.1079		
.100	-.561	-.669	-.561	-.710	-.596	-.276	-.911	-.342	-.141	-.1031	-.163	-.082
.150	-.589	-.685	-.453	-.900	-.537	-.233	-.593	-.231	-.120	-.1139	-.201	-.066
.200	-.534	-.751	-.355	-.845	-.379	-.194	-.927	-.277	-.109	-.1068	-.179	-.065
.250	-.648	-.713	-.334	-.872	-.260	-.173	-.992	-.226	-.103	-.1128	-.131	-.066
.300	-.648	-.599	-.301	-.878	-.249	-.173	-.1009	-.179	-.103	-.1123	-.131	-.075
.350	-.653			-.878			-.1025			-.1123		
.400	-.713	-.269	-.247	-.855	-.173	-.157	-.1068	-.131	-.103	-.1121	-.093	-.082
.450												
.500	-.805	-.172	-.152	-.927	-.130	-.152	-.1128	-.076	-.103	-.689	-.060	-.093
.550												
.600	-.507	-.101	-.117	-.558	-.081	-.099	-.532	-.044	-.055	-.548	-.038	-.066
.650				-.314			-.309		.016	-.391	.010	-.011
.700	-.377	-.031	-.058		-.027	-.022		-.011				
.750				-.152	.016	.054	-.136	.055	.081	-.267	.054	.054
.800	-.193	.023	.045	-.038	.119	.114	-.032	.086	.113	-.097	.097	.065
.900	.029	.059	.094	.034			.130			-.006		
.975	.083					.092	.157	.075	.151	.048		
1.000	.137	.094	.142	-.125	.038							

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(j) $M = 0.825$; $p_{te}/p_{t,\infty} = 1.3$

x/c	C_p at $\alpha =$											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.245	.369	-.269	-.476	.546	.068	-.599	.652	.145	-.689	.727	.232
.025	-.458	.108	-.302	-.674	.350	-.081	-.786	.465	.034	-.855	.543	-.886
.050	-.210	-.224	-.453	-.248	.028	-.232	-.225	.130	-.132	-.439	.278	-.1041
.075	-.520			-.767			-.900			-.990		-.527
.100	-.598	-.790	-.411	-.793	-.456	-.268	-.921	-.277	-.194	-.1026	-.148	-.1140
.150	-.520	-.816	-.521	-.788	-.544	-.440	-.947	-.385	-.220	-.1037	-.148	-.1171
.200	-.562	-.857	-.593	-.788	-.461	-.409	-.947	-.334	-.215	-.1052	-.252	-.1208
.250	-.556	-.845	-.427	-.715	-.430	-.243	-.947	-.298	-.179	-.1058	-.226	-.1234
.300	-.556	-.375	-.385	-.747	-.341	-.253	-.942	-.262	-.225	-.1068	-.205	-.1122
.350	-.619			-.715			-.838			-.1084		-.1223
.400	-.650	-.406	-.416	-.767	-.258	-.243	-.884	-.215	-.277	-.1084	-.174	-.1218
.450												-.901
.500	-.717	-.393	-.364	-.825	-.232	-.211	-.893	-.154	-.199	-.943	-.153	-.678
.550												-.106
.600	-.712	-.214	-.178	-.825	-.159	-.175	-.895	-.158	-.179	-.803	-.169	-.636
.650												-.091
.700	-.383	-.157	-.115	-.404	-.392	-.066	-.385	-.090	-.070	-.387	-.073	-.574
.750												-.143
.800	-.183	-.017	-.022	-.159	-.068	.037	-.215	-.007	-.007	-.237	-.098	-.496
.900	.035	.077	.066	.007	.064	.083	.013	.065	.060	-.101	.063	-.428
.975	.134			.090			.063			.002		-.340
1.000	.154	.154	.144	.116	.127	.121	.063	.091	.065	.028	.075	-.257
												-.304
Outboard station												
.010	-.191	-.167	-.274	-.397	.202	.013	-.500	.356	.133	-.559	.485	.221
.025	-.478	-.264	-.431	-.590	.016	-.264	-.793	.168	-.098	-.841	.283	-.961
.050	-.431	-.426	-.494	-.784	-.162	-.251	-.929	-.061	-.171	-.1003	.116	-.1128
.075	-.489			-.627			-.865			-.961		-.1123
.100	-.509	-.671	-.562	-.659	-.622	-.268	-.824	-.422	-.223	-.940	-.276	-.135
.150	-.603	-.645	-.562	-.831	-.795	-.251	-.923	-.464	-.182	-.1024	-.313	-.109
.200	-.473	-.692	-.384	-.795	-.849	-.209	-.871	-.344	-.166	-.977	-.266	-.114
.250	-.588	-.682	-.374	-.821	-.828	-.193	-.939	-.250	-.166	-.1029	-.193	-.103
.300	-.635	-.645	-.327	-.847	-.846	-.204	-.963	-.229	-.166	-.1045	-.188	-.103
.350	-.666			-.873			-.981			-.1060		-.123
.400	-.697	-.400	-.264	-.905	-.178	-.178	-.1023	-.176	-.166	-.726	-.151	-.135
.450												-.605
.500	-.812	-.181	-.212	-.552	-.136	-.167	-.777	-.150	-.166	-.822	-.114	-.167
.550												-.553
.600	-.854	-.092	-.113	-.567	-.369	-.135	-.463	-.129	-.124	-.470	-.099	-.543
.650												-.104
.700	-.306	-.024	-.050	-.272	-.047	-.042	-.375	-.056	-.061	-.376	-.062	-.109
.750												-.454
.800	-.159	.033	.043	-.120	.011	.037	-.276	-.041	-.004	-.323	-.036	-.169
.900	.039	.107	.096	-.047	.068	.079	-.176	.027	.033	-.261	-.010	-.088
.975	.127			.021			-.108			-.224		-.355
1.000	.133	.065	.128	.021	-.052	.063	-.041	-.182	-.046	-.182	-.381	-.203
												-.307
												-.396
												-.088
												-.078
												-.093
												-.057
												-.488
												-.114
												-.135
												-.093
												-.412
												-.386
												-.355
												-.307
												-.281

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(k) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.0$

x/c	C _p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.312	.144	-.215	-.782	.532	.150	-.937	.661	.287	-1.134	.733	.381
.025	-.511	-.069	-.261	-.910	.321	.000	-1.174	.461	.147	-1.345	.565	.193
.050	-.652	-.409	-.409	-.865	.007	-.160	-1.001	.199	-.090	-1.172	.315	.045
.075	-.575	-.665	-.421	-.955	-.320	-.186	-.988	-.141	-.077	-1.409	-.038	-.025
.100	-.375	-.537	-.421	-.827	-.333	-.224	-.892	-.206	-.122	-1.236	-.128	-.072
.150	-.530	-.441	-.377	-.686	-.282	-.218	-.911	-.193	-.129	-.903	-.147	-.070
.200	-.530	-.377	-.325	-.763	-.256	-.186	-.821	-.174	-.116	-.833	-.140	-.091
.250	-.511	-.332	-.287	-.622	-.230	-.173	-.699	-.167	-.097	-.858	-.128	-.025
.300	-.543	-.255	-.229	-.654	-.173	-.141	-.699	-.122	-.090	-.781	-.096	-.063
.350	-.543	-.197	-.184	-.577	-.128	-.115	-.667	-.090	-.071	-.820	-.051	-.012
.400	-.524	-.133	-.114	-.487	-.077	-.064	-.610	-.045	-.032	-.647	-.044	.007
.450	-.441	-.075	-.069	-.423	-.038	-.025	-.481	-.013	-.000	-.518	-.031	.033
.500	-.400	-.005	-.001	-.263	.026	.032	-.417	.045	.051	-.435	.001	.046
.550	-.274	.072	.072	-.319	.090	.103	-.250	.102	.051	-.249	.039	.091
.600	-.142	.181	.168	-.141	.167	.167	-.000	.134	.115	-.001	.097	.110
.650	.168	.181	.168	-.167	.167	.167	.154	.160	.160	.135	.142	.129
.700												
.750												
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010	-.321	-.281	-.242	-.584	.245	.137	-.878	.493	.307	-1.018	.609	.408
.025	-.517	-.418	-.405	-.793	.058	-.110	-1.214	.245	.038	-1.341	.439	.129
.050	-.538	-.527	-.424	-.813	-.116	-.148	-1.304	.090	-.026	-1.541	.239	.033
.075	-.521	-.650	-.431	-.793	-.368	-.206	-.859	-.168	-.097	-1.470	-.045	-.025
.100	-.372	-.727	-.353	-.832	-.329	-.174	-.988	-.181	-.085	-1.360	-.090	-.013
.150	-.489	-.540	-.289	-.558	-.258	-.148	-.769	-.168	-.085	-.838	-.070	-.033
.200	-.495	-.340	-.276	-.551	-.200	-.148	-.762	-.130	-.095	-.793	-.077	-.019
.250	-.547	-.276	-.276	-.684	-.187	-.148	-.781	-.130	-.104	-.851	-.084	-.033
.300	-.540	-.276	-.276	-.654	-.142	-.135	-.743	-.104	-.188	-.799	-.058	-.007
.350	-.553	-.315	-.231	-.658	-.090	-.110	-.730	-.104	-.085	-.773	-.032	-.001
.400	-.553	-.147	-.166	-.645	-.045	-.045	-.717	-.059	-.033	-.741	.000	.027
.450	-.521	-.134	-.089	-.580	-.006	-.013	-.614	-.026	-.033	-.632	.033	.065
.500	-.489	-.050	-.036	-.510	.006	.013	-.530	-.001	-.012	-.535	.039	.085
.550	-.386	-.014	-.018	-.387	.032	.032	-.388	.070	.077	-.387	.084	.130
.600	-.231	.078	.078	-.226	.064	.064	-.267	.090	.122	-.200	.097	.117
.650	-.016	.078	.078	-.103	.064	.064	-.096	.025	.070	-.097	.026	.078
.700	.117	.078	.078	.103	.064	.064	.070	.025	.070	.052	.026	.078
.750	.130	.078	.078	.135	.064	.064	.070	.025	.070	.104	.026	.078
.800												
.850												
.900												
.950												
1.000												

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(1) $M = 0.750$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -														
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$			$\alpha = 4^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station															
.010	-.281	.193	-.200	-.698	.513	.093	-.805	.657	.230	-.955	.729	.342	-1.112	.814	.515
.025	-.491	-.013	-.322	-.345	.330	-.026	-1.056	.459	.113	-1.142	.553	.184	-1.334	.716	.302
.050	-.491	-.404	-.439	-.381	.003	-.190	-.752	.184	.062	-.902	.313	.021	-1.059	.529	.161
.075	-.555			-.886			-1.126			-1.276			-1.457		
.100	-.550	-.766	-.427	-.868	-.400	-.225	-1.126	-.179	.115	-1.288	-.055	-.020	-1.486	.161	.115
.150	-.538	-.731	-.456	-.751	-.400	-.260	-.974	-.343	.161	-1.271	-.149	-.084	-1.509	.027	.080
.200	-.550	-.462	-.615	-.441	-.400	-.266	-.752	-.232	.155	-1.224	-.154	-.079	-1.509	-.026	.056
.250	-.655	-.435	-.421	-.751	-.283	-.207	-.822	-.214	-.138	-.768	-.149	-.067	-1.509	-.049	.056
.300	-.655	-.363	-.310	-.739	-.248	-.196	-.799	-.191	-.132	-.791	-.137	-.061	-1.509	-.049	.062
.350	-.561			-.739			-.799			-.850			-1.515		
.400	-.631	-.281	-.246	-.734	-.190	-.167	-.735	-.144	.109	-.867	-.108	-.055	-.989	-.049	.045
.450															
.500	-.748	-.223	-.154	-.598	-.149	-.131	-.717	-.109	-.091	-.663	-.073	-.049	-.656	-.014	.027
.550				-.594			-.507		-.050	-.499	-.026	-.008	-.464	-.027	.050
.600	-.462	-.147	-.124	-.494	-.091	-.079		-.062							
.650				-.430	-.050	-.038	-.430	-.027	-.021	-.429	.009	.015	-.394	.045	.062
.700	-.427	-.083	-.071												
.750				-.254	.015	.026	-.237	.026	.037	-.225	.050	.062	-.236	.068	.097
.800	-.270	-.007	-.001	-.003	.085	.091	.008	.090	.102	.021	.108	.120	-.002	.138	.144
.900	-.019	.075	.069	.132			.125			.126			.144		
.975	.145			.149			.137	.149	.143	.149	.155	.149	.185	.196	.202
1.000	.168	-.174	.157		.155	.155									
Outboard station															
.010	-.266	-.277	-.277	-.615	.258	.124	-.728	.449	.244	-.832	.598	.372	-.938	.725	.435
.025	-.577	-.366	-.407	-.904	.049	-.110	-1.040	.219	.019	-1.132	.373	.120	-1.243	.573	.197
.050	-.501	-.519	-.448	-.721	-.116	-.186	-1.211	.072	-.058	-1.326	.208	.002	-1.449	.438	.108
.075	-.513			-.762			-1.158			-1.285			-1.437		
.100	-.577	-.624	-.472	-.792	.439	-.233	-1.052	-.210	.116	-1.232	-.086	-.051	-1.431	.138	.073
.150	-.472	-.754	-.378	-.874	-.380	-.198	-1.075	-.228	-.099	-1.315	-.145	-.027	-1.513	.085	.085
.200	-.513	-.695	-.307	-.668	-.286	-.169	-.952	-.210	-.099	-1.220	-.133	-.045	-1.466	.004	.044
.250	-.577	-.460	-.302	-.751	-.216	-.169	-.781	-.152	-.093	-1.256	-.098	-.039	-1.508	.014	.056
.300	-.577	-.319	-.307	-.745	-.198	-.163	-.787	-.146	-.099	-1.220	-.098	-.051	-1.508	-.015	.050
.350	-.583			-.751			-.793			-.874			-1.402	-.021	.026
.400	-.595	-.284	-.225	-.739	-.151	-.157	-.822	-.110	-.099	-.703	-.080	-.051	-.932	-.021	.026
.450															
.500	-.560	-.155	-.166	-.627	-.104	-.122	-.634	-.069	-.081	-.621	-.045	-.051	-.767	-.003	.014
.550				-.545			-.540		-.040	-.527	.002	-.004	-.532	.020	.032
.600	-.507	-.096	-.090	-.392	-.051	-.063		-.029			.043	.049	-.321	.056	.061
.650				-.392			-.381	.019	.019	-.380	.065	.085	-.174	.079	.097
.700	-.431	-.037	-.043	-.210	-.004	.008		.072	.084	-.186	.132	.102	-.044	.120	.103
.750				-.210	.049	.067	-.187	.078	.101	.014	.079	.073	-.003	.079	.097
.800	-.219	.021	.035	.036	.114	.114	.107	.078	.101	.049	.073	.073	-.003	.079	.097
.900	.304	.062	.080	.114			.113	.049	.031	.049			-.003	.079	.097
.975	.096			.133									-.003	.079	.097
1.000	.068	.098	.133		.049	.123	.113	.049	.031	.049	.073	.073	-.003	.079	.097

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(m) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$			
	Row A		Row B		Row A		Row B		Row A		Row B	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
Inboard station												
.010	-.274	.222	-.246	-.629	-.077	.530	-.077	.643	-.737	.463	.196	.813
.025	-.521	.006	-.314	-.309	-.052	.307	-.052	.469	-.950	.469	.088	.405
.050	-.534	-.588	-.470	-.534	-.265	.015	-.265	.177	-.614	.177	-.103	.257
.075	-.561			-.959					-.1085			.117
.100	-.577	-.913	-.470	-.893	-.248	-.405	-.248	-.204	-.1074	-.204	.131	.089
.150	-.590	-.555	-.510	-.348	-.282	-.405	-.282	-.260	-.1063	-.260	.170	.064
.200	-.566	-.510	-.458	-.708	-.265	-.377	-.265	-.238	-.1023	-.238	.170	.005
.250	-.561	-.426	-.381	-.725	-.248	-.315	-.248	-.265	-.1076	-.265	.159	.038
.300	-.605	-.381	-.330	-.738	-.209	-.315	-.209	-.193	-.760	-.193	.142	.027
.350	-.672			-.764					-.799		-.858	.022
.400	-.567	-.302	-.269	-.792	-.191	-.220	-.191	-.154	-.832	-.154	-.114	.005
.450												
.500	-.622	-.241	-.212	-.725	-.152	-.170	-.152	-.120	-.810	-.120	-.092	-.001
.550	-.627	-.162	-.134	-.590	-.091	-.108	-.091	-.064	-.501	-.064	-.041	.016
.600												
.650	-.482	-.101	-.078	-.444	-.052	-.057	-.052	-.025	-.406	-.025	-.013	.027
.700												
.750	-.265	-.011	.000	-.237	.015	.064	.015	.032	-.727	.032	.048	.061
.800	-.304	.067	.073	.004	.088	.077	.088	.093	-.026	.093	.110	.061
.850	.140			.128	.133		.133		.133		.134	.106
.900	.162	.162	.174	.144	.144	.150	.144	.149	.149	.149	.149	.178
.950												
1.000												
Outboard station												
.010	-.228	-.256	-.256	-.539	-.054	.231	-.054	.427	-.658	.427	.223	.380
.025	-.555	-.561	-.558	-.838	-.149	.014	-.149	.234	-.980	.234	.037	.133
.050	-.497	-.514	-.474	-.930	-.200	-.132	-.200	.059	-.127	.059	-.099	.088
.075	-.531			-.551					-.1082			
.100	-.581	-.632	-.503	-.781	-.245	-.465	-.245	-.269	-.1014	-.269	.145	.037
.150	-.480	-.756	-.418	-.956	-.206	-.426	-.206	-.237	-.1070	-.237	.122	.037
.200	-.536	-.767	-.373	-.787	-.183	-.307	-.183	-.212	-.986	-.212	.111	.032
.250	-.610	-.598	-.378	-.787	-.178	-.268	-.178	-.161	-.1036	-.161	.109	.026
.300	-.627	-.407	-.305	-.304	-.178	-.211	-.178	-.156	-.1048	-.156	.111	.002
.350	-.646			-.304					-.1031		-.105	
.400	-.560	-.210	-.242	-.309	-.161	-.155	-.161	-.116	-.878	-.116	-.105	-.019
.450												
.500	-.615	-.165	-.182	-.546	-.138	-.090	-.138	-.077	-.653	-.077	-.088	-.030
.550												
.600	-.531	-.102	-.091	-.533	-.265	-.070	-.265	-.032	-.506	-.032	-.026	-.025
.650												
.700	-.390	-.041	-.018	-.375	.003	.003	.003	.019	-.359	.019	.030	-.008
.750												
.800	-.242	-.012	.046	-.183	.076	.054	.076	.070	-.161	.070	.093	.032
.850	.010	.072	.100	-.121	.116	.121	.116	.132	-.039	.132	.138	.032
.900	.134	.082	.151	.121	.126	.042	.126	.042	.126	.042	.115	
.950												
1.000												

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(n) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.0$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.262	-.252	-.207	-.528	-.529	-.045	-.679	-.638	-.184	-.770	-.717	-.257
.025	-.495	-.050	-.305	-.739	-.324	-.048	-.857	-.453	-.059	-.953	-.540	-.152
.050	-.321	-.289	-.499	-.312	.011	-.210	-.394	.162	-.108	-.587	-.276	-.010
.075	-.352	-.849	-.488	-.825	-.447	-.248	-.982	-.232	-.152	-.107	-.064	-.064
.100	-.477	-.817	-.531	-.852	-.480	-.291	-.1003	-.297	-.194	-.115	-.118	-.139
.150	-.547	-.542	-.542	-.798	-.366	-.302	-.1019	-.270	-.189	-.1126	-.199	-.134
.200	-.558	-.472	-.499	-.738	-.318	-.242	-.0971	-.248	-.167	-.1151	-.183	-.112
.250	-.536	-.510	-.510	-.695	-.280	-.215	-.814	-.221	-.156	-.1115	-.166	-.123
.300	-.585	-.510	-.510	-.695	-.280	-.215	-.814	-.221	-.156	-.1115	-.166	-.123
.350	-.650	-.510	-.510	-.749	-.249	-.205	-.820	-.167	-.124	-.1142	-.134	-.102
.400	-.677	-.327	-.285	-.776	-.221	-.205	-.847	-.167	-.124	-.926	-.102	-.080
.450	-.693	-.257	-.235	-.830	-.172	-.151	-.901	-.140	-.108	-.953	-.096	-.091
.500	-.564	-.187	-.154	-.584	-.113	-.091	-.733	-.081	-.065	-.813	-.048	-.048
.550	-.558	-.101	-.090	-.372	-.059	-.048	-.372	-.038	-.032	-.344	-.015	-.021
.600	-.242	-.014	-.014	-.193	.011	.011	-.189	.027	.032	-.172	.044	.033
.650	.707	.066	.066	.333	.087	.092	.022	.086	.103	.022	.109	.098
.700	.136	.158	.153	.130	.151	.151	.119	.140	.135	.125	.152	.130
.750	.153	.158	.153	.151	.151	.151	.140	.140	.135	.146	.152	.130
.800	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012
.850	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012
.900	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012
.950	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012
1.000	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012	-.012
Outboard station												
.010	-.216	-.242	-.288	-.462	.199	.043	-.582	.381	.173	-.646	.512	.252
.025	-.552	-.341	-.460	-.761	.026	-.164	-.880	.194	-.067	-.928	.265	.026
.050	-.471	-.460	-.504	-.896	-.148	-.224	-.1032	.025	-.115	-.102	.129	-.028
.075	-.504	-.726	-.536	-.598	-.555	-.262	-.973	.311	.164	-.1069	-.180	-.093
.100	-.563	-.699	-.655	-.712	-.511	-.224	-.918	.303	.132	-.1118	-.191	-.077
.150	-.547	-.769	-.346	-.837	-.332	-.186	-.940	.240	.126	-.1064	-.164	-.082
.200	-.639	-.682	-.325	-.844	-.234	-.180	-.905	.181	.121	-.1107	-.115	-.082
.250	-.544	-.542	-.314	-.859	-.218	-.180	-.822	.175	.126	-.1113	-.115	-.093
.300	-.561	-.542	-.314	-.859	-.218	-.180	-.822	.175	.126	-.1113	-.115	-.093
.350	-.561	-.542	-.314	-.859	-.218	-.180	-.822	.175	.126	-.1113	-.115	-.093
.400	-.720	-.232	-.260	-.837	-.169	-.164	-.1676	.126	-.115	-.1161	-.092	-.088
.450	-.912	-.200	-.189	-.934	-.104	-.131	-.146	-.077	-.099	-.652	-.050	-.093
.500	-.550	-.103	-.103	-.479	-.061	-.092	-.512	-.034	-.061	-.549	-.050	-.039
.550	-.379	-.038	-.021	-.310	.004	.010	-.305	.020	.015	-.413	.032	-.001
.600	-.195	-.011	.049	-.142	.048	.059	-.137	.069	.080	-.229	.075	.049
.650	-.022	.096	.087	-.042	.118	.118	-.036	.107	.134	-.093	.135	.070
.700	-.125	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
.750	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
.800	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
.850	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
.900	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
.950	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033
1.000	-.130	.076	.141	.134	.048	.145	.167	.091	.167	.059	-.006	-.033

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(c) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.0$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.245	.291	-.233	-.461	.534	.038	-.554	.644	.145	-.690	.704	.225
.025	-.453	.082	-.349	-.690	.324	-.082	-.776	.444	.024	-.856	.527	.116
.050	-.220	-.235	-.453	-.134	.027	-.238	-.200	.164	-.111	-.614	.267	-.055
.075	-.531			-.779			-.904			-.981		-.126
.100	-.614	-.832	-.479	-.800	-.513	-.249	-.927	-.277	-.163	-.1022	-.133	-.092
.150	-.505	-.816	-.572	-.805	-.508	-.378	-.947	-.355	-.220	-.1038	-.242	-.144
.200	-.562	-.899	-.593	-.800	-.383	-.383	-.947	-.314	-.210	-.1048	-.227	-.190
.250	-.557	-.438	-.432	-.779	-.435	-.389	-.935	-.283	-.189	-.1053	-.222	-.133
.300	-.578	-.401	-.427	-.747	-.409	-.264	-.937	-.252	-.179	-.1043	-.201	-.149
.350	-.635			-.706			-.828			-.1079		-.125
.400	-.566	-.386	-.375	-.758	-.264	-.222	-.859	-.200	-.163	-.1079	-.170	-.175
.450												
.500	-.739	-.318	-.261	-.831	-.201	-.196	-.890	-.158	-.137	-.913	-.159	-.154
.550												
.600	-.728	-.199	-.168	-.826	-.139	-.123	-.890	-.122	-.085	-.856	-.081	-.076
.650												
.700	-.417	-.111	-.105	-.383	-.077	-.066	-.470	-.054	-.044	-.404	-.071	-.045
.750												
.800	-.194	-.022	-.033	-.165	-.004	-.004	-.184	.008	.008	-.211	-.029	.012
.900	.314	.061	.056	.001	.074	.084	.080	.076	.076	.066	.084	.084
.975	.118			.085			.039			.038		
1.000	.139	.144	.133	.116	.126	.105	.086	.107	.081	.064	.101	.084
Outboard station												
.010	-.170	-.230	-.305	-.398	.170	-.000	-.495	.385	.158	-.554	.484	.220
.025	-.489	-.301	-.456	-.581	-.006	-.184	-.772	.163	-.077	-.547	.267	-.010
.050	-.432	-.432	-.505	-.780	-.168	-.252	-.924	-.015	-.145	-.1004	.115	-.094
.075	-.489			-.613			-.882			-.967		-.104
.100	-.505	-.729	-.583	-.665	-.607	-.320	-.809	-.433	-.187	-.936	-.244	.141
.150	-.630	-.681	-.567	-.838	-.649	-.257	-.924	-.405	-.161	-.1025	-.257	.110
.200	-.489	-.708	-.385	-.796	-.466	-.231	-.871	-.297	-.150	-.378	-.250	.120
.250	-.592	-.708	-.379	-.822	-.278	-.215	-.929	-.218	-.140	-.1025	-.178	.115
.300	-.635	-.651	-.369	-.853	-.231	-.225	-.955	-.203	-.161	-.1041	-.178	.125
.350	-.687			-.864			-.981			-.1056		
.400	-.724	-.353	-.306	-.900	-.184	-.199	-.1012	-.161	-.150	-.884	-.146	.136
.450												
.500	-.823	-.165	-.212	-.947	-.136	-.184	-.903	-.103	-.140	-.528	-.110	.141
.550												
.600	-.891	-.097	-.113	-.529	-.095	-.105	-.480	-.093	-.082	-.486	-.078	.099
.650												
.700	-.317	-.035	-.040	-.273	-.037	-.037	-.380	-.029	-.025	-.382	-.047	.047
.750												
.800	-.144	.028	.044	-.131	.015	.026	-.250	.017	.027	-.298	-.016	.026
.900	.023	.085	.065	-.032	.057	.068	-.135	.084	.059	-.246	.005	.010
.975	.101			.026			-.077			-.199		
1.000	.127	.049	.132	.362	-.058	.068	.006	-.119	-.015	-.167	-.350	.162

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(p) M = 0.700; windmilling

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.326	.165	-.216	-.705	.510	.107	-.905	.632	.258	-1.102	.712	.359
.025	-.506	-.089	-.298	-.897	.282	-.032	-1.174	.449	.128	-1.326	.539	.193
.050	-.653	-.518	-.525	-.929	-.013	-.211	-.988	.154	-.058	-1.153	.270	.026
.075	-.557			-.936			-1.001			-1.397		
.100	-.518	-.762	-.422	-.910	-.327	-.231	-.873	-.186	-.103	-1.179	-.057	-.025
.150	-.512	-.557	-.448	-.769	-.352	-.237	-.930	-.238	-.148	-.916	-.157	-.064
.200	-.518	-.448	-.377	-.737	-.301	-.218	-.770	-.218	-.129	-.897	-.157	-.064
.250	-.518	-.377	-.294	-.673	-.253	-.173	-.834	-.193	-.097	-.833	-.147	-.038
.300	-.308	-.313	-.249	-.634	-.224	-.147	-.776	-.193	-.077	-.852	-.115	-.076
.350	-.538			-.847			-.885			-.769		
.400	-.531	-.224	-.192	-.622	-.160	-.115	-.732	-.103	-.064	-.736	-.076	-.025
.450												
.500	-.518	-.185	-.159	-.583	-.122	-.102	-.610	-.077	-.058	-.640	-.057	-.025
.550												
.600	-.422	-.127	-.102	-.468	-.077	-.064	-.481	-.039	-.026	-.589	-.025	-.006
.650												
.700	-.409	-.063	-.057	-.423	-.038	-.019	-.417	-.007	.006	-.429	.007	.013
.750												
.800	-.275	.001	.007	-.269	.019	.032	-.250	.038	.051	-.249	.045	.052
.900	-.038	.071	.071	-.013	.084	.056	-.007	.102	.109	.001	.103	.110
.975	.142			.135			.134			.122		
1.000	.174	.180	.174	.161	.161	.161	.154	.160	.154	.135	.142	.142
Outboard station												
.010	-.309	-.312	-.235	-.658	.245	.136	-.859	.438	.291	-.999	.578	.392
.025	-.618	-.386	-.406	-.993	-.039	-.110	-1.194	.251	.032	-1.334	.361	.091
.050	-.502	-.560	-.470	-.819	-.123	-.174	-1.240	.077	-.059	-1.528	.194	-.013
.075	-.509			-.806			-.839			-1.438		
.100	-.541	-.644	-.431	-.832	-.374	-.232	-.981	-.207	-.110	-1.328	-.058	-.038
.150	-.470	-.728	-.361	-.664	-.316	-.181	-.756	-.214	-.091	-.799	-.103	-.026
.200	-.477	-.573	-.277	-.664	-.258	-.148	-.756	-.175	-.085	-.806	-.056	-.032
.250	-.548	-.348	-.264	-.690	-.206	-.142	-.781	-.130	-.072	-.870	-.094	-.032
.300	-.541	-.264	-.245	-.671	-.187	-.129	-.743	-.123	-.072	-.806	-.084	-.032
.350	-.541			-.664			-.736			-.773		
.400	-.548	-.187	-.180	-.645	-.123	-.110	-.698	-.078	-.059	-.748	-.051	-.019
.450												
.500	-.515	-.141	-.141	-.574	-.071	-.090	-.607	-.039	-.052	-.638	-.026	-.019
.550												
.600	-.477	-.109	-.096	-.503	-.026	-.032	-.523	-.007	-.007	-.548	.007	.000
.650												
.700	-.380	-.077	-.006	-.387	.006	.013	-.381	.032	.019	-.399	.026	.032
.750												
.800	-.232	-.019	.020	-.219	.032	.084	-.201	.077	.070	-.213	.052	.091
.900	-.070	.007	-.077	.006	.110	-.039	.012	.070	.006	-.006	.065	.071
.975	.033			.103			.070			.071		
1.000	.116	-.012	.116	.129	.058	.114	.122	-.020	.109	-.013	.013	.103

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(a) M = 0.750; windmilling

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.288	.186	-.208	-.658	.512	.086	-.782	.630	.230	-.909	.703	.309
.025	-.481	-.037	-.305	-.833	.301	-.050	-1.045	.447	.084	-1.143	.529	.166
.050	-.492	-.411	-.457	-.664	-.009	-.220	-.729	.137	-.086	-.866	.283	-.021
.075	-.592			-.898			-1.098			-1.254		
.100	-.568	-.837	-.486	-.869	-.395	.249	-1.104	-.226	.133	-1.289	.085	-.056
.150	-.545	-.667	-.481	-.757	-.389	-.284	-.905	-.273	.179	-1.284	-.184	-.079
.200	-.545	-.557	-.416	-.722	-.366	-.272	-.753	-.244	.150	-1.272	-.167	-.073
.250	-.545	-.405	-.381	-.722	-.296	-.190	-.817	-.214	.115	-.781	-.161	-.038
.300	-.557	-.340	-.265	-.699	-.243	-.155	-.800	-.195	.097	-.787	-.132	-.038
.350	-.580			-.734			-.788			-.869		
.400	-.574	-.241	-.206	-.781	-.167	-.132	-.741	-.121	-.086	-.986	-.085	-.032
.450												
.500	-.580	-.194	-.171	-.740	-.132	-.114	-.700	-.092	-.074	-.705	-.056	-.032
.550				-.594			-.554			-.518		
.600	-.457	-.136	-.107		-.079	-.062		-.051	-.033		-.021	.003
.650				-.430			-.425					
.700	-.422	-.078	-.066		-.038	-.027		-.016	-.010	-.418	.009	.014
.750												
.800	-.265	.004	-.002	-.249	.020	.026	-.238	.037	.043	-.231	.055	.055
.900	-.019	.074	.068	.003	.090	.090	.014	.096	.096	.014	.108	.120
.975	.144			.137			.125			.125		
1.000	.168	.179	.168	.135	.161	.155	.145	.154	.148	.159	.149	.149
Outboard station												
.010	-.273	-.256	-.270	-.575	.230	.088	-.705	.434	.236	-.822	.548	.329
.025	-.578	-.396	-.420	-.804	.037	-.139	-1.041	.219	-.005	-1.139	.355	.078
.050	-.502	-.537	-.479	-.875	-.163	-.204	-1.206	.077	-.081	-1.316	.190	-.028
.075	-.514			-.775			-1.147			-1.281		
.100	-.555	-.602	-.484	-.816	.451	.245	-1.053	-.252	.134	-1.228	-.104	-.075
.150	-.467	-.743	-.408	-.963	.375	.210	-1.064	-.246	.105	-1.328	-.151	-.051
.200	-.508	-.731	-.308	-.663	.292	.181	-.905	-.199	.093	-1.228	-.134	-.045
.250	-.567	-.461	-.291	-.751	.222	.157	-.758	-.146	.087	-1.251	-.092	-.034
.300	-.573	-.332	-.255	-.729	.204	.145	-.746	-.160	.087	-1.222	-.092	-.034
.350	-.584			-.743			-.782			-1.034		
.400	-.590	-.173	-.185	-.734	.139	-.116	-.811	-.087	.076	-.722	-.063	-.045
.450												
.500	-.561	-.126	-.156	-.628	-.081	-.092	-.641	-.046	.064	-.628	-.028	-.028
.550				-.534			-.529			-.534		
.600	-.508	-.126	-.114		-.051	-.045		-.011	-.017		.008	.008
.650				-.387			-.376		.019	-.387	.055	.049
.700	-.385	-.015	-.067		.068	.013		.030			.037	.037
.750				-.192			-.187		.089	-.192	.090	.096
.800	-.220	.009	.038	-.008	.055	.066	-.005	.377	.130	-.019	.119	.137
.900	.009	.109	.068	.108	.055	.108	-.005	.101	.130	.043	.066	.155
.975	.091			.108			.113			.137		
1.000	.121	.091	.132	.066	.055	.084	-.011	.042	.107	.019	-.028	.031

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(s) M = 0.800; windmilling

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.258	.245	-.220	-.533	.505	.039	-.652	.520	.160	-.744	.686	.214
.025	-.473	.033	-.323	-.722	.308	-.081	-.951	.432	.049	-.933	.496	.102
.050	-.312	-.323	-.479	-.285	-.027	-.242	-.355	.135	-.134	-.561	.248	-.059
.075	-.549	-.899	-.527	-.819	-.512	-.285	-.975	-.280	-.237	-.1073	-.124	-.124
.100	-.479	-.862	-.597	-.851	-.539	-.319	-.991	-.339	-.274	-.1106	-.1283	-.086
.150	-.527	-.824	-.581	-.814	-.501	-.302	-.1013	-.359	-.237	-.1122	-.1305	-.048
.200	-.560	-.624	-.555	-.823	-.329	-.221	-.1013	-.291	-.237	-.1122	-.1305	-.086
.250	-.565	-.463	-.355	-.729	-.329	-.221	-.959	.301	-.140	-.1122	-.1305	-.102
.300	-.570	-.430	-.296	-.690	-.280	-.178	-.797	.210	-.107	-.1111	-.1305	-.086
.350	-.641	-.279	-.242	-.744	-.183	-.151	-.797	.150	-.107	-.1144	-.1348	-.006
.400	-.668	-.236	-.220	-.771	-.151	-.134	-.835	.123	-.102	-.912	-.1154	-.032
.450	-.678	-.155	-.129	-.825	-.097	-.081	-.889	.070	-.048	-.939	-.086	-.021
.500	-.527	-.085	-.080	-.690	-.048	-.043	-.765	-.032	-.021	-.701	-.049	-.011
.550	-.490	-.005	-.015	-.409	-.011	.022	-.350	-.027	-.033	-.340	-.016	-.016
.600	-.236	.071	.065	-.199	.087	.087	-.177	.027	.033	-.173	.038	.043
.650	.012	.141	.157	.027	.130	.151	.027	.092	.092	.016	.097	.102
.700	.163	.168	.157	.146	.151	.151	.141	.146	.130	.108	.129	.129
.750												
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010	-.207	-.250	-.289	-.451	.180	.043	-.560	.362	.160	-.637	.499	.225
.025	-.516	-.348	-.467	-.761	.015	-.175	-.863	.145	-.066	-.930	.237	.009
.050	-.456	-.483	-.500	-.858	-.175	-.251	-.1015	.004	-.137	-.1092	.134	-.078
.075	-.496	-.630	-.576	-.587	.555	.289	-.845	.337	-.185	-.1054	-.186	-.132
.100	-.559	-.562	-.478	-.717	-.535	-.240	-.912	.321	-.147	-.1022	-.186	-.103
.150	-.565	-.727	-.348	-.891	-.359	.202	-.977	.285	-.120	-.1103	-.181	-.083
.200	-.527	-.651	-.326	-.820	-.283	.185	-.923	.185	-.115	-.1060	-.132	-.078
.250	-.635	-.527	-.283	-.858	-.213	-.164	-.904	.169	-.104	-.1098	-.132	-.072
.300	-.630	-.646	-.218	-.853	-.142	-.137	-.1021	.142	-.093	-.1114	-.088	-.067
.350	-.646	-.218	-.218	-.837	-.098	-.104	-.1064	.061	-.082	-.1168	-.045	-.045
.400	-.711	-.109	.163	-.829	-.039	-.055	-.1135	.023	-.028	-.712	-.045	-.067
.450	-.776	-.088	-.098	-.479	-.004	.010	-.538	.026	-.021	-.555	-.018	-.029
.500	-.516	-.033	-.012	-.315	.064	.069	-.316	.064	.021	-.403	.009	.009
.550	-.375	.021	.059	-.148	.064	.119	-.147	.064	.069	-.262	.058	.058
.600	-.191	.102	.097	-.020	.124	.124	.004	.140	.124	-.105	.096	.096
.650	-.001	.119	.086	.124	.015	.140	.129	.064	.151	-.050	.050	.050
.700	.119	.086	.119	.102	.015	.140	.162	.064	.151	-.023	.137	.018
.750												
.800												
.850												
.900												
.950												
1.000												

TABLE 4.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Concluded

(t) $M = 0.825$; windmilling

x/c	C_p at --											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.237	.290	-.235	-.455	.517	.015	-.580	.616	.097	-.664	.580	.202
.025	-.430	.080	-.336	-.685	.313	-.113	-.773	.423	-.019	-.856	.506	.074
.050	-.201	-.263	-.476	-.173	.001	-.254	-.159	.127	-.217	-.456	.220	-.102
.075	-.528			-.758			-.903			-.981		-.106
.100	-.589	.845	-.523	-.795	-.462	.306	-.918	.341	-.227	-.1017	.154	.118
.150	-.497	-.840	-.611	-.800	-.514	-.415	-.950	.383	-.315	-.1038	-.243	-.184
.200	-.559	-.877	-.679	-.795	-.436	-.456	-.950	-.409	-.248	-.1049	-.237	-.123
.250	-.544	-.440	-.378	-.711	-.404	-.248	-.955	.310	-.191	-.1049	-.217	-.144
.300	-.565	-.357	-.305	-.732	-.363	-.202	-.944	-.253	-.165	-.1049	-.180	-.133
.350	-.622			-.701			-.825			-.1080		-.121
.400	-.663	-.289	-.258	-.763	-.237	-.181	-.851	-.185	-.154	-.1064	-.154	-.076
.450				-.821			-.882			-.898		-.087
.500	-.715	-.248	-.216	-.821	-.170	-.160	-.882	-.165	-.149	-.898	-.113	-.087
.550				-.821			-.892			-.956		-.050
.600	-.700	-.170	-.144	-.821	-.118	-.098	-.892	-.118	-.102	-.956	-.066	-.050
.650				-.841			-.909			-.967		-.019
.700	-.378	-.092	-.092	-.441	-.066	-.061	-.409	-.071	-.071	-.367	-.024	-.019
.750				-.155			-.180			-.196		.028
.800	-.196	-.008	-.019	-.022	.006	.006	-.040	.034	-.009	-.024	.028	.022
.850	.028	.075	.064	.100	.074	.074	.038	.038	.049	-.024	.080	.080
.900	.137			.131			.069			.043		-.253
.950	.153	.153	.147		.137	.116		.085	.038	.069	.116	.069
1.000												
Outboard station												
.010	-.172	-.232	-.308	-.383	.176	-.019	-.476	.340	.101	-.539	.453	.195
.025	-.491	-.340	-.476	-.670	.010	-.205	-.764	.152	-.131	-.832	.251	-.026
.050	-.424	-.465	-.518	-.729	-.163	-.262	-.915	.047	-.214	-.983	.089	-.126
.075	-.455			-.537			-.873			-.957		-.1047
.100	-.502	-.617	-.627	-.655	-.587	.330	-.876	.434	-.251	-.926	.256	.157
.150	-.617	-.622	-.586	-.822	-.608	.278	-.910	.403	.225	-.1010	.256	.120
.200	-.512	-.669	-.376	-.780	-.466	.220	-.848	.369	.188	-.1078	.120	.100
.250	-.607	-.654	-.356	-.812	-.262	.215	-.926	.235	-.178	-.1015	.137	.100
.300	-.843	-.596	-.298	-.833	-.215	.189	-.947	.220	-.167	-.1020	.152	.089
.350	-.869			-.859			-.968			-.1041		-.633
.400	-.711	-.340	-.230	-.901	-.147	-.158	-.1009	-.146	-.141	-.978	-.105	-.100
.450				-.937			-.989			-.528		-.089
.500	-.816	-.136	-.172	-.937	-.121	-.126	-.989	-.089	-.141	-.528	-.084	-.089
.550				-.644			-.481			-.466		-.079
.600	-.858	-.063	-.089	-.644	-.058	-.074	-.481	-.120	-.094	-.466	-.052	-.079
.650				-.273			-.377			-.382		-.042
.700	-.324	-.010	-.021	-.273	-.011	.016	-.377	-.037	-.042	-.382	-.021	-.042
.750				-.116			-.261	.010	-.005	-.277	.005	-.011
.800	-.157	.037	.047	-.004	.036	.047	-.136	.037	.016	-.220	.026	-.032
.850	.024	.105	.099	.057	.089	.062	-.094			-.183		-.309
.900	.120			.083			-.016	-.183	-.068	-.141	-.335	-.152
.950												
1.000												

TABLE 5.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.364	.151	-.208	-.653	.517	.100	-.906	.676	.260	-1.075	.755	.360
.025	-.441	-.057	-.294	-.871	.327	.020	-1.078	.483	.137	-1.254	.597	.219
.050	-.284	-.518	-.499	-.557	.013	-.249	-.923	.227	-.036	-.908	.366	.059
.075	-.305	-.838	-.358	-.826	-.339	-.172	-.938	-.061	-.048	-1.305	.059	.027
.100	-.838	-.620	-.358	-.833	-.448	-.211	-.906	-.189	-.087	-1.075	-.101	.165
.150	-.499	-.544	-.524	-.756	-.448	-.211	-.950	-.183	-.087	-.895	-.101	.165
.200	-.473	-.492	-.351	-.634	-.281	-.185	-.701	-.144	-.061	-.825	-.114	.101
.250	-.473	-.358	-.287	-.615	-.236	-.153	-.746	-.119	-.055	-.825	-.108	.018
.300	-.473	-.287	-.249	-.595	-.192	-.140	-.758	-.074	-.055	-.716	-.088	-.018
.350	-.473	-.217	-.204	-.595	-.147	-.121	-.650	-.061	-.048	-.697	-.063	-.024
.400	-.480	-.178	-.172	-.615	-.115	-.102	-.567	-.016	-.004	-.825	-.044	-.024
.500	-.473	-.108	-.102	-.661	-.063	-.057	-.439	-.022	-.028	-.594	-.005	.008
.600	-.383	-.044	-.044	-.390	-.019	-.012	-.375	.073	.086	-.460	.027	.027
.650	-.403	.033	.033	-.236	.045	.052	-.208	.137	.150	-.383	.072	.085
.750	-.230	.110	.116	-.012	.116	.129	-.028	.137	.142	-.204	.123	.136
.800	-.005	.212	.200	-.142	.180	.186	.137	.195	.195	.142	.149	.168
.900	.174	.200	.200	-.186	.180	.186	.175	.195	.195	.161	.149	.168
1.000	.200	.212	.200	.186	.180	.186	.175	.195	.195	.161	.149	.168
Outboard station												
.010	-.251	-.359	-.281	-.638	.253	.129	-.776	.510	.293	-.940	.648	.439
.025	-.560	-.444	-.393	-.573	.045	-.103	-1.149	.273	.041	-1.307	.420	.130
.050	-.457	-.579	-.425	-.773	-.142	-.154	-1.149	.099	-.017	-1.455	.252	.085
.075	-.567	-.663	-.386	-.631	-.335	-.174	-1.008	-.133	-.055	-1.307	.014	.001
.100	-.547	-.670	-.315	-.670	-.303	-.161	-.943	-.165	-.043	-1.384	.083	.014
.150	-.451	-.457	-.251	-.625	-.258	-.159	-.744	-.152	-.043	-.766	-.089	.001
.200	-.438	-.302	-.238	-.657	-.193	-.122	-.705	-.107	-.043	-.811	-.063	.001
.250	-.489	-.251	-.218	-.644	-.174	-.122	-.705	-.101	-.043	-.772	-.063	-.005
.300	-.496	-.251	-.218	-.644	-.174	-.122	-.705	-.101	-.043	-.772	-.063	-.005
.350	-.612	-.180	-.193	-.748	-.129	-.122	-.789	-.062	-.049	-.862	-.051	-.025
.400	-.509	-.122	-.148	-.748	-.129	-.122	-.660	-.030	-.043	-.862	-.051	-.025
.450	-.476	-.064	-.070	-.554	-.077	-.103	-.570	-.009	-.002	-.605	-.018	-.031
.500	-.438	.001	.001	-.490	-.032	-.045	-.487	.054	.054	-.502	.020	.014
.600	-.341	.065	.078	-.367	.020	.020	-.351	.099	.112	-.360	.059	.059
.700	-.199	.136	.136	-.290	.078	.084	-.171	.150	.157	-.173	.098	.111
.800	.014	.162	.162	-.026	.136	.136	-.047	.138	.138	-.040	.149	.149
.900	.162	.136	.136	.142	.084	.168	.138	.067	.170	.130	.065	.156
1.000	.188	.136	.188	.168	.084	.168	.163	.067	.170	.156	.065	.156

TABLE 5.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.253	.205	-.228	-.605	.519	.067	-.773	.664	.231	-.913	.736	.336
.025	-.429	.004	-.277	-.833	.325	-.026	-.995	.488	.097	-1.077	.605	.202
.050	-.125	-.376	-.466	-.324	.003	-.231	-.522	.268	-.090	-.651	.354	.039
.075	-.586			-.792			-1.030			-1.176		
.100	-.610	-.781	-.405	-.827	-.365	-.266	-1.030	-.107	-.119	-1.194	.010	.015
.150	-.621	-.639	-.466	-.759	-.441	-.248	-.779	-.268	-.112	-1.147	.137	.049
.200	-.522	-.481	-.388	-.710	-.330	-.242	-.709	-.207	-.113	-1.147	.137	.031
.250	-.598	-.393	-.329	-.658	-.283	-.184	-.779	-.183	-.090	-.703	.142	.031
.300	-.580	-.329	-.329	-.716	-.301	-.166	-.756	-.148	-.084	-.762	.113	.031
.350	-.598			-.716			-.727			-.849		
.400	-.610	-.259	-.247	-.728	-.231	-.143	-.692	-.107	-.078	-.797	.061	.037
.450												
.500	-.639	-.212	-.201	-.675	-.178	-.126	-.697	-.090	-.067	-.656	-.049	.043
.550												
.600	-.434	-.136	-.125	-.587	-.085	-.067	-.499	-.055	-.026	-.464	.008	.002
.650												
.700	-.393	-.066	-.055	-.412	-.026	-.020	-.382	.004	.009	-.376	.033	.027
.750												
.800	-.236	.015	.021	-.231	.038	.044	-.195	.062	.068	-.207	.080	.080
.850	.004	.057	.103	.015	.102	.120	.044	.126	.138	.050	.144	.144
.900	.975	.161		.149			.109			.144		
.950	.185	.191	.185	.167	.178	.178	.173	.179	.179	.173	.185	.179
1.000												
Outboard station												
.010	-.215	-.306	-.285	-.551	.258	.082	-.662	.485	.245	-.755	.612	.372
.025	-.544	-.362	-.444	-.868	.008	-.121	-1.038	.261	.026	-1.108	.420	.138
.050	-.432	-.526	-.450	-.845	-.174	-.192	-1.155	.085	-.056	-1.255	.244	.032
.075	-.420			-.527			-.814			-1.020		
.100	-.567	.714	.432	-.792	-.427	-.210	-1.026	.209	.080	-1.214	.074	.003
.150	-.467	.785	.362	-.374	-.374	.174	-.985	.215	.074	-1.237	.121	.003
.200	-.667	.708	.273	-.656	-.298	-.145	-.891	-.185	-.062	-1.131	.109	.009
.250	-.538	.644	.268	-.733	-.227	-.133	-.714	-.133	-.062	-1.167	.080	.009
.300	-.550	-.297	-.244	-.727	-.198	-.133	-.714	-.127	-.062	-1.084	.015	.015
.350	-.702			-.851			-.855			-.749		
.400	-.573	-.191	-.203	-.727	-.139	-.127	-.791	-.086	-.074	-.667	.050	.033
.450												
.500	-.532	-.132	.162	-.615	-.086	-.110	-.603	-.050	-.062	-.585	.015	.033
.550												
.600	-.479	-.074	-.080	-.521	-.039	-.051	-.497	-.009	-.015	-.485	.020	.009
.650												
.700	-.362	-.069	-.003	-.368	.020	.014	-.344	.044	.044	-.344	.073	.061
.750												
.800	-.191	.056	.073	-.180	.073	.084	-.156	.091	.108	-.156	.114	.114
.850	.026	.132	.132	.037	.137	.137	.055	.155	.150	.056	.173	.161
.900	.975	.156		.137			.144			.155		
.950	.179	.114	.179	.155	.079	.161	.161	.067	.167	.179	.103	.185
1.000												

TABLE 5.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.274	.248	-.210	-.589	.530	-.059	-.740	.663	.204	-.815	.757	.310
.025	-.577	.034	-.280	-.791	.330	-.051	-.891	.487	.084	-.978	.587	.183
.050	-.306	-.314	-.554	-.158	.061	-.231	-.353	.213	-.073	-.473	.346	.037
.075	-.706			-.780			-.964			-.1085		
.100	-.610	-.778	-.543	-.802	-.315	-.192	-.975	-.213	-.084	-.1113	-.148	.032
.150	-.577	-.874	-.644	-.814	-.550	-.276	-.981	-.269	-.286	-.1113	-.131	-.036
.200	-.639	-.482	-.610	-.685	-.343	-.231	-.885	-.224	-.118	-.1129	-.137	-.041
.250	-.594	-.398	-.347	-.685	-.287	-.180	-.672	-.196	-.107	-.1085	-.114	-.030
.300	-.594	-.342	-.308	-.685	-.231	-.163	-.745	-.157	-.090	-.821	-.103	-.030
.350	-.599			-.724			-.751			-.754		
.400	-.610	-.274	-.269	-.752	-.175	-.158	-.779	-.107	-.090	-.787	-.069	-.030
.450												
.500	-.638	-.235	-.224	-.679	-.147	-.180	-.784	-.090	-.084	-.793	-.052	-.047
.550												
.600	-.610	-.157	-.140	-.483	-.096	-.163	-.443	-.045	-.039	-.557	-.019	.004
.650												
.700	-.560	-.078	-.073	-.388	-.035	-.029	-.364	-.000	.005	-.344	.026	.032
.750												
.800	-.230	.006	.011	-.214	.038	.038	-.180	.061	.067	-.159	.082	.088
.900	.017	.095	.101	.638	.111	.117	.050	.123	.140	.037	.133	.155
.975	.146			.150			.151			.155		
1.000	.179	.190	.185	.173	.178	.173	.173	.179	.173	.172	.200	.189
Outboard station												
.010	-.215	-.229	-.303	-.493	.265	-.062	-.593	.482	.238	-.652	.630	.359
.025	-.548	-.362	-.446	-.797	.020	-.155	-.926	.235	.004	-.996	.403	.104
.050	-.463	-.497	-.480	-.871	-.121	-.250	-.1067	.049	-.080	-.1137	.239	.031
.075	-.300			-.312			-.576			-.714		
.100	-.587	-.858	-.469	-.713	-.521	-.233	-.965	-.221	-.103	-.1104	-.054	.008
.150	-.486	-.745	-.407	-.927	-.391	-.183	-.988	-.238	-.086	-.1149	-.099	.008
.200	-.508	-.756	-.305	-.775	-.318	-.155	-.909	-.199	-.075	-.1092	-.105	.002
.250	-.581	-.581	-.283	-.752	-.228	-.143	-.971	-.142	-.075	-.1126	-.071	-.003
.300	-.581	-.435	-.255	-.763	-.205	-.138	-.993	-.131	-.075	-.1115	-.071	-.009
.350	-.734			-.893			-.1084			-.1200		
.400	-.627	-.193	-.221	-.792	-.138	-.132	-.875	-.086	-.089	-.1149	-.043	-.020
.450												
.500	-.576	-.131	.170	-.634	-.087	-.115	-.644	-.047	-.075	-.647	-.014	-.026
.550												
.600	-.497	-.086	-.086	-.504	-.042	-.053	-.469	-.007	-.024	-.421	.025	.025
.650												
.700	-.362	-.018	-.007	-.352	.015	.015	-.323	.044	.038	-.285	.076	.076
.750												
.800	-.182	.044	.072	-.155	.071	.082	-.131	.094	.106	-.116	.132	.138
.900	.038	.123	.128	.054	.144	.139	.066	.156	.156	.070	.200	.189
.975	.157			.144			.151			.183		
1.000	.173	.111	.173	.167	.071	.167	.173	.083	.179	.217	.149	.223

TABLE 5.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Concluded

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.223	.283	-.224	-.517	.535	.015	-.639	.651	-.161	-.773	.730	.234
.025	-.449	.094	-.272	-.738	.324	-.058	-.817	.481	-.066	-.891	.573	.169
.050	.283	-.239	-.487	.028	.033	-.236	-.138	.212	-.095	-.256	.331	.008
.075	-.509			-.721			-.882			-.999		
.100	-.525	-.686	-.417	-.721	.323	-.220	-.898	-.149	-.101	-.1015	-.014	-.014
.150	-.525	-.826	-.611	-.765	.484	.317	-.919	-.295	-.165	-.1042	-.202	-.132
.200	-.550	-.896	-.535	-.727	.441	.283	-.930	-.257	-.149	-.1042	-.364	-.100
.250	-.578	-.433	-.417	-.738	.317	.220	-.908	-.219	-.117	-.1042	-.154	-.057
.300	-.562	-.315	-.315	-.684	.263	-.204	-.758	-.176	-.101	-.1037	-.122	-.057
.350	-.605			-.705			-.725			-.1037		
.400	-.632	-.288	-.299	-.754	.209	-.193	-.774	-.133	-.203	-.854	-.084	-.170
.450												
.500	-.659	-.261	-.239	-.808	.177	-.172	-.833	-.106	-.171	-.875	-.095	-.100
.550												
.600	-.562	-.164	-.148	-.716	.112	-.102	-.747	-.063	-.052	-.708	-.030	-.030
.650												
.700	-.385	-.078	-.062	-.355	-.048	-.037	-.295	-.014	-.009	-.288	.013	.008
.750												
.800	-.202	.013	.019	-.172	-.022	.033	-.133	.050	-.066	-.116	.078	.067
.900	.040	.100	.105	.044	.098	.109	.061	.120	.131	.061	.142	.137
.975	.159			.141			.152			.158		
1.000	.175	-.186	.180	.157	.163	.163	.174	.174	.169	.110	.185	.169
Outboard station												
.010	-.183	-.189	-.280	-.435	.232	.030	-.504	.435	.187	-.590	.572	.285
.025	-.497	-.345	-.448	-.733	.043	-.191	-.840	.228	-.049	-.909	.353	.071
.050	-.427	-.459	-.492	-.647	-.174	-.234	-.970	.027	-.114	-.1055	.185	-.016
.075	-.053			.032			-.211			-.416		
.100	-.556	-.843	-.475	-.679	.608	-.245	-.899	-.298	-.130	-.1001	-.124	-.054
.150	-.486	-.697	-.470	-.857	.413	.223	-.926	-.287	-.108	-.1061	-.183	-.043
.200	-.502	-.746	-.335	-.820	.348	.180	-.867	-.238	-.092	-.1001	-.167	-.043
.250	-.616	-.676	-.313	-.841	-.256	-.174	-.943	-.173	-.087	-.1055	-.119	-.043
.300	-.616	-.540	-.275	-.868	-.223	-.163	-.948	-.157	-.081	-.1055	-.113	-.048
.350	-.744			-.950			-.1083			-.1058		
.400	-.692	-.237	-.232	-.830	.153	-.158	-.1029	-.103	-.087	-.1110	-.070	-.070
.450												
.500	-.795	-.123	-.178	-.917	-.109	-.142	-.1073	-.059	-.081	-.741	-.037	-.081
.550												
.600	-.464	-.064	-.086	-.532	-.055	-.071	-.493	-.016	-.032	-.508	-.005	-.037
.650												
.700	-.340	-.010	.001	-.299	.005	.005	-.271	.033	.027	-.357	.044	.011
.750												
.800	-.161	.050	.071	-.131	.064	.070	-.103	.087	.103	-.189	.087	.066
.900	.040	.120	.126	.048	.129	.129	.157	.152	.152	-.032	.147	.087
.975	.153			.129			.190	.087	.190	.082		.017
1.000	.169	.109	.174	.156	.048	.156	.190					

TABLE 6.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.271	.197	-.155	-.786	.534	.140	-.886	.669	.303	-1.087	.741	.374
.025	-.693	-.028	-.220	-.947	.330	.017	-1.101	.483	.131	-1.318	.597	.238
.050	-.610	-.335	-.431	-.839	.030	-.168	-.912	.240	-.023	-1.023	.367	.066
.075	-.546			-.896			-.956			-1.401		
.100	-.571	.693	-.362	-.928	-.328	-.200	-.944	-.125	-.106	-1.177	.008	.002
.150	-.565	-.661	-.399	-.883	-.372	-.226	-.873	-.234	-.119	-.895	-.120	-.037
.200	-.603	-.584	-.405	-.717	-.328	-.200	-.793	-.221	-.106	-.914	-.139	-.043
.250	-.629	-.450	-.284	-.762	-.283	-.168	-.924	-.262	-.099	-.978	-.139	-.037
.300	-.520	-.507	-.245	-.634	-.238	-.149	-.700	-.182	-.093	-.889	-.127	-.037
.350	-.584			-.647			-.694			-.889		
.400	-.565	-.245	-.213	-.628	-.168	-.136	-.841	-.125	-.080	-.793	-.063	-.030
.450												
.500	-.584	-.154	-.188	-.590	-.136	-.130	-.617	-.099	-.080	-.626	-.063	-.042
.550												
.600	-.424	-.143	-.124	-.539	-.098	-.091	-.502	-.061	-.055	-.485	-.024	-.018
.650												
.700	-.399	-.079	-.072	-.660	-.053	-.053	-.425	-.023	-.016	-.408	.002	.008
.750												
.800	-.265	.010	.010	-.270	.017	.024	-.253	.035	.035	-.235	.059	.053
.850	-.028	.063	.063	-.015	.094	.100	-.003	.105	.112	-.008	.110	.117
.900												
.950	.151			-.139			.131			.130		
1.000	.183	.189	.185	.158	.170	.164	.155	.163	.156	.149	.155	.149
Outboard station												
.010	-.241	-.308	-.193	-.638	.302	.179	-.840	.502	.332	-.997	.649	.440
.025	-.544	-.395	-.350	-.952	.075	-.086	-1.188	.292	.067	-1.332	.433	.162
.050	-.485	-.543	-.382	-.793	-.131	-.137	-1.207	.112	-.017	-1.513	.272	.072
.075	-.582			-.786			-1.278			-1.593		
.100	-.543	.710	-.466	-1.050	-.381	-.221	-1.040	-.145	-.120	-1.410	-.005	-.005
.150	-.479	-.678	-.344	-.658	-.343	-.182	-.750	-.197	-.094	-.778	.001	.001
.200	-.479	-.485	-.254	-.651	-.258	-.137	-.737	-.184	-.087	-.772	-.095	-.025
.250	-.537	-.337	-.247	-.656	-.234	-.131	-.821	-.152	-.068	-.843	-.089	-.083
.300	-.537	-.292	-.215	-.677	-.208	-.118	-.744	-.145	-.062	-.810	-.089	-.018
.350	-.537			-.671			-.750			-.765		
.400	-.537	-.209	-.183	-.651	-.137	-.112	-.699	-.100	-.055	-.727	-.050	-.037
.450												
.500	-.517	-.144	-.151	-.587	-.092	-.099	-.615	-.062	-.055	-.630	-.031	-.031
.550												
.600	-.472	-.086	-.059	-.516	-.047	-.073	-.525	-.023	-.023	-.527	.001	-.018
.650												
.700	-.376	-.022	-.022	-.394	-.002	-.015	-.377	.016	.016	-.385	-.005	-.005
.750												
.800	-.228	-.086	.042	-.214	-.054	.068	-.157	.073	.080	-.192	.079	.091
.850												
.900	-.009	.105	.094	-.002	.011	.075	.003	.112	.106	.001	.098	.124
.950	.087			.088			.003			.058		
1.000	.145	.061	.132	.113	.043	.055	.106	.022	.138	.098	.021	.085
.010												
.025												
.050												
.075												
.100												
.150												
.200												
.250												
.300												
.350												
.400												
.450												
.500												
.550												
.600												
.650												
.700												
.750												
.800												
.850												
.900												
.950												
1.000												

TABLE 6.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.250	.207	-.172	-.689	.559	.127	-.791	.657	.257	-.913	.749	.343
.025	-.472	.024	-.297	-.847	.332	-.013	-1.048	.288	.103	-1.158	.588	.220
.050	-.262	-.326	-.454	-.491	.046	-.263	-.668	.220	-.084	-.767	.354	.039
.075	-.687			-.835			-1.124			-1.257		
.100	-.594	-.862	-.440	-.859	-.345	.217	-1.107	-.166	-.107	-1.275	.004	-.008
.150	-.571	-.722	-.446	-.742	-.409	-.234	-1.067	-.359	-.143	-1.257	-.154	-.054
.200	-.728	-.635	-.356	-.742	-.450	.223	-.750	-.254	-.131	-1.205	-.165	-.049
.250	-.658	-.431	-.337	-.719	-.328	.176	-.809	-.230	-.107	-.746	-.148	-.136
.300	-.716	-.367	-.314	-.707	-.261	-.164	-.779	-.183	-.113	-.779	-.136	-.049
.350	-.705			-.777			-.762			-.855		
.400	-.652	-.262	-.250	-.748	-.182	-.153	-.762	-.131	-.107	-.872	-.113	-.049
.450												
.500	-.710	-.244	-.221	-.678	-.153	-.147	-.698	-.113	-.107	-.679	-.066	-.054
.550				-.544	-.106	-.106	-.587	-.072	-.072	-.650	-.031	-.031
.600	-.442	-.168	-.163				-.417	-.032	-.026	-.516	.004	.004
.650	-.407	-.087	-.087	-.427	-.053	-.053						
.700				-.252	.017	.022	-.230	.033	.039	-.224	.057	.057
.750	-.250	.001	-.005	-.011	.092	.058	-.021	.057	.103	.027	.121	.121
.800	-.005	.088	.068	.139			.132			.138		
.875	.158			.163	.168	.157	.150	.155	.144	.156	.162	.150
1.000	.181	.181	.181									
Outboard station												
.010	-.217	-.275	-.218	-.577	.288	.154	-.703	.471	.287	-.820	.640	.414
.025	-.552	-.346	-.370	-.868	.045	-.096	-1.026	.132	.026	-1.113	.426	.115
.050	-.481	-.493	-.428	-.753	-.125	-.160	-1.214	.102	-.044	-1.295	.214	.050
.075	-.458			-.607			-1.161			-1.290		
.100	-.604	-.798	-.459	-.953	-.407	-.272	-1.067	-.203	-.144	-1.254	-.026	-.026
.150	-.458	-.792	-.376	-.982	-.401	-.190	-1.067	-.244	-.109	-1.290	-.126	-.015
.200	-.475	-.669	-.288	-.630	-.325	-.155	-.897	-.227	-.086	-1.196	-.132	-.021
.250	-.552	-.411	-.270	-.730	-.248	-.149	-.726	-.168	-.080	-1.207	-.103	-.021
.300	-.552	-.317	-.223	-.736	-.248	-.113	-.756	-.150	-.080	-1.184	-.103	-.021
.350	-.549			-.742			-.797			-.902		
.400	-.575	-.200	-.206	-.730	-.143	-.119	-.850	-.103	-.080	-.714	-.062	-.021
.450												
.500	-.546	-.141	-.165	-.624	-.137	-.113	-.632	-.074	-.074	-.620	-.026	-.025
.550				-.530	-.061	-.090	-.526	-.027	-.044	-.514	.009	-.021
.600	-.481	-.094	-.112		-.020	-.020	-.385	.020	.014	-.367	.050	.032
.650				-.389	-.020							
.700	-.370	-.024	-.036	-.254	.039	.063	-.168	.073	.079	-.173	.097	.097
.750				-.004	.127	.022	-.026	.051	.057	.032	.150	.138
.800	-.200	.041	.046	-.098			.097			.138		
.850	.017	.123	.111	.057	.069	.104	-.102	.050	.138	.103	.079	.150
.975	.076											
1.000	.140	.111	.052									

TABLE 6.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.342	-.261	-.154	-.618	-.555	-.121	-.685	-.675	.241	-.851	.751	.317
.025	-.521	-.061	-.251	-.809	.357	-.029	-.932	.487	.083	-1.081	.571	.213
.050	-.157	-.364	-.482	-.366	.060	-.214	-.461	.234	-.085	-.605	.358	.022
.075	-.644			-.831			-1.061			-1.154		
.100	-.605	-.857	-.476	-.855	-.355	-.220	-1.061	-.287	-.125	-1.193	-.028	-.034
.150	-.616	-.868	-.661	-.753	-.551	-.287	-1.061	-.287	-.164	-1.198	-.174	-.062
.200	-.611	-.504	-.437	-.680	-.366	-.270	-.865	-.388	-.192	-1.187	-.185	-.067
.250	-.644	-.443	-.336	-.702	-.327	-.192	-.741	-.237	-.214	-1.165	-.174	-.050
.300	-.628	-.375	-.257	-.651	-.366	-.158	-.747	-.192	-.248	-.980	-.146	-.050
.350	-.611			-.736			-.797			-.818		
.400	-.689	-.275	-.331	-.820	-.186	-.158	-.820	-.136	-.108	-.834	-.095	-.056
.450				-.702			-.814			-.885		
.500	-.633	-.258	-.230	-.523	-.181	-.153	-.523	-.074	-.096	-.650	-.045	-.028
.550	-.560	-.174	-.157	-.411	-.198	-.102	-.399	-.029	-.063	-.381	-.006	.006
.600	-.415	-.090	-.084	-.224	-.052	-.052	-.203	.038	-.024	-.196	.056	.062
.700	-.247	.005	.011	.027	.027	.105	.032	.100	.038	.034	.118	.123
.800	.011	.095	.095	.144	.100		.139		.111	.140		
.900	.157			.161	.172	.167	.156	.156	.156	.162	.168	.162
.975												
1.000	-.179	.185	.175									
Outboard station												
.010	-.216	-.215	-.195	-.488	.258	.156	-.617	.488	.278	-.711	.624	.387
.025	-.537	-.340	-.365	-.798	.042	-.082	-.927	.138	.014	-1.010	.382	.117
.050	-.469	-.447	-.420	-.922	-.104	-.178	-1.102	.116	-.098	-1.184	.247	.033
.075	-.402			-.499			-1.029			-1.167		
.100	-.740	-.841	-.503	-.877	-.488	-.268	-1.029	-.217	-.166	-1.145	-.063	-.052
.150	-.469	-.762	-.350	-.945	-.409	-.194	-1.074	-.251	-.115	-1.190	-.131	-.035
.200	-.503	-.734	-.295	-.730	-.341	-.161	-.961	-.245	-.104	-1.128	-.148	-.035
.250	-.599	-.554	-.283	-.764	-.262	-.149	-1.018	-.172	-.093	-1.167	-.108	-.035
.300	-.599	-.390	-.221	-.798	-.228	-.104	-1.029	-.155	-.081	-1.167	-.108	-.029
.350	-.616			-.752			-1.029			-1.162		
.400	-.621	-.199	-.167	-.798	-.183	-.115	-.888	-.058	-.081	-1.173	-.063	-.029
.450							-.868			-.756		
.500	-.593	-.221	-.165	-.646	-.093	-.104	-.668	-.076	-.065	-.474	-.035	-.035
.550	-.509	-.092	-.109	-.510	-.059	-.082	-.493	-.031	-.042	-.010	.010	-.024
.600							-.369			-.305		
.650	-.373	-.030	-.035	-.358	.009	-.014	-.155	.031	.009	-.153	.055	.044
.700							-.161			-.076	.106	
.750	-.199	.032	.045	-.161	.065	.026	-.037	.076	.038	-.145	.173	.157
.800	.027	.122	.117	.121	.133	.121	.133	.138	.138	.162	.134	.185
.900	.145			.121								
.975												
1.000	.162	.100	.138									

TABLE 6.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.246	.306	-.153	-.540	.561	.120	-.625	.662	.197	-.748	.747	.294
.025	-.526	.099	-.241	-.745	.364	-.040	-.956	.475	.076	-.985	.583	.112
.050	-.036	-.214	-.429	-.325	.068	-.255	-.274	.211	-.080	-.452	.335	-.117
.075	-.596			-.798			-.964			-.1071		
.100	-.510	-.806	-.499	-.841	-.524	-.400	-.997	-.350	-.129	-.1109	-.064	-.021
.150	-.542	-.833	-.574	-.815	-.529	-.277	-.997	-.328	-.188	-.1109	-.204	-.096
.200	-.537	-.871	-.521	-.739	-.562	-.271	-.986	-.344	-.181	-.1120	-.215	-.074
.250	-.585	-.515	-.424	-.712	-.341	-.271	-.975	-.285	-.145	-.1109	-.295	-.064
.300	-.564	-.391	-.294	-.680	-.282	-.336	-.792	-.280	-.129	-.1093	-.177	-.064
.350	-.612			-.745			-.797			-.1120		
.400	-.666	-.284	-.273	-.766	-.207	-.191	-.819	-.188	-.123	-.904	-.107	-.074
.450												
.500	-.661	-.257	-.278	-.825	-.191	-.228	-.883	-.129	-.123	-.942	-.085	-.080
.550												
.600	-.569	-.273	-.171	-.718	-.148	-.110	-.835	-.086	-.080	-.883	-.112	-.053
.650												
.700	-.504	-.095	-.090	-.363	-.067	-.045	-.344	-.037	-.042	-.344	-.015	-.015
.750												
.800	-.224	.002	.007	-.217	.014	.025	-.210	.033	.033	-.166	.049	.055
.900	-.029	.093	.099	.041	.095	.105	.033	.103	.103	.033	.114	.120
.975	.158	.138	.138	.138			.130			.130		
1.000	.174	.180	.174	.159	.159	.159	.152	.157	.152	.157	.168	.152
Outboard station												
.010	-.157	-.177	-.197	-.442	.273	.136	-.543	.634	.245	-.613	.617	.356
.025	-.487	-.203	-.352	-.745	.039	-.134	-.836	.211	-.006	-.917	.373	.075
.050	-.422	-.422	-.422	-.870	-.118	-.193	-.1009	.032	-.088	-.1096	.200	-.017
.075	-.530			-.356			-.906			-.087		
.100	-.693	-.818	-.520	-.789	-.540	-.285	-.934	-.288	.158	-.102	-.114	-.066
.150	-.466	-.742	-.422	-.886	-.431	-.226	-.999	-.315	.126	-.101	-.179	-.044
.200	-.487	-.758	-.314	-.821	-.366	-.177	-.912	.277	.109	-.107	-.179	-.044
.250	-.607	-.655	-.292	-.832	-.269	-.188	-.977	.196	.104	-.1096	-.142	-.044
.300	-.601	-.514	-.244	-.837	-.242	-.128	-.988	.185	.093	-.101	-.131	-.039
.350	-.612			-.821			-.1009			-.101		
.400	-.688	-.222	-.249	-.821	-.150	-.123	-.1042	-.109	-.088	-.1155	-.071	-.044
.450												
.500	-.709	-.141	-.173	-.913	-.118	-.118	-.1118	-.071	-.088	-.678	-.044	-.049
.550												
.600	-.504	-.097	-.119	-.523	-.063	-.085	-.538	-.028	-.044	-.543	-.011	-.066
.650												
.700	-.352	-.021	-.032	-.307	-.004	-.009	-.310	.015	-.005	-.402	.027	-.006
.750												
.800	-.173	.033	.049	-.134	.056	.066	-.125	.080	.086	-.250	.075	.059
.900	-.038	.125	.119	.050	.126	.125	.042	.151	.135	-.039	.135	.075
.975	.146	.131	.131	.131			.135			.039		
1.000	.163	.071	.166	.148	.045	.104	.173	.051	.167	.039	-.022	.005
$\alpha = 4^\circ$												
.010												
.025												
.050												
.075												
.100												
.150												
.200												
.250												
.300												
.350												
.400												
.450												
.500												
.550												
.600												
.650												
.700												
.750												
.800												
.900												
.975												
1.000												

TABLE 6.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Concluded

(e) $M = 0.825$

x/c	C_p at											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.209	.323	-.188	-.460	.564	.086	-.550	.657	.178	-.691	.728	.256
.025	-.447	.153	-.240	-.678	.387	-.018	-.784	.490	.069	-.878	.565	.129
.050	-.096	-.204	-.478	-.003	.086	-.205	-.051	.220	-.129	-.281	.321	-.219
.075	-.509			-.751			-.898			-.982		
.100	-.592	.722	-.503	-.792	-.449	.242	-.924	-.217	.170	-.1023	-.094	-.079
.150	-.509	.784	-.520	-.752	-.486	.361	-.924	-.513	.227	-.1023	-.271	.245
.200	-.551	.856	-.523	-.777	-.475	.418	-.930	-.368	.222	-.1039	-.260	.177
.250	-.535	.773	-.525	-.773	-.340	.210	-.930	-.435	.217	-.1039	-.245	.125
.300	-.556	-.478	-.530	-.704	-.356	.252	-.846	-.238	.149	-.1028	-.276	.187
.350	-.618			-.688			-.805			-.1054		
.400	-.654	-.499	-.427	-.730	-.247	.210	-.826	-.165	.290	-.1054	-.214	.162
.450												
.500	-.716	-.478	-.515	-.787	-.179	.221	-.857	-.170	.170	-.893	-.250	.157
.550												
.600	-.711	-.225	-.205	-.782	-.153	.184	-.857	-.097	.123	-.795	-.125	.219
.650												
.700	-.463	-.199	-.111	-.351	-.075	-.055	-.352	-.056	.061	-.364	-.193	.058
.750												
.800	-.173	-.018	-.018	-.205	.023	.023	-.144	.012	.012	-.224	.064	.001
.850	.034	.075	.080	.055	.101	.101	.012	.050	.074	-.063	.072	.061
.900	.132			.148			.095			.015		
.950												
1.000	.153	.158	.158	.164	.164	.148	.126	.131	.074	.066	.087	.045
Outboard station												
.010	-.140	-.181	-.206	-.396	.234	.095	-.482	.415	.182	-.540	.542	.291
.025	-.462	-.249	-.374	-.668	.069	-.125	-.789	.183	.027	-.827	.284	.028
.050	-.400	-.395	-.447	-.815	-.104	-.208	-.921	.031	-.121	-.974	.154	-.050
.075	-.244			-.182			.717			.948		
.100	-.639	.811	-.572	-.710	-.517	.308	-.848	-.372	.225	-.863	.201	.144
.150	-.577	.644	-.572	-.820	-.847	.229	-.911	-.377	.168	-.1005	-.295	.118
.200	-.447	.707	-.374	-.783	-.360	.187	-.848	-.335	.152	-.958	.274	.118
.250	-.562	.702	-.353	-.809	-.255	.177	-.916	-.241	.142	-.1010	-.212	.107
.300	-.634	-.650	-.280	-.820	-.234	.135	-.942	-.210	.116	-.1026	.151	.086
.350	-.670			-.835			-.958			-.1036		
.400	-.691	-.400	-.239	-.851	-.140	.130	-.995	-.131	.116	-.1021	.128	.097
.450												
.500	-.801	-.176	-.197	-.856	-.099	.125	-.1083	-.100	.110	-.535	.107	.123
.550												
.600	-.858	-.098	-.129	-.642	-.062	-.099	-.487	-.074	.100	-.452	-.076	.123
.650												
.700	-.301	-.036	-.041	-.250	-.004	-.057	-.356	-.011	.042	-.368	.039	.076
.750												
.800	-.140	.037	.053	-.088	.053	.048	-.204	.015	.010	-.295	.029	.055
.850	.042	.053	.103	.048	.116	.095	-.063	.020	.015	-.243	.002	.055
.900	.120			.111			-.000			-.217		
.950												
1.000	.141	.053	.152	.131	-.015	.111	.036	-.100	-.011	-.154	-.368	.170
.010												
.025												
.050												
.075												
.100												
.150												
.200												
.250												
.300												
.350												
.400												
.450												
.500												
.550												
.600												
.650												
.700												
.750												
.800												
.850												
.900												
.950												
1.000												

TABLE 7.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.318	.195	-.121	-.736	.555	.169	-.941	.668	.309	-1.139	.755	.410
.025	-.523	-.222	-.388	-.889	.330	.024	-1.217	.481	.155	-1.351	.590	.238
.050	-.491	-.408	-.388	-.934	.017	-.161	-.967	.225	-.031	-1.082	.347	.059
.075	-.587			-.870			-.992			-1.409		
.100	-.548	-.740	-.388	-.870	-.308	.191	-.967	.166	-.070	-1.325	.012	.001
.150	-.510	-.728	-.440	-.723	.417	-.244	-1.056	-.288	-.230	-.896	-.191	-.063
.200	-.536	-.568	-.414	-.704	-.391	-.244	-.980	-.300	-.159	-.896	-.198	-.082
.250	-.515	-.523	-.408	-.666	-.347	-.219	-.858	-.320	-.147	-.922	-.255	-.076
.300	-.510	-.408	-.420	-.629	-.295	-.187	-.704	-.236	-.127	-.749	-.178	-.063
.350	-.606			-.647			-.813			-.742		
.400	-.595	-.267	-.222	-.628	-.193	-.142	-.710	.153	-.102	-.704	-.114	-.057
.450				-.589			-.621	-.108	-.095	-.640	-.082	-.063
.500	-.568	-.195	-.184	-.589	-.136	-.123	-.621	-.108	-.095	-.640	-.082	-.063
.550				-.532			-.493	-.063	-.057	-.524	-.031	-.031
.600	-.420	-.126	-.107	-.532	-.078	-.059	-.493	-.063	-.057	-.524	-.031	-.031
.650				-.423			-.429	-.018	-.006	-.422	.007	.014
.700	-.401	-.062	-.056	-.423	-.021	-.021	-.429	-.018	-.006	-.422	.007	.014
.750				-.264			-.256	.039	.039	-.242	-.018	.059
.800	-.260	.015	.015	-.264	.037	.037	-.256	.039	.039	-.242	-.018	.059
.850	-.024	.098	.092	-.014	.100	.107	-.018	.097	.103	.001	.110	.110
.900	-.156			.075			.123			.116		
.950				.171			.155			.142		
1.000	.198	.194	.188	.171	.164	.171	.155	.155	.155	.142	.123	.129
Outboard station												
.010	-.314	-.303	-.156	-.696	.295	.194	-.915	.509	.369	-1.037	.664	.470
.025	-.661	-.404	-.391	-.837	.049	-.047	-1.250	.258	.058	-1.379	.407	.130
.050	-.520	-.565	-.404	-.826	-.124	-.150	-1.334	.084	-.019	-1.553	.220	.065
.075	.279			-.426			-.883			-1.101		
.100	-.552	-.706	-.443	-.831	.407	-.233	-1.005	.219	-.109	-1.417	-.057	-.032
.150	-.468	-.752	-.378	-.657	-.394	-.195	-.754	-.244	-.103	-.818	-.154	-.025
.200	-.468	-.629	-.314	-.657	-.375	-.182	-.754	-.244	-.109	-.779	-.167	-.038
.250	-.533	-.430	-.288	-.656	-.285	-.176	-.780	-.212	-.096	-.824	-.154	-.038
.300	-.565	-.430	-.282	-.670	-.278	-.169	-.747	-.212	-.109	-.799	-.154	-.038
.350	-.552			-.683			-.741			-.792		
.400	-.546	-.275	-.211	-.651	-.182	-.131	-.696	.141	-.090	-.734	-.122	-.038
.450				-.587			-.605	-.077	-.070	-.637	-.057	-.038
.500	-.507	-.159	-.146	-.587	-.105	-.098	-.605	-.077	-.070	-.637	-.057	-.038
.550				-.529			-.522	-.019	-.019	-.534	.001	-.019
.600	-.468	-.127	-.075	-.529	-.034	-.041	-.522	-.019	-.019	-.534	.001	-.019
.650				-.388			-.380	.033	.026	-.386	.039	.033
.700	-.372	-.030	-.056	-.388	.004	.011	-.380	.033	.026	-.386	.039	.033
.750				-.214			-.199	.078	.065	-.199	.091	.091
.800	-.217	.034	.040	-.214	.017	.056	-.199	.078	.065	-.199	.091	.091
.850	-.088	.111	-.011	.011	.125	-.015	.020	.071	.046	.117	.110	.136
.900				.133			.071			.117		
.950	.111			-.015			.136			.136		
1.000	.143	.073	.131	-.015	.081	.114	.136	.046	.058	.136	.033	.130

TABLE 7.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	-.378	.253	-.140	-.751	.547	.162	-.814	.670	.257	-.586	.757	.352
.025	-.629	.024	-.256	-.867	.356	.012	-.1.060	.483	.132	-.1.178	.596	.224
.050	-.314	-.326	-.396	-.500	.036	-.157	-.1.060	.185	-.107	-.806	.340	.043
.075	-.647	-.833	-.396	-.850	-.506	-.174	-.1.159	-.265	-.090	-.1.271	-.044	.014
.100	-.500	-.792	-.353	-.722	-.477	-.250	-.1.141	-.341	-.154	-.1.295	-.201	-.056
.150	-.577	-.728	-.466	-.751	-.430	-.285	-.1.048	-.394	-.458	-.1.283	-.224	-.085
.200	-.588	-.728	-.466	-.751	-.430	-.285	-.1.048	-.394	-.458	-.1.283	-.224	-.085
.250	-.670	-.753	-.419	-.704	-.378	-.238	-.808	-.324	-.466	-.876	-.108	-.108
.300	-.612	-.437	-.402	-.693	-.326	-.197	-.779	-.324	-.466	-.876	-.108	-.108
.350	-.641	-.612	-.402	-.722	-.326	-.197	-.779	-.324	-.466	-.876	-.108	-.108
.400	-.629	-.291	-.239	-.716	-.197	-.145	-.750	-.166	-.107	-.876	-.108	-.044
.450	-.658	-.227	-.198	-.763	-.145	-.133	-.692	-.107	-.137	-.866	-.067	-.061
.500	-.654	-.151	-.110	-.483	-.087	-.063	-.499	-.055	-.049	-.509	-.021	-.003
.550	-.413	-.070	-.052	-.413	-.028	-.017	-.417	-.014	-.008	-.422	.020	.026
.600	-.256	.012	.012	-.232	.036	.042	-.224	.045	.045	-.242	.087	.067
.650	.000	.088	.088	.018	.106	.111	.021	.109	.109	.020	.131	.131
.700	.152	.187	.181	.152	.176	.176	.138	.167	.161	.136	.183	.166
.750	.181	.187	.181	.170	.176	.176	.156	.167	.161	.148	.183	.166

Outboard station												
.010	-.311	-.211	-.148	-.622	.282	.148	-.744	.478	.308	-.847	.635	.424
.025	-.436	-.335	-.405	-.826	.058	-.082	-.1.067	.220	.050	-.1.153	.417	.113
.050	-.528	-.499	-.446	-.862	-.123	-.170	-.1.232	.067	-.086	-.1.344	.148	.037
.075	-.488	-.821	-.476	-.850	.463	.229	-.1.137	-.250	-.162	-.829	-.080	-.039
.100	-.559	-.757	-.423	-.944	-.428	-.184	-.1.161	-.285	-.115	-.1.315	-.158	-.028
.150	-.481	-.757	-.423	-.944	-.428	-.184	-.1.161	-.285	-.115	-.1.315	-.158	-.028
.200	-.505	-.763	-.347	-.651	-.391	-.182	-.1.008	-.279	-.115	-.1.257	-.191	-.033
.250	-.569	-.599	-.323	-.739	-.299	-.176	-.867	-.232	-.121	-.1.286	-.156	-.033
.300	-.569	-.505	-.311	-.721	-.293	-.176	-.732	-.232	-.121	-.1.239	-.168	-.039
.350	-.610	-.270	-.223	-.739	-.182	-.129	-.779	-.150	-.086	-.1.245	-.110	-.033
.400	-.599	-.176	-.212	-.715	-.100	-.100	-.802	-.074	-.080	-.818	-.039	-.033
.450	-.558	-.106	-.135	-.616	-.030	-.035	-.638	-.015	-.027	-.584	.007	.007
.500	-.493	-.065	.017	-.510	.023	.029	-.532	.032	-.015	-.496	.066	.066
.550	-.393	-.006	.064	-.370	.070	.035	-.373	.079	-.097	-.373	.101	.101
.600	-.223	.012	.070	-.182	.129	.140	-.180	.138	.097	-.191	.148	.086
.650	-.012	.117	.070	.041	.140	.103	-.038	.103	.103	.025	.148	.086
.700	.134	.035	.011	.117	.076	.088	.103	.038	.150	.171	.095	.078

TABLE 7.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.310	.307	-.113	-.616	.562	.110	-.737	.576	.262	-.895	.757	.342
.025	-.690	.070	-.232	-.807	.370	-.006	-.006	.482	.085	-.126	.594	.179
.050	-.210	-.265	-.405	-.353	.061	-.174	-.558	.214	-.071	-.567	.342	.033
.075	-.645			-.869			-.084			-.182		
.100	-.645	-.763	-.405	-.896	-.353	-.219	-.189	-.189	-.099	-.210	-.056	-.028
.150	-.606	-.863	-.573	-.812	-.693	-.280	-.078	-.346	-.172	-.221	-.230	-.079
.200	-.601	-.813	-.589	-.667	-.465	-.297	-.051	-.351	-.189	-.227	-.275	-.118
.250	-.623	-.690	-.539	-.706	-.387	-.275	-.782	-.340	-.167	-.182	-.280	-.135
.300	-.634	-.433	-.366	-.678	-.342	-.219	-.743	-.284	-.144	-.042	-.235	-.079
.350	-.662			-.723			-.799			-.840		
.400	-.612	-.299	-.254	-.745	-.202	-.163	-.832	-.172	-.105	-.868	-.118	-.062
.450												
.500	-.601	-.237	-.215	-.689	-.157	-.191	-.804	-.122	-.105	-.924	-.084	-.084
.550												
.600	-.472	-.159	-.126	-.471	-.095	-.079	-.491	-.077	-.055	-.482	-.045	-.028
.650												
.700	-.511	-.075	-.064	-.398	-.034	-.028	-.396	-.021	-.010	-.381	.005	.011
.750												
.800	-.243	.008	.008	-.219	.039	.039	-.211	.040	.052	-.202	.056	.061
.900	.008	.098	.087	.039	.106	.106	.029	.102	.119	.033	.112	.123
.975	.154			.157			.141			.140		
1.000	.176	.182	.176	.173	.173	.151	.183	.158	.158	.157	.162	.162
Outboard station												
.010	-.291	-.171	-.151	-.565	.272	.157	-.673	.476	.294	-.762	.624	.381
.025	-.606	-.319	-.386	-.841	.077	-.120	-.979	.242	.011	-.1050	.404	.111
.050	-.521	-.448	-.437	-.954	-.125	-.204	-.159	.045	-.096	-.1235	.224	-.018
.075	.665			.015			-.439			-.661		
.100	-.639	.797	-.516	-.819	-.520	-.266	-.1064	-.298	-.146	-.168	-.120	-.080
.150	-.471	.690	-.454	-.926	-.447	-.216	-.103	-.326	-.146	-.1230	-.193	-.052
.200	-.527	.757	-.364	-.751	-.402	-.204	-.024	-.310	-.130	-.185	-.216	-.063
.250	-.622	.690	-.358	-.779	-.362	-.199	-.053	-.253	-.118	-.1213	-.187	-.063
.300	-.606	.628	-.347	-.768	-.317	-.204	-.1047	-.253	-.118	-.1202	-.187	-.063
.350	-.651			-.785			-.1047			-.1202		
.400	-.645	-.353	-.240	-.745	-.210	-.137	-.1036	-.163	-.090	-.1213	-.109	-.047
.450												
.500	-.600	-.162	-.184	-.650	-.103	-.125	-.675	-.079	-.079	-.796	-.041	-.041
.550												
.600	-.516	-.077	-.100	-.503	-.035	-.047	-.490	-.068	-.028	-.492	.010	-.007
.650												
.700	-.375	-.015	-.004	-.357	.015	.021	-.349	.034	.034	-.311	.061	.066
.750												
.800	-.195	.041	.086	-.221	-.013	.077	-.163	.079	.107	-.159	.106	.128
.900	.024	.125	.125	.010	.094	.100	.045	.146	.158	.032	.179	.179
.975	.148			.139			.152			.162		
1.000	.153	.108	.119	.089	.077	.122	.163	.073	.174	.184	.134	.184

TABLE 7.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.292	.338	-.109	-.530	.579	.120	-.673	.687	.251	-.801	.747	.294
.025	-.529	.143	-.217	-.740	.352	-.014	-.905	.507	.108	-.985	.589	.168
.050	-.077	-.212	-.497	-.229	.078	-.234	-.371	.222	-.064	-.510	.325	-.015
.075	-.567	-.567	-.843	-.843	-.385	.207	-.1007	-.183	-.091	-.1092	-.090	-.026
.100	-.551	-.701	-.443	-.864	-.525	.288	-.1013	-.420	-.177	-.1125	-.279	-.117
.150	-.557	-.852	-.545	-.832	-.638	.325	-.1002	-.431	-.177	-.1125	-.304	.139
.200	-.594	-.836	-.642	-.794	-.638	.325	-.1024	-.431	-.177	-.1125	-.304	.139
.250	-.604	-.830	-.691	-.729	-.478	.320	-.932	.339	.188	-.1119	-.279	-.168
.300	-.599	-.701	-.432	-.586	-.369	.283	-.808	-.274	.134	-.1098	-.246	-.106
.350	-.637	-.740	-.440	-.566	-.224	.340	-.781	.156	.113	-.1085	-.149	-.096
.400	-.663	-.287	-.255	-.762	-.224	.340	-.819	.156	.113	-.893	-.106	-.096
.450	-.685	-.239	-.217	-.821	-.229	.207	-.867	.118	.102	-.931	-.106	-.096
.500	-.577	-.163	-.126	-.713	-.111	.089	-.786	-.059	-.042	-.780	-.047	-.047
.550	-.405	-.077	-.061	-.385	-.041	.024	-.334	-.005	.006	-.333	-.009	.001
.600	-.222	.014	.009	-.186	.029	.035	-.156	.055	.065	-.155	.050	.050
.650	-.025	.100	.089	.040	.099	.105	-.049	.125	.135	.039	.114	.109
.700	.149	.181	.175	.142	.164	.164	.152	.173	.173	.136	.168	.147
.750	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
.800	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
.850	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
.900	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
.950	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
1.000	.185	.181	.175	.159	.164	.164	.152	.173	.173	.136	.168	.147
Outboard station												
.010	-.252	-.097	-.130	-.476	.266	.136	-.576	.460	.271	-.678	.571	.324
.025	-.561	-.263	-.344	-.784	.055	-.118	-.890	.243	-.023	-.965	.347	.032
.050	-.501	-.393	-.463	-.898	-.113	-.210	-.1048	.048	-.688	-.1133	.162	-.044
.075	.884	-.696	-.528	.207	.540	-.297	-.218	-.310	.147	-.510	-.212	-.119
.100	.615	-.599	-.474	.752	-.476	-.243	-.1004	-.348	.131	-.1095	-.271	-.092
.150	.637	-.637	-.393	-.844	-.454	-.221	-.955	-.332	.120	-.1100	-.277	-.109
.200	.523	-.637	-.377	-.871	-.351	-.216	-.999	-.240	.115	-.1128	-.228	-.109
.250	.647	-.653	-.366	-.871	-.329	-.216	-.1004	-.256	.115	-.1133	-.233	-.109
.300	.647	-.637	-.366	-.871	-.329	-.216	-.1004	-.256	.115	-.1133	-.233	-.109
.350	.691	-.463	-.280	-.875	-.308	-.140	-.1031	-.142	.077	-.1144	-.157	-.098
.400	.707	-.463	-.280	-.811	-.308	-.140	-.1031	-.142	.077	-.1144	-.157	-.098
.450	.804	-.182	-.166	-.898	-.134	-.113	-.1118	-.099	-.061	-.667	-.076	-.087
.500	.550	-.063	-.085	-.497	-.042	-.042	-.522	-.012	-.050	-.553	-.027	-.060
.550	.650	.013	.002	-.302	.006	.023	-.299	.053	.053	-.407	.005	-.000
.600	.750	.067	.045	-.140	.066	.088	-.126	.097	.129	-.222	.054	.054
.650	.800	.126	.045	-.050	.120	.142	.059	.173	.183	-.082	.119	.059
.700	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
.750	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
.800	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
.850	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
.900	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
.950	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033
1.000	.895	.137	.099	.115	.115	.173	.173	.173	.205	-.027	-.065	-.033

TABLE 7.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Concluded

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.258	.372	-.136	-.484	.576	.087	-.509	.671	.212	-.729	.724	.266
.025	-.460	.184	-.211	-.691	.393	.017	-.842	.488	.064	-.900	.562	.123
.050	.065	-.154	-.507	-.032	.097	-.214	-.215	.214	-.091	-.363	.319	-.058
.075	-.533	.637	.507	-.780	-.359	.224	.919	.226	.184	-1.003	.120	-.146
.100	-.595	.798	.512	-.816	.510	.468	.935	.412	.215	-1.055	.347	-.171
.150	-.507	.845	.585	-.795	.593	.484	.956	.526	.252	-1.075	.471	-.208
.200	-.580	.819	.653	-.712	.660	.390	.940	.505	.241	-1.050	.430	-.208
.250	-.564	.876	.580	-.728	.463	.276	.919	.366	.200	-1.045	.419	-.192
.300	-.564	.876	.580	-.728	.463	.276	.919	.366	.200	-1.045	.419	-.192
.350	-.621	.856	.546	-.692	.421	.230	.821	.215	.153	-1.075	.213	-.156
.400	-.653	.856	.546	-.692	.421	.230	.821	.215	.153	-1.075	.213	-.156
.450	-.720	.268	-.278	-.801	-.193	.250	.873	.174	.159	-.915	.177	-.182
.500	.550	.180	-.148	-.795	-.147	.183	.873	.117	.112	-.843	.130	-.166
.600	.704	-.091	-.076	-.349	-.043	.030	.360	.050	.039	-.373	.084	-.068
.700	-.528	-.003	-.013	-.131	.030	.030	.184	.023	.023	-.228	.032	-.022
.750	-.169	.044	.085	.045	.102	.030	.039	.090	.085	-.042	.035	.020
.800	.137	.163	.158	.128	.160	.139	.043	.126	.090	.040	.014	-.022
.900	.137	.163	.158	.128	.160	.139	.043	.126	.090	.040	.014	-.022
.975	.137	.163	.158	.128	.160	.139	.043	.126	.090	.040	.014	-.022
1.000	.158	.163	.158	.149	.160	.139	.080	.126	.090	.066	.014	-.022
Outboard station												
.010	-.230	.100	-.156	-.410	.260	.109	-.505	.419	.193	-.569	.526	.270
.025	-.507	.224	-.376	-.693	.070	-.139	-.797	.208	.036	-.866	.330	.002
.050	-.454	.386	-.465	-.849	-.118	-.217	-.958	.015	-.146	-1.021	.137	-.112
.075	1.114	.627	.522	.430	-.577	.327	.036	.401	.203	-.346	.258	.206
.100	.548	.522	.569	.687	.802	.269	.975	.479	.187	-.975	.356	.174
.150	.627	.522	.569	.687	.802	.269	.975	.479	.187	-1.021	.356	.174
.200	.696	.548	.538	.802	.614	.228	.885	.437	.177	-1.006	.403	.174
.250	.601	.564	.486	.844	-.394	.222	.942	.364	.177	-1.032	.330	.164
.300	.622	.585	.407	.839	-.332	.222	.942	.364	.177	-1.032	.330	.164
.350	.674	.522	.329	.870	-.207	.149	.979	.323	.182	-1.053	.325	.185
.400	.705	.522	.329	.880	-.207	.149	.979	.323	.182	-1.053	.325	.185
.450	.810	.360	.240	-.922	-.108	.128	-1.026	.203	.135	-.892	.226	.164
.500	.857	.146	.125	-.630	-.050	.071	.744	.104	.114	-.512	.143	.164
.600	-.857	-.031	-.010	-.316	.012	.002	-.458	-.052	-.078	-.429	-.102	-.168
.700	-.292	.058	.074	-.092	.049	.070	-.359	.010	-.026	-.346	.076	-.091
.750	-.135	.136	.136	-.028	.117	.112	-.255	.036	.031	-.289	.060	-.055
.800	.047	.136	.136	-.014	.117	.112	-.078	.094	.078	-.232	.029	-.060
.900	.157	.110	.152	.117	.002	.101	-.010	-.099	-.010	-.200	-.434	-.221
1.000	.157	.110	.152	.117	.002	.101	-.010	-.099	-.010	-.169	-.434	-.221

TABLE 8.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.333	.194	-.152	-.703	.547	.152	-.932	.683	.288	-1.135	.753	.400
.025	-.487	-.019	-.243	-.914	.228	.021	-1.207	.489	.162	-1.215	.588	.261
.050	-.586	-.394	-.384	-.754	.014	-.132	-.868	.201	-.023	-.993	.370	.093
.075	-.545			-.308			-.990			-1.313		
.100	-.545	-.679	-.416	.792	-.319	-.171	-.551	-.139	-.094	-.974	-.016	.010
.150	-.512	-.647	-.429	.920	-.389	-.220	-.894	.267	-.139	-.842	-.164	-.035
.200	-.512	-.545	-.461	.996	-.376	-.235	-.887	.331	-.131	-.864	-.190	-.061
.250	-.506	-.455	-.539	.792	-.331	-.197	-.740	.267	-.132	-.788	-.190	-.170
.300	-.480	-.378	-.269	.620	-.280	-.158	-.734	.215	-.094	-.723	-.151	-.035
.350	-.512			.671			-.747			-.723		
.400	-.506	-.220	-.186	.620	-.158	-.114	-.657	-.119	-.075	-.678	-.067	-.029
.450												
.500	-.453	-.186	-.160	.575	-.120	-.107	-.606	-.087	-.075	-.614	-.061	-.029
.550												
.600	-.404	-.121	-.102	.368	-.082	-.069	-.491	-.055	-.043	-.511	-.016	-.003
.650												
.700	-.384	-.064	-.057	.415	-.037	-.024	-.446	-.017	-.011	-.402	-.016	.023
.750												
.800	-.250	.020	.020	.255	.027	.034	-.247	.534	.047	-.222	.061	.068
.900	-.019	.103	.057	-.011	.104	.104	-.009	.105	.111	-.023	.113	.125
.975	-.137			.142			.137			-.132		
1.000	-.195	.155	.153	.168	.168	.168	.156	.156	.162	.151	.158	.158
Outboard station												
.010	-.271	-.266	-.197	-.649	.308	.192	-.854	.525	.339	-.567	.662	.452
.025	-.580	-.400	-.255	-.1010	.021	-.044	-1.234	.311	.047	-1.336	.424	.171
.050	-.484	-.574	-.415	-.785	-.128	-.140	-1.305	.060	-.030	-1.485	.275	.074
.075	-.006			-.488			-.829			-1.032		
.100	-.509	.716	-.438	-.804	-.411	-.186	-.944	-.191	-.095	-1.362	-.015	.003
.150	-.484	-.774	-.342	.701	-.353	-.160	-.790	.224	-.082	-.785	-.094	.010
.200	-.497	-.542	-.280	.584	-.308	-.160	-.790	.224	-.095	-.812	-.113	.010
.250	-.509	-.574	-.284	.584	-.250	-.160	-.784	.191	-.095	-.806	-.107	.016
.300	-.503	-.516	-.264	.562	-.237	-.153	-.745	.185	-.095	-.786	-.107	.023
.350	-.574			.688			-.751			-.786		
.400	-.520	-.219	-.174	.543	-.166	-.108	-.694	-.127	-.075	-.715	-.075	.010
.450												
.500	-.497	-.129	-.122	.572	-.082	-.082	-.603	-.063	-.056	-.611	-.016	-.010
.550												
.600	-.445	-.064	-.056	.508	-.031	-.031	-.513	-.017	-.017	-.514	.016	.016
.650												
.700	-.355	-.006	.013	.379	-.057	-.037	-.378	.028	.028	-.366	.061	.061
.750												
.800	-.219	-.006	.071	.211	.040	.046	-.158	.079	.073	-.185	.087	.107
.900	-.012	-.229	.110	.008	.111	.117	.068	.118	.118	-.029	.145	.152
.975	-.155			.111			.111			-.019		
1.000	-.174	.135	.168	.117	.014	.143	.124	.066	.118	-.113	.055	.145

TABLE 8.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.306	.231	-.156	-.774	.546	.106	-.806	.677	.271	-.662	.755	.381
.025	-.328	.044	-.210	-.896	.342	.004	-.1.040	.489	.133	-1.120	.587	.213
.050	-.358	-.341	-.400	-.540	.021	-.224	-.619	.221	-.048	-.752	.348	.061
.075	-.593		-.832	-.832			-1.115			-1.261		
.100	-.516	-.750	-.417	-.896	-.347	-.189	-1.127	-.182	-.088	-1.284	-.027	-.009
.150	-.375	-.768	-.487	-.739	-.458	-.265	-1.698	-.316	-.147	-1.255	-.190	-.079
.200	-.610	-.698	-.458	-.762	-.435	-.422	-.765	-.456	-.398	-1.255	-.226	-.109
.250	-.546	-.517	-.386	-.698	-.376	-.318	-.824	-.298	-.141	-.752	-.208	-.095
.300	-.622	-.422	-.300	-.703	-.300	-.178	-.812	-.240	-.112	-.793	-.179	-.050
.350	-.910		-.207	-.750			-.906			-.845		
.400	-.569	-.260	-.207	-.698	-.172	-.119	-.765	-.129	-.077	-.628	-.079	-.027
.450												
.500	-.558	-.213	-.189	-.756	-.137	-.125	-.695	-.094	-.088	-.641	-.050	-.038
.550												
.600	-.453	-.146	-.125	-.487	-.096	-.078	-.608	-.053	-.048	-.489	-.027	-.015
.650												
.700	-.417	-.084	-.078	-.417	-.049	-.043	-.415	-.024	-.018	-.401	.008	.008
.750												
.800	-.354	.009	.005	-.236	.027	.032	-.228	.040	.046	-.214	.061	.067
.850	-.002	.057	.051	.015	.103	.102	.022	.104	17.621	.032	.125	.125
.900	-.161			.144			.133			.137		
.975	-.185	-.165	-.185	.167	.173	.173	.151	.168	.157	.155	.166	.160
1.000												
Outboard station												
.010	-.262	-.226	-.287	-.568	.273	.146	-.712	.493	.295	-.793	.654	.406
.025	-.309	-.356	-.380	-.920	.073	-.097	-1.047	.245	.039	-1.134	.413	.125
.050	-.457	-.505	-.405	-.885	-.133	-.168	-1.217	.080	-.055	-1.287	.231	.049
.075	.156			-.256			-.577			-.769		
.100	-.338	-.773	-.450	-.832	-.479	-.221	-1.059	-.237	-.113	-1.222	-.099	-.028
.150	-.302	-.762	-.357	-.938	-.415	-.180	-1.123	-.260	-.084	-1.287	-.134	-.022
.200	-.544	-.750	-.327	-.715	-.362	-.180	-1.035	-.254	-.102	-1.257	-.163	-.040
.250	-.556	-.538	-.315	-.756	-.297	-.174	-.806	-.201	-.102	-1.228	-.128	-.040
.300	-.556	-.386	-.257	-.732	-.256	-.174	-.724	-.201	-.102	-1.199	-.134	-.046
.350	-.632			-.762			-.812			-.881		
.400	-.597	-.233	-.186	-.720	-.180	-.109	-.813	-.137	-.078	-.663	-.081	-.022
.450												
.500	-.350	-.145	-.145	-.526	-.097	-.092	-.630	-.055	-.061	-.593	-.016	-.016
.550												
.600	-.497	-.080	-.074	-.521	-.039	-.044	-.536	-.014	-.020	-.499	.019	.013
.650												
.700	-.397	-.015	.008	-.374	.014	.020	-.372	.027	.027	-.363	.066	.066
.750												
.800	-.203	.044	.075	-.180	.067	.073	-.184	.080	.092	-.181	.102	.025
.850	.014	.126	.126	.032	.126	.138	.016	.127	.063	.037	.154	.154
.900	.155			.138			.127			.143		
.975	.167	.126	.167	.149	.073	.155	.133	.069	.133	.166	.090	.078
1.000												

TABLE 8.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.275	-.260	-.153	-.423	.556	.128	-.712	.663	.235	-.887	.751	.329
.025	-.548	-.072	-.251	-.807	.336	-.017	-.954	.481	.134	-1.043	.582	-.206
.050	-.505	-.268	-.380	-.382	.050	-.214	-.404	.195	-.090	-.589	.335	-.055
.075	-.604			-.859			-1.033			-1.172		
.100	-.586	-.800	-.397	-.385	.348	-.214	-1.082	-.191	-.118	-1.206	-.040	-.018
.150	-.610	-.828	-.520	-.807	-.527	-.275	-1.082	-.343	-.175	-1.184	-.242	-.085
.200	-.610	-.828	-.498	-.584	-.527	-.252	-1.045	-.359	-.186	-1.223	-.259	-.107
.250	-.521	-.587	-.455	-.718	-.387	-.264	-.757	-.326	-.163	-1.172	-.270	-.085
.300	-.543	-.414	-.230	-.595	-.326	-.202	-.741	-.270	-.124	-1.111	-.197	-.057
.350	-.592	-.592		-.740			-.819			-.847		
.400	-.521	-.268	-.218	-.763	-.174	-.135	-.819	-.135	-.085	-.870	-.085	-.035
.450												
.500	-.604	-.223	-.201	-.579	-.146	-.141	-.836	-.107	-.102	-.915	-.074	-.046
.550												
.600	-.775	-.162	-.134	-.477	-.107	-.050	-.494	-.068	-.062	-.472	-.035	-.023
.650												
.700	-.415	-.084	-.078	-.404	-.051	-.045	-.399	-.034	-.023	-.376	.005	.005
.750												
.800	-.312	.006	.006	-.214	.027	.027	-.214	.033	.044	-.192	.066	.066
.850	-.012	.056	.050	.033	.100	.100	.027	.100	.111	.044	.122	.134
.900	-.152			.145			.139			.150		
.950												
1.000	-.180	.191	.190	.158	.173	.162	.151	.162	.162	.156	.167	.162
Outboard station												
.010	-.243	-.174	-.221	-.315	.265	.102	-.622	.482	.265	-.713	.617	.360
.025	-.575	-.322	-.384	-.336	.043	-.120	-.949	.229	.025	-1.040	.392	.127
.050	-.485	-.480	-.418	-.315	-.132	-.188	-1.107	.054	-.081	-1.192	.189	-.032
.075	-.345			-.109			-.335			-.466		
.100	-.575	.795	-.468	-.769	.537	.239	-.983	.278	.137	-1.124	-.121	-.036
.150	-.502	-.623	-.418	-.721	-.459	-.216	-1.056	.301	.098	-1.181	-.205	-.025
.200	-.556	-.829	-.335	-.774	-.380	-.199	-1.028	-.295	-.109	-1.186	-.194	-.047
.250	-.352	-.649	-.332	-.763	-.301	-.194	-1.034	-.233	-.104	-1.181	-.155	-.042
.300	-.567	-.491	-.328	-.780	-.278	-.199	-1.034	-.216	-.109	-1.181	-.149	-.047
.350	-.571			-.847			-1.056			-1.192		
.400	-.646	-.237	-.156	-.774	-.182	-.126	-1.017	-.137	-.081	-1.203	-.098	-.030
.450												
.500	-.592	-.136	-.175	-.539	-.098	-.103	-.611	-.059	-.070	-.797	-.036	-.030
.550												
.600	-.508	-.080	-.074	-.315	-.041	-.052	-.487	-.030	-.019	-.482	.025	.020
.650												
.700	-.378	-.018	.010	-.357	.003	-.041	-.352	.020	.020	-.318	.065	.065
.750												
.800	-.198	.044	.072	-.171	.060	.066	-.166	.071	.004	-.155	.122	.127
.850	-.027	.122	.123	-.043	.128	.123	.043	.155	.150	.043	.178	.178
.900												
.950	-.151			.133			.150			.189		
1.000	-.168	.122	.168	.156	.060	.150	.139	.065	.167		.150	.195

TABLE 8.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.262	.316	-.183	-.539	.572	.064	-.677	.675	.204	-.799	.754	.367
.025	-.286	.120	-.225	-.764	.284	-.021	-.876	.486	.088	-.955	.584	.175
.050	-.305	-.220	-.408	-.231	.066	-.172	-.235	.223	-.068	-.444	.331	.040
.075	-.597	-.597	-.797	-.361	-.355	.215	-.1016	.187	-.133	-.1117	-.267	-.040
.100	-.351	-.158	-.454	-.856	-.549	-.201	-.985	.370	-.203	-.1117	-.245	-.034
.150	-.357	-.756	-.573	-.635	-.301	-.011	-.439	-.219	-.219	-.1133	-.283	-.121
.200	-.370	-.988	-.681	.737	-.419	-.279	-.573	.349	-.208	-.1122	-.277	-.089
.250	-.380	-.950	-.546	-.383	-.344	-.215	-.801	-.284	-.165	-.1095	-.218	-.073
.300	-.375	-.527	-.365	-.727	-.199	-.139	-.774	-.235	-.112	-.685	-.100	-.046
.350	-.516	-.268	-.224	-.759	-.319	-.166	-.887	-.122	-.122	-.528	-.078	-.052
.400	-.545	-.236	-.225	-.727	-.112	-.056	-.861	-.182	-.112	-.767	-.046	-.030
.450	-.557	-.263	-.155	-.365	-.058	-.042	-.238	-.042	-.042	-.215	-.003	.003
.500	-.510	-.096	-.086	-.188	.022	.032	-.165	.029	.034	-.153	.051	.067
.550	-.405	-.004	-.065	-.044	.103	.109	-.142	.045	.104	-.142	.121	.132
.600	-.220	-.028	-.088	-.157	.163	.168	-.152	.152	.147	-.164	.169	.175
.650	-.175	.179	.174									
.700												
.750												
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010	-.205	-.156	-.217	-.645	.291	.115	-.553	.467	.233	-.627	.598	.331
.025	-.325	-.252	-.456	-.775	.059	-.136	-.894	.217	.065	-.536	.301	.077
.050	-.485	-.445	-.495	-.912	-.120	-.207	-.1035	.049	-.091	-.1037	.185	.001
.075	-.304	-.354	-.483	-.048	-.532	.256	-.152	.331	-.132	-.454	.151	-.059
.100	-.355	-.716	-.483	-.368	-.505	-.222	-.954	.379	-.141	-.1022	-.226	-.042
.150	-.457	-.781	-.472	-.874	-.445	.212	-.992	.331	-.146	-.1103	-.210	-.089
.200	-.554	-.781	-.404	-.957	-.321	-.255	-.996	.255	-.152	-.1109	-.184	-.064
.250	-.513	-.726	-.380	-.957	-.283	.207	-.1013	.244	-.169	-.1103	-.183	-.059
.300	-.318	-.634	-.363	-.374	-.362	.201	-.1057	.157	-.103	-.1141	-.118	-.059
.350	-.557	-.330	-.235	-.814	-.201	-.131	-.1067	.157	-.103	-.1141	-.118	-.059
.400	-.555	-.330	-.235	-.814	-.201	-.131	-.1067	.157	-.103	-.1141	-.118	-.059
.450	-.744	-.141	.163	-.512	-.098	-.109	-.1143	.070	-.091	-.769	-.042	-.059
.500	-.504	-.071	-.082	-.505	-.044	-.077	-.536	.027	-.038	-.557	.001	-.021
.550	-.365	-.017	-.006	-.310	.005	.016	-.320	.011	.011	-.394	.044	.044
.600	-.190	.043	.065	-.142	.064	.081	-.146	.070	.081	-.210	.071	.088
.650	-.232	.114	.168	-.048	.135	.140	-.045	.135	.141	-.075	.158	.071
.700	-.140	.114	.157	.135	.064	.157	.163	.103	.173	.032	.055	.093
.750												
.800												
.850												
.900												
.950												
1.000												

TABLE 8.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Concluded

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.238	-.353	-.137	-.492	-.564	-.085	-.585	-.669	-.202	-.702	-.739	-.273
.025	-.456	-.168	-.217	-.721	-.376	-.030	-.819	-.891	-.085	-.863	-.575	-.154
.050	-.352	-.170	-.508	-.366	.080	-.180	-.050	.210	-.091	-.282	-.320	-.012
.075	-.325		-.784	-.784			-.923		-.995			
.100	-.512	-.590	-.436	-.320	-.357	-.222	-.938	-.195	-.133	-.100	-.085	
.150	-.508	-.763	-.525	-.836	-.508	-.378	-.544	-.382	-.216	-.100	-.303	-.240
.200	-.360	-.878	-.612	-.804	-.633	-.357	-.959	-.517	-.237	-.100	-.380	-.178
.250	-.534	-.836	-.659	-.737	-.650	-.342	-.944	-.455	-.299	-.100	-.375	-.173
.300	-.550	-.872	-.482	-.747	-.409	-.264	-.923	-.304	-.206	-.100	-.277	-.136
.350	-.512			-.715			-.819			-.100		
.400	-.538	-.316	-.238	-.747	-.222	-.170	-.824	-.159	-.221	-.100	-.131	-.090
.450												
.500	-.706	-.254	-.243	-.820	-.186	-.196	-.871	-.159	-.174	-.894	-.183	-.168
.550												
.600	-.596	-.175	-.160	-.820	-.139	-.206	-.871	-.133	-.263	-.874	-.173	-.261
.650												
.700	-.358	-.097	-.097	-.388	-.076	-.061	-.356	-.060	-.065	-.375	-.105	-.064
.750												
.800	-.170	-.007	-.005	-.160	-.007	-.007	-.180	.013	.013	-.209	.009	-.074
.850												
.900	.048	.055	.050	.007	.085	.080	.013	.091	.080	.074	.066	.056
.950	.157			.095			.080			.035		
1.000	.178	.183	.166	.116	.147	.132	.106	.132	.091	.071	.087	.050
Outboard station												
.010	-.185	-.132	-.206	-.387	-.258	-.057	-.470	-.428	-.183	-.526	-.554	-.278
.025	-.492	-.231	-.514	-.891	-.021	-.173	-.784	.199	-.062	-.834	-.336	-.009
.050	-.431	-.409	-.424	-.795	-.141	-.230	-.930	.000	-.109	-.985	-.138	-.076
.075	.701			.251			.032			-.291		
.100	-.534	-.728	-.466	-.365	-.549	-.288	-.845	-.402	-.198	-.517	-.259	-.134
.150	-.518	-.665	-.535	-.311	-.753	-.277	-.894	-.418	-.172	-.995	-.301	-.103
.200	-.308	-.581	-.461	-.827	-.685	-.241	-.909	-.413	-.172	-.006	-.327	-.134
.250	-.587	-.655	-.450	-.848	-.377	-.225	-.925	-.598	-.172	-.016	-.270	-.129
.300	-.823	-.655	-.277	-.849	-.288	-.236	-.941	-.266	-.183	-.102	-.254	-.144
.350	-.856			-.926			-.1009			-.1063		
.400	-.596	-.458	-.262	-.916	-.215	-.152	-.1005	-.183	-.125	-.1063	-.176	-.103
.450												
.500	-.801	-.225	-.173	-.368	-.110	-.121	-.1045	-.078	-.125	-.531	-.092	-.113
.550												
.600	-.327	-.065	-.085	-.576	-.068	-.147	-.496	-.068	-.078	-.447	-.066	-.129
.650												
.700	-.305	.010	-.001	-.289	-.016	-.011	-.381	-.015	-.041	-.374	-.050	-.061
.750												
.800	-.147	.057	.068	-.136	.026	.052	-.245	.027	.021	-.296	-.040	-.103
.850												
.900	.036	.125	.115	-.011	.054	.054	-.094	.095	.047	-.249	-.003	-.071
.950	.141			.042			.004			-.157		
1.000	.152	.130	.156	.083	-.021	-.059	.037	-.041	-.005	-.144	-.353	-.170

TABLE 9.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	.040	.044	-.201	-.056	.504	.145	-.132	.652	.310	-.119	.770	.411
.025	-.248	-.216	-.236	-.529	.245	.021	-.772	.437	.159	-.855	.565	.290
.050	-.127	-.607	-.389	-.773	-.126	-.120	-.663	.105	.009	-.708	.245	.105
.075	-.524			-.913			-.990			-1.348		
.100	-.549	-.697	-.415	-.824	-.312	.184	-.919	.158	-.074	-.926	-.036	.009
.150	-.511	-.575	-.517	-.741	-.350	.216	-.951	.247	-.119	-.970	-.138	-.036
.200	-.511	-.472	-.357	-.741	-.318	.190	-.855	.222	-.100	-.958	-.138	-.042
.250	-.511	-.383	-.280	-.741	-.260	.152	-.996	.183	-.081	-.823	-.126	-.023
.300	-.517	-.312	-.248	-.651	-.209	.126	-.745	.145	-.068	-.823	-.119	-.030
.350	-.543			-.651			-.759			-.894		
.400	-.549	-.229	-.264	-.632	-.145	-.120	-.758	-.054	-.062	-.772	-.062	-.030
.450												
.500	-.524	-.191	-.184	-.593	-.126	-.120	-.525	-.081	-.074	-.644	-.055	-.049
.550	-.440	-.152	-.152	-.504	-.100	-.100	-.484	-.068	-.074	-.484	-.049	-.042
.600												
.650	-.408	-.101	-.088	-.427	-.068	-.056	-.439	-.042	-.042	-.439	-.023	-.017
.700												
.750	-.274	-.011	.001	-.267	.008	.021	-.254	.022	.028	-.247	.034	-.074
.800	-.031	.072	.051	-.024	.079	.098	-.023	.052	.105	-.017	.092	.111
.850	.149			.136			.130			.130		
.900												
.975												
1.000	.181	.168	.187	.162	.149	.168	.156	.137	.162	.162	.130	.156
Outboard station												
.010	-.327	-.559	-.312	-.565	.192	.130	-.758	.463	.332	-.983	.626	.479
.025	-.327	-.727	-.379	-.571	-.056	-.037	-.790	.124	.086	-.989	.363	.169
.050	-.315	-.940	-.385	-.533	-.288	-.134	-.661	-.011	-.004	-1.112	.073	.124
.075	-.469			-.443			-.691			-1.504		
.100	-.456	-.740	-.431	-.810	-.385	.211	-1.073	.191	.101	-1.356	-.056	-.017
.150	-.424	-.572	-.347	-.591	-.327	.153	-.700	.210	.082	-1.356	-.088	.008
.200	-.450	-.443	-.289	-.617	-.288	.134	-.593	.191	.089	-.751	-.107	-.037
.250	-.553	-.327	-.244	-.681	-.224	.124	-.764	.146	.069	-.841	-.088	-.011
.300	-.534	-.282	-.244	-.574	-.198	.121	-.751	.146	-.069	-.803	-.088	-.017
.350				-.662			-.725			-.763		
.400	-.547	-.205	-.173	-.649	-.121	-.114	-.703	-.082	-.056	-.745	-.043	-.017
.450												
.500	-.527	-.154	-.147	-.591	-.076	-.088	-.622	-.037	-.056	-.635	-.017	-.024
.550												
.600	-.476	-.076	-.070	-.520	-.031	-.037	-.532	-.064	-.011	-.539	.015	-.004
.650												
.700	-.398	-.031	-.005	-.410	.015	.015	-.404	.015	.015	-.397	.028	.034
.750												
.800	-.237	.027	.027	-.237	.053	.060	-.230	.073	.066	-.210	.086	.079
.850	-.025	.072	.045	-.024	.079	.098	-.011	.111	.028	-.017	.124	.099
.900				.098			.092			.105		
.975	.111			.143			.124			.060		
1.000	.059	.072	.111		.072	.105		.041	.124		.034	.124

TABLE 9.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(b) $M = 0.750$

x/c	C_p at $\alpha =$											
	-2°			0°			1°			2°		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	.045	.100	-.221	-.030	.507	.088	-.095	.664	.297	-.054	.762	.363
.025	-.230	-.195	-.253	-.567	.238	.045	-.678	.461	.162	-.696	.553	.267
.050	.039	-.668	-.411	-.778	-.106	-.147	-.492	.059	-.019	-.457	.203	.086
.075	-.574	-.720	-.458	-.748	-.381	.211	-1.099	.206	-.106	-1.250	-.077	-.007
.100	-.522	-.767	-.487	-.748	-.410	-.252	-.918	-.322	-.159	-1.256	-.182	-.054
.150	-.574	-.516	-.395	-.725	-.404	.223	-.789	-.270	-.136	-1.233	-.182	-.048
.200	-.574	-.417	-.370	-.748	-.357	.171	-.848	-.223	-.106	-.748	-.147	-.042
.250	-.569	-.358	-.271	-.713	-.235	-.153	-.819	-.171	-.089	-.783	-.147	-.035
.300	-.615	-.756	-.271	-.708	-.708	-.141	-.848	-.112	-.101	-.877	-.095	-.048
.350	-.662	-.247	-.230	-.708	-.165	.141	-.754	-.112	-.101	-.906	-.065	-.060
.400	-.621	-.218	-.206	-.678	-.147	.141	-.696	-.112	-.106	-.690	-.054	-.065
.450	-.458	-.171	-.171	-.579	-.118	-.124	-.603	-.089	-.095	-.538	-.030	-.030
.500	-.423	-.107	-.095	-.433	-.077	-.077	-.433	-.066	-.054	-.416	-.030	-.030
.550	-.259	-.014	-.002	-.252	.005	.010	-.241	.016	.016	-.264	.040	.040
.600	-.008	.068	.086	.005	.080	.092	-.007	.086	.098	-.007	.098	.116
.650	.144	.162	.179	.127	.139	.156	.121	.127	.145	.139	.127	.162
.700	.173			.156			.145			.168		
Outboard station												
.010	-.314	-.461	-.306	-.554	.168	.111	-.783	.436	.288	-.860	.598	.415
.025	-.309	-.661	-.409	-.566	-.102	-.096	-.789	.062	.068	-.871	.327	.133
.050	-.275	-.684	-.432	-.437	-.366	-.114	-.789	-.073	-.026	-.977	.109	.068
.075	-.338	-.255	-.255	-.255	-.255	-.255	-.554	-.273	-.126	-1.259	-.037	-.037
.100	-.532	-.102	-.475	-.889	-.407	.225	-1.083	.273	-.126	-1.218	-.126	-.037
.150	-.426	-.497	-.397	-.631	-.378	-.178	-1.012	-.261	-.096	-1.171	-.173	-.008
.200	-.485	-.426	-.285	-.637	-.308	-.155	-.848	-.237	-.090	-1.130	-.149	-.020
.250	-.555	-.332	-.273	-.754	-.225	-.143	-.760	-.173	-.085	-1.229	-.108	-.025
.300	-.579	-.297	-.238	-.742	-.214	-.131	-.760	-.161	-.102	-1.212	-.102	-.026
.350	-.551	-.591	-.226	-.742	-.126	-.114	-.819	-.162	-.085	-.942	-.069	-.037
.400	-.591	-.215	-.226	-.730	-.090	-.108	-.836	-.162	-.085	-.701	-.037	-.037
.450	-.555	-.168	-.179	-.631	-.090	-.108	-.672	-.049	-.108	-.625	-.014	-.037
.500	-.497	-.091	-.079	-.543	-.038	-.049	-.549	-.014	-.026	-.519	.021	.009
.550	-.426	-.015	-.050	-.396	.004	.004	-.396	.021	.015	-.378	.068	.051
.600	-.226	.026	.003	-.220	.056	.051	-.202	.074	.062	-.220	.092	.085
.650	-.015	.085	.097	-.002	.115	.091	-.002	.121	.109	-.009	.121	.068
.700	.120	.079	.020	.127	.051	.056	.086	.021	.139	.056	.074	.133
.750	.079			.144			.127			.056		
.800												
.850												
.900												
.950												
1.000												

TABLE 9.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$			
	Row A	Row B	Row C		Row A	Row B	Row C		Row A	Row B	Row C	
Inboard station												
.010	.044	.153	-.205		-.021	.500	.111		-.072	.657	.261	
.025	-.225	-.152	-.270		-.480	.236	.012		-.615	.415	.141	
.050	.134	-.611	-.404		-.648	-.105	-.156		-.380	.062	-.027	
.075	-.555				-.827				-.1.034			
.100	-.561	-.745	-.527		-.833	-.391	-.234		-.1.062	-.223	.105	
.150	-.600	-.864	-.572		-.738	-.548	-.273		-.1.062	-.424	.161	
.200	-.611	-.555	-.527		-.715	-.385	-.240		-.1.012	-.285	.156	
.250	-.639	-.466	-.359		-.743	-.307	-.195		-.1.167	-.229	.117	
.300	-.617	-.348	-.286		-.721	-.273	.161		-.1.072	-.173	.100	
.350	-.611				-.771				-.816		-.809	
.400	-.645	-.281	-.286		-.799	-.173	-.156		-.844	-.122	-.105	
.450												
.500	-.667	-.303	-.253		-.715	-.161	.145		-.872	-.111	-.117	
.550					-.525	-.133	-.128		-.492	-.100	-.100	
.600	-.611	-.191	-.180									
.650					-.492	-.083	-.072		-.413	-.086	-.055	
.700	-.449	-.124	-.107			-.005	.012		-.229	.012	.029	
.750					-.234	-.012	.056		-.006	.085	.096	
.800	-.258	-.017	.083		.124	.079			.124			
.850	-.139	.072			.146	.135	.163		.152	.124	.152	
.900												
.950												
1.000	.168	.156	.173									
Outboard station												
.010	-.312	-.351	-.290		-.597	.151	.077		-.755	.408	.266	
.025	-.306	-.371	-.365		-.563	-.130	-.107		-.743	.135	.033	
.050	-.244	-.815	-.413		-.423	-.361	-.164		-.732	-.102	-.040	
.075	-.312				-.147				-.423			
.100	-.577	-.135	-.481		-.828	-.496	-.259		-.1.014	.253	.141	
.150	-.408	-.881	-.357		-.834	-.406	-.192		-.1.003	.271	.102	
.200	-.481	-.442	-.301		-.670	-.349	-.164		-.1.014	.283	.096	
.250	-.560	-.312	-.284		-.749	-.259	-.164		-.1.036	.158	.091	
.300	-.583	-.267	-.222		-.789	-.259	.141		-.1.014			
.350	-.599				-.789				-.811	-.051	-.085	
.400	-.639	-.188	-.188		-.817	-.158	-.124					
.450												
.500	-.583	-.154	-.171		-.682	-.096	-.107		-.710	-.051	-.085	
.550					-.558	-.051	-.051		-.513	-.012	-.029	
.600	-.515	-.092	-.064									
.650					-.383	-.012	-.000		-.366	.028	.028	
.700	-.385	-.013	-.036									
.750					-.197	.050	.011		-.186	.056	-.000	
.800	-.205	-.041	.033		.005	.084	.062		.016	.112	.106	
.850	-.002	.105	.105		.078				.123			
.900												
.950	.139	.071	.156		.140	.005	.090		.146	.022	.152	
1.000	.156											

TABLE 9.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.060	.191	-.220	-.004	.523	.100	-.046	.657	.284	-.019	.760	.337
.025	-.209	-.102	-.231	-.500	.238	.023	-.535	.406	.100	-.547	.530	.223
.050	.281	-.533	-.452	-.527	-.134	-.166	-.272	.057	-.062	-.267	.191	.035
.075	-.527	-.813	-.829	-.813	-.425	-.236	-.1009	.256	.153	-.069	.095	.030
.100	-.511	-.716	-.829	-.829	-.533	-.279	-.1009	.350	.261	-.1090	-.235	.122
.150	-.538	-.829	-.829	-.829	-.533	-.279	-.1009	.350	.261	-.1090	-.235	.122
.200	-.544	-.932	-.844	-.716	-.419	-.295	-.1009	.309	.229	-.1101	-.218	-.084
.250	-.571	-.425	-.366	-.721	-.333	-.188	-.887	.245	.137	-.1117	-.240	-.089
.300	-.560	-.317	-.290	-.694	-.258	-.204	-.821	.213	.121	-.1117	-.132	-.046
.350	-.619	-.743	-.743	-.743	-.204	-.166	-.810	.132	.126	-.1096	-.095	.111
.400	-.641	-.780	-.780	-.780	-.171	-.177	-.848	.132	.132	-.0956	-.100	-.078
.450	-.668	-.840	-.840	-.840	-.145	-.139	-.821	.121	.121	-.0848	-.084	-.078
.500	-.479	-.193	-.209	-.737	-.096	-.080	-.347	.062	.062	-.321	-.057	-.041
.550	-.479	-.123	-.118	-.371	-.004	.012	-.218	.008	.013	-.170	.024	.045
.600	-.226	-.015	-.010	-.198	.076	.098	.013	.089	.100	.008	.083	.121
.650	.028	.076	.082	.023	.136	.157	.115	.132	.159	.126	.110	.185
.700	.152	.162	.168	.120	.146	.157	.153	.132	.159	.126	.110	.185
.750	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
.800	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
.850	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
.900	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
.950	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
1.000	.173	.162	.168	.146	.146	.157	.153	.132	.159	.126	.110	.185
Outboard station												
.010	-.299	-.335	-.335	-.575	.141	.063	-.685	.383	.227	-.752	.546	.345
.025	-.299	-.335	-.335	-.575	.141	.063	-.685	.383	.227	-.752	.546	.345
.050	-.229	-.765	-.429	-.434	-.391	-.179	-.692	.058	-.010	-.747	.174	.055
.075	-.261	-.819	-.434	-.434	-.391	-.179	-.692	.058	-.010	-.747	.174	.055
.100	-.586	-.107	-.543	-.760	-.586	-.277	-.286	.351	.172	-.925	.173	-.086
.150	-.402	-.107	-.423	-.798	-.440	-.217	-.941	.359	.129	-.1012	-.200	-.053
.200	-.494	-.668	-.259	-.765	-.369	-.174	-.881	.280	.123	-.1012	-.205	-.053
.250	-.586	-.353	-.294	-.814	-.266	-.169	-.584	.199	.123	-.1088	-.146	-.043
.300	-.592	-.261	-.267	-.819	-.245	-.158	-.1017	.172	.107	-.1104	-.129	-.043
.350	-.624	-.174	-.156	-.781	-.152	-.125	-.1038	.102	.102	-.1104	-.086	-.048
.400	-.684	-.174	-.156	-.819	-.109	-.114	-.1087	.102	.102	-.1142	-.086	-.048
.450	-.662	-.147	-.174	-.938	-.109	-.114	-.1087	.102	.102	-.1142	-.086	-.048
.500	-.505	-.126	-.093	-.521	-.049	-.049	-.529	.042	.037	-.866	-.053	-.053
.550	-.364	-.023	-.023	-.375	-.001	.005	-.307	.001	.012	-.584	.001	-.070
.600	-.180	-.005	-.001	-.165	.021	.010	-.140	.050	.044	-.378	.028	.022
.650	-.021	.108	.086	.016	.092	.108	-.023	.120	.115	-.189	.060	.066
.700	.140	.064	.151	.108	.010	.113	.115	.034	.174	.033	.022	.071
.750	.173	.064	.151	.130	.010	.113	.120	.034	.174	.066	-.059	.131

TABLE 9.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Concluded

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	.051	.251	-.203	.005	.525	.054	-.012	.658	.175	-.021	.751	.315
.025	-.192	-.052	-.265	-.430	.244	-.026	-.468	.404	.108	-.509	.508	.171
.050	.425	-.477	-.415	-.451	-.150	-.207	-.162	.040	-.063	-.104	.150	.036
.075	-.509			-.752			-.883			-.975		
.100	-.540	-.675	-.493	-.788	-.472	-.295	-.925	-.261	-.193	-.101	-.151	-.058
.150	-.540	-.778	-.566	-.793	-.560	-.383	-.946	-.421	-.214	-.001	-.327	-.182
.200	-.540	-.882	-.529	-.793	-.643	-.383	-.946	-.458	-.193	-.027	-.301	-.125
.250	-.540	-.934	-.503	-.772	-.684	-.284	-.940	-.512	-.198	-.033	-.275	-.089
.300	-.566	-.953	-.301	-.762	-.295	-.218	-.904	-.224	-.209	-.103	-.177	-.073
.350	-.633			-.746			-.826			-.106		
.400	-.669	-.317	-.384	-.772	-.244	-.238	-.868	-.167	-.178	-.106	-.198	-.165
.450												
.500	-.737	-.405	-.343	-.855	-.223	-.223	-.894	-.162	-.152	-.824	-.156	-.166
.550												
.600	-.726	-.229	-.218	-.845	-.192	-.197	-.899	-.146	-.152	-.825	-.244	-.141
.650												
.700	-.379	-.146	-.120	-.415	-.181	-.124	-.453	-.146	-.089	-.374	-.161	-.068
.750												
.800	-.182	-.026	-.016	-.197	-.031	-.021	-.204	-.012	-.001	-.249	-.016	.005
.900	.036	.067	.677	-.036	.052	.067	-.043	.061	.082	-.120	.062	.067
.975	.129			.067			.077			.010		
1.000	.150	.150	.160	.119	.104	.104	.144	.108	.129	.051	.057	.062
Outboard station												
.010	-.279	-.273	-.265	-.549	.129	.010	-.640	.354	.184	-.712	.530	.292
.025	-.294	-.482	-.435	-.544	-.153	-.184	-.650	.070	-.040	-.701	.212	.055
.050	-.169	-.712	-.446	-.403	-.414	-.226	-.598	-.191	-.076	-.743	-.044	-.034
.075	-.206			.212			-.102			.754		
.100	-.576	-.119	-.371	-.690	-.659	-.356	-.838	-.442	-.212	-.905	-.284	-.117
.150	-.440	-.119	-.628	-.758	-.721	-.278	-.854	-.415	-.154	-.910	-.310	-.086
.200	-.472	-.1061	-.347	-.748	-.654	-.211	-.812	-.337	-.139	-.905	-.289	-.091
.250	-.592	-.581	-.326	-.831	-.273	-.226	-.917	-.243	-.134	-.994	-.216	-.086
.300	-.633	-.383	-.247	-.836	-.247	-.190	-.943	-.217	-.118	-.109	-.185	-.081
.350	-.675			-.862			-.969			-.1030		
.400	-.707	-.164	-.190	-.825	-.169	-.174	-.1016	-.139	-.118	-.1030	-.112	-.091
.450												
.500	-.821	-.133	-.164	-.577	-.132	-.169	-.1110	-.097	-.118	-.623	-.091	-.101
.550												
.600	-.873	-.101	-.096	-.675	-.096	-.106	-.562	-.092	-.076	-.441	-.086	-.096
.650												
.700	-.836	-.039	-.039	-.315	-.044	-.080	-.368	-.019	-.024	-.295	-.044	-.034
.750												
.800	-.154	.024	.024	-.164	-.007	-.033	-.196	.018	.002	-.248	-.034	-.060
.900	.029	.060	.060	-.054	.045	.035	-.060	.060	.065	-.206	.013	-.039
.975	.118			-.002			.013			-.190		
1.000	.149	.019	.154	.009	-.132	.061	.049	-.081	.101	-.169	-.216	-.013

TABLE 10.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.130	.555	.247	-.215	.712	.375	-.313	.798	.510
.025				-.539	.292	.139	-.842	.489	.239	-.857	.603	.347
.050				-.864	-.053	-.028	-.593	.143	.086	-.595	.296	.168
.075				-.871			-.977			-.1325		
.100				-.845	.232	.143	-.855	-.074	-.042	-.859	.020	.052
.150				-.852	-.315	-.206	-.810	-.196	-.068	-.981	-.012	-.052
.200				-.756	.283	-.135	-.823	-.196	-.062	-.825	-.121	.001
.250				-.634	-.226	-.098	-.714	-.158	-.030	-.793	-.101	.027
.300				-.609	-.162	-.072	-.670	-.106	-.017	-.780	-.063	.027
.350				-.609			-.670			-.793		
.400				-.602	-.111	-.079	-.638	-.062	-.055	-.652	-.031	.001
.450												
.500				-.570	-.104	-.085	-.586	-.062	-.055	-.761	-.031	-.025
.550				-.583	-.079	-.072	-.452	-.042	-.049	-.499	-.018	-.025
.600												
.650												
.700				-.404	-.040	-.028	-.388	-.017	-.017	-.377	.001	.001
.750												
.800				-.245	.035	.049	-.247	.047	.002	-.210	.059	.065
.900				-.002	.113	.132	-.017	.118	.130	.001	.116	.116
.975				.151			.156			.129		
1.000				.186	.170	.196	.188	.162	.181	.187	.155	.155
Outboard station												
.010				-.234	.264	.256	-.391	.517	.417	-.515	.687	.540
.025				-.279	.017	.030	-.442	.279	.157	-.618	.433	.259
.050				-.529	-.189	-.034	-.725	.047	.079	-.947	.233	.162
.075				-.645			-.655			-.831		
.100				-.781	.311	.118	-.938	-.146	-.011	-.1359	-.025	.072
.150				-.684	-.253	-.112	-.764	-.146	-.024	-.818	.052	.052
.200				-.684	-.227	-.099	-.764	-.146	-.043	-.818	-.070	.014
.250				-.671	-.176	-.092	-.758	-.114	-.037	-.818	-.064	.020
.300				-.645	-.157	-.079	-.725	-.107	-.030	-.760	-.057	.020
.350				-.664			-.732			-.766		
.400				-.632	-.086	-.067	-.680	-.056	-.030	-.702	-.019	.007
.450												
.500				-.568	-.054	-.054	-.590	-.024	-.037	-.605	.001	-.006
.550				-.497	-.009	-.009	-.500	.038	.002	-.508	.033	.020
.600												
.650				-.369	.036	.043	-.371	.054	.041	-.367	.065	.059
.700												
.750				-.202	.081	.088	-.198	.392	.086	-.193	.097	.097
.800				-.004	.133	.139	.002	.137	.124	.001	.136	.136
.900				.152			.144			.149		
.975				.184	.113	.190	.176	.099	.163	.181	.110	.162
1.000												

TABLE 10.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.106	.566	.212	-.280	.710	.377	-.224	.891	.434
.025				-.456	.303	.116	-.636	.478	.239	-.668	.600	.323
.050				-.614	-.065	-.194	-.414	.153	.070	-.387	.284	.185
.075				-.765			-1.038			-1.199		
.100				-.719	-.264	-.176	-.997	.268	-.035	-1.164	.010	.039
.150				-.719	-.357	-.188	-.962	-.228	-.076	-1.199	-.347	-.002
.200				-.719	-.503	-.170	-.747	-.403	-.076	-1.071	-.130	-.014
.250				-.666	-.258	-.106	-.805	-.181	-.035	-.726	-.107	.010
.300				-.678	-.176	-.088	-.770	-.123	-.023	-.773	-.066	.021
.350				-.672			-.747			-.837		
.400				-.760	-.123	-.106	-.706	-.076	-.047	-.855	-.031	-.014
.450				-.602	-.123	-.106	-.671	-.076	-.058	-.603	-.037	-.031
.500				-.456	-.088	-.088	-.461	-.082	-.053	-.446	-.031	-.031
.550												
.600				-.392	-.053	-.048	-.379	-.035	-.018	-.364	-.008	-.008
.650												
.700				-.211	.028	.040	-.198	.041	.052	-.189	.062	.062
.750				-.028	.110	.040	.017	.116	.128	.016	.121	.132
.800				.110			.157			.162		
.850				.186	.168	.186	.186	.169	.181	.185	.144	.185
.900												
.950												
1.000												
Outboard station												
.010				-.207	.239	.210	-.347	.493	.387	-.391	.654	.506
.025				-.237	.022	.010	-.412	.233	.151	-.461	.408	.244
.050				-.466	-.213	-.067	-.729	.089	.063	-.802	.209	.150
.075				-.460			-.424			-.555		
.100				-.748	-.372	-.125	-1.004	-.183	-.025	-1.231	-.044	.068
.150				-.730	-.295	-.137	-1.051	-.171	-.025	-1.237	-.068	.038
.200				-.683	-.254	-.119	-.957	-.171	-.048	-1.190	-.085	.003
.250				-.730	-.196	-.108	-.776	-.130	-.042	-1.184	-.068	.003
.300				-.695	-.178	-.090	-.752	-.124	-.036	-1.149	-.068	.015
.350				-.718			-.805			-.790		
.400				-.695	-.090	-.078	-.776	-.054	-.030	-.649	-.015	.003
.450												
.500				-.595	-.061	-.072	-.617	-.030	-.036	-.596	.003	-.015
.550				-.501	-.020	-.020	-.506	.005	.005	-.491	.026	.021
.600												
.650				-.354	.027	.033	-.353	.052	.052	-.344	.079	.068
.700												
.750				-.172	.080	.080	-.177	.093	.087	-.179	.103	.103
.800				.022	.133	.133	.016	.145	.140	.015	.156	.150
.850				.162			.157			.173		
.900				.186	.098	.186	.187	.122	.181	.203	.144	.191
.950												
1.000												

TABLE 10.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.085	-.562	-.223	-.117	.707	.368	-.136	.801	.449
.025				-.499	.268	.112	-.577	.471	.235	-.556	.599	.296
.050				-.538	-.057	-.057	-.297	.123	.067	-.242	.274	.156
.075				-.791	-.387	-.292	-.997	-.129	-.061	-.1061	-.029	-.033
.100				-.740	-.382	-.264	-.963	-.359	-.101	-.1049	-.130	.007
.150				-.802	-.561	-.342	-.1025	-.302	-.140	-.1077	-.225	-.012
.200				-.684	-.269	-.113	-.025	-.722	-.045	-.1083	-.113	.005
.250				-.673	-.180	-.085	-.761	-.117	-.033	-.1066	-.052	.010
.300				-.679	-.124	-.113	-.773	-.084	-.173	-.926	-.071	-.010
.350				-.757	-.124	-.113	-.773	-.084	-.173	-.741	-.018	-.023
.400				-.751	-.124	-.113	-.801	-.089	-.078	-.763	-.040	-.067
.450				-.662	-.124	-.113	-.829	-.073	-.061	-.797	-.035	-.521
.500				-.538	-.101	-.101	-.437	-.039	-.028	-.388	.007	.005
.550				-.387	-.051	-.051	-.353	-.045	.051	-.304	.061	.072
.600				-.241	.033	.033	-.179	.034	.112	-.147	.122	.150
.650				-.044	.117	.123	.034	.095	.129	-.001	.173	.073
.700				.162	.168	.140	.095	.163	.185	.206	.139	.201
.750				.179			.179					
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010				-.188	-.238	.211	-.288	.489	.347	-.296	.651	.475
.025				-.239	-.019	.009	-.356	.230	.123	-.369	.415	.229
.050				-.475	-.239	-.086	-.666	.005	.044	-.596	.178	.139
.075				-.295	-.402	.143	-.232	-.232	-.041	-.324	-.042	.048
.100				-.706	-.318	-.143	-.1009	-.193	-.046	-.1108	-.076	.037
.150				-.870	-.272	-.137	-.987	-.187	-.057	-.1125	-.087	.003
.200				-.774	-.210	-.126	-.1009	-.148	-.057	-.1096	-.064	-.002
.250				-.735	-.188	-.098	-.1004	-.131	-.046	-.1102	-.064	.003
.300				-.740	-.092	-.081	-.1026	-.057	-.041	-.1113	-.002	.003
.350				-.780	-.058	-.081	-.914	-.035	-.046	-.1108	-.002	-.002
.400				-.763	-.024	-.024	-.626	-.001	-.001	-.651	.009	-.014
.450				-.622	-.024	-.024	-.474	.044	.050	-.397	.037	.026
.500				-.498	.026	.026	-.328	.095	.089	-.262	.077	.077
.550				-.340	.077	.077	-.164	.033	.145	-.126	.116	.116
.600				-.165	.133	.133	-.033	.168	.196	-.054	.167	.172
.650				.026	.105	.105	.168	.202	.196	.201	.234	.234
.700				.190			.202			.234		
.750												
.800												
.850												
.900												
.950												
1.000												

TABLE 10.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Concluded

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.182	.560	.191	-.096	.693	.336	-.106	.790	.421
.025				-.446	.314	.120	-.587	.437	.216	-.516	.287	-.260
.050				-.505	-.058	-.058	-.204	.109	.049	-.133	.250	-.287
.075				-.742			-.932			-1.022		-.805
.100				-.758	.284	.311	-.942	-.134	-.069	-.995	.007	-1.145
.150				-.812	-.516	-.214	-.959	-.296	-.231	-.1049	-.015	-1.081
.200				-.699	-.435	-.214	-.959	-.258	-.355	-1.049	-.219	-1.194
.250				-.699	-.300	-.133	-.921	-.209	-.118	-1.049	-.155	-1.221
.300				-.667	-.241	-.209	-.802	-.139	-.042	-1.049	-.079	-1.221
.350				-.699			-.781			-1.060		-1.221
.400				-.731	-.139	-.209	-.808	-.129	-.382	-.839	-.074	-1.156
.450				-.823	-.150	-.128	-.867	-.220	-.096	-.893	-.069	-.735
.500				-.699	-.106	-.171	-.802	-.177	-.274	-.613	-.052	-.654
.550												-.595
.600				-.381	-.150	-.093	-.312	-.042	-.042	-.262	-.031	-.503
.650				-.182	.028	.023	-.150	.012	.049	-.133	.045	-.476
.700				.061	.114	.120	.038	.109	.136	.045	.109	-.249
.750				.152	.168	.195	.146	.152	.189	.158	.136	-.152
.800				.174			.162			.190	.195	-.060
.850												
.900												
.950												
1.000												
Outboard station												
.010				-.168	.219	.187	-.250	.454	.304	-.276	.604	.421
.025				-.217	-.022	-.017	-.294	.178	.086	-.325	.336	.179
.050				-.439	-.250	-.087	-.597	-.044	.021	-.840	.135	.103
.075				-.141			-.012			-.130		-.345
.100				-.662	.445	.147	-.933	-.250	-.066	-1.046	-.098	-1.137
.150				-.808	.342	.152	-.961	-.234	-.066	-1.062	-.125	-.011
.200				-.830	.293	.147	-.928	-.207	-.088	-1.046	-.130	-1.213
.250				-.813	.223	.141	-.961	-.163	-.077	-1.057	-.109	-1.154
.300				-.840	-.206	-.109	-.966	-.153	-.060	-1.079	-.103	-1.110
.350				-.857			-1.026			-1.100	-.033	-.038
.400				-.797	-.098	-.092	-1.036	-.066	-.055	-1.116	-.033	-.958
.450							-1.102			-.981	-.049	-.768
.500				-.905	-.065	-.087		-.044	-.260		-.022	-.594
.550				-.472	-.027	-.027	-.538	-.012	-.006	-.510	-.005	-.426
.600												-.334
.650				-.288	.032	.027	-.288	.037	.037	-.282	.043	-.301
.700												
.750				-.136	.081	.076	-.126	.091	.086	-.130	.076	-.274
.800				.043	.135	.135	.037	.140	.146	-.000	.108	-.241
.850				.162			.151			.097		-.198
.900				.189	.097	.195	.194	.091	.221	.103	-.090	-.176
.950												
1.000												

TABLE 11.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.242	.105	-.160	-.318	.504	.152	-.427	.669	.317	-.498	.784	-.432
.025	-.211	-.243	-.179	-.401	.226	.073	-.485	.418	.200	-.594	.565	-.296
.050	-.136	-.569	-.352	-.190	-.126	-.100	-.325	.059	.021	-.402	.284	.130
.075	-.478	-.810	-.810	-.810	-.810	-.810	-.552	-.552	-.552	-.158	-.552	-.552
.100	-.499	-.588	-.371	-.874	-.279	-.170	-.895	-.133	-.062	-.953	.001	.027
.150	-.530	-.562	-.390	-.855	-.350	-.247	-.978	-.248	-.133	-.953	-.127	-.037
.200	-.524	-.479	-.371	-.570	-.311	-.209	-.754	-.216	-.120	-.857	-.152	-.043
.250	-.524	-.396	-.231	-.644	-.266	-.138	-.773	-.190	-.088	-.767	-.127	-.018
.300	-.505	-.307	-.205	-.525	-.209	-.126	-.747	-.145	-.075	-.748	-.101	-.018
.350	-.524	-.205	-.148	-.512	-.132	-.106	-.792	-.081	-.069	-.748	-.043	-.018
.400	-.524	-.175	-.135	-.574	-.113	-.106	-.594	-.069	-.075	-.697	-.031	-.056
.500	-.422	-.135	-.103	-.458	-.087	-.087	-.466	-.056	-.069	-.472	-.024	-.037
.600	-.403	-.077	-.052	-.414	-.055	-.049	-.402	-.069	-.037	-.402	-.005	.001
.750	-.262	.012	.037	-.254	.022	.028	-.235	.034	.021	-.229	.053	.059
.800	-.020	.101	.120	-.017	.098	.111	-.024	.098	.104	-.018	.110	.130
.975	.146	.184	.210	.143	.156	.175	.143	.136	.155	.130	.130	.162
1.000	.176	.184	.210	.175	.156	.175	.175	.136	.155	.174	.130	.162
Outboard station												
.010	-.264	-.530	-.237	-.442	.154	.138	-.630	.463	.347	-.701	.664	-.509
.025	-.213	-.759	-.329	-.391	-.114	-.043	-.552	.214	.079	-.688	.375	.220
.050	-.226	-.983	-.316	-.823	-.294	-.088	-.555	-.018	.001	-.785	.149	.072
.075	.301	-.649	-.341	.028	.346	-.152	-.218	.185	-.050	-.379	-.031	.027
.100	-.489	-.483	-.256	-.719	-.301	-.146	-.939	-.179	-.050	-.152	-.076	.020
.150	-.502	-.412	-.219	-.593	-.262	-.140	-.771	-.172	-.069	-.875	-.102	-.005
.200	-.540	-.322	-.207	-.574	-.210	-.127	-.765	-.140	-.069	-.823	-.083	-.005
.250	-.534	-.284	-.168	-.561	-.185	-.114	-.726	-.127	-.057	-.772	-.083	-.005
.300	-.540	-.168	-.123	-.855	-.120	-.088	-.694	-.069	-.057	-.740	-.037	-.005
.350	-.547	-.117	-.091	-.577	-.062	-.082	-.681	-.031	-.057	-.720	-.001	-.018
.450	-.528	-.053	-.020	-.513	-.017	-.024	-.604	.001	-.011	-.617	.033	.014
.550	-.478	.005	.012	-.397	.002	-.049	-.527	.040	.027	-.521	.040	.040
.650	-.373	.018	.012	-.217	.060	.041	-.379	.072	.066	-.385	.066	.078
.750	-.232	.114	-.072	-.011	.059	.092	-.211	.117	.111	-.218	.104	.117
.800	-.020	.114	-.072	.111	.054	.137	.104	.021	.137	.104	.104	.117
.975	.164	-.085	.121	.054	.047	.137	.130	.021	.137	.104	.091	.078
1.000	.164	-.085	.121	.054	.047	.137	.130	.021	.137	.104	.091	.078

TABLE 11.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.250	.115	-.166	-.352	.520	.166	-.405	.683	.284	-.435	.775	.441
.025	-.192	-.180	-.186	-.370	.226	.062	-.458	.436	.150	-.488	.553	.278
.050	-.046	-.629	-.367	-.014	-.107	-.119	-.131	-.031	.004	-.207	.219	.114
.075	.600			-.755			-1.001			-1.189		
.100	.530	.652	-.356	-.744	-.317	-.182	-1.001	.154	-.090	-1.148	-.038	.015
.150	-.558	.787	-.485	-.749	-.353	.265	.779	.265	-.166	-1.207	-.166	-.067
.200	-.564	-.513	-.414	-.709	-.367	-.224	.791	.259	-.142	-1.218	.178	-.055
.250	-.598	-.437	-.275	-.709	-.294	-.194	.820	.242	-.096	-1.257	.166	-.020
.300	-.711	-.338	-.244	-.709	-.230	-.136	.761	.253	-.090	-.815	.108	-.020
.350	-.625			-.697			.750			-.868		
.400	-.746	-.227	-.204	-.720	-.148	-.130	.721	.125	-.078	-.868	-.055	-.026
.450												
.500	-.552	-.204	-.186	-.744	-.130	-.115	-.674	-.096	-.090	-.634	-.049	-.049
.550												
.600	-.445	-.163	-.145	-.516	-.107	-.167	-.475	-.078	-.078	-.488	-.038	-.038
.650												
.700	-.414	-.095	-.081	-.417	-.060	-.049	-.395	-.049	-.037	-.400	-.014	-.003
.750												
.800	-.350	.066	.012	-.230	.021	.027	-.224	.027	.033	-.213	.050	.062
.900	-.006	.088	.095	-.002	.086	.109	.004	.091	.109	-.003	.108	.126
.975	.152			.136			.138			.143		
1.000	.181	.124	.187	.167	.150	.173	.167	.132	.167	.179	.132	.179
Outboard station												
.010	-.265	-.453	-.303	-.467	.167	.146	-.614	.442	.329	-.627	.619	.421
.025	-.206	-.540	-.376	-.402	-.103	-.079	-.573	.150	.067	-.592	.355	.167
.050	-.255	-.898	-.405	-.414	-.326	-.115	-.602	-.080	-.021	-.652	.332	.096
.075	.457			.185			.027			-.186		
.100	.495	-.1027	-.405	-.790	-.402	-.179	-.979	.226	-.080	-1.180	-.092	.020
.150	-.534	.467	-.370	-.784	-.332	.167	-.973	-.215	-.085	-1.203	-.110	.002
.200	-.534	-.395	-.282	-.743	-.291	-.156	-.808	-.215	-.085	-1.180	.121	-.015
.250	-.552	-.317	-.270	-.761	-.226	-.144	-.779	-.162	-.080	-1.197	-.098	-.015
.300	-.558	-.282	-.212	-.726	-.209	-.115	-.808	-.150	-.074	-1.168	-.092	-.004
.350	-.364			-.720			-.868			-.957		
.400	-.593	-.177	-.171	-.726	-.120	-.103	-.868	-.085	-.062	-.727	-.051	-.027
.450												
.500	-.552	-.130	-.136	-.614	-.079	-.091	-.620	-.050	-.069	-.615	-.010	-.015
.550												
.600	-.495	-.071	-.059	-.526	-.026	-.026	-.520	-.009	-.015	-.521	.032	.020
.650												
.700	-.382	-.067	.005	-.385	-.050	.026	-.403	.038	.009	-.368	.067	.067
.750												
.800	-.225	.052	.040	-.197	.068	.050	-.191	.079	.073	-.192	.096	.102
.900	-.007	.122	.111	.003	.097	.067	.065	.109	.103	.120	.137	.114
.975	.146			.125			.085			.043		
1.000	.165	.074	.158	.138	.062	.044	.167	.085	.156	.043	.108	.155

TABLE 11.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.241	.160	-.173	-.347	.531	.122	-.399	.675	.304	-.410	.776	.417
.025	-.179	-.129	-.190	-.347	.246	.011	-.404	.425	.151	-.455	.537	.263
.050	.056	-.627	-.358	.106	-.140	-.135	-.006	.067	-.006	-.074	.229	.100
.075	.521			.801			.959			.110		
.100	.538	-.650	-.560	.756	.314	-.196	-.970	-.180	.102	-.071	-.040	-.006
.150	.594	-.806	.571	.734	.493	.297	-1.021	-.275	.158	-1.138	-.174	-.074
.200	.566	-.510	-.588	.717	.381	.252	.959	.275	.146	-1.155	-.191	-.057
.250	.610	-.448	-.508	.717	.325	-.174	.724	-.230	.090	-1.127	-.158	-.023
.300	.610	-.353	-.269	.717	.247	-.157	-.768	-.180	-.085	-1.054	-.107	-.023
.350	.605			.745			-.802			-.796		
.400	.550	-.241	-.230	.751	.157	-.140	-.830	-.102	-.079	-.841	-.062	-.029
.450												
.500	.516	-.218	-.213	.689	.146	-.140	-.808	-.102	-.085	-.903	-.051	-.034
.550												
.600	.459	-.174	-.168	.510	.118	-.118	-.460	-.079	-.079	-.449	-.040	-.034
.650												
.700	.414	-.106	-.055	.437	.073	-.062	-.387	-.046	-.034	-.359	-.018	-.006
.750												
.800	.241	.000	.006	.219	.011	.022	-.202	.033	.050	-.197	.050	.067
.900	.011	.050	.095	.011	.050	.106	.010	.100	.123	.005	.117	.139
.975	.151			.140			.139			.151		
1.000	.179	.162	.185	.162	.151	.174	.179	.139	.179	.184		
Outboard station												
.010	-.255	-.405	-.276	-.486	.137	.110	-.577	.428	.306	-.583	.590	.407
.025	-.193	-.627	-.367	-.407	.108	-.131	-.526	.156	.043	-.543	.235	.167
.050	.513	-.824	-.379	.356	.351	-.148	-.566	-.075	-.030	-.611	.111	.083
.075				.359			.105			-.041		
.100	.497	-1.094	-.418	.751	.418	-.204	-.955	-.267	.081	-1.084	-.092	.009
.150	.457	-.903	-.401	.880	.362	-.187	-.989	-.228	.092	-1.135	-.132	-.002
.200	.555	-.424	-.306	.779	.311	-.165	-.972	-.216	-.087	-1.118	-.132	-.019
.250	.574	.306	-.289	.745	.249	-.154	-.989	-.166	-.087	-1.135	-.098	-.013
.300	.565	-.277	-.272	.773	.210	-.125	-1.006	-.160	-.070	-1.130	-.092	-.036
.350	.552			.768			-.572			-1.146		
.400	.632	-.182	-.187	.785	.131	-.103	-.808	-.081	-.058	-1.169	-.036	-.008
.450												
.500	.582	-.131	-.136	.666	.086	-.114	-.707	-.087	-.075	-.915	-.008	-.013
.550				.514			-.493			-.447		
.600	.508	-.075	-.069		.041	-.035	-.013	-.013	-.030	-.306	.032	.032
.650				.362	.016	-.041	-.352	.032	.043	-.306	.077	.077
.700	.379	-.035	-.013				-.182	.083	.060	-.160	.116	.116
.750				.182	.066	.032	-.182	.083	.060	-.160	.116	.116
.800	.204	.049	.044	.182	.128	.117	-.156	.139	.139	-.178	.161	.161
.900	.010	.123	.089	.128			.173			.150		
.975	.151			.168	.072	.168						
1.000	.179	.076	.156	.168								

TABLE 11.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Concluded

(d) $M = 0.800$

κ/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.242	.206	-.174	-.445	-.330	-.161	-.514	.663	-.271	-.462	.753	-.361
.025	-.162	-.058	-.221	-.311	-.249	-.045	-.444	.417	.148	-.451	.529	-.233
.050	-.180	-.506	-.366	-.244	-.133	-.128	-.132	.056	-.008	-.050	.195	-.066
.075	-.488	-.608	-.452	-.742	-.419	-.214	-.907	-.186	-.094	-.116	-.065	-.015
.100	-.548	-.861	-.580	-.774	-.526	-.337	-.932	-.320	-.207	-.1054	-.203	-.101
.150	-.538	-.888	-.603	-.720	-.472	-.219	-.972	-.310	-.164	-.1082	-.218	-.085
.200	-.549	-.469	-.334	-.704	-.408	-.182	-.945	-.272	-.116	-.1076	-.203	-.047
.250	-.571	-.383	-.280	-.588	-.273	-.185	-.794	-.191	-.100	-.1086	-.138	-.047
.300	-.630	-.280	-.259	-.731	-.176	-.160	-.772	-.116	-.100	-.855	-.079	-.058
.350	-.678	-.259	-.237	-.906	-.165	-.160	-.880	-.105	-.110	-.919	-.079	-.074
.400	-.522	-.205	-.184	-.599	-.138	-.133	-.842	-.089	-.100	-.752	-.068	-.074
.450	-.415	-.130	-.098	-.354	-.090	-.074	-.331	-.051	-.062	-.311	-.042	-.031
.500	-.216	-.017	-.001	-.187	.007	.018	-.175	.029	.035	-.155	.029	.045
.550	-.015	-.080	.090	.029	.082	.104	-.013	.099	.116	.018	.099	.120
.600	.133	.155	.182	.136	.136	.169	.142	.126	.175	.142	.109	.185
.650	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.700	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.750	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.800	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.850	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.900	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
.950	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
1.000	.150	.155	.182	.169	.136	.169	.173	.126	.175	.190	.109	.185
Outboard station												
.010	-.273	-.362	-.304	-.477	-.135	-.082	-.530	.377	.273	-.536	.539	.324
.025	-.192	-.587	-.392	-.390	-.141	-.114	-.470	.104	.039	-.488	.255	.103
.050	-.176	-.787	-.415	-.309	-.385	-.152	-.497	-.134	-.042	-.547	.065	.049
.075	.760	.760	.760	.498	.498	.218	.218	.087	.087	.087	.087	.087
.100	.500	.500	.451	.451	.558	.211	.838	-.291	.107	.981	-.168	-.022
.150	.472	-.1041	-.451	-.325	-.396	-.211	-.925	-.275	-.107	-.1024	-.179	-.038
.200	.570	.916	-.332	.813	-.341	-.179	-.914	-.270	-.107	-.1024	-.173	-.038
.250	.608	-.343	-.316	.824	-.260	-.179	-.957	-.194	-.096	-.1062	-.135	-.043
.300	.614	-.252	-.230	.829	-.244	-.141	-.968	-.178	-.086	-.1062	-.130	-.033
.350	.514	-.181	-.176	.733	-.135	-.114	-.985	-.140	-.069	-.1073	-.070	-.038
.400	.700	-.143	-.145	-.807	-.052	-.108	-.1044	-.059	-.069	-.1122	-.033	-.060
.450	.803	-.143	-.145	-.821	-.052	-.108	-.1093	-.059	-.069	-.1187	-.033	-.060
.500	.550	-.089	-.068	-.488	-.043	-.043	-.551	-.004	-.021	-.585	.005	-.022
.600	.650	-.030	-.003	-.309	.011	.016	-.308	.033	.033	-.336	.043	-.038
.700	.750	-.041	.046	-.146	.060	.054	-.140	.082	.077	-.152	.060	.060
.800	.800	.116	.105	-.033	.119	.119	-.033	.136	.136	-.011	.060	.060
.850	.975	.142	.115	.119	.119	.119	.136	.136	.136	.087	.060	.060
.900	.975	.142	.115	.119	.119	.119	.136	.136	.136	.087	.060	.060
.950	.975	.142	.115	.119	.119	.119	.136	.136	.136	.087	.060	.060
1.000	.975	.142	.115	.119	.119	.119	.136	.136	.136	.087	.060	.060

TABLE 12.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.802	.555	.153	-.927	.676	.274	-1.128	.762	.409
.025				-.949	.329	-.150	-1.164	.488	.155	-1.295	.635	.257
.050				-1.049	.035	-.144	-1.016	.239	-.050	-1.019	.359	.083
.075				-.937	-.853	-.114	-.856	-.050	-.1346	-1.346	-.1664	-.1664
.100				-.824	-.284	-.355	-.920	-.114	-.146	-1.385	.013	-.090
.150				-.790	-.451	-.463	-.805	-.229	-.101	-.923	-.115	-.019
.200				-.825	-.284	-.169	-.907	-.197	-.088	-.840	-.115	-.026
.250				-.745	-.246	-.137	-.779	-.171	-.075	-.827	-.109	-.013
.300				-.604	-.201	-.125	-.760	-.139	-.062	-.853	-.083	-.013
.350				-.604	-.201	-.125	-.831	-.089	-.058	-.825	-.021	-.091
.400				-.585	-.144	-.118	-.639	-.101	-.075	-.731	-.058	-.026
.450				-.553	-.118	-.105	-.594	-.082	-.075	-.603	-.045	-.038
.500				-.483	-.061	-.054	-.555	-.043	-.024	-.571	.000	.000
.600				-.406	-.010	-.019	-.523	.002	.008	-.430	.026	.032
.700				-.240	.048	.048	-.229	.053	.059	-.205	.071	.077
.800				.010	.118	.118	.021	.117	.123	.039	.115	.122
.900				.157	.157	.157	.149	.168	.168	.103	.110	.128
.975				.182	.195	.170	.168	.168	.168	.160	.160	.160
1.000												
Outboard station												
.010				-.692	.332	.216	-.842	.509	.347	-1.000	.671	.470
.025				-1.013	.054	-.048	-1.229	.291	.066	-1.361	.438	.200
.050				-.745	-.126	-.151	-1.268	.130	-.031	-1.497	.284	.077
.075				-.917	-.820	-.171	-1.003	-.173	-.063	-1.361	-.039	.019
.100				-.820	-.344	-.171	-.958	-.205	-.057	-1.368	-.039	.019
.150				-.666	-.332	-.159	-.759	-.205	-.057	-1.368	-.039	.019
.200				-.672	-.267	-.119	-.772	-.179	-.057	-1.368	-.039	.019
.250				-.666	-.203	-.119	-.759	-.134	-.057	-1.368	-.039	.019
.300				-.647	-.184	-.119	-.720	-.127	-.063	-1.368	-.039	.019
.350				-.660	-.126	-.106	-.675	-.089	-.063	-1.368	-.039	.019
.400				-.627	-.081	-.094	-.591	-.057	-.063	-1.368	-.039	.019
.450				-.563	-.029	-.042	-.501	-.011	-.024	-1.368	-.039	.019
.500				-.492	-.029	-.042	-.501	-.011	-.024	-1.368	-.039	.019
.600				-.370	.029	.022	-.366	.034	.034	-1.368	-.039	.019
.700				-.203	.074	.087	-.179	.079	.092	-1.368	-.039	.019
.800				.022	.138	.138	.034	.137	.137	-.174	.097	.103
.900				.144	.144	.144	.130	.130	.130	.129	.148	.148
.975				.164	.093	.164	.150	.066	.150	.148	.148	.148
1.000												

TABLE 12.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.691	.552	.120	-.810	.690	.224	-.938	.755	.335
.025				-.843	.360	.004	-1.026	.500	.149	-1.119	.605	.184
.050				-.931	.056	-.183	-.792	.219	-.073	-.769	.359	.050
.075				-.785			-1.108			-1.201		
.100				-.808	-.393	-.241	-1.079	-.143	.149	-1.271	-.015	-.120
.150				-.738	-.387	-.212	-.915	-.260	.120	-1.254	-.155	-.050
.200				-.726	-.370	-.189	-.716	-.231	.108	-1.160	-.149	-.032
.250				-.726	-.370	-.148	-.763	-.196	.085	-.722	-.132	-.015
.300				-.732	-.241	-.165	-.769	-.155	-.067	-.816	-.108	-.026
.350				-.691			-.751			-.821	-.061	-.050
.400				-.691	-.160	-.136	-.693	-.108	-.079	-.816	-.050	.003
.450												
.500				-.650	-.130	-.154	-.634	-.091	-.085	-.646	-.050	.003
.550				-.592	-.078	-.066	-.488	-.050	-.032	-.494	-.020	.021
.600												
.650				-.399	-.019	-.014	-.401	-.003	.003	-.383	.020	.062
.700												
.750				-.212	.051	.039	-.208	.055	.067	-.243	.067	.109
.800				.027	.121	.121	.038	.120	.126	.038	.131	.167
.850				.156			.135			.126		
.900				.179	.185	.179	.166	.178	.178	.167	.178	.225
.950												
1.000												
Outboard station												
.010				-.579	.294	.167	-.710	.498	.315	-.810	.647	.414
.025				-.902	.085	-.097	-1.080	.284	.031	-1.151	.419	.131
.050				-.814	-.115	-.173	-1.192	.072	-.033	-1.286	.261	.049
.075				-.749			-.757			-1.074		
.100				-.773	-.461	-.185	-1.092	.216	.075	-1.227	-.063	.002
.150				-.925	-.367	-.156	-1.092	.245	-.075	-1.280	-.139	-.008
.200				-.684	-.303	-.126	-.933	.216	.069	-1.162	-.133	-.010
.250				-.731	-.226	-.120	-.780	.157	-.069	-1.180	-.092	-.004
.300				-.714	-.197	-.115	-.722	.139	-.063	-1.139	-.086	-.010
.350				-.737			-.763			-.874		
.400				-.708	-.132	-.109	-.792	-.092	-.063	-.857	-.045	-.022
.450												
.500				-.602	-.085	-.097	-.628	-.051	-.069	-.592	-.016	-.039
.550				-.508	-.038	-.044	-.510	-.010	-.022	-.492	.020	.008
.600												
.650				-.355	.021	.021	-.357	.043	.037	-.345	.067	.055
.700												
.750				-.173	-.079	.091	-.169	.090	.102	-.157	.108	.120
.800				-.044	.150	.138	.049	.149	.149	-.049	.167	.161
.850				-.144			.137			.155		
.900				-.162	-.085	.167	.161	.078	.161	.178	.096	.184
.950												
1.000												

TABLE 12.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				-.584	.556	.103	-.769	.675	.222	-.870	.751	.342
.025				-.808	.358	-.107	-.926	.569	.106	-.1021	.599	.184
.050				-.858	.083	-.214	-.663	.223	-.069	-.612	.341	.038
.075				-.780			-.1011			-.1133		-.1287
.100				-.830	-.371	.275	-.1033	-.175	-.170	-.1172	-.029	-.130
.150				-.802	-.623	-.371	-.1039	-.287	-.130	-.1172	-.175	-.063
.200				-.668	-.365	-.264	-.938	-.321	-.130	-.1150	-.163	-.051
.250				-.679	-.393	-.331	-.736	-.332	-.091	-.1128	-.141	-.029
.300				-.668	-.225	-.354	-.730	-.242	-.113	-.870	-.113	-.023
.350				-.701			-.764			-.797		
.400				-.741	-.175	-.214	-.792	-.170	-.108	-.814	-.074	-.046
.450				-.673	-.203	-.141	-.786	-.192	-.097	-.859	-.057	-.051
.500				-.511	-.107	-.074	-.461	-.052	-.046	-.455	-.018	-.007
.550				-.483	-.029	-.023	-.377	-.001	-.007	-.348	.010	.027
.600				-.387	.044	-.001	-.186	.049	.060	-.163	.072	.083
.650				-.038	-.117	.117	.049	.116	.128	-.049	.128	.145
.700				.162			.150			.162		
.750				.027	.179	.078	.173	.173	.156	.178	.178	.178
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010				-.526	.285	.123	-.646	.495	.265	-.724	.631	.394
.025				-.825	.082	-.109	-.961	.285	.031	-.1045	.409	.116
.050				-.961	-.104	-.188	-.1102	.093	-.053	-.1186	.206	.020
.075				-.628			-.550			-.910		-.868
.100				-.729	-.515	.199	-.955	-.251	-.104	-.1130	-.087	-.025
.150				-.910	-.402	.171	-.1032	.262	-.087	-.1186	-.160	-.008
.200				-.729	-.312	.137	-.956	.211	-.082	-.1113	-.132	-.008
.250				-.741	-.228	.137	-.995	.155	-.070	-.1158	-.092	-.014
.300				-.752	-.205	.126	-.1012	-.155	-.070	-.1153	-.092	-.014
.350				-.786			-.1029			-.1147		
.400				-.763	-.132	.115	-.883	.093	-.076	-.1175	-.047	-.019
.450				-.622	-.081	-.104	-.589	.048	-.070	-.724	-.019	-.030
.500				-.493	-.030	-.047	-.482	-.008	-.025	-.453	.020	.009
.550				-.340	.026	.020	-.330	.048	.037	-.290	.065	.065
.600				-.149	.082	.094	-.144	.093	.105	-.132	.122	.133
.650				-.066	.150	.144	-.059	.161	.155	-.054	.184	.184
.700				.156			.150			.184		
.750				.178	.088	.173	.172	.088	.178	.212	.150	.218
.800												
.850												
.900												
.950												
1.000												

TABLE 12.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Concluded

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010				.515	.560	.089	-.682	.657	.210	-.785	.741	.294
.025				-.742	.296	-.058	-.855	.503	.055	-.930	.578	.152
.050				-.790	.066	-.208	-.510	.222	-.208	-.472	.282	.002
.075				-.779			-.930			-.1038		
.100				-.812	-.408	-.381	-.973	-.225	.332	-.1070	-.112	-.208
.150				-.812	-.548	-.451	-.989	-.451	-.295	-.1092	-.214	-.149
.200				-.785	-.435	-.241	-.979	-.284	-.289	-.1081	-.219	-.095
.250				-.731	-.431	-.327	-.909	-.235	-.106	-.1070	-.273	-.063
.300				-.688	-.257	-.246	-.785	-.262	-.101	-.1065	-.149	-.058
.350				-.715			-.758			-.1038		
.400				-.742	-.219	-.241	-.801	-.262	-.117	-.849	-.106	-.079
.450												
.500				-.801	-.203	-.321	-.855	-.208	-.122	-.903	-.208	-.101
.550												
.600				-.715	-.106	-.165	-.769	-.305	-.063	-.823	-.101	-.090
.650												
.700				-.483	-.090	-.031	-.322	-.025	-.015	-.311	-.009	.002
.750												
.800				-.278	.034	.039	-.144	.039	.045	-.133	.050	.061
.850				-.050	.104	.115	-.045	.115	.109	.039	.120	.120
.900				.120			.136			.142		
.950												
1.000				.169	.174	.174	.158	.169	.163	.163	.169	.158
Outboard station												
.010				-.455	.285	.096	-.564	.454	.213	-.639	.591	.291
.025				-.764	.081	-.173	-.883	.260	-.022	-.943	.374	.049
.050				-.878	-.146	-.195	-.102	.043	-.103	-.1078	.184	-.011
.075				-.466			-.287			-.748		
.100				-.693	-.515	-.233	-.937	-.320	.135	-.1008	-.146	-.070
.150				-.856	-.433	-.200	-.986	-.341	.108	-.100	-.217	-.049
.200				-.834	-.352	-.168	-.905	-.271	.098	-.1030	-.209	-.054
.250				-.834	-.249	-.162	-.959	-.190	.092	-.1073	-.146	-.054
.300				-.856	-.222	-.146	-.970	-.173	.087	-.1078	-.135	-.054
.350				-.861			-.1002			-.1095		
.400				-.823	-.146	-.130	-.1046	-.114	-.087	-.1111	-.087	-.060
.450												
.500				-.899	-.097	-.125	-.1089	-.065	-.092	-.629	-.054	-.081
.550												
.600				-.531	-.049	-.059	-.488	-.027	-.038	-.515	-.022	-.043
.650												
.700				-.292	.011	.011	-.287	.033	.027	-.379	.033	.016
.750												
.800				-.119	.071	.087	-.114	.081	.098	-.238	.065	.065
.850				.054	.141	.136	.040	.152	.146	-.103	.114	.087
.900												
.950				.146			.157			-.011		
1.000				.168	.065	.168	-.184	.103	.190	.034	-.076	-.016

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	-.313	-.203	-.145	-.694	.556	.165	-.934	.708	.305	-1.125	.775	.440
.025	-.513	-.439	-.238	-.850	.348	.036	-1.159	.485	.122	-1.325	.592	.261
.050	-.894	-.394	-.376	-.844	.036	-.139	-.947	.216	-.003	-1.063	.342	.092
.075	-.544			-.900			-1.034			-1.344		
.100	-.501	-.663	-.351	-.838	-.282	-.133	-.866	-.128	-.040	-1.069	-.014	.049
.150	-.513	-.551	-.363	-.782	-.320	-.137	-.847	-.222	-.090	-.888	-.126	-.008
.200	-.513	-.438	-.313	-.676	-.276	-.151	-.791	-.197	-.090	-.857	-.126	-.014
.250	-.513	-.382	-.263	-.651	-.245	-.126	-.734	-.178	-.072	-.819	-.120	-.014
.300	-.507	-.313	-.232	-.619	-.207	-.114	-.703	-.159	-.072	-.907	-.107	-.014
.350	-.532	-.259	-.216	-.632	-.167	-.123	-.678	-.125	-.081	-.794	-.086	-.049
.400	-.522	-.232	-.201	-.619	-.139	-.114	-.637	-.103	-.078	-.907	-.076	-.041
.450	-.515	-.210	-.185	-.591	-.129	-.117	-.622	-.094	-.081	-.866	-.055	-.043
.500	-.513	-.195	-.176	-.576	-.120	-.108	-.600	-.090	-.084	-.844	-.058	-.045
.550	-.497	-.154	-.142	-.553	-.092	-.080	-.581	-.069	-.057	-.891	-.030	-.030
.600	-.419	-.132	-.114	-.437	-.070	-.084	-.484	-.053	-.040	-.895	-.033	-.026
.650	-.415	-.104	-.080	-.440	-.055	-.056	-.455	-.032	-.019	-.859	-.005	-.001
.700	-.407	-.076	-.064	-.420	-.033	-.025	-.428	-.022	-.019	-.832	-.001	-.001
.750	-.346			-.346			-.348			-.846		
.800	-.270	-.001	.005	-.257	.030	.036	-.259	.035	.041	-.825	.049	.049
.850	-.039	.086	.131	-.008	.098	.111	-.009	.091	.110	-.800	.099	.111
.900	.149	.150	.131	.148	.143	.137	.128	.130	.123	.117	.131	.125
.950	.186	.186	.180	.173	.173	.173	.147	.153	.153	.142	.148	.142
1.000												

Outboard station												
.010	-.021	-.170	-.064	-.492	.319	.388	-.752	.494	.576	-.919	.608	.664
.025	-.561	-.247	-.247	-.139	.155	.149	-1.374	.298	.273	-1.503	.431	.413
.050	-.410	-.253	-.272	-.181	.029	.023	-1.311	.179	.173	-1.534	.281	.281
.075	-.441			-.774			-.840			-1.434		
.100	-.473	-.265	-.278	-.768	-.058	-.058	-.984	.041	.041	-1.421	.092	.143
.150	-.492	-.247	-.240	-.743	-.071	-.058	-.840	.015	.041	-1.132	.086	.143
.200	-.454	-.234	-.215	-.624	-.077	-.077	-.727	-.003	.041	-1.132	.036	.030
.250	-.529	-.209	-.209	-.762	-.077	-.084	-.783	-.010	.016	-.825	.030	.030
.300	-.529	-.222	-.222	-.680	-.102	-.115	-.758	-.029	.047	-.812	.011	.004
.350	-.548	-.214	-.227	-.674	-.120	-.127	-.745	-.072	.084	-.787	-.020	-.032
.400	-.542	-.222	-.222	-.655	-.121	-.128	-.720	-.079	.085	-.749	-.040	-.040
.450	-.534	-.221	-.227	-.642	-.139	-.139	-.656	-.097	.109	-.679	-.064	-.057
.500	-.517	-.209	-.215	-.580	-.121	-.134	-.620	-.085	.110	-.642	-.058	-.058
.550	-.465	-.183	-.202	-.522	-.120	-.145	-.543	-.084	.116	-.553	-.057	-.076
.600	-.473	-.152	-.152	-.517	-.102	-.109	-.525	-.073	.079	-.542	-.046	-.046
.650	-.421	-.120	-.126	-.440	-.076	-.083	-.442	-.065	.053	-.453	-.026	-.032
.700	-.379	-.083	-.090	-.391	-.052	-.052	-.393	-.022	.022	-.397	-.002	-.008
.750	-.302			-.302			-.296			-.296		
.800	-.234	-.002	.005	-.222	.017	.023	-.217	.028	.009	-.213	.036	.036
.850	-.014	.092	.092	.004	.099	.099	-.047	.097	.097	.004	.099	.092
.900	.975	.125	.137	.124	.118	.111	-.003	.098	.111	.092	.106	.118
.950	.136	.074	.162	.124	.099	.130	-.122	.047	.148	.099	.036	.117
1.000												

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.291	.249	-.150	-.650	.565	.149	-.820	.693	.299	-.956	.777	.366
.025	-.494	.017	-.273	-.821	.352	.016	-1.030	.472	.102	-1.138	.210	.576
.050	-.680	-.404	-.382	-.593	.033	-.121	-.706	.216	-.023	-.814	.756	.232
.075	-.763			-.793			-1.104			-1.234	.562	
.100	-.566	-.837	-.370	-.632	.343	.149	-1.081	-.154	-.052	-1.274	.221	.186
.150	-.518	-.620	-.387	-.730	.366	.195	-.740	-.251	-.109	-1.246	.044	.112
.200	-.598	-.478	-.347	-.696	.309	.182	-.768	-.217	-.109	-1.195	.004	.095
.250	-.632	-.421	-.285	-.696	.289	.144	-.820	-.200	-.086	-.745	.007	.090
.300	-.660	-.330	-.251	-.679	.229	.138	-.785	-.171	-.086	-.785	.007	.078
.350	-.711	-.274	-.235	-.690	.185	.138	-.740	-.144	-.094	-.870	.004	.041
.400	-.689	-.245	-.222	-.696	.161	.128	-.710	-.116	-.094	-.859	.004	.038
.450	-.567	-.229	-.212	-.660	.145	.121	-.683	-.103	-.097	-.772	.001	.018
.500	-.615	-.211	-.199	-.667	.138	.111	-.636	-.082	-.071	-.641	.004	.010
.550	-.550	-.178	-.161	-.602	.111	.100	-.601	-.057	-.057	-.512	.046	.027
.600	-.467	-.148	-.137	-.576	.087	.081	-.507	-.037	-.048	-.469	.030	.047
.650	-.447	-.116	-.099	-.459	.066	.044	-.464	-.048	-.031	-.437	.044	.044
.700	-.421	-.086	-.080	-.428	.047	.035	-.427	-.029	-.023	-.432	.017	.000
.750	-.355			-.345			-.338			-.334		
.800	-.268	-.000	-.000	-.240	.022	.027	-.234	.034	.039	-.228	.045	.051
.900	-.012	-.085	.085	.004	.090	.101	.011	.091	.108	.011	.102	.114
.975	.147	.144	.121	.141	.137	.132	.130	.127	.121	.125	.121	.116
1.000	.170	.176	.176	.158	.164	.164	.147	.147	.147	.148	.192	.196
Outboard station												
.010	.033	-.189	-.132	-.375	.290	.348	-.562	.468	.554	-.717	.583	.777
.025	-.539	-.259	-.253	-.687	.095	.112	-1.181	.222	.279	-1.266	.365	.564
.050	-.396	-.271	-.271	-.695	.043	.009	-1.203	.164	.096	-1.323	.282	.364
.075	-.442			-.690			-1.100			-1.272		
.100	-.488	-.288	-.345	-.833	.071	.071	-1.089	.033	.038	-1.260	.107	.244
.150	-.528	-.271	-.265	-.913	.094	.077	-1.089	.016	.021	-1.357	.073	.204
.200	-.488	-.259	-.242	-.707	.094	.094	-.889	.013	.025	-1.083	.044	.112
.250	-.557	-.242	-.236	-.764	.094	.088	-.877	.019	.030	-1.283	.016	.106
.300	-.574	-.246	-.253	-.741	.117	.134	-.797	.047	.070	-1.260	.001	.078
.350	-.602	-.247	-.258	-.753	.122	.139	-.797	.081	.092	-1.192	.001	.089
.400	-.602	-.248	-.248	-.735	.150	.162	-.797	.093	.105	-.942	.044	.044
.450	-.590	-.247	-.252	-.694	.140	.152	-.738	.109	.121	-.722	.035	.003
.500	-.557	-.236	-.242	-.615	.140	.146	-.637	.105	.122	-.675	.081	.013
.550	-.510	-.207	-.218	-.545	.127	.140	-.561	.098	.121	-.574	.089	.026
.600	-.505	.168	-.168	-.529	.111	.111	-.540	.076	.082	-.516	.059	.042
.650	-.435	.127	-.132	-.448	.082	.082	-.447	.058	.058	-.441	.041	.020
.700	-.391	-.087	-.093	-.380	.054	.054	-.385	.030	.030	-.373	.019	.014
.750	-.304			-.282			-.281			-.292		
.800	-.219	-.007	-.002	-.091	.009	.015	-.196	.027	.033	-.202	.039	.032
.900	-.004	.096	.090	-.020	.101	.095	.016	.107	.107	.016	.119	.032
.975	.141	.125	.131	.118	.107	.124	.096	.108	.114	.153	.120	.004
1.000	.130	.107	.164	.095	.066	.089	.101	.061	.101	.142	.079	.014

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	-.277	.269	-.173	-.583	.579	.120	-.742	.690	.280	-.894	.781	.381
.025	-.528	.667	-.277	-.785	.335	-.020	-.950	.693	.127	-.1031	.564	.198
.050	-.605	-.310	-.419	-.490	.029	-.151	-.606	.214	-.048	-.659	.329	.029
.075	-.567			-.839			-.1037			-.1167		
.100	-.567	.851	-.370	-.850	-.364	-.200	-.1037	-.185	-.059	-.1195	-.042	.018
.150	-.616	.741	-.457	-.812	-.408	-.200	-.1043	-.294	-.108	-.1195	-.168	-.042
.200	-.621	.468	-.370	-.692	-.331	-.189	-.1004	-.245	-.119	-.1195	-.168	-.059
.250	-.567	.430	-.288	-.703	-.282	-.157	-.715	-.223	-.087	-.1151	-.162	-.037
.300	-.567	.332	-.250	-.697	-.239	-.146	-.759	-.185	-.087	-.954	-.135	-.031
.350	-.594	.277	-.245	-.757	-.190	-.147	-.802	-.148	-.104	-.823	-.104	-.055
.400	-.616	.255	-.233	-.774	-.168	-.146	-.841	-.125	-.103	-.856	-.097	-.059
.450	-.598	.239	-.218	-.712	-.158	-.147	-.840	-.121	-.104	-.878	-.082	-.071
.500	-.599	.222	-.206	-.719	-.146	-.140	-.846	-.114	-.103	-.856	-.080	-.075
.550	-.592	.185	-.163	-.685	-.120	-.109	-.730	-.094	-.077	-.751	-.066	-.050
.600	-.517	.157	-.135	-.490	-.097	-.086	-.562	-.076	-.059	-.545	-.037	-.037
.650	-.449	.120	-.104	-.460	-.071	-.060	-.456	-.050	-.039	-.444	-.028	-.042
.700	-.577	-.086	-.075	-.424	-.047	-.047	-.447	-.032	-.021	-.408	-.015	-.009
.750	-.345			-.334	.018	.024	-.318	.028	.034	-.307	.045	.051
.800	-.250	.001	.001	-.233	.089	.100	-.218	.023	.059	-.212	.105	.116
.900	-.007	.689	.059	.018	.089	.130	.023	.127	.123	.029	.118	.132
.975	.149	.146	.118	.138	.135	.130	.127	.129	.138	.127	.135	.143
1.000	.176	.182	.176	.144	.166	.160	.138	.153	.138	.143	.154	.159

Outboard station												
.010	.082	-.213	-.125	-.307	.281	.341	-.490	.434	.522	-.615	.577	.627
.025	-.483	-.297	-.286	-.917	.061	.099	-.1067	.203	.225	-.1143	.368	.374
.050	-.385	-.286	-.330	-.714	.006	-.005	-.1111	.168	.131	-.1231	.247	.209
.075	-.423			-.620			-.1017			-.1187		
.100	-.472	.297	-.335	-.796	-.093	.093	-.1017	.005	.016	-.1203	.093	.121
.150	-.522	.286	-.280	-.972	-.099	.088	-.1083	-.039	.005	-.1247	.066	.071
.200	-.494	.280	-.242	-.725	-.110	.104	-.1019	-.028	.045	-.1016	.033	.011
.250	-.577	.242	-.242	-.802	-.104	.104	-.1056	-.039	.045	-.1231	.027	.017
.300	-.588	.247	-.244	-.802	-.121	.137	-.1067	-.072	.077	-.1209	.011	.016
.350	-.632	.263	-.268	-.835	-.142	.153	-.1067	-.082	.093	-.1203	.038	.043
.400	-.632	.253	-.275	-.818	-.159	.159	-.1050	-.110	.110	-.1231	.071	.045
.450	-.647	.257	-.268	-.822	-.169	.175	-.1050	.116	.132	-.1231	.081	.092
.500	-.582	.236	-.247	-.659	-.159	.170	-.1050	.116	.127	-.1231	.077	.082
.550	-.526	.213	-.230	-.537	-.142	.164	-.1050	.104	.126	-.1231	.077	.082
.600	-.516	.170	-.176	-.522	-.121	.126	-.1050	.099	.094	-.1231	.077	.082
.650	-.444	.131	-.142	-.433	-.087	.087	-.1050	.060	.060	-.1231	.077	.082
.700	-.379	-.088	-.088	-.379	-.066	.066	-.1050	.034	.034	-.1231	.077	.082
.750	-.296			-.273			-.1050			-.1231		
.800	-.203	.005	.000	-.192	.006	.011	-.1050	.001	.032	-.1231	.055	.060
.900	.022	.093	.093	.028	.094	.088	-.1050	.109	.104	-.1231	.137	.132
.975	.099	.122	.133	.121	.111	.122	-.1050	.116	.138	-.1231	.149	.160
1.000	.170	.093	.181	.143	.066	.143	-.1050	.137	.153	-.1231	.176	.181

(d) $M = 0.800$

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TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7.- Continued

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.241	.332	-.182	-.492	-.674	-.182	-.599	.682	.187	-.692	.327	-.882
.025	-.439	.134	-.256	-.674	.358	-.047	-.797	.797	.061	-.843	.564	-.721
.050	-.439	-.186	-.449	-.765	.194	-.204	-.883	.213	-.091	-.408	.296	-.518
.075	-.500			-.765						-.970		-.110
.100	-.586	.722	-.434	-.795	-.446	-.194	-.908	-.223	-.091	-.1005	-.094	-.018
.150	-.495	.753	-.439	.785	.457	-.285	-.923	.446	.152	-.1021	-.231	-.084
.200	-.335	.659	.510	.775	.441	-.249	-.913	.304	.157	-.1036	-.226	-.084
.250	-.535	-.646	-.459	.704	.355	-.194	-.939	.269	.127	-.1026	-.210	-.074
.300	-.555	.383	-.272	.730	.300	-.173	-.862	.218	.122	-.1021	-.180	-.079
.350	-.611	.296	-.276	.694	.218	-.183	-.807	.183	.152	-.1051	-.150	-.086
.400	-.657	.403	-.287	.745	.214	-.194	-.812	.157	.157	-.1051	-.135	-.114
.450	-.682	.301	-.266	.775	.198	-.183	-.822	.168	.152	-.895	-.140	-.800
.500	-.707	.282	-.251	.821	.199	-.178	-.868	.147	.157	-.894	-.135	-.889
.550	-.732	.231	-.196	.851	.158	-.149	-.929	.137	.137	-.956	-.120	-.838
.600	-.687	.196	-.165	.800	.138	-.123	-.842	.102	.122	-.894	-.099	-.800
.650	-.732	.145	-.115	.857	.108	-.088	-.623	.087	.132	-.472	-.085	-.587
.700	-.393	-.115	-.099	-.401	.087	-.072	-.421	-.061	-.081	-.367	-.064	-.572
.750	-.284		.262	-.262			-.261			-.284		-.525
.800	-.191	-.013	-.013	-.188	-.012	.004	-.167	.010	-.015	-.200	-.008	.002
.850	-.032	.078	.068	.004	.069	.074	-.000	.076	.066	-.089	.058	.063
.900	-.134	.136	.101	.090	.113	.098	.081	.094	.034	.033	.061	.021
.975	-.154	.164	.154	.110	.120	.120	.101	.106	.045	.043	.068	.017
1.000												
Outboard station												
.010	.153	-.244	-.177	-.156	.180	.262	-.293	.335	.397	-.387	.469	.520
.025	-.377	-.357	-.342	-.721	.032	-.003	-.049	.182	.166	-.922	.280	.245
.050	-.316	-.367	-.377	-.604	-.034	-.064	-.505	.069	.049	-.968	.148	.148
.075	-.347									-.937		
.100	-.443	.352	-.423	-.670	.146	-.156	-.854	-.028	-.043	-.947	.026	.036
.150	-.545	.352	-.342	-.838	.151	-.146	-.931	.048	.038	-.1034	.010	.010
.200	-.459	.336	-.301	.731	.161	-.151	-.803	.074	.034	-.861	.030	.046
.250	-.606	.436	-.291	.838	.156	-.151	-.951	.079	.034	-.1049	.035	.046
.300	-.642	.311	-.321	.848	.181	-.187	-.972	.109	.145	-.1064	-.076	.102
.350	-.657	.310	-.310	.868	.186	-.201	-.992	.159	.164	-.1064	-.111	.116
.400	-.703	.311	-.321	.878	.217	-.222	-.1028	.165	.191	-.973	.153	.158
.450	-.743	.310	-.315	.902	.216	-.226	-.1057	.210	.205	-.544	.157	.182
.500	-.810	.285	.301	.950	.217	-.222	-.1023	.170	.222	-.524	.173	.188
.550	-.819	.249	.264	.918	.191	-.216	-.552	.195	.195	-.462	.157	.197
.600	-.861	.209	.209	.568	.171	-.171	-.492	.145	.186	-.453	.163	.168
.650	-.452	.152	-.152	.348	.130	-.140	-.435	.134	.124	-.406	.131	.152
.700	-.316	.112	-.112	.283	.110	-.105	-.395	.094	.135	-.387	.127	.132
.750	-.223			.201			-.343			-.345		
.800	-.143	-.015	-.010	.151	.044	-.024	-.262	.033	.068	-.321	-.086	.081
.850	.036	.086	.086	.044	.037	.042	-.140	.039	.022	-.275	-.041	.056
.900	.137	.123	.138	.002	.043	.074	-.038	.009	.027	-.234	.131	.116
.975												
1.000												

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	.051	-.166	-.346	.280	-.572	.471	-.767	.577	-1.104	.769
.025	-.327	-.179	-.799	.143	-1.018	.297	-1.207	.410	-1.476	.601
.050	-.458	-.303	-.842	-.012	-1.217	.123	-1.437	.218	-1.749	.403
.075	-.489		-.799		-1.024		-1.307		-1.737	
.100	-.532	-.303	-.811	-.092	-.956	.024	-1.350	.106	-1.718	.266
.150	-.532		-.761		-.907		-.922		-1.693	
.200	-.526	-.259	-.706	-.098	-.814	-.032	-.867	.032	-1.669	.155
.250	-.526		-.687		-.776		-.829		-1.526	
.300	-.532	-.222	-.675	-.098	-.751	-.044	-.805	.007	-.906	.111
.350	-.532		-.656		-.708		-.761		-.745	
.400	-.550	-.222	-.656	-.129	-.708	-.081	-.755	-.043	-.714	.043
.450	-.550		-.650		-.689		-.718		-.707	
.500	-.532	-.216	-.606	-.136	-.640	-.094	-.662	-.061	-.664	.012
.550	-.513		-.563		-.590		-.606		-.608	
.600	-.495	-.166	-.532	-.111	-.553	-.081	-.557	-.055	-.565	.006
.650	-.464		-.495		-.503		-.507		-.509	
.700	-.420	-.098	-.439	-.061	-.441	-.044	-.445	-.024	-.447	.031
.750	-.358		-.365		-.354		-.352		-.354	
.800	-.272	-.011	-.259	.007	-.255	.018	-.241	.032	-.255	.074
.900	-.042	.069	-.024	.075	-.013	.074	-.005	.081	-.019	.118
.975	.131	.125	.112	.112	.105	.105	.100	.100	.124	.130
1.000	.150		.137		.123		.112		.149	
$M = 0.750$										
.010	.104	-.184	-.253	.267	-.432	.426	-.584	.556	-.841	.736
.025	-.302	-.218	-.688	.126	-.867	.273	-.980	.387	-1.202	.567
.050	-.438	-.336	-.897	-.044	-1.104	.098	-1.273	.195	-1.462	.374
.075	-.494		-.812		-1.042		-1.194		-1.462	
.100	-.540	-.336	-.818	-.106	-1.110	.002	-1.239	.082	-1.468	.244
.150	-.545		-.789		-1.087		-1.251		-1.468	
.200	-.545	-.280	-.761	-.117	-.811	-.037	-1.256	.014	-1.457	.148
.250	-.551		-.750		-.833		-1.222		-1.462	
.300	-.568	-.246	-.744	-.117	-.805	-.054	-.889	-.003	-1.462	.086
.350	-.568		-.699		-.777		-.827		-1.457	
.400	-.596	-.252	-.705	-.151	-.788	-.099	-.901	-.054	-1.163	.018
.450	-.602		-.722		-.777		-.782		-.880	
.500	-.579	-.240	-.654	-.151	-.681	-.110	-.675	-.071	-.778	-.004
.550	-.545		-.592		-.607		-.618		-.677	
.600	-.523	-.184	-.558	-.123	-.568	-.088	-.562	-.065	-.541	-.004
.650	-.494		-.507		-.506		-.505		-.434	
.700	-.438	-.110	-.439	-.072	-.438	-.048	-.438	-.031	-.383	.013
.750	-.353		-.349		-.342		-.336		-.321	
.800	-.257	-.014	-.241	.007	-.235	.019	-.229	.037	-.264	.064
.900	-.014	.070	-.004	.075	.002	.081	.003	.093	-.078	.109
.975	.127	.121	.115	.115	.104	.104	.099	.110	.098	.109
1.000	.149		.132		.121		.121		.160	

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(g) $M = 0.775$ and 0.800 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.775$										
.010	.146	-.185	-.179	.216	-.343	.405	-.483	.530	-.707	.698
.025	-.266	-.212	-.608	.108	-.771	.259	-.906	.362	-1.087	.530
.050	-.424	-.326	-.814	-.055	-.999	.080	-1.145	.194	-1.325	.356
.075	-.462		-.824		-.983		-1.101		-1.320	
.100	-.521	-.353	-.814	-.114	-1.053	-.007	-1.145	.375	-1.315	.226
.150	-.554		-.857		-1.075		-1.172		-1.342	
.200	-.554	-.288	-.743	-.125	-1.053	-.050	-1.177	.015	-1.347	.123
.250	-.559		-.732		-1.053		-1.177		-1.353	
.300	-.586	-.256	-.765	-.125	-.788	-.066	-1.192	-.012	-1.358	.074
.350	-.592		-.776		-.804		-1.183		-.946	
.400	-.630	-.256	-.814	-.158	-.847	-.110	-1.080	-.060	-.788	-.007
.450	-.662		-.846		-.907		-.911		-.745	
.500	-.640	-.250	-.770	-.163	-.929	-.121	-.911	-.077	-.718	-.040
.550	-.565		-.586		-.782		-.602		-.685	
.600	-.548	-.185	-.543	-.136	-.517	-.099	-.510	-.066	-.620	-.045
.650	-.499		-.494		-.462		-.456		-.555	
.700	-.440	-.109	-.429	-.076	-.408	-.050	-.386	-.028	-.490	-.029
.750	-.353		-.331		-.310		-.299		-.425	
.800	-.250	-.012	-.223	.005	-.207	.015	-.196	.043	-.376	.009
.900	-.006	.070	.010	.075	.015	.080	.015	.097	-.278	.009
.975	.124	.118	.113	.108	.102	.107	.113	.118	-.213	-.078
1.000	.151		.124		.123		.129		-.105	
$M = 0.800$										
.010	.175	-.211	-.120	.223	-.258	.388	-.367	.493	-.593	.664
.025	-.242	-.242	-.547	.098	-.680	.243	-.774	.342	-.969	.508
.050	-.414	-.367	-.760	-.068	-.915	.065	-1.024	.165	-1.204	.315
.075	-.461		-.760		-.889		-.993		-1.193	
.100	-.529	-.377	-.823	-.141	-.962	-.013	-1.039	.055	-1.199	.189
.150	-.570		-.833		-.993		-1.076		-1.235	
.200	-.560	-.310	-.823	-.141	-.993	-.055	-1.097	-.013	-1.246	.101
.250	-.581		-.766		-.993		-1.097		-1.251	
.300	-.607	-.268	-.708	-.141	-.988	-.075	-1.123	-.034	-1.084	.048
.350	-.638		-.739		-.972		-1.128		-.760	
.400	-.648	-.278	-.797	-.177	-.826	-.117	-1.128	-.086	-.708	-.025
.450	-.690		-.849		-.847		-1.149		-.672	
.500	-.711	-.268	-.870	-.183	-.910	-.128	-1.128	-.101	-.840	-.072
.550	-.721		-.911		-.930		-.883		-.619	
.600	-.575	-.205	-.880	-.146	-.967	-.107	-.643	-.086	-.583	-.082
.650	-.487		-.464		-.477		-.450		-.552	
.700	-.424	-.122	-.360	-.084	-.347	-.060	-.351	-.039	-.505	-.077
.750	-.330		-.276		-.258		-.268		-.479	
.800	-.226	-.018	-.177	.005	-.159	.018	-.169	.029	-.432	-.040
.900	.013	.066	.020	.072	.018	.081	.008	.081	-.369	-.051
.975	.118	.112	.109	.109	.091	.107	.097	.107	-.306	-.145
1.000	.139		.124		.117		.107		-.270	

TABLE 13.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Concluded

(h) $M = 0.825$; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.825$										
.010	.192	-.236	-.038	.163	-.168	.351	-.281	.458	-.489	.654
.025	-.206	-.251	-.469	.073	-.590	.220	-.678	.307	-.836	.493
.050	-.382	-.392	-.670	-.103	-.807	.019	-.919	.132	-1.088	.301
.075	-.432		-.690		-.807		-.899		-1.083	
.100	-.492	-.417	-.756	-.153	-.857	-.047	-.949	.031	-1.098	.170
.150	-.532		-.781		-.903		-.979		-1.128	
.200	-.542	-.331	-.801	-.163	-.923	-.072	-1.009	-.029	-1.153	.080
.250	-.552		-.796		-.928		-1.019		-1.158	
.300	-.588	-.296	-.730	-.163	-.943	-.092	-1.049	-.054	-1.052	.030
.350	-.623		-.740		-.938		-1.060		-.675	
.400	-.658	-.311	-.781	-.208	-.953	-.147	-1.060	-.110	-.635	-.046
.450	-.708		-.801		-.968		-1.110		-.605	
.500	-.743	-.296	-.841	-.213	-.883	-.152	-1.115	-.140	-.579	-.091
.550	-.764		-.886		-.898		-.723		-.554	
.600	-.774	-.221	-.916	-.178	-.893	-.137	-.582	-.130	-.534	-.101
.650	-.784		-.590		-.429		-.542		-.509	
.700	-.382	-.130	-.364	-.118	-.344	-.087	-.492	-.095	-.489	-.101
.750	-.271		-.269		-.273		-.411		-.459	
.800	-.170	-.030	-.188	-.028	-.203	-.011	-.331	-.019	-.444	-.076
.900	.021	.061	-.053	.038	-.097	.044	-.200	.016	-.398	-.076
.975	.106	.105	.012	.058	-.042	.049	-.095	.021	-.363	-.182
1.000	.126		.053		.034		.011		-.323	

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	.007	.158	.084	-.575	.468	.432	-.949	.596	.575	-.588	.675	.675
.025	.040	.040	-.005	.027	.250	.290	.007	.417	.423	.006	.487	.500
.050	.052	-.121	-.140	-.552	.123	.117	-.314	.225	.225	-.524	.321	.301
.075	.435			.793			.984			-.1424		
.100	.114	-.377	.185	-.530	-.082	.014	.981	.083	.109	-.1257	.160	.173
.150	.422	.223	.242	.539	-.063	.088	.910	.019	.019	-.936	.070	.070
.200	.441	.223	.242	.620	-.088	.101	.974	.013	.006	-.847	.026	.032
.250	.447	.230	.230	.601	-.107	.319	.692	.045	-.032	-.956	.000	.000
.300	.441	.236	.223	.620	-.139	.120	.641	.077	.051	-.815	.039	.019
.350	.447			.575			.641			-.834		
.400	.447	.242	.230	.556	-.159	.146	.609	.109	.083	-.763	.083	.077
.450				.703			.581	.122	.102	-.590	.103	.090
.500	.460	.223	.217	-.703	.165	.152				-.628	.038	.019
.550	.492	.153	.140	-.549	.114	.101	.571	.083	.122	-.622	.083	.058
.600										-.648	.019	.006
.650	.432	.063	.057	.396	-.037	.031	.372	.019	.006	-.391	.019	.039
.700												
.750	.204	.027	.040	-.280	.046	.046	.378	.058	.064	-.212	.006	.058
.800	.020	.129	.136	.027	.123	.130	.019	.128	.109	-.000	.115	.128
.900	.200			.181			.154			.147		
.975										-.148		
1.000	.232	.238	.225	.200	.200	.200	.192	.199	.192	.167	.173	.160

Outboard station												
.010	.257	-.134	-.134	-.682	.362	.192	-.852	.594	.346	-.1033	.702	.423
.025	.595	.289	.328	-.965	.162	-.051	-.1200	.367	.077	-.1394	.490	.174
.050	.450	-.450	-.375	-.772	-.044	-.121	-.1222	.187	-.020	-.1536	.322	.071
.075	.618			.785			-.1045			-.1536		
.100	.455	.631	.321	.798	.269	.147	.890	.110	.045	-.1413	.025	.012
.150	.412	.611	.276	.637	.263	.121	.729	.149	.020	-.755	.065	.019
.200	.431	.352	.205	.637	.218	.108	.748	.136	.026	-.787	.071	.006
.250	.457	.263	.199	.630	.166	.096	.729	.103	.026	-.794	.058	.006
.300	.457	.225	.180	.611	.154	.096	.690	.103	.032	-.768	.065	.007
.350	.483			.617			.690			-.775		
.400	.470	.147	.141	.592	-.096	.082	.652	.065	.032	-.710	.046	.026
.450												
.500	.437	.089	.109	.527	.057	.076	.561	.032	.045	-.607	.020	.039
.550				.463			.478	.000	.000	-.504	.012	.000
.600	.399	.031	.038		.012	.025				-.362	.051	.051
.650				.340	.040	.033	.336	.051	.051			
.700	.302	.032	.039		.091			.097	.109	-.175	.096	.103
.750				-.173	.051	.104	.161	.051	.155	.032	.148	.148
.800	.160	.061	.104	.040	.156	.145	.148	.148	.168	.129	.148	.148
.900	.035	.168	.168	.162	.111	.121						
.975	.188			.181								
1.000	.220	.168	.213	.181	.111	.121	.168	.084	.168	.148	.148	.148

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(d) M = 0.800

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.006	-.155	.112	-.242	.457	.397	-.447	.559	.541	-.523	.632	.614
.025	-.010	.035	.012	-.010	.254	.238	.038	.356	.415	-.038	.475	.475
.050	.346	-.107	-.231	-.139	.093	.060	.033	.221	.189	-.512	.302	.281
.075	-.527			-.753			-.917			-.992		
.100	.389	-.204	-.209	.001	.071	-.031	-.259	.275	.059	-.533	.356	.168
.150	-.506	-.436	-.403	-.732	-.118	-.118	-.922	-.102	-.021	-.1019	.076	.054
.200	-.527	-.500	-.473	-.721	-.145	-.145	-.927	-.043	-.048	-.1051	.022	.011
.250	-.517	-.482	-.326	-.721	-.141	-.231	-.925	.075	.035	-.1054	.010	-.043
.300	-.538	-.405	-.358	-.700	-.198	-.209	-.971	-.102	-.097	-.1035	-.086	-.043
.350	-.560			-.732			-.149			-.965		
.400	-.581	-.473	-.436	-.743	-.252	-.242	-.787	-.248	-.140	-.819	-.296	-.113
.450												
.500	-.557	-.409	-.425	-.802	-.247	-.220	-.830	-.286	-.259	-.873	-.129	-.194
.550												
.600	-.552	-.209	-.215	-.759	-.247	-.150	-.809	-.124	-.199	-.830	-.080	-.296
.650												
.700	-.457	-.134	-.188	-.328	-.091	-.052	-.280	-.086	-.027	-.275	-.016	-.010
.750												
.800	-.166	.028	.028	-.161	.028	.033	-.118	.059	.054	-.102	-.059	.065
.900	.066	.125	.130	.060	.114	.130	.065	.124	.119	.065	.141	.141
.975	.200			.163			.167			.178		
1.000	.222	.227	.211	.184	.184	.184	.189	.194	.184	.200	.200	.173
Outboard station												
.010	-.224	-.061	-.172	-.487	.356	.121	-.544	.518	.219	-.620	.636	.297
.025	-.554	-.174	-.364	-.754	.108	-.125	-.859	.302	-.007	-.896	.444	.302
.050	-.440	-.326	-.407	-.890	-.060	-.185	-.983	.134	-.056	-.1037	.286	.010
.075	-.326			-.223			-.321			-.728		
.100	-.527	-.678	.402	-.651	-.478	-.223	-.907	.213	.099	-.994	-.055	-.023
.150	-.494	-.665	-.364	-.814	-.413	-.174	-.962	.240	.067	-.1064	-.131	-.007
.200	-.500	-.711	-.277	-.908	-.326	-.142	-.902	.208	.067	-.1010	-.126	-.017
.250	-.581	-.608	-.272	-.846	-.234	-.142	-.935	.159	.061	-.1070	-.093	-.017
.300	-.570	-.483	-.239	-.835	-.207	-.125	-.945	.137	.056	-.1070	-.088	-.017
.350	-.592			-.852			-.983			-.1086		
.400	-.635	-.190	-.185	-.798	-.131	-.114	-.1000	-.077	-.061	-.1091	-.039	-.028
.450												
.500	-.695	-.093	-.147	-.883	-.082	-.109	-.1081	.039	.061	-.609	-.017	-.055
.550												
.600	-.451	-.050	-.060	-.510	-.039	-.044	-.468	.007	-.018	-.489	.021	-.017
.650												
.700	-.310	.016	.021	-.272	.027	.027	-.267	.053	.047	-.337	.064	.042
.750												
.800	-.131	.081	.097	-.109	.086	.102	-.083	.113	.118	-.164	.107	.096
.900	-.075	.151	.146	-.070	.157	.157	-.075	.167	.172	-.004	.178	.124
.975	.184			.157			.172			.107		
1.000	-.200	.146	.154	.173	.097	.184	.199	.129	.210	.129	.037	.075
Row C												
.753							-.807					
.589							.001					
.400							-.559					
.249							-.1291					
.082							-.575					
.009							-.1259					
-.031							-.1335					
-.026							-.1324					
-.004							-.1265					
-.052							-.828					
-.079							-.774					
-.096							-.710					
-.133							-.796					
-.193							-.516					
-.007							-.435					
-.058							-.284					
-.155							-.155					
.460							-.835					
.189							-.111					
.087							-.1236					
.038							-.558					
.043							-.1214					
.016							-.1290					
.016							-.160					
.000							-.019					
-.038							-.846					
-.071							-.694					
-.076							-.591					
-.065							-.493					
-.038							-.417					
-.000							-.342					
-.005							-.304					
-.163							-.244					

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	.117	.173	.103	-.17C	.406	.371	-.362	.535	.529	-.446	.622	.610
.025	.C34	-.075	.026	.023	.257	.251	.039	.356	.366	-.C43	.443	.443
.050	.444	-.142	-.185	.033	.C85	.054	.080	.179	.179	-.456	.267	.246
.075	-.455	-.455	-.185	-.710	.127	.064	-.824	.340	.065	-.925	-.438	.126
.100	.558	-.226	-.295	.179	.127	.165	-.169	.340	.065	-.342	.438	.126
.150	.425	-.403	.403	-.595	.191	.165	-.855	.019	.050	-.951	.012	.022
.200	.486	-.403	.403	-.773	.154	.165	-.881	-.085	.076	-.977	-.077	-.020
.250	.481	-.382	.435	-.590	.191	.165	-.861	-.107	.021	-.982	-.248	.326
.300	.507	-.340	.403	-.716	.227	.243	-.751	-.201	.027	-.977	-.170	.264
.350	.534	-.354	.403	-.590	.227	.243	-.772	-.206	.027	-.982	-.176	.264
.400	.559	-.460	.476	-.726	.357	.331	-.762	-.216	.206	-.925	-.176	.264
.450	.595	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.500	.637	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.550	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.600	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.650	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.700	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.750	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.800	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.850	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.900	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
.950	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
1.000	.616	-.512	.470	-.778	.383	.419	-.824	-.253	.216	-.836	-.217	.202
Outboard station												
.010	.158	-.C18	-.162	-.403	.283	.C13	-.455	.478	.177	-.529	.603	.264
.025	.496	-.151	-.386	-.285	.136	-.162	-.773	.283	.036	-.817	.376	.005
.050	.397	-.287	-.412	-.816	-.058	-.209	-.883	.105	-.089	-.937	.209	.048
.075	.224	-.684	-.423	.031	-.518	-.246	-.026	-.277	.130	-.398	-.152	.090
.100	.470	-.684	-.423	-.554	.518	-.246	-.810	-.277	.130	-.901	-.241	.058
.150	.558	-.658	.501	-.769	.663	.209	-.899	.319	.109	-.973	-.226	.053
.200	.445	-.684	.339	-.753	.466	.178	-.831	.266	.104	-.927	-.173	.063
.250	.527	-.632	.323	-.805	.258	.173	-.878	.199	.094	-.969	-.168	.063
.300	.574	-.565	-.271	-.816	.225	.162	-.883	.183	.094	-.989	-.173	.063
.350	.621	-.574	-.271	-.842	.141	.141	-.951	.109	.104	-.1005	-.116	.079
.400	.658	-.574	-.271	-.842	.141	.141	-.951	.109	.104	-.1005	-.116	.079
.450	.757	-.130	.161	-.910	-.099	-.141	-.868	-.073	.120	-.461	-.090	.110
.500	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.550	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.600	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.650	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.700	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.750	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.800	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.850	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.900	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
.950	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079
1.000	.784	-.052	-.062	-.328	-.063	-.084	-.422	-.047	.073	-.393	-.074	.079

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	.040	-.047	-.418	.371	-.656	.544	-.941	.623	-1.312	.773
.025	-.171	-.110	-.836	.221	-1.091	.263	-1.352	.465	-1.652	.623
.050	-.434	-.268	-.891	.048	-1.256	.173	-1.620	.244	-1.904	.441
.075	-.442		-.804		-1.170		-1.510		-1.896	
.100	-.465	-.229	-.765	-.023	-1.027	.094	-1.447	.141	-1.888	.307
.150	-.445		-.725		-.838		-1.257		-1.849	
.200	-.457	-.252	-.662	-.110	-.767	-.017	-.870	.031	-1.809	.149
.250	-.465		-.638		-.719		-.799		-1.715	
.300	-.473	-.260	-.631	-.126	-.688	-.040	-.767	-.017	-1.028	.094
.350	-.465		-.599		-.656		-.728		-.828	
.400	-.481	-.308	-.599	-.141	-.656	-.080	-.720	-.072	-.641	.023
.450	-.481		-.591		-.633		-.638		-.641	
.500	-.457	-.205	-.544	-.134	-.577	-.080	-.632	-.072	-.601	.007
.550	-.434		-.512		-.530		-.586		-.585	
.600	-.410	-.118	-.481	-.086	-.490	-.056	-.530	-.056	-.562	.007
.650	-.386		-.481		-.443		-.491		-.554	
.700	-.347	-.079	-.394	-.031	-.380	-.001	-.428	-.009	-.467	.031
.750	-.284		-.386		-.356		-.396		-.427	
.800	-.205	.048	-.262	.040	-.277	.062	-.341	.046	-.356	.086
.900	.024	.125	-.110	.111	.015	.118	-.143	.102	-.119	.126
.975	.182	.174	.150	.143	.141	.149	.110	.117	.141	.149
1.000	.505		.174		.157		.125		.165	
$M = 0.750$										
.010	-.019	-.046			-.500	.515	-.727	.593	-1.010	.753
.025	-.312	-.111			-.932	.349	-1.124	.442	-1.355	.595
.050	-.435	-.262			-1.163	.169	-1.369	.247	-1.607	.408
.075	-.471				-1.112		-1.326		-1.607	
.100	-.485	-.255			-1.084	.083	-1.304	.124	-1.600	.278
.150	-.507				-1.120		-1.362		-1.593	
.200	-.499	-.298			-1.048	-.032	-1.311	.002	-1.571	.127
.250	-.499				-.756		-1.297		-1.564	
.300	-.514	-.284			-.760	-.068	-1.052	-.034	-1.535	.062
.350	-.514				-.688		-.756		-1.384	
.400	-.535	-.276			-.672	-.184	-.705	-.085	-.924	-.010
.450	-.543				-.680		-.691		-.980	
.500	-.514	-.226			-.637	-.104	-.677	-.092	-.787	-.024
.550	-.478				-.544		-.568		-.729	
.600	-.456	-.147			-.500	-.068	-.532	-.063	-.744	-.024
.650	-.420				-.450		-.475		-.441	
.700	-.370	-.095			-.385	-.011	-.417	-.020	-.355	.005
.750	-.312				-.277		-.316		-.326	
.800	-.226	-.060			-.227	.061	-.222	-.056	-.290	.055
.900	-.125	.012			.004	.119	-.114	.103	-.067	.127
.975	.062	-.003			.148	.148	.016	.131	-.053	.098
1.000	.164				.162		.146		.113	

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(g) M = 0.775 and 0.800; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
M = 0.775										
.010	-.007	-.117	-.256	.317	-.415	.450	-.549	.506	-.881	.731
.025	-.290	-.117	-.677	.186	-.815	.338	-.929	.447	-1.206	.578
.050	-.421	-.276	-.891	-.042	-1.030	.158	-1.178	.253	-1.462	.385
.075	-.462		-.850		-1.015		-1.164		-1.462	
.100	-.483	-.276	-.878	-.166	-1.022	.062	-1.130	.163	-1.455	.260
.150	-.524		-.871		-1.057		-1.178		-1.455	
.200	-.517	-.324	-.719	-.153	-1.036	-.097	-1.178	.032	-1.455	.108
.250	-.517		-.746		-1.022		-1.164		-1.448	
.300	-.545	-.317	-.719	-.166	-.933	-.139	-1.150	-.003	-1.337	.039
.350	-.538		-.746		-.781		-1.144		-.943	
.400	-.566	-.310	-.788	-.222	-.829	-.111	-1.123	-.148	-.819	-.037
.450	-.593		-.774		-.884		-.881		-.777	
.500	-.566	-.243	-.657	-.194	-.663	-.104	-.687	-.141	-.736	-.072
.550	-.497		-.560		-.477		-.459		-.504	
.600	-.476	-.152	-.512	-.111	-.456	-.063	-.404	-.023	-.629	-.065
.650	-.425		-.456		-.415		-.362		-.518	
.700	-.379	-.069	-.387	-.077	-.346	-.001	-.314	.025	-.521	-.065
.750	-.290		-.298		-.277		-.238		-.438	
.800	-.207	-.041	-.222	-.042	-.155	.068	-.141	.094	-.348	.004
.900	-.221	.007	-.159	.027	-.070	.131	.004	.150	-.251	-.003
.975	.055	.028	.006	.055	.082	.165	.115	.177	-.168	-.079
1.000	.090		.006		.165		.053		-.175	
M = 0.800										
.010	.055	-.071	-.130	.015	-.344	.460	-.457	.566	-.734	.699
.025	-.230	-.117	-.595	.042	-.749	.320	-.829	.413	-1.053	.540
.050	-.423	-.290	-.814	-.005	-.968	.128	-1.081	.221	-1.331	.348
.075	-.456		-.794		-.942		-1.034		-1.338	
.100	-.496	-.283	-.828	-.078	-.942	-.012	-1.034	.123	-1.331	.228
.150	-.542		-.854		-.988		-1.094		-1.344	
.200	-.529	-.350	-.848	-.170	-.998	-.085	-1.101	-.065	-1.338	.076
.250	-.549		-.738		-.975		-1.114		-1.278	
.300	-.562	-.350	-.758	-.197	-.988	-.098	-1.108	-.078	-.900	-.017
.350	-.589		-.748		-.955		-1.108		-.747	
.400	-.622	-.350	-.794	-.217	-.836	-.145	-1.108	-.138	-.734	-.117
.450	-.642		-.861		-.862		-1.147		-.701	
.500	-.642	-.264	-.861	-.204	-.909	-.125	-1.134	-.118	-.674	-.137
.550	-.622		-.861		-.895		-.802		-.601	
.600	-.476	-.157	-.721	-.117	-.729	-.085	-.516	-.085	-.595	-.143
.650	-.423		-.396		-.377		-.364		-.555	
.700	-.363	-.131	-.323	-.051	-.291	-.019	-.271	-.078	-.522	-.077
.750	-.270		-.250		-.205		-.191		-.482	
.800	-.177	-.044	-.170	-.024	-.118	.061	-.105	.075	-.402	-.037
.900	-.071	-.028	-.078	.042	-.072	.121	.041	.128	-.343	-.037
.975	.102	.082	-.018	.029	.001	.108	.075	.114	-.283	-.124
1.000	.128		-.011		.035		.048		-.243	

TABLE 14.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Concluded

(h) $M = 0.825$; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.825$										
.010	-.602	-.060	-.144	.125	-.239	.420	-.356	.522	-.630	.689
.025	-.720	-.117	-.528	.042	-.649	.279	-.724	.369	-.957	.520
.050	-.886	-.277	-.739	-.041	-.847	.100	-.964	.144	-1.207	.337
.075	-.925		-.739		-.841		-.951		-1.220	
.100	-.969	-.284	-.771	-.099	-.860	-.060	-.958	.085	-1.207	.209
.150	-.921		-.803		-.924		-1.003		-1.222	
.200	-.921	-.367	-.803	-.208	-.924	-.150	-1.009	-.055	-1.232	.061
.250	-.921		-.784		-.924		-1.016		-1.188	
.300	-.972	-.373	-.779	-.240	-.937	-.150	-1.035	-.106	-.880	-.016
.350	-.991		-.759		-.937		-1.035		-.888	
.400	-.949	-.521	-.767	-.285	-.937	-.194	-1.046	-.183	-.656	-.092
.450	-.900		-.822		-.969		-1.049		-.924	
.500	-.713	-.264	-.854	-.220	-.999	-.175	-1.003	-.183	-.579	-.163
.550	-.713		-.861		-.892		-.999		-.554	
.600	-.725	-.169	-.891	-.144	-.515	-.130	-.546	-.151	-.522	-.185
.650	-.681		-.483		-.342		-.503		-.496	
.700	-.309	-.072	-.310	-.105	-.271	-.060	-.445	-.090	-.477	-.099
.750	-.213		-.070		-.207		-.388		-.470	
.800	-.117	-.002	-.144	-.073	-.137	-.000	-.232	-.010	-.413	-.060
.900	-.021	-.021	-.048	-.028	-.066	-.041	-.163	.029	-.438	-.073
.975	.011	.011	-.060	-.041	-.060	.030	-.074	.035	-.310	-.189
1.000	-.009		-.039		-.034		-.035		-.304	

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.301	-.232	-.163	-.715	-.556	-.190				-1.116	-.775	-.439
.025	-.477	-.003	-.215	-.871	.356	-.076				-1.335	-.612	-.295
.050	-.484	-.325	-.307	-.794	.076	-.070				-1.371	-.387	-.137
.075	-.550			-.812						-1.414		
.100	-.532	-.532	-.288	-.752	-.198	-.107				-1.085	-.052	.070
.150	-.519	-.434	-.313	-.703	-.247	-.155				-.897	-.064	.009
.200	-.501	-.345	-.321	-.654	-.216	-.155				-.842	-.015	-.015
.250	-.501	-.319	-.264	-.642	-.204	-.143				-.781	-.094	-.021
.300	-.495	-.288	-.246	-.612	-.192	-.143				-.739	-.088	-.027
.350	-.519	-.281	-.245	-.612	-.189	-.140				-.714	-.111	-.068
.400	-.507	-.228	-.203	-.587	-.149	-.125				-.678	-.076	-.033
.450	-.517	-.238	-.203	-.588	-.147	-.133				-.659	-.104	-.068
.500	-.495	-.185	-.161	-.557	-.125	-.107				-.611	-.045	-.033
.550	-.496	-.174	-.139	-.547	-.111	-.090				-.594	-.068	-.047
.600	-.404	-.118	-.054	-.441	-.076	-.058				-.471	-.015	-.003
.650	-.422	-.111	-.057	-.447	-.062	-.040				-.464	-.033	-.012
.700	-.386	-.069	-.051	-.395	-.003	-.021				-.398	-.003	-.022
.750	-.346	.016	.016	-.352	.039	.039				-.340	.064	.070
.800	-.252	.055	.089	-.247	.106	.106				-.222	.119	.125
.900	-.021	.137	.102	-.003	.158	.130				-.022	.116	.094
.975	.162	.137	.102	.149	.179	.173				.137	.161	.155
1.000	.192	.158	.152	.173	.179	.173				.155		
Outboard station												
.010	-.293	-.189	-.147	-.687	.323	.213				-.999	.677	.488
.025	-.630	-.341	-.306	-1.116	.123	.006				-1.446	.454	.221
.050	-.502	-.483	-.342	-.773	-.043	-.092				-1.520	.319	.129
.075	-.488			-.742						-1.452		
.100	-.502	-.624	-.318	-.729	-.282	-.147				-1.342	.007	.037
.150	-.508	-.522	-.283	-.674	-.239	-.110				-.809	-.048	.037
.200	-.502	-.515	-.259	-.650	-.202	-.104				-.796	-.048	.001
.250	-.526	-.450	-.214	-.668	-.141	-.104				-.833	-.042	.001
.300	-.520	-.422	-.220	-.662	-.159	-.116				-.790	-.055	-.018
.350	-.520	-.416	-.210	-.631	-.146	-.128				-.735	-.069	-.027
.400	-.538	-.381	-.208	-.637	-.141	-.116				-.729	-.061	-.048
.450	-.534	-.374	-.186	-.612	-.128	-.104				-.682	-.069	-.057
.500	-.502	-.146	.152	-.552	-.092	-.104				-.612	-.036	-.030
.550	-.466	-.122	.133	-.517	-.081	-.093				-.558	-.027	-.039
.600	-.459	-.085	-.079	-.490	-.043	-.049				-.514	.001	-.005
.650	-.411	-.057	-.045	-.429	-.016	-.016				-.440	.020	
.700	-.361	-.018	-.011	-.392	-.025	.019				-.367	.031	.019
.750	-.304			-.296						-.293		
.800	-.214	.044	.074	-.216	.068	.043				-.183	.007	.062
.892	.001	.123	.123	.018	.080	.129				-.042	.154	.142
.975	.145	.150	.150	.074	.137	.149				-.062	.132	.138
1.000	.172	.154	.156	.105	.141	.135				.050	.074	.148

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$			
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.289	.272	-.120	-.632	.584	.137	-.821	.685	.304	-.938	.767	.379
.025	-.465	.062	-.559	-.860	.377	.045	-1.016	.510	.155	-1.169	.605	.261
.050	-.493	-.310	-.326	-.793	.072	-.077	-1.099	.244	.005	-1.198	.361	.105
.075	-.548			-.793			-1.132			-1.282		
.100	-.526	-.632	-.304	-.793	-.227	-.105	-1.054	-.049	-.028	-1.248	.039	.072
.150	-.532	-.465	-.343	-.738	-.272	-.155	-.982	-.172	-.072	-1.254	-.082	.006
.200	-.521	-.371	-.326	-.693	-.233	-.161	-.738	-.167	-.089	-1.198	-.094	-.028
.250	-.532	-.338	-.292	-.677	-.222	-.150	-.798	-.167	-.089	-.743	-.105	-.028
.300	-.524	-.304	-.271	-.660	-.205	-.150	-.794	-.150	-.089	-.777	-.100	-.039
.350	-.554	-.298	-.266	-.666	-.208	-.169	-.755	-.157	-.111	-.827	-.111	-.060
.400	-.548	-.258	-.227	-.654	-.166	-.138	-.699	-.128	-.089	-.843	-.083	-.050
.450	-.567	-.244	-.214	-.632	-.176	-.150	-.686	-.124	-.099	-.702	-.092	-.060
.500	-.548	-.204	-.182	-.605	-.138	-.116	-.649	-.106	-.083	-.638	-.067	-.050
.550	-.551	-.189	-.152	-.584	-.124	-.105	-.606	-.079	-.066	-.600	-.053	-.034
.600	-.437	-.132	-.105	-.469	-.083	-.061	-.477	-.056	-.039	-.477	-.017	-.006
.650	-.448	-.111	-.071	-.455	-.074	-.053	-.460	-.047	-.027	-.460	-.021	-.001
.700	-.410	-.010	-.060	-.393	.000	-.032	-.411	-.000	-.000	-.394	.006	.017
.750	-.357			-.342	.033	.039	-.337	.044	.050	-.326	.067	.072
.800	-.243	.012	.306	-.222	.106	.111	-.222	.022	.111	-.205	.125	.133
.900	.001	.065	.084	.028	.134	.115	-.022	.105	.128	.033	.134	.121
.975	.156	.148	.103	.150	.148	.148	.139	.134	.144	.144	.177	.166
1.000	.184	.189	.184	.172	.172	.172	.161	.166	.161	.167	.216	.210
Outboard station												
.010	-.265	-.137	-.137	-.579	.325	.197	-.730	.503	.319	-.809	.636	.425
.025	-.662	-.282	-.328	-1.015	.081	.045	-1.156	.321	.019	-1.228	.450	.170
.050	-.536	-.450	-.355	-.887	-.081	-.104	-1.195	.136	-.003	-1.306	.243	.047
.075	-.489			-.857			-1.128			-1.267		
.100	-.489	-.556	-.344	-.747	.339	-.121	-1.010	-.132	-.059	-1.211	-.005	.025
.150	-.517	-.685	-.253	-.987	-.277	-.104	-1.044	-.149	-.037	-1.273	.076	.041
.200	-.517	-.489	-.237	-.966	-.216	-.098	-.932	-.154	-.043	-1.200	-.082	.002
.250	-.550	-.253	-.232	-.713	-.171	-.098	-.736	-.099	-.043	-.222	-.054	.008
.300	-.545	-.243	-.243	-.713	-.165	-.115	-.703	-.099	-.059	-.200	-.059	-.020
.350	-.550	-.223	-.228	-.703	-.153	-.137	-.747	-.116	-.084	-.965	-.073	-.035
.400	-.584	-.226	-.215	-.713	-.154	-.121	-.792	-.116	-.071	-.697	-.065	-.065
.450	-.594	-.154	-.201	-.675	-.137	-.137	-.735	-.105	-.094	-.627	-.067	-.057
.500	-.539	-.154	-.165	-.585	-.098	-.109	-.630	-.065	-.071	-.579	-.054	-.054
.550	-.502	-.137	-.142	-.535	-.083	-.099	-.546	-.051	-.073	-.541	-.024	-.040
.600	-.489	-.098	-.087	-.455	-.042	-.087	-.512	-.031	-.017	-.501	.002	-.009
.650	-.427	-.067	-.051	-.428	-.019	-.024	-.428	.002	-.003	-.422	.024	.024
.700	-.366	-.025	-.014	-.355	-.037	-.020	-.354	.030	.019	-.361	.047	.047
.750	-.294			-.282			-.272			-.272		
.800	-.204	.042	.046	-.177	.056	.086	-.166	.064	.103	-.171	.109	.058
.900	.020	.126	.126	.047	.131	.142	.047	.069	.142	.041	.159	.153
.975	.154	.145	.146	.148	.148	.148	.120	.137	.148	.144	.164	.164
1.000	.176	.171	.171	.148	.148	.159	.131	.125	.108	.164	.164	.164

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.261	.253	-.156	-.597	.561	.168	-.841	.762	.354	-.1.082	.855	.509
.025	-.447	.085	-.186	-.816	.377	.073	-.1.054	.617	.244	-.1.231	.744	.377
.050	-.479	-.282	-.309	-.985	.063	-.086	-.1.102	.244	.106	-.1.289	.563	.249
.075	-.559			-.842			-.1.177			-.1.359		
.100	-.522	-.639	-.309	-.778	-.246	-.118	-.1.171	.042	.042	-.1.359	.232	.180
.150	-.522	.279	-.341	.794	-.289	-.172	-.1.193	-.080	-.017	-.1.374	.057	.105
.200	-.522	-.373	-.350	-.653	-.241	-.182	-.1.171	-.094	-.038	-.1.380	.020	.073
.250	-.543	-.341	-.303	-.699	-.235	-.172	-.1.134	-.137	-.027	-.1.385	.028	.057
.300	-.554	-.304	-.277	-.699	-.219	-.166	-.1.047	-.107	-.049	-.1.369	.022	.041
.350	-.569	-.215	-.272	-.720	-.211	-.180	-.1.009	-.123	-.068	-.1.374	.026	.006
.400	-.575	-.256	-.229	-.746	-.182	-.155	-.1.049	-.091	-.049	-.1.374	.028	.004
.450	-.594	-.253	-.222	-.669	-.185	-.155	-.1.049	-.091	-.049	-.1.374	.028	.004
.500	-.575	-.208	-.181	-.663	-.150	-.129	-.1.049	-.091	-.049	-.1.374	.028	.004
.550	-.589	-.191	-.160	-.647	-.130	-.105	-.1.049	-.091	-.049	-.1.374	.028	.004
.600	-.447	-.128	-.107	-.470	-.081	-.070	-.1.049	-.091	-.049	-.1.374	.028	.004
.650	-.455	-.123	-.058	-.456	-.080	-.056	-.1.049	-.091	-.049	-.1.374	.028	.004
.700	-.399	-.000	-.059	-.395	-.004	-.028	-.1.049	-.091	-.049	-.1.374	.028	.004
.750	-.347			-.332			-.1.049	-.091	-.049	-.1.374	.028	.004
.800	-.234	.016	.016	-.209	.031	.036	-.1.049	-.091	-.049	-.1.374	.028	.004
.900	-.026	.106	.050	.031	.103	.103	-.1.049	-.091	-.049	-.1.374	.028	.004
.975	.165	.150	.112	.143	.136	.118	-.1.049	-.091	-.049	-.1.374	.028	.004
1.000	.186	.151	.186	.163	.169	.169	-.1.049	-.091	-.049	-.1.374	.028	.004
Outboard station												
.010	-.245	-.116	-.164	-.547	.267	.155	-.718	.620	.401	-.938	.791	.518
.025	-.637	-.256	-.331	-.960	.119	-.053	-.1.120	.446	.140	-.1.260	.617	.280
.050	-.492	-.444	-.379	-.971	-.075	-.118	-.1.206	.285	.065	-.1.330	.440	.172
.075	-.497			-.654			-.1.168			-.1.319		
.100	-.535	-.647	-.352	-.675	-.370	-.144	-.1.120	-.010	-.004	-.1.355	.183	.076
.150	-.519	-.674	-.306	-.934	-.300	-.128	-.1.211	-.074	.012	-.1.373	.054	.070
.200	-.556	-.578	-.251	-.772	-.241	-.128	-.1.147	-.079	-.004	-.1.217	.017	.043
.250	-.572	-.369	-.256	-.767	-.177	-.119	-.1.173	-.063	.001	-.1.217	.005	.043
.300	-.578	-.261	-.261	-.767	-.162	-.136	-.1.173	-.063	.001	-.1.217	.005	.043
.350	-.599	-.218	-.233	-.772	-.144	-.139	-.1.136	-.074	.043	-.1.217	.005	.043
.400	-.537	-.202	-.245	-.772	-.144	-.139	-.1.136	-.074	.043	-.1.217	.005	.043
.450	-.636	-.192	-.208	-.766	-.147	-.141	-.1.132	-.074	.043	-.1.217	.005	.043
.500	-.572	-.170	-.165	-.622	-.112	-.123	-.1.132	-.074	.043	-.1.217	.005	.043
.550	-.512	-.135	-.146	-.534	-.095	-.105	-.1.132	-.074	.043	-.1.217	.005	.043
.600	-.503	-.055	-.090	-.509	-.048	-.059	-.1.132	-.074	.043	-.1.217	.005	.043
.650	-.424	-.063	-.053	-.420	-.028	-.029	-.1.132	-.074	.043	-.1.217	.005	.043
.700	-.369	-.025	-.015	-.354	-.011	.000	-.1.132	-.074	.043	-.1.217	.005	.043
.750	-.285			-.245			-.1.132	-.074	.043	-.1.217	.005	.043
.800	-.186	.039	.066	-.160	.070	.043	-.1.132	-.074	.043	-.1.217	.005	.043
.900	.034	.114	.119	.043	.135	.135	-.1.132	-.074	.043	-.1.217	.005	.043
.975	.151	.148	.143	.140	.137	.148	-.1.132	-.074	.043	-.1.217	.005	.043
1.000	.173	.173	.173	.156	.145	.156	-.1.132	-.074	.043	-.1.217	.005	.043

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.237	-.322	-.134	-.555	-.744	.133	-.662	.689	.232	-.770	.763	.361
.025	-.451	-.111	-.206	-.744	.367	.003	-.831	.495	.131	-.959	.592	.203
.050	-.472	-.242	-.354	-.836	.060	-.125	-.913	.213	-.017	-.1010	.362	.070
.075	-.544			-.831			-.1000			-.1097		
.100	-.487	-.738	-.313	-.773	-.299	-.130	-.975	-.125	-.033	-.1077	.003	.034
.150	-.544	-.549	-.370	-.785	-.324	-.191	-.980	-.207	-.089	-.1092	-.114	-.038
.200	-.533	-.375	-.358	-.784	-.273	-.202	-.975	-.181	-.099	-.1087	-.120	-.053
.250	-.554	-.349	-.329	-.724	-.253	-.196	-.934	-.196	-.104	-.1067	-.125	-.055
.300	-.564	-.339	-.313	-.693	-.248	-.196	-.795	-.181	-.120	-.1087	-.125	-.058
.350	-.620	-.349	-.313	-.729	-.242	-.195	-.770	-.183	-.141	-.1062	-.141	-.093
.400	-.646	-.283	-.267	-.744	-.212	-.176	-.806	-.150	-.115	-.0857	-.104	-.075
.450	-.645	-.283	-.253	-.761	-.207	-.171	-.820	-.159	-.129	-.0845	-.123	-.093
.500	-.661	-.231	-.211	-.811	-.171	-.150	-.857	-.120	-.099	-.0898	-.089	-.068
.550	-.715	-.212	-.182	-.815	-.153	-.129	-.885	-.123	-.087	-.0924	-.087	-.064
.600	-.538	-.150	-.129	-.724	-.104	-.089	-.806	-.063	-.048	-.0729	-.038	-.037
.650	-.462	-.134	-.116	-.634	-.093	-.076	-.453	-.064	-.046	-.0428	-.040	-.028
.700	-.400	-.011	-.079	-.355	-.007	-.043	-.324	.003	.017	-.309	.008	.003
.750	-.333			-.289			-.254			-.244		
.800	-.216	.009	-.001	-.181	.019	.029	-.156	.039	.049	-.145	.049	.065
.850	.029	.051	.081	.034	.090	.095	.044	.111	.116	.044	.116	.131
.900	.152	.128	.098	.136	.127	.109	.141	.133	.121	.141	.133	.121
.950				.152		.152	.152		.162		.167	.167
1.000	.168	.178	.168	.152	.157	.157						
Outboard station												
.010	-.222	-.108	-.169	-.475	-.261	.107	-.563	.477	.272	-.646	.364	.364
.025	-.608	-.237	-.351	-.868	.102	-.104	-.955	.262	-.011	-.1038	.108	.108
.050	-.454	-.387	-.352	-.873	-.109	-.161	-.1012	.087	-.058	-.1105	.211	.035
.075	-.438			-.661			-.945			-.1069		
.100	-.505	-.732	-.371	-.651	-.481	-.171	-.904	-.243	-.088	-.1033	-.098	-.021
.150	-.567	-.375	-.325	-.862	-.372	-.145	-.697	-.217	-.068	-.1105	-.140	-.011
.200	-.536	-.640	-.268	-.852	-.269	-.140	-.935	-.176	-.068	-.1069	-.125	-.032
.250	-.619	-.438	-.253	-.842	-.181	-.140	-.960	-.135	-.063	-.1094	-.088	-.032
.300	-.629	-.335	-.273	-.857	-.212	-.158	-.971	-.135	-.094	-.1100	-.099	-.047
.350	-.639	-.329	-.253	-.837	-.180	-.153	-.1002	.125	.100	-.1100	-.100	-.066
.400	-.691	-.286	-.242	-.811	-.176	-.156	-.1048	.119	.109	-.1120	-.094	-.088
.450	-.735	-.154	-.219	-.820	-.160	-.160	-.1024	.125	.125	-.1004	-.098	-.090
.500	-.794	-.165	-.196	-.909	-.123	-.140	-.1095	-.083	-.094	-.094	-.078	-.088
.550	-.566	-.139	-.154	-.880	-.103	-.120	-.835	-.076	-.090	-.092	-.051	-.080
.600	-.480	-.108	-.108	-.517	-.063	-.068	-.517	-.063	-.047	-.052	-.042	-.040
.650	-.412	-.070	-.060	-.363	-.036	-.036	-.413	-.011	-.011	-.011	-.001	-.016
.700	-.382	-.031	-.016	-.290	-.001	-.004	-.305	.015	.020	-.393	-.001	.004
.750	-.268			-.220			-.220			-.299		
.800	-.175	.036	.062	-.135	.051	.046	-.114	.077	.097	-.228	.041	.056
.850	-.041	.119	.108	-.035	.133	.128	-.046	.149	.087	-.099	.118	.056
.900	-.149	.143	.143	.139	.133	.143	.149	.148	.168	-.006	.098	.078
.950				.154	.139	.154	.175	.133	.139	-.001	.025	.025
1.000	.170	.160	.165	.154	.139	.154						

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(e) $M = 0.825$

x/c	C_p at -								
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station									
.010	-.244	.330	-.158	-.477	.574	.119	-.597	.674	.219
.025	-.412	.151	-.204	-.679	.377	.032	-.765	.505	.100
.050	-.451	-.264	-.352	-.753	.076	-.111	-.834	.219	-.038
.075	-.500			-.783			-.937		
.100	-.530	-.712	-.313	-.709	-.309	-.126	-.883	-.137	-.058
.150	-.505	-.727	-.387	-.778	-.334	-.200	-.919	-.241	-.122
.200	-.535	-.318	-.382	-.773	-.269	-.205	-.919	-.211	-.137
.250	-.525	-.338	-.347	-.704	-.255	-.195	-.918	-.211	-.137
.300	-.550	-.347	-.328	-.724	-.250	-.200	-.878	-.211	-.152
.350	-.599	-.363	-.340	-.654	-.267	-.232	-.804	-.210	-.164
.400	-.634	-.328	-.288	-.724	-.220	-.185	-.814	-.187	-.152
.450	-.665	-.366	-.277	-.742	-.221	-.192	-.773	-.193	-.164
.500	-.658	-.254	-.229	-.784	-.185	-.166	-.844	-.147	-.142
.550	-.727	-.226	-.157	-.814	-.157	-.150	-.864	-.141	-.124
.600	-.692	-.165	-.140	-.789	-.106	-.092	-.848	-.093	-.083
.650	-.735	-.134	-.122	-.800	-.106	-.077	-.528	-.089	-.078
.700	-.446	-.012	-.066	-.353	.002	-.042	-.340	.001	-.043
.750	-.268			-.245			-.250		
.800	-.195	-.002	-.007	-.141	.022	.017	-.152	.021	.021
.900	.042	.082	.067	.027	.096	.091	.001	.085	.080
.975	.131	.125	.050	.106	.113	.090	.090	.106	.077
1.000	.151	.156	.151	.133	.155	.135	.120	.125	.100
Outboard station									
.010	-.265	-.085	-.194	-.395	.266	.092	-.471	.438	.210
.025	-.568	-.200	-.264	-.778	.087	-.102	-.864	.251	-.003
.050	-.419	-.234	-.394	-.788	-.087	-.161	-.909	.052	-.088
.075	-.424			-.569			-.834		
.100	-.484	-.732	-.394	-.594	-.524	-.181	-.834	-.302	-.127
.150	-.609	-.673	-.359	-.783	-.430	-.161	-.909	-.282	-.103
.200	-.484	-.707	-.290	-.778	-.286	-.151	-.854	-.217	-.103
.250	-.563	-.588	-.280	-.823	-.196	-.146	-.904	-.162	-.098
.300	-.613	-.454	-.300	-.923	-.191	-.161	-.924	-.162	-.142
.350	-.638	-.287	-.277	-.843	-.187	-.173	-.934	-.159	-.131
.400	-.683	-.200	-.260	-.873	-.221	-.171	-.984	-.157	-.152
.450	-.727	-.181	-.239	-.848	-.168	-.177	-.959	-.155	-.155
.500	-.732	-.151	-.195	-.932	-.136	-.166	-.944	-.123	-.157
.550	-.764	-.138	-.162	-.828	-.110	-.130	-.943	-.107	-.126
.600	-.852	-.116	-.121	-.770	-.087	-.092	-.841	-.073	-.093
.650	-.443	-.076	-.071	-.531	-.048	-.043	-.413	-.049	-.064
.700	-.280	-.046	-.021	-.251	-.017	-.017	-.352	-.023	-.033
.750	-.261			-.173			-.293		
.800	-.146	.028	.053	-.107	.043	.053	-.217	.032	.027
.900	.048	.104	.117	-.007	.117	.077	-.168	.080	.052
.975	.138	.130	.130	.058	.067	.105	.002	.090	.032
1.000	.148	.132	.148	.073	.073	.092	.047	.037	-.013
$\alpha = 4^\circ$									
Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row C
-.861	.852	.301	-.861	.852	.301	-.861	.852	.301	.488
-1.033	.738	.175	-1.033	.738	.175	-1.033	.738	.175	.338
-1.073	.541	.057	-1.073	.541	.057	-1.073	.541	.057	.215
-1.132			-1.132			-1.132			
-1.127	.166	.017	-1.127	.166	.017	-1.127	.166	.017	.141
-1.167	.032	-.052	-1.167	.032	-.052	-1.167	.032	-.052	.082
-1.172	-.012	-.047	-1.172	-.012	-.047	-1.172	-.012	-.047	.047
-1.191	-.037	.086	-1.191	-.037	.086	-1.191	-.037	.086	.022
-1.191	-.066	.003	-1.191	-.066	.003	-1.191	-.066	.003	.003
-1.198	-.111	-.048	-1.198	-.111	-.048	-1.198	-.111	-.048	-.048
-1.037	-.061	-.111	-1.037	-.061	-.111	-1.037	-.061	-.111	-.037
-.650	-.099	-.123	-.650	-.099	-.123	-.650	-.099	-.123	-.059
-.629	-.061	-.116	-.629	-.061	-.116	-.629	-.061	-.116	.066
-.617	-.088	.115	-.617	-.088	.115	-.617	-.088	.115	.059
-.589	-.037	-.072	-.589	-.037	-.072	-.589	-.037	-.072	-.053
-.569	-.065	-.059	-.569	-.065	-.059	-.569	-.065	-.059	-.053
-.530	-.022	-.042	-.530	-.022	-.042	-.530	-.022	-.042	-.037
-.521			-.521			-.521			
-.461	.008	.012	-.461	.008	.012	-.461	.008	.012	-.017
-.367	.027	.076	-.367	.027	.076	-.367	.027	.076	.012
-.279	-.048	.056	-.279	-.048	.056	-.279	-.048	.056	-.145
-.219	-.150	.071	-.219	-.150	.071	-.219	-.150	.071	-.269

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	.059	-.192	-.352	.267			-.812	.574		
.025	-.344	-.210	-.806	.149			-1.273	.397		
.050	-.486	-.310	-.855	-.034			-1.520	.197		
.075	-.493		-.813				-1.393			
.100	-.557	-.352	-.813	-.146			-1.372	.067		
.150	-.550		-.763				-.940			
.200	-.542	-.281	-.699	-.134			-.869	.002		
.250	-.543		-.578				-.833			
.300	-.543	-.239	-.671	-.128			-.805	-.022		
.350	-.543		-.657				-.770			
.400	-.550	-.234	-.650	-.146			-.755	-.057		
.450	-.550		-.629				-.720			
.500	-.536	-.204	-.593	-.140			-.663	-.063		
.550	-.514		-.551				-.607			
.600	-.493	-.151	-.522	-.105			-.564	-.045		
.650	-.465		-.480				-.508			
.700	-.422	-.080	-.423	-.046			-.444	-.010		
.750	-.344		-.339				-.352			
.800	-.259	.014	-.239	.031			-.246	.049		
.900	-.040	.085	-.012	.050			-.012	.091		
.975	.123	.132	.122	.126			.087	.120		
1.000	.151		.144				.102			
$M = 0.750$										
.010	.056	-.190	-.260	.239	-.454	.422	-.589	.535	-.894	.739
.025	-.304	-.228	-.718	.127	-.900	.277	-1.035	.379	-1.269	.562
.050	-.459	-.341	-.393	-.051	-1.120	.078	-1.294	.186	-1.535	.373
.075	-.498		-.854		-1.068		-1.242		-1.528	
.100	-.550	-.375	-.802	-.153	-1.100	-.035	-1.236	.062	-1.502	.217
.150	-.563		-.809		-1.126		-1.287		-1.509	
.200	-.556	-.309	-.777	-.142	-.855	-.067	-1.287	.003	-1.509	.136
.250	-.556		-.744		-.842		-1.268		-1.515	
.300	-.552	-.266	-.744	-.137	-.842	-.078	-.900	-.030	-1.522	.088
.350	-.569		-.599		-.790		-.822		-1.522	
.400	-.595	-.260	-.699	-.159	-.758	-.111	-.861	-.062	-.991	.029
.450	-.595		-.699		-.725		-.673		-.894	
.500	-.582	-.223	-.641	-.148	-.680	-.111	-.641	-.067	-.810	.007
.550	-.543		-.583		-.603		-.589		-.688	
.600	-.524	-.163	-.550	-.110	-.557	-.078	-.551	-.051	-.578	.013
.650	-.485		-.499		-.493		-.499		-.410	
.700	-.434	-.083	-.434	-.051	-.422	-.035	-.421	-.014	-.351	.040
.750	-.327		-.331		-.325		-.318		-.280	
.800	-.246	.009	-.227	.030	-.221	.040	-.208	.051	-.209	.099
.900	-.001	.084	.012	.094	.018	.094	.024	.105	-.015	.142
.975	.135	.132	.121	.127	.108	.121	.121	.126	.134	.190
1.000	.154		.134		.134		.134		.172	

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(g) $M = 0.775$ and 0.800 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.775$										
.010	.125	-.197	-.198	.230			-.507	.520	-.756	.705
.025	-.290	-.213	-.657	.122			-.641	.370	-1.122	.524
.050	-.451	-.342	-.262	-.059			-1.177	.174	-1.382	.333
.075	-.488		-.837				-1.133		-1.388	
.100	-.556	-.378	-.849	-.162			-1.154	.045	-1.376	.189
.150	-.587		-.899				-1.226		-1.401	
.200	-.575	-.321	-.728	-.162			-1.226	-.002	-1.401	.106
.250	-.575		-.756				-1.220		-1.412	
.300	-.606	-.280	-.787	-.162			-1.220	-.027	-1.240	.044
.350	-.606		-.806				-1.214		-.837	
.400	-.625	-.245	-.827	-.178			-1.195	-.074	-.781	-.023
.450	-.656		-.812				-.941		-.750	
.500	-.631	-.228	-.594	-.162			-.848	-.084	-.719	-.049
.550	-.556		-.583				-.545		-.669	
.600	-.533	-.161	-.545	-.121			-.483	-.064	-.601	-.059
.650	-.488		-.490				-.458		-.530	
.700	-.420	-.094	-.465	-.059			-.408	-.017	-.477	-.033
.750	-.340		-.322				-.291		-.452	
.800	-.222	.009	-.211	.024			-.185	.060	-.496	.008
.900	.013	.086	.025	.091			.032	.112	-.254	.019
.975	.121	.132	.118	.127			.131	.133	-.222	-.074
1.000	.156		.136				.156		-.149	
$M = 0.800$										
.010	.146	-.214	-.135	.207	-.284	.361	-.415	.485	-.623	.669
.025	-.265	-.219	-.552	.068	-.690	.227	-.821	.332	-.993	.495
.050	-.408	-.397	-.761	-.105	-.940	.039	-1.065	.138	-1.255	.307
.075	-.480		-.779		-.910		-1.035		-1.231	
.100	-.528	-.397	-.839	-.180	-.976	-.071	-1.065	.034	-1.231	.158
.150	-.593		-.845		-1.023		-1.125		-1.273	
.200	-.587	-.353	-.868	-.180	-1.023	-.090	-1.149	-.036	-1.279	.079
.250	-.611		-.797		-1.005		-1.142		-1.279	
.300	-.635	-.303	-.767	-.175	-1.035	-.105	-1.166	-.056	-.969	.029
.350	-.665		-.767		-1.017		-1.166		-.743	
.400	-.718	-.252	-.839	-.195	-.874	-.145	-1.172	-.090	-.707	-.040
.450	-.742		-.880		-.868		-1.220		-.677	
.500	-.736	-.243	-.880	-.180	-.934	-.150	-1.184	-.105	-.629	-.070
.550	-.754		-.922		-.940		-.880		-.617	
.600	-.617	-.165	-.845	-.130	-.839	-.100	-.588	-.080	-.582	-.080
.650	-.468		-.439		-.433		-.457		-.510	
.700	-.420	-.050	-.362	-.066	-.338	-.041	-.350	-.031	-.480	-.070
.750	-.325		-.272		-.248		-.248		-.469	
.800	-.218	.010	-.171	.019	-.153	.039	-.159	.049	-.397	-.031
.900	.021	.084	.026	.088	.026	.093	.020	.108	-.349	-.050
.975	.116	.129	.109	.118	.091	.113	.109	.133	-.427	-.140
1.000	.134		.127		.121		.139		-.272	

TABLE 15.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Concluded

(h) $M = 0.825$; center station

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
M = 0.825										
.010	.171	-.188	-.060	.172	-.181	.315	-.301	.450	-.524	.641
.025	-.220	-.320	-.485	.062	-.601	.200	-.709	.297	-.886	.474
.050	-.386	-.387	-.692	-.120	-.849	.023	-.956	.110	-1.145	.278
.075	-.450		-.715		-.843		-.921		-1.145	
.100	-.513	-.387	-.773	-.192	-.901	-.083	-.979	-.019	-1.128	.144
.150	-.576		-.819		-.952		-1.025		-1.191	
.200	-.570	-.363	-.819	-.201	-.952	-.116	-1.054	-.062	-1.202	.053
.250	-.570		-.842		-.953		-1.054		-1.214	
.300	-.639	-.344	-.907	-.201	-.993	-.140	-1.077	-.086	-.996	.005
.350	-.639		-.778		-.993		-1.088		-.674	
.400	-.708	-.320	-.818	-.211	-.993	-.174	-1.094	-.139	-.634	-.072
.450	-.737		-.847		-1.021		-1.151		-.616	
.500	-.782	-.263	-.893	-.192	-.935	-.164	-1.163	-.139	-.582	-.100
.550	-.783		-.893		-.947		-.669		-.559	
.600	-.800	-.186	-.933	-.144	-.670	-.131	-.605	-.120	-.553	-.110
.650	-.823		-.514		-.400		-.559		-.501	
.700	-.386	-.100	-.347	-.077	-.331	-.078	-.502	-.072	-.484	-.091
.750	-.254		-.261		-.251		-.404		-.461	
.800	-.151	-.004	-.163	.009	-.170	-.001	-.347	-.000	-.456	-.057
.900	.021	.072	-.037	.062	-.061	.051	-.105	.038	-.410	-.072
.975	.096	.110	.021	.076	-.015	.047	-.025	.043	-.369	-.177
1.000	.118		.067		.020		.050		-.346	

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.332	.258	-.123	-.795	.604	.143	-.955	.711	.330	-1.177	.784	.432
.025	-.006	.058	-.224	.000	.411	.064	-.013	.564	.186	-.005	.648	.283
.050	-.378	-.231	-.301	-.186	.167	-.077	-.910	.308	.007	-.517	.431	.130
.075	-.526	-.075	-.827	-.827	-.865	-.865	-1.433	.007	.007	-1.433	.007	.007
.100	-.795	-.583	-.288	-1.077	.263	-.109	-1.307	.321	.070	-1.382	.405	.066
.150	-.513	-.355	-.295	-.699	-.179	-.141	-.846	-.122	.045	-.889	-.011	.008
.200	-.506	-.295	-.276	-.673	-.160	-.141	-.763	-.128	.044	-.857	-.037	.008
.250	-.454	-.256	-.237	-.647	-.154	-.122	-.711	-.128	.051	-.793	-.043	.058
.300	-.494	-.237	-.212	-.622	-.147	-.109	-.679	-.089	.051	-.754	-.056	.011
.350	-.464	-.222	-.192	-.602	-.128	-.109	-.640	-.077	.057	-.728	-.056	.030
.400	-.500	-.205	-.186	-.602	-.122	-.102	-.602	-.077	.070	-.696	-.056	.043
.450	-.454	-.192	-.160	-.570	-.122	-.102	-.602	-.077	.070	-.696	-.056	.043
.500	-.454	-.141	-.135	-.590	-.096	-.083	-.622	-.057	.037	-.664	-.043	.037
.550	-.454	-.083	-.070	-.404	-.057	-.057	-.397	-.038	.019	-.415	-.011	.005
.600	-.378	.006	.000	-.250	.019	.026	-.237	.045	.007	-.229	.040	.046
.650	-.250	.013	.090	.000	.096	.103	.007	.109	.109	-.069	.104	.110
.700	-.160	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
.750	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
.800	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
.850	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
.900	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
.950	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
1.000	-.192	.199	.192	.154	.173	.161	.103	.167	.167	-.104	.104	.103
Outboard station												
.010	-.000	-.197	-.135	-.007	.361	.222	-.000	.578	.361	-.012	.703	.471
.025	-.000	-.323	-.323	-.013	.148	-.026	-1.309	.200	.129	-.012	.491	.227
.050	-.490	-.439	-.336	-.306	-.007	-.077	-.419	-.052	.026	-1.545	.323	.104
.075	-.174	-.574	-.323	-.284	-.226	-.136	-.039	-.084	.019	-.546	.059	.033
.100	-.155	-.478	-.252	-.058	-.206	-.102	-.761	-.077	.019	-.141	.012	.033
.150	-.478	-.490	-.220	-.677	-.206	-.110	-.832	-.077	.019	-.832	-.031	.005
.200	-.536	-.310	-.210	-.729	-.161	-.110	-.767	-.058	.045	-.856	-.031	.005
.250	-.523	-.213	-.213	-.584	-.123	-.123	-.729	-.322	.039	-.817	-.031	.001
.300	-.516	-.207	-.213	-.658	-.129	-.123	-.709	-.071	.058	-.791	-.031	.018
.350	-.516	-.207	-.213	-.658	-.129	-.123	-.709	-.071	.058	-.791	-.031	.018
.400	-.525	-.181	-.194	-.651	-.116	-.123	-.697	-.071	.071	-.765	-.044	.046
.450	-.525	-.181	-.194	-.651	-.116	-.123	-.697	-.071	.071	-.765	-.044	.046
.500	-.497	-.142	-.155	-.581	-.084	-.102	-.522	-.052	.065	-.643	-.031	.044
.550	-.458	-.091	-.084	-.510	-.052	-.058	-.387	-.019	.032	-.540	-.005	.012
.600	-.361	-.032	-.020	-.387	-.000	-.000	-.200	.019	.019	-.392	.033	.027
.650	-.220	.032	.045	-.213	.045	.058	.019	.064	.064	-.205	.072	.072
.700	-.000	.109	.090	-.006	.116	.102	.064	.116	.110	-.014	.124	.104
.750	-.142	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117
.800	-.161	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117
.850	-.161	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117
.900	-.161	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117
.950	-.161	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117
1.000	-.161	.109	.155	.142	.064	.148	.355	.039	.116	.085	.046	.117

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$			
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.389	.302	-.150	-.692	.591	.139	-.857	.716	.296	-1.014	.775	.388
.025	-.005	.090	-.208	-.003	.418	.038	.003	.352	.161	-.003	.652	.184
.050	-.249	-.249	-.424	-.061	.126	-.114	-.734	.301	.036	-.383	.518	.091
.075	-.658	-.711	-.611	-.891	.360	-.114	-1.161	.406	-.015	-1.172	.476	.067
.100	-.655	-.401	-.506	-.716	.196	.149	-.985	.097	-.138	-1.236	.009	.067
.150	-.524	-.319	-.442	-.716	.178	.155	-.978	.097	-.067	-1.242	-.026	.009
.200	-.518	-.250	-.272	-.575	.213	.330	-.781	.108	-.062	-.751	-.055	-.009
.250	-.518	-.266	-.243	-.575	.155	.125	-.786	.097	-.062	-.774	-.126	-.009
.300	-.518	-.266	-.243	-.575	.155	.125	-.786	.097	-.062	-.774	-.126	-.009
.350	-.522	-.243	-.208	-.675	.137	.120	-.769	.108	-.073	-.839	-.050	-.032
.400	-.541	-.225	-.190	-.616	.131	.120	-.640	.108	-.132	-.646	-.061	-.050
.450	-.511	-.167	-.149	-.722	.102	.155	-.699	.073	-.155	-.704	-.050	-.038
.500	-.553	-.167	-.149	-.722	.102	.155	-.699	.073	-.155	-.704	-.050	-.038
.550	-.553	-.167	-.149	-.722	.102	.155	-.699	.073	-.155	-.704	-.050	-.038
.600	-.553	-.167	-.149	-.722	.102	.155	-.699	.073	-.155	-.704	-.050	-.038
.650	-.553	-.167	-.149	-.722	.102	.155	-.699	.073	-.155	-.704	-.050	-.038
.700	-.401	-.163	-.085	-.406	.055	.049	-.406	.038	-.021	-.400	.009	.003
.750	-.243	-.003	-.005	-.225	.027	.021	-.225	.073	.014	-.207	.050	.061
.800	.008	.090	-.005	-.027	.102	.026	.026	.026	.102	-.026	.114	.114
.850	.143	.143	.143	.149	.149	.149	.131	.131	.131	.126	.126	.126
.900	.184	.184	.184	.161	.178	.167	.161	.161	.161	.167	.167	.167
.950	.184	.184	.184	.161	.178	.167	.161	.161	.161	.167	.167	.167
1.000	.184	.184	.184	.161	.178	.167	.161	.161	.161	.167	.167	.167
Outboard station												
.010	-.004	-.152	-.152	-.010	.329	.188	-.010	.315	.117	-.004	.661	.407
.025	-.004	-.304	-.304	-.010	.155	.045	-.127	.166	.019	-.004	.484	.167
.050	-.004	-.457	-.457	-.010	.010	-.045	-.228	.092	-.039	-1.345	.308	.078
.075	-.005	-.553	-.553	-.121	.274	.145	.056	.110	-.028	-.333	.020	.031
.100	-.005	-.553	-.553	-.121	.274	.145	.056	.110	-.028	-.333	.020	.031
.150	-.481	-.604	-.281	-.997	.221	.115	-.185	.092	-.051	-1.321	.016	.083
.200	-.561	-.410	-.246	-.750	.174	.115	-.780	.075	-.045	-1.292	.002	.002
.250	-.569	-.240	-.240	-.727	.133	.115	-.704	.081	-.069	-1.262	.016	.004
.300	-.537	-.216	-.240	-.715	.139	.127	-.757	.081	-.081	-1.215	.039	.016
.350	-.569	-.216	-.240	-.721	.121	.127	-.798	.083	-.075	-.833	.039	.039
.400	-.587	-.198	-.204	-.721	.121	.127	-.633	.069	-.075	-.680	.039	.039
.450	-.551	-.157	-.169	-.515	.092	.110	-.522	.033	-.028	-.604	.027	.045
.500	-.551	-.157	-.169	-.515	.092	.110	-.522	.033	-.028	-.604	.027	.045
.550	-.498	-.104	-.098	-.521	.051	.057	-.369	.014	.019	-.515	.004	.037
.600	-.498	-.104	-.098	-.521	.051	.057	-.369	.014	.019	-.515	.004	.037
.650	-.381	-.040	-.028	-.374	.002	.004	-.175	.061	.072	-.368	.043	.037
.700	-.381	-.040	-.028	-.374	.002	.004	-.175	.061	.072	-.368	.043	.037
.750	-.204	.025	.043	-.186	.055	.055	.031	.119	.108	-.186	.090	.084
.800	.013	.107	.084	.026	.120	.102	.125	.031	.125	.020	.137	.126
.850	.142	.142	.142	.131	.055	.142	.102	.102	.102	.137	.067	.161
.900	.142	.142	.142	.131	.055	.142	.102	.102	.102	.137	.067	.161
.950	.142	.142	.142	.131	.055	.142	.102	.102	.102	.137	.067	.161
1.000	.160	.078	.143	.137	.055	.143	.137	.090	.119	.137	.067	.161

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(c) $M = 0.775$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.288	.304	-.156	-.618	.600	.122	-.776	.719	.285	-.920	.782	.367
.025	-.131	.105	-.187	-.001	.414	-.021	.004	.548	.015	-.005	.644	.262
.050	-.305	-.260	-.344	-.012	.128	-.091	-.664	.234	.015	-.332	.431	.111
.075	-.585	-.585	-.831	-.831	-.408	-.130	-.1.036	.464	-.024	-.1.195	.520	.049
.100	-.647	-.652	-.305	-.864	-.220	-.163	-.1.034	-.092	-.075	-.1.010	-.023	-.012
.150	-.565	-.383	-.349	-.825	-.192	-.163	-.1.000	-.114	-.080	-.1.156	-.046	-.018
.200	-.585	-.316	-.316	-.657	-.186	-.147	-.731	-.116	-.069	-.1.173	-.057	-.018
.250	-.596	-.288	-.282	-.596	-.169	-.135	-.745	-.108	-.069	-.1.122	-.063	-.018
.300	-.596	-.276	-.417	-.702	-.169	-.135	-.745	-.108	-.069	-.1.122	-.063	-.018
.350	-.624	-.248	-.220	-.730	-.163	-.130	-.782	-.103	-.080	-.1.392	-.057	-.035
.400	-.624	-.248	-.220	-.763	-.163	-.130	-.804	-.103	-.080	-.1.016	-.057	-.035
.450	-.541	-.226	-.192	-.679	-.152	-.130	-.759	-.120	-.080	-.836	-.074	-.051
.500	-.550	-.164	-.159	-.574	-.119	-.107	-.810	-.086	-.092	-.870	-.051	-.035
.600	-.502	-.103	-.086	-.399	-.068	-.051	-.372	-.075	-.024	-.371	-.018	-.012
.700	-.400	.004	-.007	-.208	.021	.027	-.187	.032	.049	-.186	.049	.061
.800	-.349	.021	.094	-.094	.100	.100	-.150	.110	.043	-.044	.111	.117
.900	-.021	.054	.094	.145	.167	.167	.150	.166	.172	.145	.162	.150
.975	.161	.178	.178	.145	.167	.167	.150	.166	.172	.145	.162	.150
1.000	.185	.178	.178	.145	.167	.167	.150	.166	.172	.145	.162	.150
Outboard station												
.010	-.003	-.170	-.156	-.008	.326	.170	-.003	.291	.054	-.002	.665	.394
.025	-.003	-.279	-.347	-.008	.139	-.064	-.1.143	.160	-.031	-.002	.471	.178
.050	-.483	-.426	-.370	-.772	-.047	-.132	-.110	-.099	-.048	-.1.215	.325	.082
.075	.048	.048	.025	.025	.312	.155	.1.098	.116	.037	.211	.032	.009
.100	.330	.578	.364	.246	-.250	-.121	-.1.053	-.110	.060	.082	.042	.032
.150	-.477	.618	.291	-.344	-.183	-.121	-.1.025	-.082	.054	-.1.232	.031	.008
.200	-.618	.440	.246	-.848	-.183	-.121	-.1.025	-.082	.054	-.1.232	.031	.008
.250	-.578	.251	.234	-.769	-.143	-.132	-.1.019	-.082	.054	-.1.192	.019	-.002
.300	-.578	.221	.240	-.792	-.143	-.132	-.1.019	-.082	.054	-.1.175	.042	.025
.350	-.601	-.195	-.212	-.780	-.132	-.132	-.1.019	-.065	.071	-.1.164	.047	.036
.400	-.618	-.195	-.212	-.769	-.132	-.132	-.1.019	-.065	.071	-.1.187	.047	.036
.450	-.578	-.155	-.172	-.634	-.104	-.109	-.489	-.037	.031	-.685	.031	.036
.500	-.550	-.165	-.099	-.515	-.059	-.059	-.336	.019	.019	-.465	.002	.002
.600	-.505	-.165	-.099	-.515	-.059	-.059	-.336	.019	.019	-.465	.002	.002
.650	-.370	-.037	-.031	-.358	-.008	-.002	-.156	.065	.076	-.312	.048	.043
.700	-.370	-.037	-.031	-.358	-.008	-.002	-.156	.065	.076	-.312	.048	.043
.750	-.195	.031	.042	-.166	.048	.060	.042	.132	.121	-.155	.094	.099
.800	-.031	.116	.087	-.087	.122	.165	.138	.048	.166	-.167	.156	.139
.900	.144	.093	.132	.150	.054	.150	.330	.053	.053	.178	.122	.189
.975	.161	.093	.132	.150	.054	.150	.330	.053	.053	.178	.122	.189
1.000	.161	.093	.132	.150	.054	.150	.330	.053	.053	.178	.122	.189

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.275	.323	-.167	-.587	-.608	-.106	-.716	.717	-.233	-.819	.777	.330
.025	-.005	.135	-.227	-.305	-.416	-.010	-.006	.529	.130	-.066	.637	.205
.050	-.151	-.205	-.345	-.038	.146	-.140	-.581	.297	-.010	-.512	.410	.076
.075	-.529	-.674	-.324	-.330	-.470	-.150	-.996	-.507	-.042	-.1.084	-.1.212	-.1.212
.100	-.604	-.674	-.324	-.797	-.470	-.150	-.921	.507	-.091	-.1.078	-.1.142	.631
.130	-.507	-.648	-.394	-.803	-.226	-.172	-.959	.129	-.096	-.1.078	-.1.255	.103
.200	-.512	-.329	-.475	-.824	-.210	-.166	-.686	-.129	-.096	-.1.078	-.1.282	.044
.250	-.539	-.351	-.313	-.722	-.312	-.167	-.959	-.129	-.096	-.1.078	-.1.298	.011
.300	-.539	-.302	-.286	-.584	-.188	-.210	-.813	-.129	-.096	-.1.078	-.1.298	.001
.350	-.595	-.302	-.286	-.722	-.188	-.210	-.813	-.129	-.096	-.1.078	-.1.298	.001
.400	-.626	-.275	-.275	-.749	-.183	-.145	-.819	-.123	-.102	-.900	-.070	-.075
.450	-.547	-.259	-.221	-.803	-.172	-.145	-.878	-.123	-.107	-.922	-.091	-.172
.500	-.595	-.329	-.302	-.787	-.129	-.123	-.851	-.107	-.096	-.873	-.075	-.070
.600	-.399	-.108	-.097	-.355	-.075	-.064	-.328	-.096	-.042	-.318	-.037	-.032
.700	-.210	-.000	-.011	-.183	.001	.017	-.150	.033	-.042	-.156	-.036	-.075
.800	-.038	.092	-.086	-.044	.092	.087	-.038	.103	.103	.038	.103	.097
.975	-.182	.189	.178	.146	.157	.151	.130	.157	.092	.124	.162	.097
1.000	-.182	.189	.178	.146	.157	.151	.146	.157	.092	.151	.162	.097
Outboard station												
.010	-.002	-.134	-.212	-.001	.310	.115	-.001	.232	.056	.004	.623	.342
.025	-.002	-.257	-.376	-.001	.124	-.077	-.1.047	.135	-.039	-.004	.438	.107
.050	-.436	-.398	-.392	-.912	-.039	-.142	-.006	.158	-.088	-.1.113	.275	.026
.075	.125	-.734	-.327	.075	-.348	-.180	.227	-.169	-.055	-.099	-.023	-.018
.100	.389	-.653	-.327	.303	-.267	-.137	1.020	.126	-.071	.151	.064	.064
.150	-.441	-.561	-.268	-.891	-.196	-.142	1.004	.098	-.077	-.1.113	-.061	-.001
.200	-.604	-.561	-.268	-.891	-.196	-.142	1.004	.098	-.077	-.1.113	-.061	-.001
.250	-.620	-.316	-.263	-.858	-.147	-.137	1.009	.109	-.104	-.1.119	-.050	-.039
.300	-.604	-.235	-.268	-.859	-.153	-.147	1.009	.104	-.109	-.1.108	-.061	-.061
.350	-.604	-.197	-.235	-.777	-.142	-.142	1.080	.077	-.098	-.1.102	-.066	-.066
.400	-.585	-.159	.192	-.820	-.110	-.115	1.150	-.039	-.050	-.1.146	-.066	-.066
.450	-.734	-.159	.192	-.918	-.110	-.115	1.150	-.039	-.050	-.1.146	-.066	-.066
.500	-.550	-.110	-.110	-.473	-.066	-.066	1.150	-.039	-.050	-.1.146	-.066	-.066
.600	-.506	-.110	-.110	-.473	-.066	-.066	1.150	-.039	-.050	-.1.146	-.066	-.066
.650	-.360	-.040	-.034	-.299	-.006	-.001	1.150	.064	.070	-.397	.010	-.012
.700	-.360	-.040	-.034	-.299	-.006	-.001	1.150	.064	.070	-.397	.010	-.012
.750	-.175	.031	.036	-.137	.048	.053	.032	.129	.124	-.245	.048	.031
.800	-.036	.112	.085	-.048	.118	.102	.129	.081	.167	-.007	.102	.042
.900	-.145	.134	.134	-.135	.135	.156	.146	.129	.167	-.007	.102	.042
.975	-.145	.134	.134	-.135	.135	.156	.146	.129	.167	-.007	.102	.042
1.000	-.161	.085	.134	-.145	.135	.156	.146	.129	.167	-.007	.102	.042

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.248	.342	-.195	-.508	.604	.079	-.637	.704	.214	-.732	.762	.283
.025	-.007	.168	-.222	-.012	.423	.007	-.002	.532	.101	.007	.615	.163
.050	-.092	-.160	-.404	-.100	.137	-.118	-.512	.272	-.024	-.238	.381	.043
.075	-.492	-.638	-.498	-.742	.532	-.144	-.897	.558	-.045	-.1002	.615	.007
.100	-.524	-.388	-.513	-.643	.238	-.212	-.809	.148	-.076	-.633	.615	.007
.150	-.472	-.388	-.472	-.747	.238	-.212	-.897	.148	-.076	-.633	.615	.007
.200	-.518	-.326	-.472	-.768	.217	-.201	-.923	.154	-.128	-.676	.615	.007
.250	-.508	-.326	-.347	-.690	.206	-.186	-.912	.154	-.128	-.676	.615	.007
.300	-.544	-.326	-.357	-.726	.201	-.180	-.881	.148	-.128	-.676	.615	.007
.350	-.591	-.342	-.492	-.700	.201	-.180	-.798	.206	-.206	-.676	.615	.007
.400	-.528	-.342	-.492	-.732	.201	-.180	-.824	.206	-.206	-.676	.615	.007
.450	-.685	-.305	-.258	-.789	.201	-.170	-.850	.148	-.315	-.872	-.181	-.285
.500	-.664	-.212	-.196	-.758	.149	-.154	-.803	.143	-.148	-.888	-.253	-.155
.550	-.347	-.123	-.113	-.352	-.092	-.076	-.351	-.081	-.216	-.363	-.077	-.071
.600	-.170	-.009	-.024	-.134	.007	.007	-.164	.002	-.003	-.207	-.030	-.004
.650	-.054	-.085	-.080	-.038	.090	.069	-.013	.085	.065	-.061	.059	.048
.700	-.152	-.168	.163	-.116	.142	-.132	-.080	.096	.096	-.022	.064	.007
.750	-.178	-.168	.163	-.132	.142	-.132	-.117	.096	.096	-.048	.064	.007
.800	-.170	-.009	-.024	-.134	.007	.007	-.164	.002	-.003	-.207	-.030	-.004
.850	-.054	-.085	-.080	-.038	.090	.069	-.013	.085	.065	-.061	.059	.048
.900	-.152	-.168	.163	-.116	.142	-.132	-.080	.096	.096	-.022	.064	.007
.950	-.178	-.168	.163	-.132	.142	-.132	-.117	.096	.096	-.048	.064	.007
1.000	-.178	-.168	.163	-.132	.142	-.132	-.117	.096	.096	-.048	.064	.007
Outboard station												
.010	-.005	-.094	-.220	.305	.277	-.082	-.005	.214	.051	.005	.585	.258
.025	-.000	-.230	-.413	.005	.104	-.126	-.005	.074	-.078	.005	.387	.068
.050	-.429	-.361	-.413	-.842	-.073	-.173	-.541	-.230	-.120	-.010	.209	.011
.075	.204	-.780	-.424	.178	.094	.094	.094	.209	.083	.015	.094	.015
.100	.439	-.670	-.372	.361	.492	-.194	.283	.209	.083	.214	.110	.063
.150	-.581	-.696	-.319	.811	.319	-.162	-.930	.156	-.109	-.1015	.131	.042
.200	-.539	-.696	-.319	.842	.220	-.162	-.930	.125	-.104	-.1015	.131	.042
.250	-.570	-.502	-.304	.837	.162	-.157	-.925	.125	-.125	-.1026	.110	.079
.300	-.617	-.304	-.298	.821	.168	-.178	-.935	.130	-.141	-.1026	.110	.079
.350	-.633	-.837	-.262	.837	.168	-.178	-.951	.130	-.141	-.1026	.110	.079
.400	-.685	-.194	-.262	-.889	-.162	-.178	-.951	.130	-.141	-.1026	.110	.079
.450	-.801	-.168	-.215	-.931	-.136	-.157	-.1014	.109	-.141	-.947	.121	.126
.500	-.827	-.121	-.126	-.586	-.094	-.100	-.1019	-.083	-.089	-.508	.115	.142
.550	-.827	-.121	-.126	-.586	-.094	-.100	-.475	-.036	-.041	-.424	-.095	.115
.600	-.293	-.047	-.047	-.262	-.037	-.032	-.366	.011	.021	-.367	-.058	.074
.650	-.131	.021	.031	-.110	.021	.026	-.235	.074	.047	-.299	-.042	.042
.700	-.052	.104	.078	-.005	.083	.068	-.036	-.094	.006	-.262	-.011	.047
.750	-.141	.057	.157	.057	.057	.089	.011	-.057	.006	-.225	-.025	.042
.800	-.157	.057	.157	.057	.057	.089	.262	-.036	.053	-.178	-.424	.163
.850	-.157	.057	.157	.057	.057	.089	.262	-.036	.053	-.178	-.424	.163
.900	-.157	.057	.157	.057	.057	.089	.262	-.036	.053	-.178	-.424	.163
.950	-.157	.057	.157	.057	.057	.089	.262	-.036	.053	-.178	-.424	.163
1.000	-.157	.057	.157	.057	.057	.089	.262	-.036	.053	-.178	-.424	.163

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	.007	-.159	-.380	.299	-.593	.481	-.796	.569	-1.153	.765
.025	-.356	-.190	-.306	.157	-1.043	.299	-1.246	.419	-1.531	.608
.050	-.451	-.372	-.854	-.016	-1.217	.110	-1.490	.229	-1.824	.418
.075	-.483		-.814		-1.059		-1.411		-1.800	
.100	-.514	-.340	-.790	-.072	-.988	.039	-1.372	.119	-1.768	.284
.150	-.522		-.759		-.909		-1.009		-1.752	
.200	-.522	-.372	-.703	-.119	-.822	-.032	-1.143	.032	-1.729	.166
.250	-.522		-.680		-.759		-.938		-1.626	
.300	-.530	-.230	-.672	-.111	-.727	-.040	-.922	-.007	-.979	.110
.350	-.522		-.696		-.703		-.867		-1.058	
.400	-.546	-.230	-.680	-.143	-.703	-.087	-.749	-.055	-.695	.047
.450	-.546		-.625		-.688		-.717		-.663	
.500	-.538	-.206	-.648	-.143	-.648	-.087	-.678	-.062	-.647	.016
.550	-.491		-.640		-.617		-.607		-.624	
.600	-.475	-.151	-.593	-.111	-.593	-.072	-.599	-.055	-.647	.008
.650	-.451		-.546		-.593		-.630		-.624	
.700	-.412	-.127	-.506	-.056	-.490	-.032	-.599	-.015	-.521	.039
.750	-.340		-.388		-.395		-.362		-.387	
.800	-.254	.007	-.388	.015	-.340	.039	-.331	.040	-.332	.087
.900	-.088	.086	-.364	.086	-.174	.094	-.126	.087	-.174	.126
.975	.141	.133	.118	.118	.110	.118	.064	.103	.087	.142
1.000	.165		.141		.125		.119		.166	
$M = 0.750$										
.010	.003	-.184	-.255	.249	-.450	.443	-.530	.580	-.894	.739
.025	-.277	-.285	-.723	.126	-.896	.292	-1.047	.393	-1.247	.573
.050	-.429	-.371	-.896	-.039	-1.113	.104	-1.306	.198	-1.535	.386
.075	-.472		-.853		-1.084		-1.285		-1.535	
.100	-.508	-.349	-.817	-.090	-1.098	.018	-1.256	.112	-1.513	.264
.150	-.537		-.795		-1.149		-1.285		-1.528	
.200	-.537	-.328	-.773	-.270	-.853	-.054	-1.299	.018	-1.513	.149
.250	-.551		-.737		-.846		-1.263		-1.520	
.300	-.556	-.256	-.737	-.126	-.853	-.061	-1.018	-.003	-1.528	.106
.350	-.566		-.709		-.788		-.831		-1.520	
.400	-.602	-.263	-.701	-.154	-.774	-.112	-.788	-.054	-1.046	.034
.450	-.602		-.709		-.760		-.730		-.930	
.500	-.573	-.234	-.644	-.154	-.839	-.112	-.702	-.068	-.923	-.010
.550	-.537		-.579		-.724		-.687		-.743	
.600	-.515	-.169	-.543	-.111	-.666	-.090	-.543	-.054	-.830	-.002
.650	-.479		-.500		-.500		-.522		-.477	
.700	-.429	-.162	-.457	-.054	-.486	-.047	-.450	-.018	-.448	.034
.750	-.371		-.370		-.421		-.450		-.355	
.800	-.342	-.083	-.234	-.032	-.256	.032	-.241	.054	-.225	.084
.900	-.155	-.184	-.097	.090	-.075	.068	-.097	.105	-.189	.127
.975	-.025	.047	-.133	.119	-.061	.112	-.054	.126	.041	.156
1.000	.068		.040		-.076		-.003		.055	

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Continued

(g) $M = 0.775$ and 0.800 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.775$										
.010	-.105	-.181	-.201	.207	-.368	.413	-.512	.545	-.754	.697
.025	-.291	-.202	-.629	.117	-.810	.261	-.926	.386	-1.113	.524
.050	-.416	-.381	-.857	-.049	-1.045	.074	-1.188	.179	-1.375	.344
.075	-.464		-.850		-1.010		-1.147		-1.389	
.100	-.506	-.333	-.864	-.152	-1.031	.005	-1.147	.096	-1.361	.227
.150	-.547		-.891		-1.086		-1.202		-1.396	
.200	-.547	-.319	-.726	-.180	-1.079	-.064	-1.209	.013	-1.410	.117
.250	-.561		-.767		-1.059		-1.209		-1.410	
.300	-.588	-.264	-.788	-.152	-.782	-.078	-1.216	-.014	-1.361	.048
.350	-.588		-.781		-.796		-1.202		-.857	
.400	-.616	-.271	-.822	-.166	-.865	-.112	-1.064	-.139	-.871	-.022
.450	-.651		-.795		-.914		-.912		-.788	
.500	-.609	-.236	-.691	-.159	-.762	-.119	-.795	-.070	-.733	-.063
.550	-.554		-.677		-.623		-.615		-.788	
.600	-.526	-.167	-.553	-.118	-.603	-.091	-.615	-.049	-.684	-.049
.650	-.478		-.477		-.554		-.443		-.615	
.700	-.423	-.139	-.429	-.056	-.437	-.043	-.387	-.014	-.574	-.077
.750	-.367		-.394		-.354		-.401		-.450	
.800	-.264	-.105	-.270	-.097	-.264	.040	-.270	.006	-.408	-.001
.900	-.091	.019	-.050	.041	-.036	-.050	-.159	.068	-.408	-.008
.975	.006	.109	.034	-.042	-.002	.033	-.028	-.021	-.305	-.077
1.000	.061		.089		-.002		.061		-.270	
$M = 0.800$										
.010	.061	-.172	-.144	.095	-.304	.400	-.410	.500	-.629	.686
.025	-.245	-.218	-.543	.048	-.695	.241	-.816	.340	-.994	.520
.050	-.391	-.364	-.789	-.071	-.941	.055	-1.055	.141	-1.253	.321
.075	-.451		-.789		-.914		-1.028		-1.253	
.100	-.491	-.358	-.815	-.125	-.954	-.018	-1.048	.088	-1.233	.214
.150	-.564		-.842		-1.021		-1.101		-1.280	
.200	-.550	-.344	-.948	-.158	-1.021	-.078	-1.134	-.005	-1.286	.095
.250	-.577		-.782		-1.034		-1.148		-1.300	
.300	-.610	-.291	-.742	-.151	-1.054	-.091	-1.148	-.052	-.914	.035
.350	-.637		-.749		-1.014		-1.148		-.768	
.400	-.670	-.291	-.815	-.184	-.981	-.131	-1.168	-.092	-.782	-.038
.450	-.677		-.882		-.921		-1.214		-.715	
.500	-.703	-.258	-.868	-.178	-.941	-.131	-1.214	-.105	-.655	-.078
.550	-.690		-.901		-.967		-.809		-.642	
.600	-.524	-.225	-.656	-.138	-.821	-.124	-.610	-.085	-.675	-.084
.650	-.477		-.516		-.556		-.523		-.655	
.700	-.411	-.112	-.357	-.111	-.350	-.045	-.351	-.038	-.582	-.191
.750	-.324		-.364		-.304		-.324		-.536	
.800	-.298	-.125	-.191	.015	-.211	.028	-.191	-.032	-.536	-.031
.900	-.099	-.045	-.111	.068	-.111	.082	-.038	.021	-.363	-.038
.975	-.006	-.006	-.012	.075	-.025	.075	-.072	.081	-.297	-.144
1.000	.008		.061		.062		.041		-.297	

TABLE 16.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10 - Concluded

(h) $M = 0.825$; center station

x/c	C _p at -									
	α = -2°		α = 0°		α = .1°		α = 2°		α = 4°	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
M = 0.825										
.010	.036	-.189	-.061	.048	-.194	.241	-.304	.465	-.508	.651
.025	-.195	-.221	-.477	.080	-.598	.209	-.708	.304	-.879	.484
.050	-.381	-.368	-.701	-.086	-.835	-.034	-.951	.131	-1.123	.292
.075	-.438		-.701		-.828		-.932		-1.135	
.100	-.470	-.381	-.752	-.201	-.867	-.047	-.945	.003	-1.129	.190
.150	-.541		-.757		-.943		-1.015		-1.180	
.200	-.541	-.381	-.816	-.195	-.956	-.111	-1.041	-.048	-1.187	.068
.250	-.567		-.804		-.963		-1.060		-1.206	
.300	-.605	-.349	-.771	-.182	-.969	-.137	-1.073	-.080	-.860	.004
.350	-.624		-.752		-.963		-1.073		-.719	
.400	-.688	-.342	-.757	-.214	-.995	-.169	-1.092	-.131	-.668	-.066
.450	-.733		-.836		-1.014		-1.150		-.623	
.500	-.765	-.285	-.874	-.208	-.943	-.175	-.919	-.163	-.611	-.111
.550	-.759		-.880		-.943		-.669		-.579	
.600	-.759	-.195	-.919	-.176	-.674	-.130	-.612	-.138	-.559	-.162
.650	-.688		-.547		-.527		-.580		-.559	
.700	-.464	-.169	-.432	-.086	-.450	-.079	-.535	-.093	-.540	-.175
.750	-.278		-.253		-.252		-.496		-.502	
.800	-.253	-.067	-.272	-.003	-.239	-.054	-.528	-.029	-.489	-.079
.900	-.029	-.125	-.048	-.009	-.111	-.009	-.221	-.080	-.591	-.092
.975	.016	-.048	-.061	-.022	-.041	-.098	-.099	-.118	-.489	-.201
1.000	.087		-.009		-.041		-.035		-.419	
										</

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.302	.222	-.225	-.755	.582	.173	-.981	.697	.301	-1.148	.763	.402
.025	.006	.031	-.225	.001	.385	.020	.007	.559	.141	.006	.641	.269
.050	-.540	-.289	-.334	-.287	.084	-.114	-1.000	.289	.020	-.430	.417	.139
.075	-.527			-.838			-.891			-1.411		
.100	-.815	-.873	-.321	-1.082	.200	.134	-1.102	.289	-.032	-1.405	.410	.045
.150	-.495	-.379	-.315	-.697	-.204	-.146	-.820	-.056	-.057	-1.405	.410	.045
.200	-.482	-.302	-.270	-.909	-.172	-.140	-.743	-.096	-.070	-.834	-.032	-.000
.250	-.482	-.264	-.239	-.640	-.159	-.121	-.711	-.096	-.057	-.796	-.045	-.007
.300	-.482	-.238	-.212	-.620	-.146	-.121	-.673	-.096	-.057	-.738	-.045	-.007
.350	-.507			-.620			-.673			-.719		
.400	-.495	-.206	-.187	-.601	-.127	-.101	-.641	-.083	-.064	-.687	-.045	-.025
.450												
.500	-.475	-.187	-.151	-.576	-.121	-.108	-.589	-.083	-.064	-.623	-.058	-.064
.550												
.600	-.482	-.135	-.123	-.582	-.089	-.082	-.615	-.064	-.051	-.648	-.037	-.032
.650												
.700	-.373	-.078	-.065	-.403	-.057	-.037	-.397	-.025	-.019	-.404	-.013	-.030
.750												
.800	-.238	.012	.012	-.249	.027	.033	-.237	.039	.052	-.218	.051	.057
.900	-.007	.102	.065	.001	.104	.110	.007	.103	.116	.019	.109	.115
.975	.166			.155			.141			.128		
1.000	.198	.198	.198	.174	.180	.180	.161	.167	.167	.147	.154	.141
Outboard station												
.010	.006	-.221	-.198	-.006	.300	.168	-.000	.547	.354	-.007	.671	.454
.025	.006	-.343	-.324	-.006	.130	-.032	-.000	.348	.097	-.007	.484	.199
.050	-.472	-.446	-.349	-.798	-.077	-.122	-1.277	.148	-.000	-1.517	.316	.095
.075	.715			.342			.039			-.304		
.100	1.187	-.550	-.337	.955	-.276	-.141	.735	-.077	-.039	.419	.032	.025
.150	-.466	-.569	-.259	-.670	-.238	-.122	-.767	-.116	-.026	-.794	-.020	.039
.200	-.485	-.395	-.214	-.676	-.173	-.109	-.767	-.090	-.045	-.820	-.026	-.001
.250	-.498	-.214	-.201	-.682	-.128	-.109	-.761	-.065	-.045	-.827	-.014	-.031
.300	-.498	-.188	-.195	-.663	-.128	-.115	-.729	-.077	-.058	-.788	-.025	-.014
.350	-.524			-.670			-.722			-.775		
.400	-.517	-.162	-.159	-.644	-.109	-.115	-.584	-.071	-.071	-.736	-.033	-.039
.450												
.500	-.485	-.136	-.143	-.579	-.083	-.109	-.600	-.052	-.071	-.633	-.026	-.052
.550												
.600	-.440	-.085	-.065	-.508	-.044	-.051	-.516	-.019	-.026	-.530	-.001	-.014
.650												
.700	-.349	-.020	-.007	-.386	.001	.007	-.374	.019	.019	-.388	.032	.032
.750												
.800	-.207	.038	.057	-.212	.052	.059	-.193	.064	.071	-.194	.070	.077
.900	.006	.115	.109	.014	.117	.110	.019	.116	.110	.012	.122	.115
.975	.154			.123			.115			.109		
1.000	.180	.122	.186	.142	.065	.155	.129	.045	.135	.157	.103	.135

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Continued

(b) $M = 0.750$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.279	.249	-.204	-.645	.578	-.100	-.851	.703	.253	-.986	.775	.368
.025	.007	.066	-.238	.003	.395	-.008	-.003	.346	.125	.008	.640	.860
.050	-.683	-.274	-.379	-.166	.103	-.114	-.845	.283	-.009	.325	.412	.780
.075	-.590			-.827			-1.143			-1.267		.513
.100	-.677	.765	-.332	-.858	.301	-.131	-.933	.383	-.038	-1.220	.488	.581
.150	-.519	-.402	-.344	-.716	-.225	-.160	-.939	.126	-.073	-1.220	-.027	.194
.200	-.519	-.320	.309	-.692	-.190	-.155	-.740	-.114	-.079	-1.179	-.047	.142
.250	-.590	-.285	-.274	-.675	-.172	-.137	-.798	-.108	-.073	-.734	-.050	.118
.300	-.525	-.268	-.244	-.681	-.166	-.155	-.775	-.108	-.067	-.775	-.055	.048
.350	-.713			-.681			-.734			-.840		.112
.400	-.683	-.238	-.209	-.675	-.143	-.125	-.699	-.097	-.073	-.793	-.050	.095
.450												.065
.500	-.549	-.215	-.186	-.681	-.137	-.125	-.635	-.108	-.085	-.670	-.068	.036
.550												.042
.600	-.566	-.157	-.145	-.692	-.108	-.096	-.570	-.073	-.067	-.693	-.050	.013
.650												.054
.700	-.402	-.092	-.080	-.418	-.055	-.049	-.401	-.032	-.021	-.401	-.015	.042
.750												.095
.800	-.244	.002	.002	-.231	.027	.027	-.220	.038	.020	-.208	.055	.048
.900	.007	.095	.083	.015	.097	.103	.032	.102	.114	.032	.108	.177
.975	.165			.143			.143			.143		.171
1.000	.189	.189	.183	.167	.173	.167	.161	.166	.161	.160	.154	.212
Outboard station												
.010	-.001	-.238	-.196	-.004	.287	.131	-.004	.520	.307	.001	.647	.823
.025	-.005	-.335	-.359	-.004	.108	-.068	-.004	.325	.049	.001	.660	.660
.050	-.470	-.476	-.388	-.821	-.045	-.139	-1.210	.149	-.039	-1.310	.519	.224
.075	.872			.473			.208			-.040		.582
.100	1.237	-.565	-.375	1.013	-.315	-.151	.819	.128	-.057	.566	.013	.233
.150	-.476	.618	-.294	-.991	-.274	-.115	-.1086	.145	-.039	-1.287	-.034	.155
.200	-.523	-.529	-.247	-.697	-.192	-.121	-.922	-.122	-.057	-1.234	-.034	.118
.250	-.553	-.282	-.235	-.727	-.139	-.115	-.745	-.081	-.057	-1.222	-.015	.082
.300	-.565	-.217	-.229	-.721	-.133	-.127	-.739	-.087	-.075	-1.193	-.028	.071
.350	-.588			-.750			-.792			-.899		.041
.400	-.576	-.182	-.200	-.733	-.121	-.127	-.792	-.081	-.081	-.681	-.028	.024
.450												.035
.500	-.541	-.153	-.164	-.627	-.098	-.121	-.628	-.063	-.087	-.605	-.022	.012
.550												.018
.600	-.488	-.105	-.054	-.533	-.057	-.062	-.522	-.028	-.034	-.504	.001	.047
.650												.065
.700	-.370	-.041	-.023	-.386	-.004	-.004	-.363	.013	.013	-.363	.043	.037
.750												.059
.800	-.206	.030	.042	-.186	.049	.055	-.175	.060	.066	-.175	.090	.092
.900	.012	.107	.095	.020	.120	.108	.031	.125	.113	.037	.143	.037
.975	.148			.126			.125			.143		.135
1.000	.165	.095	.171	.137	.061	.149	.137	.043	.143	.160	.065	.035

(c) $M = 0.775$

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TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.282	.286	-.247	-.544	.584	.094	-.678	.697	.206	-.809	.753	.287
.025	.004	.096	-.272	-.006	.405	-.032	.009	.536	-.100	.011	.631	.134
.050	-.396	-.212	-.423	-.048	.109	-.134	-.602	.268	-.018	-.254	.388	.075
.075	1.045	-.456	-.449	-.885	-.088	.431	-.983	.983		-1.079		-1.205
.100	1.284	-.661	-.411	1.100	-.364	.180	-.532	.479	-.056	-.923	.582	.021
.150	-.470	-.628	-.335	-.868	-.353	.142	-.026	-.189	-.058	-1.068	-.049	.027
.200	-.552	-.661	-.275	-.847	-.218	.126	-.966	-.129	-.069	-1.079	-.065	-.037
.250	-.622	-.465	-.264	-.825	-.136	.126	-.977	-.091	-.085	-1.084	-.065	-.043
.300	-.612	-.313	-.253	-.841	-.142	.136	-.988	-.096	-.085	-1.084	-.065	-.043
.350	-.633	-.182	-.226	-.825	-.131	.136	-.1037	-.096	-.096	-1.014	-.065	-.043
.400	-.693			-.809	-.131	.136	-1.059	-.096	-.096	-1.014	-.065	-.043
.450		-.161	.193	-.906	-.098	-.126	-1.113	-.075	-.102	-.826	-.065	-.043
.500	-.742									-.912	-.070	-.085
.550	-.503	-.117	-.106	-.494	-.055	-.055	-.482	-.031	-.042	-.858	-.054	-.070
.600		-.041	-.036	-.288	.005	.005	-.292	.012	.023	-.302	-.022	-.033
.650	-.356									-.140	.054	.043
.700										-.054	.113	.113
.750	-.177	.024	.029	-.120	.059	.064	-.118	.056	.078	-.145	.172	.156
.800	.035	.106	.089	.053	.129	.118	.050	.127	.132	.167		
.850												
.900												
.950												
1.000												
Outboard station												
.010	-.003	-.162	-.220	.005	.271	.121	.002	.478	.249	.004	.583	.310
.025	-.009	-.286	-.438	.005	.097	-.136	.002	.279	-.009	.004	.432	.118
.050	-.456	-.421	-.449	-.885	-.088	-.174	-1.021	.059	-.058	-1.098	.248	.025
.075	1.045						.431					
.100	1.284	-.661	-.411	1.100	-.364	.180	.926	.154	-.080	.709	.055	.023
.150	-.470	-.628	-.335	-.868	-.353	.142	-.026	-.189	-.058	-1.120	-.094	.043
.200	-.552	-.661	-.275	-.847	-.218	.126	-.966	-.129	-.069	-1.087	-.072	-.034
.250	-.622	-.465	-.264	-.825	-.136	.126	-.977	-.091	-.085	-1.093	-.040	-.045
.300	-.612	-.313	-.253	-.841	-.142	.136	-.988	-.096	-.085	-1.098	-.051	-.055
.350	-.633	-.182	-.226	-.825	-.131	.136	-.1037	-.096	-.096	-1.125	-.065	-.065
.400	-.693			-.809	-.131	.136	-1.059	-.096	-.096	-1.147	-.051	-.083
.450		-.161	.193	-.906	-.098	-.126	-1.113	-.075	-.102	-.805	-.040	-.105
.500	-.742									-.858	-.013	-.056
.550	-.503	-.117	-.106	-.494	-.055	-.055	-.482	-.031	-.042	-.544	-.013	-.056
.600		-.041	-.036	-.288	.005	.005	-.292	.012	.023	-.387	.025	-.013
.650	-.356									-.186	.074	.031
.700										-.034	.123	.047
.750	-.177	.024	.029	-.120	.059	.064	-.118	.056	.078	-.145	.172	.156
.800	.035	.106	.089	.053	.129	.118	.050	.127	.132	.167		
.850												
.900												
.950												
1.000												

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Continued

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	-.322	.324	-.207	-.476	.587	.045	-.644	.622	.190	-.738	.755	.277
.025	.006	.152	-.265	.007	.397	-.055	.001	.526	.084	.006	.610	.167
.050	-.291	-.202	-.415	-.003	.085	-.159	-.447	.251	-.056	-.207	.370	.027
.075	-.478	-.478	-.757	-.757	.480	-.169	-.925	.266	-.072	-.998	.625	.001
.100	-.499	-.613	-.405	-.481	.268	-.211	-.899	.166	.114	-.774	.082	-.041
.150	-.483	-.415	.410	-.736	.226	-.226	-.925	.150	.114	-.983	-.093	-.056
.200	-.554	-.322	.426	.762	.221	-.195	-.915	.150	.114	-.1019	-.093	-.056
.250	-.509	-.317	.353	.762	.221	-.195	-.889	.150	.114	-.1019	-.093	-.056
.300	-.546	-.332	-.369	.725	.221	-.195	-.889	.150	.114	-.1019	-.093	-.056
.350	-.598	-.332	-.384	.731	.216	-.185	-.825	.150	.124	-.943	-.093	-.093
.400	-.629	-.301	-.249	.793	.211	-.180	-.847	.155	.155	-.879	-.109	-.109
.450	-.722	-.223	-.181	.767	.154	-.143	-.795	.119	.103	-.873	-.082	-.093
.500	-.655	-.114	-.093	.356	.091	-.081	-.343	.067	.056	-.353	-.056	-.056
.550	-.348	-.004	-.010	-.138	.003	.007	-.145	.011	.022	-.197	.022	.011
.600	-.166	.089	.084	.028	.080	.085	.011	.089	.100	-.015	.094	.053
.650	.053	.157	.173	.111	.137	.137	.100	.100	.042	.042	.053	.053
.700	.178	.178	.173	.132	.137	.137	.126	.131	.115	.089	.115	.053
.750												
.800												
.850												
.900												
.950												
1.000												
Outboard station												
.010	-.001	-.171	-.252	.000	.233	.026	-.001	.446	.201	-.001	.559	.276
.025	-.001	-.273	-.430	.000	.074	-.167	-.005	.245	-.038	-.001	.376	.045
.050	-.420	-.394	-.451	-.815	-.130	-.214	-.933	.072	-.095	-.1001	.198	-.022
.075	1.130	-.734	-.451	.779	-.444	-.224	.533	.263	.116	.355	.132	.089
.100	1.302	-.640	-.367	1.130	-.428	-.172	.262	.242	.095	.758	.059	.059
.150	-.530	-.702	-.305	-.800	-.324	.167	-.917	.179	.095	1.022	.158	.032
.200	-.493	-.582	-.289	-.815	-.177	-.162	-.895	.127	.090	1.001	.116	.059
.250	-.577	-.420	-.284	-.831	-.167	-.177	-.927	.132	.116	1.027	.080	.053
.300	-.619	-.184	-.242	-.862	-.162	-.177	-.938	.132	.116	1.027	.090	.080
.350	-.661	-.158	-.200	-.888	-.146	-.172	-.980	.116	.132	1.048	.095	.080
.400	-.687	-.158	-.200	-.909	-.146	-.172	1.011	.116	.132	1.058	.095	.121
.450	-.807	-.111	-.111	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.500	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.550	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.600	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.650	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.700	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.750	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.800	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.850	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.900	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
.950	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143
1.000	-.839	-.043	-.038	-.956	-.099	-.099	1.090	.080	.080	1.058	.090	.143

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	.085	-.231	-.379	.284	-.624	.473	-.791	.591	-1.155	.773
.025	-.333	-.199	-.785	.142	-1.035	.299	-1.234	.410	-1.526	.607
.050	-.460	-.381	-.852	-.008	-1.193	.102	-1.494	.220	-1.818	.410
.075	-.491		-.813		-1.059		-1.400		-1.795	
.100	-.491	-.294	-.789	-.071	-.996	.039	-1.336	.125	-1.747	.291
.150	-.515		-.750		-.901		-.989		-1.739	
.200	-.507	-.270	-.702	-.118	-.838	-.032	-.965	.022	-1.700	.165
.250	-.507		-.686		-.822		-.957		-1.131	
.300	-.515	-.231	-.671	-.118	-.727	-.056	-.870	-.001	-1.068	.109
.350	-.515		-.647		-.688		-.839		-.665	
.400	-.531	-.231	-.647	-.142	-.695	-.087	-.736	-.057	-.681	.054
.450	-.531		-.639		-.688		-.712		-.673	
.500	-.507	-.207	-.615	-.142	-.617	-.095	-.649	-.064	-.641	.015
.550	-.484		-.608		-.609		-.649		-.633	
.600	-.468	-.167	-.608	-.102	-.609	-.095	-.633	-.057	-.610	.015
.650	-.436		-.568		-.609		-.531		-.531	
.700	-.397	-.144	-.568	-.055	-.467	-.032	-.523	-.017	-.467	.030
.750	-.389		-.387		-.411		-.396		-.428	
.800	-.428	.006	-.387	.016	-.451	.031	-.428	.045	-.373	.086
.900	-.081	.085	-.213	.087	-.166	.086	-.215	.086	-.191	.125
.975	.149	.141	.118	.126	.102	.118	.101	.109	.141	.141
1.000	.172		.142		.134		.117		.157	
$M = 0.750$										
.010	.031	-.228	-.233	.256	-.450	.414	-.502	.558	-.892	.738
.025	-.272	-.250	-.701	.126	-.882	.270	-1.034	.399	-1.253	.572
.050	-.437	-.336	-.867	-.053	-1.113	.090	-1.293	.205	-1.505	.391
.075	-.473		-.845		-1.070		-1.236		-1.520	
.100	-.502	-.329	-.802	-.104	-1.084	-.018	-1.228	.111	-1.505	.262
.150	-.531		-.795		-1.134		-1.279		-1.513	
.200	-.531	-.308	-.759	-.133	-.825	-.061	-1.279	.003	-1.513	.139
.250	-.538		-.737		-.846		-1.264		-1.520	
.300	-.567	-.257	-.737	-.133	-.810	-.068	-1.320	-.018	-1.520	.088
.350	-.560		-.730		-.745		-.832		-1.513	
.400	-.582	-.257	-.709	-.161	-.745	-.112	-.789	-.069	-1.152	.009
.450	-.589		-.723		-.745		-.760		-.892	
.500	-.560	-.279	-.709	-.154	-.817	-.104	-.775	-.076	-.835	-.013
.550	-.531		-.615		-.702		-.724		-.799	
.600	-.509	-.272	-.543	-.111	-.558	-.090	-.710	-.062	-.633	.009
.650	-.473		-.500		-.493		-.494		-.474	
.700	-.416	-.228	-.471	-.053	-.450	-.032	-.479	-.025	-.402	.031
.750	-.394		-.399		-.342		-.494		-.315	
.800	-.257	-.257	-.305	-.017	-.285	.004	-.227	.003	-.251	.088
.900	-.062	-.048	-.125	.011	-.061	.090	-.184	.097	-.142	.132
.975	.118	.046	-.133	.119	-.018	.119	-.112	.119	.002	.161
1.000	.161		-.053		-.104		.025		.103	

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Continued

(g) $M = 0.775$ and 0.800 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.750$										
.010	.046	-.217	-.200	.234	-.354	.372	-.547	.551	-.742	.710
.025	-.272	-.252	-.622	.110	-.809	.247	-.920	.379	-1.115	.544
.050	-.418	-2.299	-.835	-.049	-1.037	.047	-1.183	.178	-1.385	.351
.075	-.473		-.842		-1.010		-1.135		-1.385	
.100	-.508	-.342	-.822	-.111	-1.017	.006	-1.149	.088	-1.343	.233
.150	-.542		-.877		-1.086		-1.190		-1.392	
.200	-.542	-2.299	-.732	-.152	-1.058	-.063	-1.204	-.022	-1.392	.115
.250	-.563		-.773		-1.051		-1.204		-1.412	
.300	-.597	-.279	-.787	-.152	-.789	-.139	-1.211	-.071	-1.412	.060
.350	-.590		-.773		-.809		-1.218		-1.198	
.400	-.618	-.279	-.829	-.173	-.858	-.167	-1.142	-.160	-.825	.005
.450	-.646		-.829		-.941		-.920		-.776	
.500	-.618	-.238	-.718	-.166	-.885	-.153	-.824	-.133	-.742	-.037
.550	-.549		-.594		-.644		-.510		-.693	
.600	-.528	-.203	-.580	-.125	-.533	-.119	-.686	-.133	-.666	-.044
.650	-.487		-.559		-.464		-.534		-.686	
.700	-.418	-.127	-.421	-.062	-.416	-.036	-.430	-.119	-.603	-.023
.750	-.362		-.362		-.319		-.388		-.424	
.800	-.259	-.120	-.257	-.021	-.208	.033	-.271	-.098	-.382	.019
.900	-.072	-.010	-.152	-.049	-.139	.020	-.188	-.209	-.410	.026
.975	.039	.004	.014	-.035	-.119	.075	-.091	.040	-.223	-.037
1.000	.046		.089		-.008		.040		-.258	
$M = 0.800$										
.010	.086	-.207	-.171	.108	-.274	.405	-.404	.493	-.605	.672
.025	-.240	-.227	-.556	.068	-.700	.245	-.796	.346	-.977	.512
.050	-.413	-.386	-.782	-.111	-.940	.052	-1.056	.174	-1.250	.339
.075	-.466		-.775		-.920		-1.029		-1.250	
.100	-.506	-.373	-.808	-.138	-.946	-.054	-1.042	.094	-1.230	.220
.150	-.572		-.828		-1.020		-1.102		-1.290	
.200	-.553	-.366	-.835	-.164	-1.020	-.081	-1.115	.008	-1.283	.100
.250	-.579		-.755		-1.006		-1.135		-1.270	
.300	-.606	-.313	-.748	-.164	-1.020	-.108	-1.142	-.119	-1.057	.053
.350	-.646		-.748		-1.006		-1.149		-.784	
.400	-.672	-.300	-.808	-.184	-.827	-.141	-1.155	-.218	-.724	-.026
.450	-.686		-.874		-.893		-1.202		-.691	
.500	-.706	-.260	-.868	-.177	-.926	-.141	-1.202	-.092	-.658	-.073
.550	-.679		-.901		-.933		-.790		-.638	
.600	-.526	-.187	-.821	-.124	-.807	-.101	-.584	-.072	-.591	-.073
.650	-.479		-.509		-.687		-.604		-.578	
.700	-.413	-.127	-.350	-.064	-.347	-.088	-.325	-.012	-.552	-.060
.750	-.320		-.290		-.294		-.305		-.552	
.800	-.293	-.094	-.164	-.084	-.194	-.074	-.238	-.072	-.485	-.033
.900	-.080	-.074	-.084	-.005	-.074	.032	-.165	.034	-.326	-.033
.975	.026	-.027	-.104	-.031	-.101	.006	-.079	.074	-.345	-.119
1.000	.106		.009		-.008		-.105		-.319	

TABLE 17.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 10a - Concluded

(h) $M = 0.825$; center station

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
M = 0.825										
.010	.099	-.196	-.137	-.066	-.196	.035	-.299	.317	-.515	.644
.025	-.209	-.247	-.470	-.015	-.600	.086	-.590	.304	-.874	.477
.050	-.363	-.388	-.700	-.111	-.837	.028	-.939	.080	-1.124	.279
.075	-.433		-.700		-.830		-.933		-1.111	
.100	-.459	-.375	-.738	-.150	-.862	-.081	-.952	-.049	-1.111	.176
.150	-.529		-.790		-.933		-1.010		-1.169	
.200	-.542	-.382	-.815	-.220	-.946	-.113	-1.042	-.029	-1.175	.067
.250	-.568		-.815		-.959		-1.068		-1.182	
.300	-.600	-.343	-.903	-.201	-.984	-.119	-1.074	-.055	-.817	.010
.350	-.619		-.758		-.978		-1.074		-.688	
.400	-.690	-.337	-.809	-.220	-.978	-.158	-1.093	-.119	-.644	-.054
.450	-.741		-.835		-.984		-1.151		-.618	
.500	-.766	-.279	-.860	-.214	-.927	-.164	-1.170	-.132	-.592	-.106
.550	-.760		-.886		-.901		-.741		-.567	
.600	-.760	-.196	-.924	-.162	-.734	-.125	-.593	-.113	-.541	-.118
.650	-.683		-.553		-.497		-.555		-.560	
.700	-.516	-.183	-.399	-.092	-.395	-.100	-.515	-.081	-.496	-.099
.750	-.279		-.258		-.273		-.625		-.528	
.800	-.234	-.061	-.284	-.066	-.234	-.125	-.324	-.023	-.509	-.067
.900	-.049	-.068	-.073	-.060	-.138	-.042	-.158	-.029	-.599	-.080
.975	-.029	-.036	-.022	-.041	-.081	-.049	-.145	-.036	-.451	-.176
1.000	.028		-.060		-.049		-.049		-.310	
										</

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11

(a) $M = 0.700$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C

Inboard station												
.010	-.046	.137	.084	-.488	.462	.468	-.759	.597	.591	-.578	.668	.680
.025	-.356	-.003	.009	-.798	.278	.290	-.1124	.398	.410	-.1306	.489	.520
.050	-.447	-.143	-.149	-.835	.095	.083	-.1167	.221	.215	-.1459	.307	.367
.075	-.502			-.853			-.1539			-.1434		
.100	-.502	-.180	-.210	-.786	.004	-.014	-.572	.094	.087	-.1306	.179	.167
.150	-.502	-.253	-.259	-.726	-.087	-.087	-.862	.002	-.004	-.517	.063	.082
.200	-.502	-.253	-.259	-.683	-.118	-.112	-.783	.034	-.034	-.856	.021	.039
.250	-.508	-.265	-.240	-.659	-.136	-.124	-.741	.059	-.040	-.795	.010	.015
.300	-.502	-.271	-.246	-.634	-.160	-.130	-.704	.083	-.059	-.746	-.034	-.016
.350	-.520	-.295	-.274	-.640	-.195	-.174	-.658	.126	.112	-.734	-.091	-.055
.400	-.514	-.277	-.259	-.616	-.184	-.166	-.668	.132	.101	-.698	-.065	-.038
.450	-.517	-.288	-.260	-.610	-.202	-.174	-.648	.148	.126	-.678	-.112	-.084
.500	-.502	-.271	-.246	-.574	-.197	-.178	-.607	.150	.126	-.631	-.107	-.089
.550	-.506	-.253	-.218	-.569	-.188	-.160	-.595	.151	.119	-.601	-.112	-.084
.600	-.429	-.198	-.180	-.476	-.148	-.124	-.497	.113	.089	-.503	-.077	-.065
.650	-.429	-.168	-.187	-.453	-.124	-.103	-.465	.098	.070	-.471	-.069	-.041
.700	-.352	-.009	-.094	-.422	-.008	-.063	-.424	-.010	-.034	-.418	.002	-.016
.750	-.358			-.369			-.365			-.353		
.800	-.259	-.009	-.003	-.257	.004	.016	-.260	.027	.033	-.235	.045	.045
.900	-.021	.082	.094	-.008	.089	.101	-.004	.094	.106	.015	.106	.112
.975	.161	.144	.123	.150	.123	.060	.142	.129	.108	.124	.087	.108
1.000	.192	.198	.186	.174	.180	.168	.161	.161	.161	.155	.155	.142

Outboard station												
.010	-.024	-.111	-.025	-.538	.342	.427	-.786	.536	.585	-.945	.640	.683
.025	-.527	-.202	-.150	-.1040	.173	.148	-.1332	.319	.325	-.1467	.435	.460
.050	-.441	-.288	-.251	-.814	.050	.007	-.1581	.214	.159	-.1589	.300	.300
.075	-.441						-.862			-.1510		
.100	-.478	-.263	-.263	-.789	-.066	-.060	-.909	.055	.061	-.1460	.147	.129
.150	-.472	-.227	-.220	-.722	-.084	-.054	-.841	.030	.042	-.1246	.098	.122
.200	-.527	-.227	-.208	-.709	-.078	-.078	-.810	.006	.000	-.829	.055	.043
.250	-.523	-.208	-.208	-.740	-.078	-.078	-.817	.000	-.000	-.804	.049	.043
.300	-.527	-.220	-.220	-.673	-.121	-.121	-.767	.019	.025	-.822	.006	.000
.350	-.545	-.222	-.228	-.673	-.127	-.127	-.761	.070	.070	-.853	.034	.023
.400	-.545	-.220	-.223	-.666	-.140	-.140	-.792	.142	.080	-.798	.037	.037
.450	-.541	-.234	-.234	-.634	-.151	-.156	-.672	.099	.105	-.695	.070	.070
.500	-.521	-.202	-.214	-.587	-.140	-.152	-.620	.086	.099	-.651	.056	.056
.550	-.482	-.193	-.205	-.540	-.133	-.145	-.560	.088	.105	-.571	.044	.070
.600	-.505	-.153	-.153	-.519	-.097	-.097	-.553	.058	.074	-.540	.031	.049
.650	-.423	-.122	-.122	-.457	-.086	-.086	-.460	.038	.058	-.540	.028	.028
.700	-.369	-.104	-.079	-.391	-.078	-.060	-.418	.019	.019	-.448	.006	.000
.750	-.317			-.322			-.312			-.306		
.800	-.220	.000	.006	-.219	.001	.014	-.277	.024	.030	-.246	.049	.049
.900	-.012	.025	.063	.007	.063	.069	-.000	.067	.036	.012	.049	.061
.975	.135	.126	.137	.124	.109	.126	.092	.107	.119	.122	.107	.125
1.000	.141	.154	.141	.148	.148	.148	.104	.069	.098	.129	.116	.129

Outboard station												
.010	-.805	.781	.805	-.1281	.683	.683	-.1281	.683	.683	-.1281	.683	.683
.025	-.614	.607	.614	-.1716	.460	.460	-.1716	.460	.460	-.1716	.460	.460
.050	-.430	.454	.430	-.1839	.300	.300	-.1839	.300	.300	-.1839	.300	.300
.075												
.100		.295	.282	-.1845	.129	.129	-.1845	.147	.129	-.1845	.147	.129
.150		.233	.233	-.1851	.098	.098	-.1851	.098	.098	-.1851	.098	.098
.200		.178	.178	-.1778	.043	.043	-.1778	.043	.043	-.1778	.043	.043
.250		.123	.123	-.1710	.049	.049	-.1710	.049	.049	-.1710	.049	.049
.300		.092	.092	-.1097	.000	.000	-.1097	.000	.000	-.1097	.000	.000
.350		.067	.067	-.0968	.023	.023	-.0968	.023	.023	-.0968	.023	.023
.400		.043	.043	-.0809	.037	.037	-.0809	.037	.037	-.0809	.037	.037
.450		.008	.008	-.653	.070	.070	-.653	.070	.070	-.653	.070	.070
.500		.007	.007	-.551	.056	.056	-.551	.056	.056	-.551	.056	.056
.550		.016	.016	-.523	.044	.044	-.523	.044	.044	-.523	.044	.044
.600		.007	.007	-.515	.031	.031	-.515	.031	.031	-.515	.031	.031
.650		.019	.019	-.441	.028	.028	-.441	.028	.028	-.441	.028	.028
.700		.043	.043	-.398	.000	.000	-.398	.000	.000	-.398	.000	.000
.750		.074	.074	-.305	.049	.049	-.305	.049	.049	-.305	.049	.049
.800		.129	.129	-.245	.061	.061	-.245	.061	.061	-.245	.061	.061
.900		.137	.137	-.012	.135	.135	-.012	.135	.135	-.012	.135	.135
.975		.137	.137	.098	.125	.125	.098	.125	.125	.098	.125	.125
1.000		.123	.123	.105	.129	.129	.105	.129	.129	.105	.129	.129

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11 - Continued

(b) $M = 0.750$

x/c	C_p at -								
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station									
.010	.016	.175	.110	-.389	.449	.427	-.622	.567	.578
.025	-.360	.011	-.006	-.733	.272	.277	-.1117	.388	.399
.050	-.433	-.161	-.150	-.822	.111	.111	-.1099	.205	.216
.075	-.516			-.794			-.1105		
.100	-.565	-.194	-.211	-.844	.005	.017	-.1127	.088	.072
.150	-.522	-.266	-.266	-.772	-.095	-.095	-.966	-.017	-.006
.200	-.522	-.266	-.266	-.700	-.112	-.112	-.827	-.050	-.050
.250	-.517	-.283	-.272	-.678	-.139	-.123	-.822	-.089	-.061
.300	-.517	-.283	-.272	-.672	-.167	-.139	-.789	-.117	-.078
.350	-.549	-.318	-.292	-.672	-.209	-.173	-.766	-.157	-.131
.400	-.549	-.300	-.283	-.667	-.195	-.173	-.711	-.156	-.122
.450	-.557	-.318	-.286	-.660	-.222	-.196	-.703	-.183	-.151
.500	-.544	-.288	-.266	-.617	-.206	-.170	-.655	-.167	-.145
.550	-.546	-.273	-.241	-.606	-.202	-.170	-.633	-.170	-.144
.600	-.449	-.216	-.194	-.489	-.150	-.128	-.517	-.128	-.106
.650	-.449	-.182	-.157	-.472	-.131	-.112	-.477	-.112	-.086
.700	-.445	-.066	-.105	-.411	-.000	-.062	-.417	-.006	-.045
.750	-.358			-.359			-.358		
.800	-.250	-.006	-.006	-.234	.016	.022	-.228	.011	.027
.900	.000	.089	.100	.016	.099	.111	.022	.094	.105
.975	.147	.147	.134	.155	.134	.114	.138	.127	.108
1.000	.154	.154	.189	.177	.183	.177	.161	.161	.150
Outboard station									
.010	-.015	-.104	-.032	-.407	.324	.380	-.602	.497	.536
.025	-.456	-.227	-.216	-.921	.102	.125	-.1117	.287	.298
.050	-.434	-.249	-.277	-.793	.063	.007	-.1212	.175	.159
.075	-.434			-.670			-.1145		
.100	-.479	-.294	-.289	-.776	-.060	-.054	-.1111	.030	.041
.150	-.455	-.261	-.244	-.722	-.077	-.082	-.1201	.007	.019
.200	-.540	-.249	-.227	-.675	-.099	-.082	-.1089	-.026	-.026
.250	-.514	-.227	-.221	-.748	-.082	-.082	-.910	-.026	-.020
.300	-.543	-.244	-.255	-.726	-.105	-.121	-.764	-.065	-.082
.350	-.555	-.240	-.245	-.754	-.138	-.138	-.764	-.089	-.089
.400	-.556	-.244	-.255	-.742	-.133	-.138	-.764	-.099	-.093
.450	-.584	-.250	-.250	-.708	-.165	-.165	-.741	-.132	-.127
.500	-.546	-.221	-.238	-.619	-.133	-.133	-.630	-.099	-.116
.550	-.503	-.207	-.213	-.558	-.143	-.154	-.574	-.116	-.127
.600	-.501	-.171	-.171	-.530	-.105	-.105	-.546	-.116	-.127
.650	-.493	-.132	-.127	-.455	-.084	-.089	-.461	-.068	-.068
.700	-.438	-.087	-.138	-.379	-.071	-.082	-.373	-.043	-.026
.750	-.364	-.009	.002	-.299	.007	-.077	-.299	.007	-.009
.800	-.244	-.097	.092	-.194	.007	-.077	-.188	.007	-.009
.900	-.015	.132	.148	.102	.115	.131	.097	.115	.126
.975	.142	.142	.164	.130	.091	.142	.108	.131	.142
1.000	.155	.142							

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11 - Continued

(d) $M = 0.800$

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$			
	Row A			Row B	Row A			Row B	Row A			Row B
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row A	Row B	Row C	Row A
Inboard station												
.010	.014	.158	.093	.268	.441	.421	.474	.370	.550	.561	.644	.644
.025	-.278	.009	-.007	.611	.265	.249	.781	.366	.397	.878	.458	.469
.050	-.406	-.155	-.176	.739	.075	.065	.929	.392	.192	-.047	.274	.274
.075	-.458	-.177	-.177	.774	-.017	-.048	.965	.085	.064	-.083	.151	.131
.100	-.508	-.186	-.237	.826	.119	-.130	-.011	-.023	.033	-.114	.049	.038
.150	-.544	-.288	.304	.841	.145	.150	-.996	.054	.059	-.114	.003	.008
.200	-.529	.304	.314	.713	.176	.160	.878	.089	.089	-.114	.038	.028
.250	-.560	.319	.319	.723	.201	.176	.822	.130	.110	-.094	.074	.044
.300	-.565	.340	.324	.723	.201	.176	.822	.130	.110	-.094	.074	.044
.350	-.565	.340	.324	.723	.201	.176	.822	.130	.110	-.094	.074	.044
.400	-.616	.391	.349	.784	.254	.224	.806	.171	.148	-.042	.124	.100
.450	-.647	.375	.338	.774	.272	.236	.837	.182	.151	.043	.131	.100
.500	-.611	.385	.319	.790	.263	.222	.856	.201	.176	.043	.131	.100
.550	-.661	.320	.250	.815	.263	.222	.883	.192	.176	.043	.131	.100
.600	-.566	.247	.222	.820	.242	.212	.915	.159	.159	.043	.131	.100
.650	-.477	.212	.183	.842	.165	.135	.473	.130	.100	.043	.131	.100
.700	-.411	.007	.119	.308	.375	.375	.007	.003	.064	.330	.003	.044
.750	-.348	-.017	.012	.196	.002	.014	.171	.018	.028	.260	.018	.038
.800	-.019	.085	.096	.034	.085	.106	.033	.095	.110	.033	.105	.115
.850	.157	.139	.127	.142	.127	.098	.131	.133	.109	.136	.127	.115
.900	.183	.188	.172	.162	.157	.157	.151	.156	.146	.151	.161	.151
1.000												
Outboard station												
.010	.046	.155	-.088	-.274	.267	.339	-.430	.446	.502	-.512	.538	.665
.025	-.358	-.243	-.243	-.764	.103	.087	-.515	.221	.236	-.982	.314	.324
.050	-.353	-.289	-.331	-.583	.020	-.011	-.1008	.133	.118	-.056	.215	.205
.075	-.353	-.289	-.331	-.583	-.093	-.098	-.067	.006	.009	-.080	.071	.076
.100	-.444	-.331	-.336	.733	.129	-.109	-.166	.012	.006	-.183	.055	.045
.150	-.511	-.255	-.279	.903	.140	-.114	-.166	.037	.053	-.137	.019	.012
.200	-.522	-.289	-.274	.882	.119	-.114	-.166	.048	.053	-.173	.001	.012
.250	-.625	-.269	-.264	.851	.140	-.145	-.166	.079	.094	-.178	.058	.048
.300	-.620	.274	-.284	.862	.174	-.170	-.166	.101	.101	-.178	.041	.071
.350	-.656	.284	-.284	.877	.171	-.191	-.166	.130	.130	-.178	.041	.071
.400	-.657	.284	-.284	.857	.171	-.191	-.166	.130	.130	-.178	.041	.071
.450	-.740	.284	-.284	.857	.171	-.191	-.166	.130	.130	-.178	.041	.071
.500	-.774	.284	-.284	.857	.171	-.191	-.166	.130	.130	-.178	.041	.071
.550	-.562	.234	-.249	.909	.170	-.184	-.166	.135	.135	-.166	.141	.136
.600	-.501	.186	-.186	.932	.135	-.150	-.166	.104	.104	-.166	.101	.136
.650	-.433	.150	-.150	.983	.105	-.116	-.166	.076	.076	-.166	.076	.076
.700	-.372	.099	-.099	.983	.104	-.104	-.166	.037	.037	-.166	.048	.063
.750	-.264	-.006	.005	.983	.005	.020	-.166	.019	.019	-.166	.001	.007
.800	-.212	.072	.072	.983	.092	.098	-.166	.097	.107	-.166	.071	.030
.850	.005	.113	.113	.983	.113	.133	-.166	.128	.147	-.166	.038	.013
.900	.113	.139	.123	.983	.129	.123	-.166	.128	.148	-.166	.004	.068
1.000												

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11 - Continued

(e) $M = 0.825$

x/c	C_p at -											
	$\alpha = -2^\circ$			$\alpha = 0^\circ$			$\alpha = 1^\circ$			$\alpha = 2^\circ$		
	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C	Row A	Row B	Row C
Inboard station												
.010	.071	.158	.095	-.205	.407	.402	-.349	.535	.526	-.477	.617	.617
.025	-.235	.022	.012	-.565	.254	.234	-.705	.377	.377	-.774	.446	.451
.050	-.312	-.141	-.166	-.674	.071	.071	-.833	.179	.179	-.941	.259	.254
.075	-.467			-.723			-.898			-.996		
.100	-.451	.185	-.235	-.763	-.027	-.052	-.917	.070	.031	-.1040	.140	.115
.150	-.526	-.269	-.319	-.778	-.116	-.136	-.942	-.028	-.048	-.1035	.021	.021
.200	-.531	-.299	-.319	-.768	-.151	-.161	-.937	-.068	-.093	-.1045	.067	.067
.250	-.526	-.319	-.319	-.713	-.190	-.175	-.917	-.112	-.117	-.1050	.107	.107
.300	-.511	-.343	-.353	-.718	-.220	-.205	-.894	-.147	-.132	-.1035	.152	.152
.350	-.615	-.422	-.399	-.733	-.278	-.261	-.829	-.193	-.181	-.1045	.171	.171
.400	-.635	-.432	-.393	-.753	-.289	-.245	-.814	-.211	-.176	-.922	.164	.164
.450	-.666	-.473	-.410	-.771	-.307	-.272	-.811	-.239	-.199	-.858	.210	.210
.500	-.689	-.437	-.353	-.797	-.299	-.259	-.868	-.236	-.206	-.892	.201	.201
.550	-.713	-.353	-.313	-.828	-.278	-.244	-.903	-.227	-.199	-.911	.204	.204
.600	-.723	-.259	-.235	-.842	-.215	-.190	-.908	-.176	-.167	-.917	.151	.151
.650	-.723	-.221	-.192	-.795	-.180	-.163	-.590	-.153	-.135	-.470	.141	.141
.700	-.588	-.068	-.126	-.373	-.608	-.101	-.359	.001	-.083	-.359	.002	.002
.750	-.592			-.268			-.289			-.293		
.800	-.175	.018	-.013	-.156	.013	.003	-.167	-.004	-.004	-.201	.002	.002
.850	.047	.076	.051	.022	.071	.086	.013	.070	.070	.062	.066	.066
.900	.150	.136	.113	.106	.113	.078	.080	.101	.066	.026	.078	.078
.950	.175	.175	.170	.136	.136	.121	.105	.120	.070	.071	.086	.086
1.000												
Outboard station												
.010	.057	-.170	-.121	-.191	.226	.290	-.317	.394	.439	-.435	.508	.543
.025	-.355	-.266	-.315	-.674	.058	.043	-.794	.181	.156	-.878	.306	.256
.050	-.335	-.305	-.325	-.748	-.017	-.047	-.884	.127	.097	-.993	.182	.122
.075	-.330			-.544			-.819			-.978		
.100	-.325	-.330	-.400	-.869	-.171	-.151	-.856	.013	-.033	-.993	.047	.052
.150	-.355	-.325	-.320	-.833	-.141	-.161	-.859	-.028	-.063	-.1063	.012	.012
.200	-.470	-.320	-.291	-.823	-.196	-.146	-.919	-.043	-.068	-.1043	.027	.027
.250	-.585	-.291	-.266	-.863	-.196	-.151	-.959	-.058	-.073	-.1082	.042	.042
.300	-.624	-.310	-.320	-.858	-.191	-.181	-.959	-.097	-.162	-.1057	.087	.087
.350	-.664	-.316	-.311	-.878	-.197	-.201	-.984	-.130	-.145	-.1102	.111	.111
.400	-.664	-.310	-.315	-.892	-.216	-.216	-.1009	-.167	-.147	-.908	.106	.106
.450	-.733	-.321	-.316	-.876	-.230	-.235	-.1002	-.183	-.188	-.908	.177	.177
.500	-.758	-.281	-.291	-.912	-.216	-.221	-.849	-.157	-.172	-.945	.164	.164
.550	-.758	-.259	-.268	-.871	-.201	-.220	-.849	-.174	-.193	-.904	.177	.177
.600	-.858	-.201	-.206	-.559	-.171	-.166	-.476	-.152	-.152	-.904	.152	.152
.650	-.855	-.158	-.158	-.364	-.139	-.149	-.432	-.121	-.130	-.427	.155	.155
.700	-.325	-.122	-.111	-.276	-.151	-.097	-.381	-.068	-.083	-.421	.057	.057
.750	-.325			-.206			-.317			-.355		
.800	-.526	-.007	-.007	-.181	-.032	-.042	-.257	-.013	-.048	-.321	.022	.022
.850	-.053	.053	.088	-.087	.043	.048	-.132	.027	.027	-.057	.027	.027
.900	.142	.124	.134	.018	.043	.057	-.086	.028	.018	-.246	.111	.111
.950	.157	.122	.167	.048	.033	.058	-.003	.037	-.023	-.157	.207	.197
1.000												

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11 - Continued

(f) $M = 0.700$ and 0.750 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.700$										
.010	-.026	-.028	-.478	.374	-.743	.532	-.934	.633	-1.287	.775
.025	-.409	-.099	-.903	.209	-1.190	.361	-1.374	.473	-1.634	.621
.050	-.515	-.228	-.946	.032	-1.388	.178	-1.636	.284	-1.904	.427
.075	-.515		-.875		-1.225		-1.530		-1.911	
.100	-.586	-.252	-.917	-.044	-1.091	.066	-1.501	.149	-1.882	.303
.150	-.536		-.790		-.920		-1.055		-1.840	
.200	-.529	-.258	-.726	-.103	-.821	-.023	-.835	.031	-1.776	.155
.250	-.529		-.691		-.786		-.814		-1.634	
.300	-.536	-.240	-.684	-.127	-.764	-.052	-.793	-.011	-.954	.102
.350	-.536		-.662		-.729		-.764		-.791	
.400	-.551	-.246	-.662	-.162	-.722	-.105	-.750	-.070	-.671	.025
.450	-.543		-.648		-.686		-.715		-.657	
.500	-.529	-.228	-.606	-.162	-.644	-.117	-.658	-.087	-.628	-.004
.550	-.501		-.570		-.594		-.608		-.593	
.600	-.487	-.181	-.535	-.133	-.552	-.099	-.552	-.070	-.558	-.004
.650	-.458		-.492		-.502		-.502		-.508	
.700	-.416	-.110	-.443	-.074	-.445	-.052	-.438	-.028	-.437	.031
.750	-.352		-.358		-.353		-.346		-.352	
.800	-.267	-.016	-.266	.003	-.254	.013	-.240	.031	-.253	.073
.900	-.033	.067	-.032	.068	-.013	.072	-.006	.078	-.033	.120
.975	.066	.126	.003	.109	.008	.101	-.084	.101	.066	.143
1.000	.158		-.004		.065		.079		.165	
$M = 0.750$										
.010	.037	-.051	-.364	.352	-.571	.503	-.745	.621	-1.004	.750
.025	-.370	-.100	-.791	.212	-1.004	.341	-1.132	.444	-1.347	.605
.050	-.499	-.234	-.998	.029	-1.256	.148	-1.417	.261	-1.619	.416
.075	-.518		-.907		-1.146		-1.359		-1.612	
.100	-.596	-.261	-.914	-.068	-1.198	.040	-1.359	.132	-1.599	.276
.150	-.564		-.797		-1.191		-1.359		-1.580	
.200	-.551	-.277	-.778	-.122	-.997	-.046	-1.320	.024	-1.554	.136
.250	-.557		-.739		-.816		-1.275		-1.554	
.300	-.564	-.256	-.752	-.133	-.758	-.073	-1.100	-.014	-1.554	.077
.350	-.570		-.713		-.758		-.822		-1.250	
.400	-.589	-.272	-.720	-.170	-.784	-.132	-.751	-.078	-.927	.007
.450	-.589		-.713		-.790		-.725		-.823	
.500	-.570	-.250	-.655	-.176	-.693	-.132	-.680	-.094	-.707	-.014
.550	-.538		-.590		-.603		-.602		-.578	
.600	-.518	-.202	-.552	-.138	-.564	-.105	-.551	-.078	-.487	-.014
.650	-.480		-.506		-.500		-.499		-.435	
.700	-.428	-.116	-.448	-.079	-.448	-.052	-.454	-.030	-.403	.002
.750	-.344		-.345		-.338		-.338		-.351	
.800	-.253	-.014	-.235	.002	-.228	.018	-.221	.040	-.274	.056
.900	-.008	.067	.004	.072	.005	.078	.011	.094	-.132	.093
.975	.134	.126	.114	.110	.102	.104	.108	.110	.037	.093
1.000	.154		.134		.121		.134		.088	

TABLE 18.- WING SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 11 - Continued

(g) $M = 0.775$ and 0.800 ; center station

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B	Row A	Row B
$M = 0.775$										
.010	.062	-.069	-.297	.334	-.497	.503	-.620	.586	-.869	.720
.025	-.359	-.100	-.730	.200	-.900	.333	-1.023	.426	-1.210	.565
.050	-.496	-.255	-.935	.009	-1.136	.136	-1.277	.240	-1.476	.385
.075	-.502		-.885		-1.068		-1.228		-1.470	
.100	-.607	-.296	-.966	-.074	-1.130	.033	-1.253	.127	-1.464	.240
.150	-.589		-.910		-1.148		-1.253		-1.464	
.200	-.570	-.296	-.774	-.136	-1.105	-.055	-1.240	.018	-1.464	.111
.250	-.570		-.755		-1.068		-1.240		-1.452	
.300	-.595	-.276	-.755	-.146	-.857	-.080	-1.228	-.023	-1.303	.049
.350	-.595		-.780		-.838		-1.215		-.856	
.400	-.632	-.296	-.805	-.192	-.894	-.137	-1.110	-.069	-.825	-.023
.450	-.663		-.842		-.931		-.930		-.782	
.500	-.638	-.276	-.817	-.192	-.931	-.148	-.918	-.095	-.745	-.059
.550	-.570		-.619		-.658		-.583		-.676	
.600	-.545	-.214	-.544	-.156	-.522	-.117	-.496	-.085	-.621	-.070
.650	-.496		-.495		-.478		-.447		-.559	
.700	-.434	-.126	-.439	-.089	-.416	-.065	-.453	-.033	-.515	-.044
.750	-.347		-.346		-.317		-.298		-.441	
.800	-.242	-.023	-.222	.004	-.212	.018	-.192	.034	-.403	-.008
.900	.000	.060	.013	.066	.018	.075	.025	.096	-.292	-.008
.975	.124	.117	.026	.107	.111	.106	.099	.127	-.186	-.034
1.000	.143		.137		.123		.142		-.174	
$M = 0.800$										
.010	.074	-.051	-.218	.302	-.410	.475	-.518	.564		
.025	-.332	-.110	-.641	.183	-.786	.296	-.905	.396		
.050	-.457	-.284	-.856	-.001	-1.030	.118	-1.132	.187		
.075	-.504		-.832		-.982		-1.114			
.100	-.594	-.309	-.927	-.085	-1.048	.023	-1.144	.098		
.150	-.600		-.898		-1.066		-1.180			
.200	-.570	-.319	-.880	-.150	-1.054	-.066	-1.162	-.017		
.250	-.600		-.761		-1.042		-1.168			
.300	-.612	-.299	-.766	-.165	-1.042	-.091	-1.168	-.051		
.350	-.642		-.761		-1.006		-1.168			
.400	-.677	-.323	-.826	-.214	-.857	-.151	-1.162	-.101		
.450	-.695		-.868		-.899		-1.209			
.500	-.725	-.299	-.892	-.214	-.947	-.160	-1.144	-.121		
.550	-.707		-.933		-.976		-.941			
.600	-.558	-.224	-.910	-.165	-.947	-.131	-.589	-.106		
.650	-.493		-.463		-.470		-.446			
.700	-.427	-.135	-.361	-.095	-.350	-.066	-.446	-.051		
.750	-.332		-.284		-.255		-.261			
.800	-.218	-.026	-.182	-.006	-.159	.018	-.166	.018		
.900	.020	.064	.020	.069	.019	.073	.019	.078		
.975	.127	.118	.080	.108	.097	.098	.103	.108		
1.000	.145		.092		.115		.133			

(h) $M = 0.825$; center station

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TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1

(a) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.272	.117	.240	.071	.223	.068	.197	.041	.141	.054
.025	.277	.049	.279	.032	.262	.068	.221	.027	.156	.064
.050	.369	-.004	.410	-.007	.412	.053	.395	.061	.325	.059
.100	.219	.015	.521	.085	.621	.141	.672	.148	.737	.224
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.034	-.005	.018	-.011	.058	-.000	.075	-.012	.141	-.001
.025	.010	-.053	.018	-.069	.039	-.053	.032	-.065	.064	-.048
.050	-.024	-.116	.003	-.118	.034	-.102	.037	-.104	.063	-.077
.100	.000	-.145	-.007	-.137	.058	-.116	.051	-.104	.069	-.067
.200	.039	-.087	.105	-.069	.155	-.044	.177	-.046	.246	.001
.300	.073	.000	.187	.023	.262	.048	.303	.061	.418	.112
.400	-.238	.044	.037	.110	.165	.155	.255	.168	.408	.238
.500	-.568	-.268	-.321	-.036	-.184	.048	-.099	.109	.103	.224
.600										
.700	-.286	-.286	-.200	-.171	-.131	-.087	-.094	-.036	.054	.078
.800										
.900	-.179	-.189	-.103	-.123	-.049	-.078	-.026	-.051	.049	.030
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.272	.160	.269	.124	.271	.116	.254	.104	.233	.097
.025	.373	.126	.245	.114	.407	.111	.264	.109	.175	.107
.050	.276	.117	.385	.124	.440	.145	.448	.143	.450	.160
.100	-.353	.088	.119	.182	.353	.242	.482	.254	.707	.305
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.048	-.034	-.011	-.045	.024	-.039	.036	-.051	.063	-.038
.025	.063	.010	-.055	.003	.194	.014	.041	.007	-.006	.025
.050	-.077	-.106	-.045	-.098	.005	-.092	.007	-.089	.059	-.067
.100	-.082	-.135	-.021	-.122	.019	-.092	.041	-.089	.112	-.052
.200	-.019	-.048	.061	-.031	.131	-.000	.157	-.002	.247	.035
.300	-.077	.029	.080	.066	.179	.092	.235	.099	.378	.155
.400	-.499	-.043	-.229	.071	-.058	.140	.046	.177	.247	.267
.500	-.479	-.281	-.267	-.108	-.165	-.010	-.099	.046	.045	.160
.600										
.700	-.160	-.179	-.103	-.103	-.058	-.044	-.041	-.012	.039	.068
.800										
.900	-.048	-.063	-.011	-.036	.019	-.005	.032	-.002	.088	.059
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(b) $M = 0.750$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.295	.118	.265	.084	.237	.089	.221	.048	.182	.056
.025	.317	.042	.279	.040	.272	.055	.256	.048	.198	.052
.050	.401	-.028	.438	-.022	.436	.047	.429	.053	.361	.043
.100	.281	.007	.530	.062	.631	.091	.694	.146	.755	.198
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.020	.007	.038	.089	.069	-.002	.097	-.005	.153	-.008
.025	.016	-.059	.071	-.057	.042	-.060	.053	-.049	.083	-.041
.050	-.015	-.117	.062	-.114	.033	-.104	.057	-.093	.100	-.063
.100	.003	-.148	.054	-.123	.051	-.117	.061	-.058	.038	-.059
.200	.060	-.086	.133	-.057	.162	-.046	.194	-.027	.259	.012
.300	.100	.011	.217	.040	.272	.051	.318	.075	.415	.114
.400	-.205	.047	.045	.115	.157	.135	.256	.181	.405	.246
.500	-.666	-.223	-.370	-.048	-.223	.029	-.129	.101	.069	.220
.600										
.700	-.316	-.316	-.211	-.180	-.157	-.108	-.102	-.054	-.019	.056
.800										
.900	-.210	-.219	-.123	-.141	-.077	-.099	-.036	-.058	.034	.052
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.294	.161	.282	.128	.289	.122	.286	.123	.259	.104
.025	.498	.122	.340	.119	.532	.117	.384	.114	.272	.109
.050	.303	.104	.401	.115	.457	.135	.472	.141	.475	.133
.100	-.298	.095	.111	.177	.347	.219	.476	.234	.651	.281
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.046	-.037	-.008	-.052	.029	-.042	.048	-.036	.051	-.032
.025	.219	.007	.053	.005	.351	.011	.176	.013	.091	.021
.050	-.059	-.117	-.035	-.105	.024	-.095	.035	-.080	.078	-.063
.100	-.073	-.139	-.013	-.118	.029	-.099	.066	-.080	.122	-.054
.200	-.006	-.046	.080	-.026	.135	-.007	.172	.013	.254	.038
.300	-.064	.038	.093	.067	.188	.095	.251	.114	.360	.149
.400	-.536	-.046	-.268	.071	-.090	.126	.035	.176	.224	.246
.500	-.616	-.307	-.299	-.123	-.196	-.033	-.133	.035	.007	.144
.600										
.700	-.161	-.192	-.110	-.114	-.068	-.055	-.040	-.018	.003	.043
.800										
.900	-.055	-.077	-.022	-.044	.002	-.011	.035	.004	.047	.025
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(c) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	-.015	-.015	.274	.087	.259	.072	.249	.056	.202	.079
.025	-.015	-.015	.291	.028	.284	.025	.291	.041	.236	.045
.050	-.015	-.015	.443	-.018	.450	-.013	.448	.066	.407	.045
.100	-.015	-.015	.545	.058	.637	.089	.698	.147	.769	.215
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.015	-.015	.045	.087	.080	.004	.113	.019	.172	.036
.025	-.015	-.015	.037	-.061	.051	-.051	.070	-.032	.108	-.011
.050	-.015	-.015	.028	-.116	.051	-.098	.079	-.074	.130	-.036
.100	-.015	-.015	.054	-.120	.021	-.111	.041	-.078	.066	-.028
.200	-.015	-.015	.138	-.052	.174	-.039	.219	-.015	.283	.040
.300	-.015	-.015	.219	.032	.280	.055	.333	.091	.428	.147
.400	-.015	-.015	.041	.117	.157	.144	.253	.185	.411	.253
.500	-.015	-.015	-.365	-.052	-.242	.029	-.125	.104	.066	.228
.600										
.700	-.015	-.015	-.222	-.156	-.170	-.115	-.108	-.057	-.011	.057
.800										
.900	-.015	-.015	-.133	-.158	-.085	-.111	-.032	-.057	.032	.057
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.307	.171	.298	.146	.305	.135	.299	.138	.287	.134
.025	.587	.129	.404	.121	.657	.118	.472	.121	.351	.138
.050	.332	.108	.417	.117	.470	.127	.489	.146	.504	.155
.100	-.239	.078	.125	.163	.339	.199	.477	.248	.682	.295
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.032	-.040	-.001	-.048	.034	-.043	.057	-.032	.112	-.002
.025	.345	.015	.134	.007	.496	.012	.286	.015	.168	.027
.050	-.040	-.121	-.018	-.120	.038	-.089	.053	-.082	.104	-.041
.100	-.066	-.134	-.001	-.116	.042	-.089	.074	-.074	.151	-.028
.200	.006	-.040	.091	-.023	.144	-.000	.189	.019	.266	.066
.300	-.036	.044	.104	.079	.186	.097	.252	.125	.372	.172
.400	-.544	-.045	-.318	.066	-.119	.131	.002	.176	.210	.257
.500	-.667	-.328	-.314	-.128	-.208	-.038	-.133	.023	-.002	.134
.600										
.700	-.155	-.197	-.111	-.116	-.072	-.055	-.036	-.019	-.002	.036
.800										
.900	-.053	-.074	-.023	-.044	.004	-.013	.040	.011	.023	.002
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(d) $M = 0.800$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	-.015	-.015	.290	.090	.274	.082	.255	.067	.212	.057
.025	-.015	-.015	.306	.037	.319	.041	.291	.031	.257	.061
.050	-.015	-.015	.465	-.008	.466	.078	.459	.043	.420	.037
.100	-.015	-.015	.566	.058	.642	.107	.695	.137	.767	.196
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.015	-.015	.058	.058	.094	.017	.120	.023	.175	.041
.025	-.015	-.015	.054	-.048	.066	-.040	.076	-.030	.114	.077
.050	-.015	-.015	.041	-.089	.066	-.081	.084	-.071	.130	-.041
.100	-.015	-.015	.066	-.113	.037	-.094	.043	-.079	.073	-.037
.200	-.015	-.015	.155	-.044	.192	-.028	.218	-.006	.290	.037
.300	-.015	-.015	.237	.045	.286	.070	.336	.084	.424	.135
.400	-.015	-.015	.049	.123	.164	.152	.247	.181	.400	.237
.500	-.015	-.015	-.447	-.056	-.285	.029	-.161	.076	.020	.200
.600										
.700	-.015	-.015	-.240	-.215	-.171	-.138	-.124	-.079	-.045	.024
.800										
.900	-.015	-.015	-.154	-.174	-.102	-.118	-.067	-.088	-.004	-.037
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.323	.168	.313	.151	.318	.139	.315	.124	.297	.118
.025	.658	.123	.484	.110	.787	.119	.563	.120	.403	.114
.050	.355	.107	.439	.122	.481	.143	.498	.144	.513	.126
.100	-.198	.087	.147	.155	.326	.200	.453	.209	.651	.265
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.027	-.040	.009	-.044	.041	-.032	.059	-.034	.110	-.008
.025	.502	.013	.236	.009	.665	.013	.392	.010	.228	.012
.050	-.027	-.121	-.003	-.109	.057	-.085	.063	-.087	.106	-.049
.100	-.052	-.133	.005	-.113	.053	-.085	.079	-.075	.142	-.037
.200	.026	-.044	.102	-.012	.151	.009	.189	.018	.265	.053
.300	-.023	.046	.110	.082	.180	.114	.246	.120	.354	.163
.400	-.573	-.040	-.365	.066	-.167	.127	-.026	.161	.155	.232
.500	-.765	-.370	-.349	-.154	-.248	-.061	-.173	-.010	-.061	.098
.600										
.700	-.198	-.211	-.117	-.121	-.077	-.069	-.055	-.047	-.041	-.004
.800										
.900	-.052	-.076	-.032	-.048	.000	-.020	-.002	-.030	-.021	-.061
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(e) $M = 0.825$; $p_{t,e}/p_{t,\infty} = 1.5$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.330	.121	.302	.098	.284	.087	.279	.099	.231	.062
.025	.373	.039	.333	.031	.331	.036	.319	.044	.274	.030
.050	.463	-.040	.478	-.024	.484	.040	.484	.051	.447	.018
.100	.412	-.028	.565	.021	.650	.079	.712	.122	.846	.172
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.023	.019	.074	.058	.103	.024	.138	.036	.187	.050
.025	.047	-.040	.058	-.044	.075	-.035	.095	-.011	.117	.034
.050	.054	-.111	.047	-.091	.075	-.082	.099	-.055	.136	-.033
.100	.027	-.138	.035	-.110	.044	-.090	.063	-.063	.077	-.033
.200	.113	-.067	.164	-.040	.201	-.019	.240	.012	.294	.042
.300	.168	.023	.247	.043	.300	.075	.350	.095	.415	.132
.400	-.087	.062	.058	.113	.162	.146	.240	.185	.455	.231
.500	-.744	-.225	-.479	-.079	-.329	.009	-.220	.071	.062	.172
.600										
.700	-.425	-.641	-.271	-.248	-.215	-.164	-.157	-.102	-.080	-.009
.800										
.900	-.398	-.394	-.205	-.228	-.149	-.168	-.106	-.133	-.048	-.084
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.337	.156	.328	.148	.330	.138	.326	.145	.305	.120
.025	.843	.121	.606	.125	.978	.118	.683	.122	.520	.116
.050	.384	.074	.454	.117	.495	.138	.510	.149	.520	.124
.100	-.138	.054	.144	.121	.303	.185	.424	.216	.618	.250
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.028	-.052	.067	-.048	.040	-.035	.067	-.027	.105	-.017
.025	.694	-.001	.355	-.001	.899	.009	.537	.008	.359	-.001
.050	-.016	-.142	.007	-.110	.079	-.089	.083	-.074	.112	-.052
.100	-.048	-.142	.011	-.110	.060	-.086	.087	-.066	.148	-.041
.200	.031	-.048	.105	-.016	.154	.005	.192	.032	.261	.054
.300	-.016	.039	.113	.074	.185	.107	.236	.130	.336	.159
.400	-.589	-.056	-.365	.046	-.227	.107	-.098	.153	.101	.214
.500	-.726	-.440	-.651	-.189	-.301	-.093	-.255	-.043	-.119	.058
.600										
.700	-.393	-.275	-.138	-.142	-.113	-.105	-.094	-.082	-.076	-.041
.800										
.900	-.060	-.103	-.071	-.087	-.054	-.078	-.066	-.094	-.072	-.107
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(f) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.3$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.274	.104	.242	.087	.132	-.039	.100	-.041	.150	.043
	.253	.051	.276	.048	.171	-.067	.148	-.075	.169	.019
	.371	.080	.417	.062	.321	-.072	.309	-.041	.339	.058
	.240	.056	.514	.087	.516	.025	.580	.071	.731	.218
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.041	-.031	.033	-.001	-.043	-.009	-.021	-.104	.135	-.005
	.003	-.060	.024	-.059	-.067	-.155	-.055	-.148	.063	-.049
	-.031	-.123	.058	-.098	-.067	-.199	-.055	-.196	.087	-.068
	-.007	-.152	.048	-.117	-.048	-.213	-.046	-.201	.024	-.068
	.032	-.094	.111	-.059	.049	-.140	.081	-.138	.242	-.005
	.075	-.002	.198	.038	.156	-.048	.212	-.031	.407	.106
	-.220	.027	.058	.106	.074	.040	.158	.071	.412	.223
	-.549	-.215	-.287	-.035	-.262	-.053	-.162	.017	.106	.232
	-.298	-.298	-.185	-.156	-.233	-.194	-.201	-.138	-.000	.068
	-.172	-.191	-.088	-.103	-.150	-.165	-.114	-.138	.043	.039
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.273	.152	.271	.130	.267	.112	.255	.105	.222	.096
	.273	.128	.222	.116	.233	.102	.206	.109	.159	.106
	.273	.114	.392	.130	.437	.132	.453	.143	.445	.140
	-.345	.075	.120	.198	.340	.238	.487	.259	.701	.299
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.055	-.041	-.001	-.040	.020	-.043	.042	-.046	.082	-.039
	-.050	-.017	-.078	-.006	-.043	-.004	-.036	-.002	-.015	.009
	-.064	-.108	-.040	-.098	-.014	-.091	.003	-.089	.048	-.068
	-.054	-.137	-.015	-.107	.020	-.096	.046	-.089	.111	-.053
	-.026	-.050	.072	-.020	.122	-.009	.163	-.007	.237	.034
	-.075	.022	.096	.072	.185	.083	.245	.095	.367	.135
	-.481	-.055	-.185	.082	-.048	.136	.056	.172	.251	.261
	-.490	-.273	-.233	-.088	-.150	-.014	-.084	.046	.058	.159
	-.162	-.191	-.093	-.093	-.062	-.053	-.041	-.021	.034	.058
	-.050	-.070	-.006	-.025	.015	-.009	.032	.008	.082	.058

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(g) $M = 0.750$; $p_{t,e}/p_{t,\infty} = 1.3$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.295	.118	.263	.095	.154	-.019	.135	-.007	.174	.046
.025	.217	.065	.308	.033	.198	-.068	.171	-.064	.205	.028
.050	.461	-.019	.436	.046	.349	-.041	.339	-.064	.373	.037
.100	.266	-.001	.525	.077	.539	.030	.600	.082	.753	.201
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.019	.003	.037	.002	-.023	-.085	.007	-.086	.157	.020
.025	.016	-.059	.029	-.056	-.050	-.138	-.038	-.135	.086	.064
.050	-.015	-.116	.020	-.109	-.050	-.187	-.024	-.175	.108	-.055
.100	.003	-.152	.051	-.127	-.032	-.192	-.069	-.175	.046	-.047
.200	.056	-.081	.126	-.056	.074	-.125	.113	-.108	.263	.024
.300	.100	.007	.215	.037	.185	-.032	.237	-.011	.408	.112
.400	-.214	.030	.051	.099	.074	.048	.153	.082	.413	.218
.500	-.642	-.222	-.357	-.042	-.289	-.050	-.184	.016	.090	.210
.600										
.700	-.311	-.324	-.211	-.189	-.240	-.196	-.192	-.144	.073	.051
.800										
.900	-.200	-.214	-.113	-.135	-.152	-.174	-.126	-.139	.037	.020
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.254	.157	.285	.130	.291	.127	.281	.130	.258	.121
.025	.369	.126	.307	.117	.317	.123	.294	.113	.258	.117
.050	.312	.100	.408	.121	.454	.136	.475	.135	.478	.156
.100	-.284	.069	.126	.174	.357	.211	.488	.236	.685	.271
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.050	-.046	-.011	-.051	.025	-.045	.046	-.042	.090	-.016
.025	.062	-.010	.020	-.011	.056	-.001	.060	.002	.081	.006
.050	-.068	-.121	-.038	-.122	-.001	-.094	.024	-.086	.072	-.051
.100	-.076	-.138	-.016	-.122	.025	-.098	.064	-.077	.130	-.047
.200	-.006	-.046	.091	-.033	.136	-.006	.179	.011	.253	.046
.300	-.046	.034	.095	.064	.193	.096	.254	.113	.355	.152
.400	-.455	-.046	-.246	.068	-.090	.127	.024	.179	.218	.249
.500	-.615	-.310	-.281	-.126	-.178	-.023	-.117	.038	.024	.139
.600										
.700	-.151	-.196	-.117	-.122	-.072	-.059	-.042	-.029	.006	.042
.800										
.900	-.054	-.075	-.020	-.042	.008	-.010	.033	.007	.050	.024
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(h) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.3$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.301	.124	.283	.108	.253	.078	.154	-.020	.202	.366
.025	.318	.047	.321	.053	.299	.023	.192	-.063	.236	.019
.050	.420	-.020	.457	.083	.452	.019	.358	-.071	.397	.045
.100	.318	-.003	.550	.079	.635	.091	.604	.061	.761	.189
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.016	.005	.053	.011	.078	.049	.022	-.075	.168	.036
.025	.013	-.059	.045	-.040	.053	-.045	-.020	-.126	.100	.070
.050	-.012	-.122	.032	-.095	.049	-.092	-.016	-.173	.117	-.053
.100	.005	-.152	.062	-.112	.023	-.100	-.054	-.173	.062	-.027
.200	.068	-.084	.142	-.040	.172	-.032	.120	-.097	.274	.036
.300	.119	.005	.227	.049	.282	.066	.243	-.003	.414	.130
.400	-.181	.043	.066	.108	.163	.138	.162	.082	.397	.214
.500	-.864	-.232	-.363	-.040	-.232	.036	-.220	.014	.062	.202
.600										
.700	-.329	-.334	-.210	-.185	-.160	-.122	-.199	-.152	-.044	.036
.800										
.900	-.224	-.232	-.117	-.138	-.075	-.104	-.131	-.148	.015	.032
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.305	.165	.299	.142	.303	.129	.293	.120	.277	.108
.025	.448	.127	.367	.121	.379	.120	.353	.103	.320	.112
.050	.334	.098	.422	.125	.460	.129	.484	.132	.493	.134
.100	-.236	.051	.146	.168	.328	.210	.471	.238	.658	.265
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.046	-.050	.002	-.040	.031	-.041	.048	-.041	.095	-.019
.025	.155	-.012	.037	-.002	.129	-.003	.124	-.003	.138	.002
.050	-.054	-.126	-.015	-.112	.010	-.100	.035	-.088	.087	-.053
.100	-.075	-.147	-.006	-.112	.031	-.096	.069	-.092	.129	-.044
.200	.005	-.050	.095	-.023	.142	.002	.175	.005	.252	.045
.300	-.033	.039	.100	.078	.184	.095	.247	.115	.345	.150
.400	-.506	-.054	-.265	.070	-.113	.125	.001	.162	.172	.235
.500	-.675	-.337	-.230	-.129	-.206	-.041	-.135	.014	-.027	.108
.600										
.700	-.160	-.211	-.108	-.116	-.075	-.070	-.046	-.033	-.031	.007
.800										
.900	-.063	-.084	-.019	-.040	.006	-.015	.026	.005	-.002	-.031
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(i) $M = 0.800$; $p_{t,e}/p_{t,\infty} = 1.3$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.315	.128	.288	.104	.273	.102	.176	.004	.135	-.512
.025	.336	.046	.329	.035	.306	.036	.233	-.049	.171	-.041
.050	.442	-.035	.468	.035	.469	.028	.380	-.053	.355	-.045
.100	.268	-.015	.566	.063	.641	.106	.621	.041	.681	.102
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.001	.018	.059	.022	.089	.024	.041	-.053	.098	-.041
.025	.030	-.048	.047	-.043	.065	-.037	-.000	-.110	.029	-.053
.050	.042	-.117	.039	-.092	.061	-.086	.004	-.155	.049	-.110
.100	.018	-.141	.027	-.120	.036	-.086	-.033	-.143	-.012	-.114
.200	.087	-.076	.149	-.047	.191	-.017	.143	-.086	.200	-.045
.300	.144	.018	.235	.047	.289	.073	.257	.020	.331	.049
.400	-.141	.038	.059	.104	.163	.142	.167	.094	.310	.135
.500	-.773	-.239	-.374	-.063	-.262	.028	-.221	.004	-.061	.106
.600										
.700	-.325	-.349	-.231	-.214	-.168	-.127	-.204	-.159	-.138	-.081
.800										
.900	-.251	-.256	-.141	-.169	-.086	-.119	-.131	-.155	-.098	-.122
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.323	.156	.316	.145	.317	.146	.314	.147	.285	.126
.025	.559	.119	.442	.120	.456	.130	.424	.126	.395	.118
.050	.359	.095	.438	.120	.476	.126	.493	.126	.505	.151
.100	-.190	.058	.149	.161	.330	.203	.460	.236	.639	.244
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.039	-.047	.002	-.047	.036	-.033	.061	-.029	.102	-.016
.025	.351	-.011	.181	-.006	.220	-.001	.208	-.000	.216	-.004
.050	-.039	-.125	-.006	-.112	.028	-.090	.053	-.078	.098	-.057
.100	-.064	-.141	.002	-.116	.044	-.094	.081	-.073	.138	-.041
.200	.022	-.043	.104	-.018	.154	.008	.196	.024	.252	.045
.300	-.015	.038	.108	.084	.191	.105	.253	.118	.342	.155
.400	-.560	-.056	-.340	.067	-.151	.118	-.012	.167	.147	.216
.500	-.654	-.381	-.328	-.157	-.233	-.070	-.151	-.004	-.073	.082
.600										
.700	-.162	-.222	-.116	-.128	-.082	-.078	-.057	-.049	-.057	-.020
.800										
.900	-.060	-.080	-.031	-.051	.004	-.025	.004	-.016	-.036	-.069
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(j) $M = 0.825$; $p_{t,e}/p_{t,\infty} = 1.3$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.331	.119	.307	.102	.291	.102	.199	.022	.251	.070
.025	.359	.024	.342	.047	.322	.023	.247	-.037	.279	.050
.050	.465	-.047	.480	.066	.488	.039	.404	-.029	.456	.042
.100	.358	-.015	.563	.063	.645	.090	.625	.050	.779	.184
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.016	.024	.070	.027	.098	.027	.057	-.037	.196	.058
.025	.040	-.043	.063	-.036	.074	-.032	.018	-.010	.129	.015
.050	.016	-.114	.047	-.083	.066	-.083	.022	-.136	.149	-.021
.100	.028	-.137	.039	-.107	.043	-.087	-.017	-.136	.086	-.025
.200	.107	-.062	.165	-.032	.200	-.016	.156	-.069	.294	.046
.300	.162	.020	.256	.059	.303	.074	.270	.026	.428	.137
.400	-.117	.044	.074	.106	.149	.133	.160	.093	.456	.216
.500	-.763	-.224	-.477	-.068	-.320	-.001	-.262	-.006	.034	.176
.600										
.700	-.334	-.369	-.257	-.233	-.217	-.182	-.238	-.187	-.076	-.009
.800										
.900	-.258	-.294	-.174	-.209	-.142	-.170	-.171	-.199	-.040	-.060
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.342	.165	.334	.153	.325	.125	.329	.144	.313	.140
.025	.664	.122	.538	.129	.573	.121	.509	.128	.506	.113
.050	.381	.083	.456	.129	.483	.125	.505	.140	.522	.152
.100	-.141	.063	.149	.157	.290	.176	.415	.207	.612	.235
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.031	-.047	.015	-.036	.031	-.040	.065	-.029	.109	-.009
.025	.542	-.011	.310	-.004	.369	-.012	.313	-.006	.341	-.001
.050	-.019	-.133	.007	-.107	.035	-.103	.065	-.076	.113	-.048
.100	-.054	-.145	.007	-.103	.043	-.099	.089	-.072	.144	-.040
.200	.036	-.043	.113	-.004	.149	.003	.191	.022	.258	.054
.300	-.003	.040	.121	.086	.176	.101	.238	.120	.345	.156
.400	-.561	-.043	-.351	.058	-.232	.105	-.092	.152	.125	.211
.500	-.671	-.435	-.370	-.177	-.307	-.111	-.230	-.041	-.107	.062
.600										
.700	-.306	-.255	-.126	-.142	-.126	-.122	-.104	-.088	-.076	-.036
.800										
.900	-.050	-.090	-.048	-.075	-.067	-.087	-.053	-.084	-.056	-.095
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(k) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.273	.123	.243	.083	.227	.081	.104	-.032	.133	.055
.025	.283	.055	.272	.044	.251	.038	.143	-.066	.157	.050
.050	.375	.045	.418	.044	.417	.033	.304	-.061	.313	.080
.100	.229	.045	.506	.093	.616	.145	.581	.051	.711	.196
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.033	.045	.020	-.009	.057	-.006	-.012	-.100	.162	.002
.025	-.009	-.067	.010	-.068	.033	-.055	-.051	-.148	.094	-.022
.050	-.033	-.135	.000	-.107	.028	-.099	-.041	-.182	.114	-.032
.100	-.018	-.149	-.005	-.121	.052	-.108	-.036	-.182	.055	-.018
.200	.035	-.086	.107	-.058	.145	-.035	.080	-.109	.254	.041
.300	.079	-.004	.185	.029	.242	.052	.192	-.036	.386	.109
.400	-.203	.006	.049	.078	.174	.110	.143	.036	.400	.172
.500	-.504	-.213	-.272	-.043	-.147	.042	-.153	.002	.128	.196
.600										
.700	-.285	-.305	-.194	-.180	-.138	-.099	-.197	-.148	-.013	.080
.800										
.900	-.159	-.169	-.082	-.097	-.040	-.050	-.100	-.109	.060	.075
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.272	.146	.272	.126	.270	.125	.254	.114	.201	.099
.025	.282	.127	.223	.117	.236	.120	.206	.104	.162	.104
.050	.277	.103	.393	.126	.440	.130	.453	.119	.390	.142
.100	-.338	.088	.136	.194	.363	.236	.487	.245	.646	.234
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.057	-.047	-.005	-.043	.028	-.035	.051	-.046	.099	-.013
.025	-.043	-.018	-.087	-.009	-.040	-.006	-.036	-.007	-.013	.002
.050	-.081	-.130	-.043	-.092	-.006	-.084	.012	-.080	.075	-.027
.100	-.086	-.139	-.014	-.097	.023	-.079	.056	-.065	.133	-.013
.200	-.018	-.052	.068	-.024	.120	-.001	.148	.007	.234	.055
.300	-.057	.025	.102	.068	.178	.091	.230	.099	.336	.133
.400	-.425	-.047	-.169	.078	-.026	.130	.065	.162	.254	.210
.500	-.445	-.261	-.213	-.092	-.128	-.006	-.061	.041	.075	.152
.600										
.700	-.154	-.198	-.097	-.106	-.055	-.055	-.036	-.022	.031	.079
.800										
.900	-.047	-.062	.005	-.009	.028	.008	.046	.022	.054	.094
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1.- Continued

(1) $M = 0.750$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.294	.122	.265	.092	.247	.070	.139	-.016	.171	.060
.025	.307	.051	.300	.030	.283	.025	.183	-.052	.202	.051
.050	.400	-.029	.442	.079	.438	.039	.343	-.029	.353	.069
.100	.272	.007	.526	.087	.624	.127	.591	.050	.729	.175
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.024	-.002	.034	-.001	.070	-.001	.024	-.078	.175	-.002
.025	.002	-.060	.030	-.054	.039	-.059	-.021	-.127	.109	-.015
.050	-.024	-.122	.056	-.099	.043	-.094	-.016	-.153	.127	-.028
.100	-.007	-.148	.052	-.116	.021	-.094	-.052	-.153	.078	-.015
.200	.055	-.082	.127	-.045	.158	-.028	.108	-.091	.264	.047
.300	.086	.007	.207	.043	.251	.052	.210	-.012	.397	.109
.400	-.193	.011	.056	.079	.158	.110	.157	.055	.388	.171
.500	-.577	-.228	-.307	-.050	-.201	.025	-.176	.002	.087	.175
.600										
.700	-.312	-.330	-.214	-.192	-.152	-.130	-.198	-.153	-.028	.051
.800										
.900	-.179	-.193	-.094	-.116	-.059	-.068	-.100	-.109	.043	.047
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.293	.152	.282	.131	.286	.127	.280	.130	.237	.113
.025	.386	.121	.313	.114	.326	.114	.302	.103	.263	.118
.050	.311	.099	.410	.105	.454	.140	.475	.147	.427	.135
.100	-.285	.082	.127	.167	.344	.224	.475	.249	.630	.215
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.051	-.046	-.006	-.050	.034	-.041	.059	-.029	.100	-.011
.025	.095	-.020	.021	-.014	.070	-.010	.068	-.003	.087	-.002
.050	-.073	-.126	-.032	-.107	.008	-.094	.041	-.074	.095	-.037
.100	-.077	-.135	-.014	-.103	.034	-.085	.072	-.056	.140	-.011
.200	-.002	-.046	.078	-.028	.127	.003	.165	.015	.232	.051
.300	-.051	.033	.105	.070	.180	.092	.227	.108	.339	.131
.400	-.457	-.046	-.209	.070	-.072	.127	.041	.165	.215	.188
.500	-.554	-.303	-.236	-.125	-.156	-.032	-.087	.028	.038	.122
.600										
.700	-.152	-.214	-.107	-.125	-.068	-.068	-.038	-.029	.007	.051
.800										
.900	-.051	-.069	-.006	-.028	.021	.003	.046	.028	.065	.056
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(m) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.301	.123	.278	.104	.263	.092	.158	.005	.200	.060
.025	.335	.042	.308	.040	.322	.041	.209	-.042	.222	.052
.050	.420	-.034	.452	.074	.458	.075	.366	-.059	.379	.060
.100	.322	-.004	.542	.091	.637	.114	.604	.060	.731	.171
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.017	-.000	.044	.006	.084	.007	.030	-.067	.196	.013
.025	.008	-.055	.036	-.054	.058	-.039	-.008	-.123	.128	-.008
.050	-.017	-.119	.027	-.100	.054	-.082	-.008	-.152	.149	-.025
.100	-.004	-.149	.053	-.117	.033	-.086	-.042	-.152	.098	.001
.200	.068	-.077	.133	-.045	.173	-.014	.120	-.089	.281	.056
.300	.110	.008	.210	.040	.263	.067	.217	-.004	.396	.124
.400	-.191	.008	.040	.082	.156	.114	.145	.064	.387	.175
.500	-.696	-.234	-.343	-.058	-.205	.024	-.191	-.008	.073	.166
.600										
.700	-.331	-.353	-.224	-.215	-.167	-.137	-.208	-.165	-.042	.039
.800										
.900	-.200	-.212	-.105	-.126	-.056	-.078	-.118	-.110	.030	.039
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.305	.165	.299	.142	.300	.139	.293	.124	.259	.128
.025	.474	.127	.383	.129	.394	.122	.365	.119	.331	.119
.050	.334	.097	.426	.129	.466	.118	.484	.140	.458	.145
.100	-.242	.076	.137	.159	.338	.198	.458	.234	.611	.221
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.051	-.051	-.003	-.049	.037	-.039	.064	-.029	.115	-.008
.025	.224	-.021	.099	-.011	.143	-.005	.132	-.008	.153	.001
.050	-.060	-.132	-.024	-.121	.016	-.086	.047	-.088	.111	-.025
.100	-.076	-.140	-.003	-.109	.045	-.082	.073	-.059	.149	-.004
.200	.004	-.051	.086	-.024	.135	-.001	.166	.022	.251	.056
.300	-.034	.034	.112	.074	.186	.096	.234	.111	.335	.136
.400	-.475	-.047	-.266	.078	-.090	.122	.022	.153	.200	.187
.500	-.606	-.331	-.261	-.138	-.175	-.044	-.114	.013	.009	.102
.600										
.700	-.153	-.225	-.117	-.130	-.078	-.073	-.042	-.038	-.016	.026
.800										
.900	-.055	-.072	-.007	-.032	.020	-.001	.043	.030	.026	.018
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(n) $M = 0.800$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.311	.119	.289	.105	.277	.093	.260	.080	.137	.002
.025	.352	.046	.326	.036	.309	.036	.301	.023	.165	-.035
.050	.442	-.040	.469	.048	.473	.048	.464	.015	.317	-.023
.100	.352	-.015	.559	.048	.645	.109	.685	.142	.656	.075
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.007	.005	.056	.056	.093	.027	.121	.027	.125	-.051
.025	.021	-.052	.044	-.042	.072	-.030	.088	.056	.063	-.047
.050	.034	-.122	.036	-.095	.068	-.075	.097	-.055	.075	-.014
.100	.050	-.146	.027	-.107	.044	-.079	.056	-.067	.026	-.039
.200	.082	-.077	.150	-.038	.187	-.009	.207	-.001	.206	-.027
.300	.131	.009	.228	.048	.277	.072	.305	.076	.321	.047
.400	-.146	.013	.052	.081	.162	.117	.223	.137	.288	.096
.500	-.795	-.248	-.385	-.062	-.234	.003	-.140	.064	-.039	.071
.600										
.700	-.350	-.379	-.242	-.230	-.177	-.153	-.165	-.108	-.145	-.076
.800										
.900	-.228	-.228	-.116	-.136	-.071	-.087	-.050	-.059	-.084	-.080
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.319	.156	.317	.154	.317	.137	.308	.121	.275	.120
.025	.583	.119	.460	.121	.488	.125	.435	.113	.405	.124
.050	.359	.094	.439	.117	.484	.113	.488	.117	.471	.136
.100	-.195	.066	.142	.166	.333	.195	.418	.206	.585	.214
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.048	-.056	.003	-.046	.035	-.038	.064	-.022	.112	-.006
.025	.384	-.024	.203	-.009	.260	-.005	.223	-.014	.234	-.002
.050	-.044	-.138	-.005	-.107	.035	-.091	.056	-.079	.120	-.027
.100	-.064	-.162	.003	-.107	.052	-.083	.080	-.075	.153	.002
.200	.009	-.052	.097	-.009	.146	.015	.158	.015	.242	.059
.300	-.028	.037	.109	.080	.190	.105	.215	.100	.320	.132
.400	-.561	-.048	-.303	.068	-.107	.113	-.018	.141	.161	.177
.500	-.679	-.366	-.279	-.152	-.201	-.062	-.144	-.018	-.031	.075
.600										
.700	-.170	-.252	-.119	-.144	-.079	-.087	-.075	-.071	-.043	-.010
.800										
.900	-.056	-.077	-.009	-.034	.019	-.005	.003	-.010	-.006	-.018
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(o) $M = 0.825$; $p_{t,e}/p_{t,\infty} = 1.0$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.330	.126	.302	.101	.279	.106	.277	.095	.158	.020
.025	.354	.047	.337	.038	.334	.043	.316	.052	.194	-.011
.050	.464	-.040	.479	-.017	.488	.035	.485	.024	.347	-.035
.100	.397	-.024	.558	.061	.638	.094	.698	.119	.667	.076
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.016	.016	.065	.022	.102	.023	.147	.048	.146	-.031
.025	.035	-.047	.057	-.033	.078	-.028	.107	.040	.079	-.047
.050	.043	-.114	.042	-.092	.078	-.075	.115	-.047	.103	-.039
.100	.019	-.138	.034	-.100	.051	-.083	.080	-.039	.052	-.023
.200	.102	-.063	.160	-.033	.189	-.008	.233	.020	.225	-.003
.300	.153	.019	.239	.053	.275	.071	.320	.103	.332	.068
.400	-.110	.016	.053	.077	.149	.110	.229	.147	.288	.103
.500	-.779	-.244	-.451	-.092	-.284	-.008	-.153	.064	-.042	.068
.600										
.700	-.366	-.409	-.270	-.262	-.213	-.193	-.157	-.125	-.157	-.086
.800										
.900	-.272	-.268	-.155	-.179	-.114	-.126	-.062	-.090	-.094	-.090
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.337	.165	.329	.163	.318	.149	.323	.146	.292	.150
.025	.722	.114	.561	.120	.597	.129	.528	.130	.520	.138
.050	.385	.086	.451	.120	.479	.118	.508	.130	.492	.146
.100	-.157	.047	.132	.160	.283	.173	.414	.201	.567	.213
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.040	-.063	.006	-.041	.031	-.044	.068	-.023	.119	.008
.025	.597	-.028	.333	-.010	.400	-.016	.339	-.003	.358	.005
.050	-.024	-.142	.006	-.108	.035	-.099	.071	-.070	.138	-.011
.100	-.059	-.153	.010	-.104	.047	-.091	.087	-.062	.162	.001
.200	.027	-.047	.104	-.006	.141	.004	.186	.036	.252	.064
.300	-.008	.043	.112	.085	.169	.098	.229	.123	.319	.138
.400	-.581	-.063	-.368	.057	-.201	.098	-.058	.150	.119	.178
.500	-.632	-.444	-.368	-.183	-.252	-.106	-.180	-.046	-.058	.056
.600										
.700	-.275	-.287	-.139	-.163	-.118	-.130	-.094	-.102	-.062	-.027
.800										
.900	-.051	-.091	-.037	-.065	-.036	-.051	-.023	-.062	-.031	-.043
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(p) $M = 0.700$; windmilling

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.274	.118	.236	.081	.224	.073	.104	-.023	.131	.039
.025	.283	.060	.275	.032	.258	.029	.152	-.061	.141	.068
.050	.371	-.018	.411	.057	.414	.059	.298	-.061	.296	.092
.100	.210	.016	.494	.100	.589	.127	.546	.060	.690	.175
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.047	-.008	.008	.086	.049	-.014	-.003	-.105	.180	-.073
.025	-.013	-.066	.008	-.070	.029	-.053	-.047	-.149	.058	-.078
.050	-.042	-.125	-.002	-.109	.029	-.092	-.042	-.168	.102	-.068
.100	-.018	-.154	-.007	-.123	.054	-.107	-.076	-.159	.058	-.019
.200	.031	-.081	.095	-.055	.132	-.034	.065	-.110	.238	.039
.300	.050	-.008	.168	.022	.214	.044	.162	-.037	.355	.097
.400	-.212	-.008	.027	.057	.136	.088	.109	.011	.369	.151
.500	-.518	-.237	-.308	-.070	-.185	.020	-.183	-.018	.112	.170
.600										
.700	-.256	-.275	-.177	-.162	-.136	-.082	-.183	-.115	-.005	.097
.800										
.900	-.139	-.144	-.070	-.080	-.034	-.039	-.095	-.095	.058	.078
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.263	.152	.265	.129	.262	.117	.249	.113	.179	.092
.025	.278	.128	.216	.110	.233	.112	.200	.113	.160	.107
.050	.278	.098	.391	.119	.432	.122	.438	.142	.373	.131
.100	-.352	.069	.124	.178	.335	.214	.448	.234	.635	.213
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.071	-.052	-.007	-.041	.029	-.029	.055	-.027	.078	-.034
.025	-.047	-.028	-.094	-.016	-.043	-.009	-.047	-.013	-.024	-.010
.050	-.095	-.125	-.045	-.094	-.005	-.087	.016	-.061	.063	-.034
.100	-.095	-.139	-.021	-.104	.024	-.077	.055	-.052	.131	-.010
.200	-.028	-.052	.056	-.026	.102	-.009	.137	.007	.223	.044
.300	-.057	.016	.085	.056	.151	.073	.200	.084	.315	.112
.400	-.425	-.052	-.176	.061	-.048	.112	.045	.142	.233	.189
.500	-.435	-.299	-.225	-.123	-.140	-.034	-.081	.021	.058	.141
.600										
.700	-.125	-.163	-.075	-.084	-.039	-.034	-.013	.011	.039	.102
.800										
.900	-.037	-.052	.008	-.007	.034	.020	.050	.041	.097	.097
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(q) $M = 0.750$; windmilling

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.288	.120	.266	.089	.247	.092	.134	-.017	.164	.076
.025	.310	.054	.310	.045	.295	.039	.169	-.061	.191	.053
.050	.395	-.030	.439	.058	.442	.047	.333	-.052	.355	.089
.100	.266	-.004	.514	.080	.605	.114	.568	.045	.731	.173
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.039	-.008	.036	.000	.070	.043	.023	-.088	.200	-.071
.025	-.004	-.070	.031	-.049	.052	-.041	-.013	-.128	.076	-.066
.050	-.035	-.128	.062	-.102	.047	-.086	-.008	-.154	.124	-.057
.100	-.017	-.159	.053	-.106	.025	-.077	-.044	-.154	.031	-.004
.200	.040	-.088	.115	-.044	.149	-.019	.094	-.097	.271	.067
.300	.085	-.004	.186	.031	.234	.052	.187	-.021	.381	.115
.400	-.216	-.008	.031	.058	.132	.096	.120	.027	.368	.160
.500	-.606	-.265	-.337	-.088	-.210	.047	-.199	-.030	.067	.169
.600										
.700	-.278	-.305	-.190	-.177	-.134	-.103	-.190	-.132	-.018	.098
.800										
.900	-.172	-.168	-.084	-.088	-.037	-.046	-.101	-.097	.062	.067
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.288	.151	.288	.137	.286	.131	.266	.120	.221	.115
.025	.394	.115	.323	.115	.335	.118	.306	.102	.265	.115
.050	.310	.102	.407	.120	.454	.131	.456	.129	.407	.137
.100	-.291	.076	.111	.154	.322	.202	.443	.204	.619	.199
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.066	-.053	.000	-.040	.038	-.024	.058	-.030	.080	-.044
.025	.107	-.030	.031	-.013	.074	-.006	.071	-.013	.084	.000
.050	-.083	-.128	-.026	-.102	.015	-.081	.041	-.061	.093	-.035
.100	-.088	-.141	-.013	-.102	.030	-.077	.071	-.052	.151	.000
.200	-.017	-.053	.062	-.022	.114	.003	.151	.010	.243	.066
.300	-.057	.018	.089	.062	.171	.087	.209	.085	.336	.128
.400	-.441	-.053	-.221	.053	-.069	.114	.018	.133	.217	.181
.500	-.556	-.326	-.248	-.141	-.161	-.050	-.097	.001	.031	.128
.600										
.700	-.119	-.176	-.075	-.093	-.037	-.041	-.017	-.004	.027	.093
.800										
.900	-.044	-.053	.005	-.009	.038	.021	.045	.036	.080	.075
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(r) $M = 0.775$; windmilling

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.298	.119	.276	.097	.255	.086	.244	.082	.104	-.002
.025	.315	.051	.327	.051	.298	.026	.308	.044	.142	-.045
.050	.412	-.017	.455	.046	.452	.043	.448	.031	.300	-.002
.100	.285	-.017	.527	.055	.609	.082	.665	.133	.653	.070
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.030	.000	.046	.051	.082	.005	.129	.010	.125	-.159
.025	.009	-.064	.042	-.039	.060	-.038	.087	-.028	-.002	-.172
.050	-.021	-.123	.034	-.094	.056	-.080	.095	-.071	.049	-.147
.100	-.008	-.157	.025	-.094	.035	-.076	.065	-.054	-.049	-.083
.200	.055	-.076	.127	-.039	.154	-.021	.201	.010	.198	-.028
.300	.098	.000	.199	.038	.239	.052	.287	.078	.304	.032
.400	-.204	-.004	.034	.072	.133	.094	.219	.125	.279	.070
.500	-.727	-.268	-.366	-.081	-.229	.073	-.113	.065	-.045	.062
.600										
.700	-.285	-.314	-.200	-.183	-.148	-.114	-.105	-.041	-.117	-.011
.800										
.900	-.178	-.178	-.086	-.098	-.046	-.059	-.007	-.007	-.045	-.036
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.301	.166	.297	.140	.298	.128	.294	.133	.231	.117
.025	.492	.127	.386	.118	.404	.120	.371	.125	.337	.108
.050	.335	.110	.415	.123	.459	.124	.477	.133	.422	.134
.100	-.254	.064	.114	.161	.311	.192	.422	.214	.596	.189
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.064	-.055	-.000	-.039	.043	-.033	.065	-.015	.066	-.053
.025	.250	-.025	.106	-.013	.153	-.008	.142	-.007	.155	-.002
.050	-.068	-.131	-.022	-.098	.022	-.080	.057	-.066	.100	-.040
.100	-.085	-.153	-.009	-.102	.047	-.076	.082	-.058	.155	-.006
.200	-.008	-.055	.076	-.022	.120	.001	.159	.019	.240	.062
.300	-.042	.030	.093	.067	.152	.090	.214	.095	.325	.121
.400	-.450	-.068	-.264	.059	-.038	.107	.014	.142	.180	.168
.500	-.602	-.356	-.268	-.158	-.178	-.063	-.100	.002	-.006	.095
.600										
.700	-.114	-.178	-.081	-.098	-.042	-.042	-.015	.010	.002	.070
.800										
.900	-.034	-.059	.004	-.013	.035	.018	.057	.048	.040	.040
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(s) $M = 0.800$; windmilling

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.313	.121	.281	.105	.272	.088	.258	.103	.122	-.001
.025	.337	.035	.330	.036	.317	.035	.291	.038	.163	-.030
.050	.435	-.035	.457	-.021	.464	.019	.459	.042	.330	-.017
.100	.337	-.027	.527	.056	.611	.080	.671	.132	.665	.064
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.014	.006	.048	.052	.096	.019	.136	.025	.146	-.169
.025	.010	-.055	.044	-.046	.075	-.035	.107	-.020	.011	-.169
.050	-.014	-.133	.036	-.091	.068	-.079	.111	-.061	.064	-.156
.100	-.002	-.145	.028	-.111	.047	-.075	.079	-.040	-.058	-.079
.200	.072	-.076	.130	-.038	.165	-.014	.209	.013	.212	-.009
.300	.116	.002	.204	.036	.248	.064	.291	.083	.314	.040
.400	-.165	-.002	.028	.061	.121	.100	.209	.124	.273	.073
.500	-.819	-.264	-.373	-.103	-.263	-.018	-.159	.046	-.054	.048
.600										
.700	-.296	-.337	-.209	-.209	-.165	-.133	-.118	-.061	-.132	-.034
.800										
.900	-.198	-.198	-.111	-.115	-.059	-.075	-.024	-.032	-.066	-.050
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.320	.165	.313	.150	.308	.133	.315	.144	.260	.129
.025	.610	.120	.468	.126	.500	.121	.454	.131	.423	.121
.050	.353	.088	.432	.130	.467	.129	.490	.148	.444	.134
.100	-.218	.071	.117	.150	.292	.178	.413	.201	.562	.178
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.059	-.063	-.005	-.050	.043	-.026	.078	-.007	.076	-.046
.025	.426	-.027	.219	-.013	.280	-.014	.246	-.003	.248	-.001
.050	-.051	-.133	-.009	-.107	.039	-.079	.074	-.064	.109	-.042
.100	-.080	-.149	-.001	-.103	.047	-.079	.091	-.048	.154	-.001
.200	.002	-.047	.085	-.009	.121	-.002	.160	.029	.252	.068
.300	-.039	.031	.097	.073	.153	.084	.205	.103	.317	.125
.400	-.495	-.063	-.290	.052	-.153	.096	-.024	.131	.162	.150
.500	-.638	-.402	-.290	-.172	-.210	-.087	-.134	-.024	-.034	.060
.600										
.700	-.145	-.194	-.078	-.107	-.051	-.055	-.032	-.020	-.017	.040
.800										
.900	-.035	-.067	-.001	-.013	.023	.002	.029	.013	.011	.007
1.000										

TABLE 19.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Concluded

(t) $M = 0.825$; windmilling

x/c	C _p at -									
	α = -2°	α = 0°	α = 1°	α = 2°	α = 4°					
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.324	.131	.299	.102	.285	.105	.281	.087	.230	.072
.025	.364	.048	.346	.043	.353	.042	.344	.032	.262	.041
.050	.458	-.042	.473	-.036	.483	.038	.474	.028	.427	.041
.100	.367	-.023	.536	.035	.629	.089	.667	.099	.743	.124
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.001	.013	.063	.019	.109	.026	.158	.028	.234	-.101
.025	.029	-.050	.059	-.040	.093	-.025	.115	-.015	.088	-.097
.050	.040	-.109	.047	-.087	.093	-.056	.123	-.035	.151	-.077
.100	.052	-.133	.035	-.099	.066	-.068	.084	.043	.005	-.010
.200	.092	-.066	.145	-.040	.188	-.001	.221	.024	.297	.069
.300	.147	.013	.216	.039	.263	.066	.300	.084	.396	.124
.400	-.121	.013	.027	.063	.125	.101	.194	.119	.352	.147
.500	-.787	-.259	-.462	-.111	-.301	-.041	-.185	.020	-.010	.112
.600										
.700	-.306	-.342	-.237	-.233	-.187	-.155	-.145	-.094	-.085	.021
.800										
.900	-.228	-.224	-.131	-.139	-.088	-.108	-.051	-.066	-.014	.033
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.335	.170	.330	.157	.325	.148	.323	.138	.281	.127
.025	.764	.127	.578	.121	.628	.128	.567	.127	.540	.112
.050	.374	.091	.448	.121	.475	.125	.496	.134	.462	.123
.100	-.192	.052	.114	.141	.266	.168	.366	.197	.525	.147
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.046	-.062	.003	-.048	.034	-.037	.071	-.007	.061	-.081
.025	.658	-.023	.361	-.012	.439	-.009	.386	-.007	.383	-.010
.050	-.027	-.141	.003	-.115	.042	-.084	.087	-.062	.116	-.054
.100	-.062	-.145	.011	-.107	.046	-.076	.091	-.043	.155	-.014
.200	.013	-.038	.094	-.008	.135	.014	.158	.024	.249	.064
.300	-.023	.040	.098	.078	.164	.101	.189	.099	.300	.120
.400	-.557	-.058	-.359	.047	-.210	.097	-.086	.115	.116	.143
.500	-.620	-.436	-.359	-.209	-.261	-.127	-.192	-.062	-.069	.053
.600										
.700	-.231	-.204	-.091	-.119	-.075	-.092	-.062	-.058	-.046	.002
.800										
.900	-.019	-.074	-.024	-.036	-.021	-.045	-.019	-.039	-.030	-.030
1.000										

TABLE 20.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a

(a) $M = 0.700$

x/c	C _p at --									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.304	.104	.274	.070	.239	.050	.213	.023		
.025	.304	.036	.289	.061	.278	-.013	.252	-.020		
.050	.493	-.017	.337	.095	.293	-.013	.252	-.011		
.100	.134	-.036	.474	.061	.613	.089	.693	.116		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.051	-.022	.056	-.027	.050	-.037	.052	-.035		
.025	.036	-.066	.051	-.071	.050	-.066	.062	-.064		
.050	.095	-.114	.036	-.110	.050	-.110	.062	-.103		
.100	-.046	-.139	.070	-.110	.050	-.105	.077	-.093		
.200	.061	-.105	.138	-.076	.176	-.056	.198	-.045		
.300	.075	-.017	.187	.012	.244	.026	.286	.043		
.400	-.250	.012	.022	.090	.147	.128	.237	.154		
.500	-.513	-.250	-.270	-.076	-.149	.011	-.059	.082		
.600										
.700	-.406	-.294	-.348	-.168	-.241	-.095	-.200	-.050		
.800										
.900	-.168	-.187	-.090	-.105	-.056	-.076	-.020	-.035		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.259	.148	.264	.123	.253	.113	.241	.096		
.025	.269	.119	.303	.099	.311	.074	.304	.081		
.050	.613	.075	.395	.089	.326	.098	.304	.096		
.100	-.308	.065	.162	.157	.389	.200	.537	.236		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.022	-.027	-.003	-.032	-.003	-.037	.004	-.035		
.025	-.046	-.003	-.017	.007	-.008	.011	.004	.018		
.050	-.080	-.109	-.027	-.095	-.008	-.090	.009	-.083		
.100	-.095	-.129	-.027	-.100	.007	-.100	.033	-.083		
.200	-.022	-.056	.070	-.027	.118	-.013	.164	-.001		
.300	-.066	.012	.099	.056	.181	.069	.246	.086		
.400	-.497	-.046	-.187	.065	-.052	.113	.062	.159		
.500	-.453	-.284	-.235	-.104	-.153	-.023	-.083	.043		
.600										
.700	-.162	-.196	-.095	-.104	-.071	-.061	-.045	-.025		
.800										
.900	-.046	-.070	-.003	-.027	.016	-.008	.038	.014		
1.000										

TABLE 20.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.321	.117	.295	.078	.264	.047	.246	.038		
.025	.330	.033	.344	.002	.313	.060	.299	.082		
.050	.649	-.011	.446	-.015	.388	.082	.361	.051		
.100	.188	-.033	.499	.029	.609	.082	.702	.104		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.060	-.016	.060	-.024	.060	-.024	.069	-.020		
.025	.046	-.064	.060	-.068	.069	-.068	.082	-.051		
.050	.051	-.118	.042	-.113	.060	-.099	.082	-.086		
.100	-.025	-.135	.042	-.113	.065	-.095	.095	-.082		
.200	.082	-.100	.153	-.073	.189	-.055	.219	-.038		
.300	.099	-.011	.197	.020	.251	.038	.299	.060		
.400	-.237	.020	.011	.082	.144	.131	.242	.157		
.500	-.636	-.273	-.312	-.095	-.183	.003	-.082	.064		
.600										
.700	-.175	-.326	-.219	-.184	-.144	-.117	-.126	-.051		
.800										
.900	-.188	-.215	-.113	-.130	-.068	-.086	-.029	-.046		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.280	.152	.281	.131	.272	.117	.263	.104		
.025	.294	.121	.321	.109	.330	.104	.329	.100		
.050	.771	.077	.520	.095	.431	.100	.400	.091		
.100	-.232	.059	.179	.144	.369	.188	.524	.214		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.020	-.029	-.002	-.037	.003	-.037	.011	-.033		
.025	-.042	-.002	-.015	-.007	-.002	.016	.011	.016		
.050	-.078	-.117	-.029	-.099	-.002	-.095	.020	-.086		
.100	-.082	-.135	-.024	-.104	.020	-.090	.047	-.077		
.200	-.007	-.051	.086	-.024	.131	-.006	.175	.002		
.300	-.051	.020	.104	.055	.188	.082	.245	.095		
.400	-.568	-.064	-.250	.060	-.081	.126	.038	.157		
.500	-.599	-.321	-.276	-.130	-.183	-.037	-.104	.029		
.600										
.700	-.161	-.210	-.113	-.121	-.073	-.068	-.046	-.029		
.800										
.900	-.051	-.078	-.011	-.037	.016	-.015	.038	.011		
1.000										

TABLE 20.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.333	.103	.302	.089	.276	.059	.267	.054		
.025	.367	.031	.340	-.004	.323	.047	.314	.007		
.050	.834	-.024	.544	-.038	.471	.076	.488	.012		
.100	.252	-.058	.510	.038	.629	.072	.705	.084		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.069	-.011	.072	-.013	.076	-.013	.088	-.005		
.025	.052	-.067	.072	-.055	.076	-.051	.097	-.048		
.050	.061	-.113	.055	-.098	.072	-.089	.097	-.078		
.100	.078	-.134	.047	-.106	.081	-.085	.114	-.073		
.200	.095	-.096	.161	-.064	.200	-.042	.233	-.022		
.300	.120	-.003	.208	.021	.259	.042	.309	.067		
.400	-.207	.027	.034	.089	.157	.132	.237	.173		
.500	-.729	-.300	-.348	-.093	-.200	-.004	-.086	.063		
.600										
.700	.120	-.338	-.093	-.195	-.042	-.119	-.039	-.061		
.800										
.900	-.211	-.228	-.119	-.140	-.072	-.093	-.031	-.056		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.298	.150	.293	.131	.288	.127	.283	.109		
.025	.315	.116	.339	.106	.343	.102	.351	.109		
.050	.938	.061	.615	.097	.504	.093	.504	.084		
.100	-.160	.044	.187	.144	.381	.182	.517	.203		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.024	-.033	-.000	-.034	.008	-.038	.029	-.031		
.025	-.041	-.003	-.009	.008	.008	.017	.029	.020		
.050	-.066	-.126	-.021	-.115	.008	-.089	.037	-.086		
.100	-.075	-.138	-.004	-.106	.030	-.089	.067	-.082		
.200	.006	-.058	.093	-.025	.140	-.000	.181	.003		
.300	-.037	.018	.114	.064	.199	.085	.253	.101		
.400	-.609	-.049	-.280	.064	-.106	.123	.012	.160		
.500	-.693	-.350	-.288	-.136	-.195	-.047	-.116	.016		
.600										
.700	-.160	-.223	-.110	-.123	-.072	-.068	-.044	-.035		
.800										
.900	-.054	-.083	-.013	-.042	.013	-.013	.037	.012		
1.000										

TABLE 20.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Concluded

(d) $M = 0.800$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.342	.110	.314	.081	.294	.066	.273	.044		
.025	.363	.024	.351	.016	.352	.074	.334	.077		
.050	1.081	-.054	.686	-.029	.613	.074	.607	-.013		
.100	.305	-.062	.522	.028	.621	.049	.693	.097		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.073	-.009	.081	-.009	.082	.074	.093	.081		
.025	.061	-.062	.081	-.054	.082	-.053	.101	-.041		
.050	.024	-.123	.065	-.095	.082	-.089	.101	-.066		
.100	-.037	-.135	.052	-.107	.086	-.089	.114	-.066		
.200	.106	-.094	.171	-.058	.205	-.040	.236	-.025		
.300	.138	-.009	.216	.024	.262	.049	.310	.073		
.400	-.143	.028	.040	.097	.135	.123	.240	.163		
.500	-.759	-.286	-.356	-.107	-.241	-.020	-.123	.052		
.600										
.700	.591	-.372	.118	-.217	.131	-.143	.118	-.082		
.800										
.900	-.258	-.278	-.140	-.172	-.089	-.118	-.054	-.082		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.313	.154	.301	.134	.294	.127	.288	.117		
.025	.329	.101	.346	.101	.355	.102	.354	.097		
.050	1.147	.048	.737	.089	.603	.094	.594	.089		
.100	-.119	.044	.195	.121	.355	.167	.492	.170		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.021	-.045	-.001	-.042	.008	-.032	.028	-.033		
.025	-.041	-.009	-.009	.003	.008	.008	.028	.012		
.050	-.074	-.135	-.017	-.107	.008	-.097	.036	-.086		
.100	-.066	-.147	-.001	-.111	.029	-.093	.064	-.078		
.200	.016	-.058	.105	-.025	.143	-.004	.183	.016		
.300	-.021	.020	.113	.068	.184	.082	.240	.101		
.400	-.603	-.066	-.352	.052	-.163	.106	-.025	.138		
.500	-.729	-.420	-.323	-.168	-.236	-.077	-.159	-.021		
.600										
.700	-.204	-.241	-.119	-.135	-.089	-.089	-.078	-.062		
.800										
.900	-.066	-.090	-.025	-.054	.000	-.032	.003	-.033		
1.000										

TABLE 21.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2

(a) $M = 0.700$

x/c	C _p at -										
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°		
Inboard station											
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.265	.051	.258	.035	.241	.003	.118	-.081	.146	.020	
	.114	.051	.094	.079	.090	-.011	.002	-.018	.073	.035	
	.386	.046	.442	.127	.454	.144	.342	.079	.379	.229	
	.177	.061	.491	.040	.604	.047	.594	-.047	.772	.083	
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.337	-.109	.341	-.110	.328	-.089	.201	-.173	.229	-.048	
	.022	-.065	.031	-.076	.042	-.070	-.047	-.173	.069	-.053	
	.056	.100	.031	.069	.047	.052	-.032	-.057	.098	.020	
	.042	-.099	.035	-.105	.066	-.089	-.013	-.183	.141	-.057	
	.071	-.055	.137	-.042	.183	-.021	.118	-.100	.277	.030	
	.075	-.007	.195	.026	.265	.047	.215	-.037	.403	.112	
	-.220	-.007	.035	.060	.163	.110	.152	.045	.423	.214	
	-.084	-.244	-.037	-.071	.018	.023	-.061	-.003	.112	.214	
	-.258	-.288	-.202	-.168	-.142	-.099	-.152	-.129	.054	.069	
	-.186	-.191	-.110	-.124	-.050	-.060	-.115	-.125	.049	.044	
	Outboard station										
	.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
		.249	.206	.248	-.013	.245	.022	.239	-.023	.199	.010
		.254	.133	.287	.108	.293	.110	.292	.099	.257	.097
.278		.346	.369	.122	.400	.129	.418	.055	.402	.078	
-.147		.713	.253	.316	.434	.269	.564	.118	.722	.131	
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.201	-.041	.190	-.061	.376	-.045	.404	-.047	.538	-.038	
	-.046	-.021	-.032	-.023	-.011	-.016	.002	-.008	.030	.006	
	-.070	-.089	-.047	-.090	-.021	-.079	-.003	-.071	.044	-.053	
	-.065	-.118	-.023	-.100	.013	-.089	.040	-.076	.097	-.048	
	-.002	-.036	.074	-.023	.124	-.002	.157	.002	.238	.044	
	-.065	.017	.098	.050	.182	.081	.249	.099	.368	.141	
	-.524	-.055	-.197	.055	-.050	.119	.069	.162	.272	.243	
	-.437	-.292	-.269	-.124	-.171	-.031	-.090	.040	.064	.160	
	-.157	-.176	-.105	-.100	-.060	-.050	-.032	-.003	.044	.078	
	-.055	-.084	-.018	-.047	.018	-.011	.036	.002	.093	.054	

TABLE 21.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(b) $M = 0.750$

x/c	C_p at -									
	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 1^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$					
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.295	.060	.276	.033	.271	.036	.151	-.066	.189	.039
.025	.122	.056	.099	.041	.111	.045	.018	-.053	.110	.013
.050	.418	.039	.466	.103	.484	.142	.385	.062	.428	.220
.100	.259	.038	.524	.037	.643	.054	.602	-.027	.786	.105
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.361	-.107	.364	-.091	.359	-.066	.239	-.155	.265	-.027
.025	.034	-.059	.046	-.065	.067	-.061	-.022	-.146	.101	-.045
.050	.021	.113	.046	.081	.072	.072	-.009	-.044	.119	.021
.100	.047	-.094	.050	-.087	.089	-.079	.022	-.164	.167	-.049
.200	.067	-.046	.165	-.021	.209	.001	.142	-.080	.304	.057
.300	.100	.003	.214	.037	.289	.067	.235	-.009	.419	.119
.400	-.191	-.001	.050	.077	.169	.125	.164	.066	.402	.216
.500	-.065	-.262	-.025	-.078	.023	.027	-.049	-.013	.114	.203
.600										
.700	-.311	-.311	-.211	-.180	-.145	-.097	-.190	-.146	-.014	.048
.800										
.900	-.209	-.218	-.122	-.140	-.057	-.070	-.119	-.128	.039	.030
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.267	.387	.266	.121	.275	.151	.265	.075	.229	.118
.025	.285	.140	.306	.121	.319	.129	.318	.110	.299	.114
.050	.303	.497	.390	.244	.434	.244	.437	.159	.436	.193
.100	-.068	.867	.262	.461	.474	.407	.557	.243	.714	.268
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.355	-.050	.288	-.056	.460	-.035	.468	-.044	.604	-.023
.025	-.041	-.023	-.016	-.012	.005	-.004	.013	-.009	.052	.008
.050	-.076	-.094	-.034	-.087	-.004	-.070	.013	-.071	.070	-.049
.100	-.059	-.125	-.007	-.100	.032	-.079	.053	-.071	.123	-.040
.200	.003	-.037	.094	-.012	.147	.005	.172	.013	.255	.048
.300	-.050	.029	.103	.072	.200	.093	.252	.110	.374	.154
.400	-.610	-.059	-.241	.059	-.070	.129	.049	.163	.233	.238
.500	-.535	-.341	-.294	-.135	-.198	-.039	-.124	.022	.021	.132
.600										
.700	-.160	-.187	-.100	-.096	-.061	-.039	-.035	-.009	.021	.052
.800										
.900	-.066	-.094	-.021	-.051	.018	-.013	.035	.000	.061	.012
1.000										

TABLE 21.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(c) $M = 0.775$

x/c	C _p at -										
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°		
Inboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010	.307	.052	.290	.044	.283	.032	.262	.037	.119	-.059
	.025	.137	.027	.116	.052	.121	.053	.118	.050	.051	-.072
	.050	.439	.056	.482	.103	.491	.130	.492	.148	.365	.136
	.100	.299	.035	.537	.052	.640	.058	.712	.046	.704	-.025
	.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010	.383	-.056	.380	-.075	.372	-.061	.356	-.048	.208	-.110
	.025	.048	-.054	.056	-.050	.075	-.053	.084	-.048	.022	-.114
	.050	.035	.124	.061	.086	.083	.070	.101	.050	.043	-.046
	.100	.027	-.084	.069	-.084	.100	-.061	.131	-.056	.094	-.122
	.200	.116	-.041	.180	-.007	.219	.007	.250	.025	.221	-.033
	.300	.124	.014	.239	.052	.296	.075	.339	.088	.340	.034
	.400	-.169	.001	.052	.086	.168	.130	.250	.165	.306	.123
	.500	-.084	-.258	-.020	-.084	.028	.024	.050	.080	.026	.094
	.600										
	.700	-.312	-.321	-.211	-.182	-.155	-.112	-.103	-.056	-.114	-.050
	.800										
	.900	-.220	-.224	-.122	-.143	-.070	-.087	-.031	-.039	-.072	-.076
	1.000										
Outboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010	.289	.552	.285	.213	.287	.244	.279	.152	.246	.182
	.025	.306	.145	.328	.133	.333	.130	.334	.113	.318	.102
	.050	.336	.641	.404	.336	.435	.342	.457	.232	.449	.263
	.100	-.033	.589	.298	.561	.460	.494	.562	.317	.703	.326
	.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010	.578	-.045	.353	-.046	.511	-.027	.503	-.035	.622	-.025
	.025	-.033	-.016	-.007	-.003	.011	-.002	.024	-.001	.051	.000
	.050	-.062	-.092	-.024	-.084	.007	-.070	.024	-.069	.064	-.050
	.100	-.050	-.117	.005	-.092	.041	-.070	.071	-.069	.119	-.042
	.200	.031	-.033	.103	-.003	.147	.015	.185	.024	.246	.047
	.300	-.024	.031	.128	.077	.206	.100	.266	.118	.360	.136
	.400	-.065	-.050	-.262	.073	-.078	.125	.020	.169	.195	.212
	.500	-.626	-.350	-.304	-.152	-.205	-.036	-.141	.020	-.029	.098
	.600										
	.700	-.151	-.181	-.092	-.096	-.061	-.044	-.031	-.005	-.016	.026
	.800										
	.900	-.062	-.088	-.020	-.050	.019	-.019	.041	.003	.000	-.038
	1.000										

TABLE 21.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.324	.659	.302	.037	.291	.030	.270	.041	.140	-.069
.025	.137	.030	.118	.053	.120	.018	.123	.021	.074	-.056
.050	.463	.655	.494	.106	.504	.103	.499	.164	.393	.115
.100	.333	.043	.538	.053	.643	.030	.695	.058	.711	.001
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.402	-.088	.392	-.081	.385	-.056	.364	-.040	.229	-.093
.025	.659	-.039	.065	-.045	.079	-.048	.090	-.036	.033	-.097
.050	.051	.116	.065	.082	.087	.063	.107	.062	.058	-.044
.100	.039	-.076	.078	-.077	.108	-.064	.139	-.048	.107	-.105
.200	.128	-.027	.188	-.004	.226	.009	.262	.029	.233	-.020
.300	.149	.030	.241	.049	.291	.063	.340	.094	.339	.050
.400	-.121	.022	.061	.094	.157	.124	.246	.160	.352	.135
.500	-.076	-.251	-.024	-.086	.034	-.011	.045	.062	.017	.078
.600										
.700	-.313	-.333	-.232	-.208	-.174	-.134	-.130	-.065	-.130	-.069
.800										
.900	-.239	-.239	-.143	-.167	-.093	-.101	-.061	-.061	-.089	-.024
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.303	.751	.293	.322	.291	.352	.294	.245	.261	.290
.025	.324	.132	.338	.122	.344	.115	.347	.127	.326	.103
.050	.356	.817	.419	.435	.446	.438	.461	.327	.465	.363
.100	-.002	1.089	.293	.651	.438	.568	.547	.392	.693	.392
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.772	-.039	.419	-.049	.572	-.040	.547	-.024	.673	-.020
.025	-.023	-.010	-.008	-.008	.009	-.011	.033	.000	.062	.005
.050	-.051	-.088	-.024	-.081	.009	-.072	.033	-.053	.074	-.036
.100	-.035	-.116	.004	-.089	.042	-.080	.082	-.053	.127	-.032
.200	.043	-.027	.106	-.008	.152	.009	.196	.033	.253	.050
.300	-.010	.043	.114	.077	.189	.095	.257	.127	.351	.151
.400	-.577	-.043	-.317	.069	-.141	.111	-.008	.164	.172	.217
.500	-.744	-.385	-.338	-.159	-.260	-.084	-.179	-.008	-.052	.074
.600										
.700	-.165	-.186	-.106	-.110	-.076	-.064	-.053	-.020	-.036	.001
.800										
.900	-.059	-.084	-.028	-.065	.001	-.031	.005	-.028	-.020	-.073
1.000										

TABLE 21.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Concluded

(e) $M = 0.825$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.331	.044	.310	.030	.312	.044	-.019	-.019	.169	-.055
.025	.146	.041	.133	.014	.146	.036	-.019	-.019	.091	-.043
.050	.480	.025	.503	.093	.521	.134	-.019	-.019	.413	.118
.100	.286	.033	.558	.038	.643	.044	-.019	-.019	.716	-.012
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.413	-.085	.397	-.072	.402	-.039	-.019	-.019	.264	-.043
.025	.068	-.046	.074	-.049	.099	-.027	-.019	-.019	.055	-.051
.050	.056	.115	.074	.085	.107	.091	-.019	-.019	.075	-.039
.100	.044	-.081	.081	-.080	.130	-.047	-.019	-.019	.122	-.059
.200	.150	-.026	.192	-.009	.249	.032	-.019	-.019	.248	-.008
.300	.174	.029	.251	.050	.312	.087	-.019	-.019	.346	.059
.400	-.089	.025	.058	.081	.174	.130	-.019	-.019	.370	.134
.500	-.085	-.250	-.041	-.100	.028	.000	-.019	-.019	.016	.083
.600										
.700	-.387	-.599	-.249	-.222	-.193	-.146	-.019	-.019	-.153	-.086
.800										
.900	-.332	-.344	-.167	-.194	-.114	-.118	-.019	-.019	-.118	-.134
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.314	.999	.309	.451	.315	.531	.307	.350	.294	.432
.025	.338	.134	.348	.117	.362	.130	.362	.130	.345	.118
.050	.381	1.026	.427	.549	.457	.606	.472	.421	.483	.491
.100	.040	1.167	.301	.722	.429	.641	.507	.440	.679	.443
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	1.007	-.050	.502	-.048	.681	-.019	.597	-.027	.726	-.016
.025	-.026	-.018	-.009	-.013	.024	.000	.036	-.007	.071	.000
.050	-.050	-.104	-.017	-.084	.020	-.067	.036	-.066	.082	-.039
.100	-.030	-.124	.014	-.092	.055	-.067	.079	-.058	.137	-.023
.200	.048	-.030	.109	-.009	.165	.020	.193	.036	.259	.059
.300	.009	.040	.117	.077	.201	.106	.248	.114	.345	.149
.400	-.558	-.046	-.324	.058	-.177	.106	-.066	.142	.130	.200
.500	-.707	-.437	-.335	-.190	-.295	-.098	-.247	-.050	-.106	.055
.600										
.700	-.312	-.234	-.119	-.119	-.086	-.075	-.086	-.066	-.063	-.016
.800										
.900	-.057	-.108	-.048	-.080	-.023	-.063	-.050	-.094	-.059	-.110
1.000										

TABLE 22.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3

(a) $M = 0.700$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.045	.137	-.088	.087	-.093	.063	-.087	.034		
.025	.253	.069	-.010	.048	-.049	.082	-.014	.015		
.050	.549	-.004	.135	.067	.043	.048	.063	.054		
.100	-.344	-.023	.184	.106	.417	.169	.554	.185		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.098	.025	.106	.009	.125	.004	.127	.005		
.025	.098	-.053	.087	-.073	.106	-.059	.117	-.063		
.050	.093	-.116	.063	-.117	.097	-.098	.122	-.092		
.100	.093	-.135	.082	-.180	.131	-.171	.161	-.150		
.200	.093	-.043	.169	-.025	.237	-.006	.287	.015		
.300	.098	.035	.121	.087	.233	.126	.301	.151		
.400	.074	-.125	-.247	.058	-.098	.087	.010	.146		
.500	.171	-.378	.193	-.175	.228	-.069	.224	.005		
.600										
.700	-.290	-.285	-.209	-.180	-.145	-.103	-.111	-.063		
.800										
.900	-.140	-.169	-.078	-.112	-.030	-.078	.000	-.048		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.267	.277	-.005	.043	-.054	-.001	-.058	.005		
.025	.320	.727	.386	.251	.415	.091	.427	.054		
.050	.199	.892	.420	.487	.508	.217	.572	.117		
.100	.485	.998	.396	.681	.455	.377	.490	.199		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.035	1.085	.038	.826	.053	.537	.063	.310		
.025	-.009	.015	.014	.014	.048	.019	.063	.020		
.050	.262	.277	.033	.019	.014	-.039	.039	-.038		
.100	-.062	.727	-.000	.246	.053	.067	.097	.020		
.200	-.052	-.004	.091	.019	.169	.043	.238	.054		
.300	-.406	.015	-.107	.101	.053	.140	.146	.170		
.400	-.595	-.241	-.319	-.063	-.195	.024	-.106	.083		
.500	-.416	-.333	-.295	-.174	-.209	-.078	-.150	-.019		
.600										
.700	-.144	-.169	-.092	-.092	-.054	-.044	-.029	-.019		
.800										
.900	-.009	-.028	.024	.004	.048	.019	.063	.034		
1.000										

TABLE 22.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	-.022	-.022	.028	.095	.007	.060	.015	.042		
.025	-.022	-.022	.174	.046	.095	.074	.143	.051		
.050	-.022	-.022	.329	.064	.193	.038	.209	.042		
.100	-.022	-.022	.223	.095	.432	.158	.549	.170		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.022	-.022	.121	.019	.131	.012	.130	.006		
.025	-.022	-.022	.108	-.056	.122	-.064	.134	-.055		
.050	-.022	-.022	.095	-.104	.113	-.090	.134	-.091		
.100	-.022	-.022	.117	-.078	.149	-.073	.183	-.038		
.200	-.022	-.022	.196	-.011	.255	.007	.302	.020		
.300	-.022	-.022	.152	.099	.242	.122	.315	.139		
.400	-.022	-.022	-.272	.024	-.130	.087	-.024	.134		
.500	-.022	-.022	.205	-.197	.229	-.086	.227	-.024		
.600										
.700	-.022	-.022	-.219	-.188	-.174	-.126	-.126	-.082		
.800										
.900	-.022	-.022	-.082	-.118	-.046	-.090	-.016	-.069		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.441	.450	.138	.165	.060	.109	.055	.125		
.025	.375	.895	.416	.412	.436	.224	.446	.196		
.050	.265	.974	.451	.632	.528	.352	.583	.257		
.100	.635	1.054	.482	.791	.524	.502	.565	.341		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.035	1.138	.050	.910	.064	.648	.072	.451		
.025	-.009	.009	.037	.019	.056	.020	.077	.020		
.050	.441	.450	.165	.152	.126	.082	.152	.086		
.100	-.057	.900	.028	.412	.073	.206	.116	.169		
.200	-.035	-.009	.112	.033	.188	.042	.248	.059		
.300	-.428	.018	-.113	.108	.029	.144	.125	.165		
.400	-.666	-.264	-.351	-.073	-.236	.007	-.143	.068		
.500	-.542	-.383	-.311	-.183	-.249	-.095	-.192	-.046		
.600										
.700	-.145	-.181	-.086	-.095	-.059	-.055	-.038	-.033		
.800										
.900	-.013	-.035	.024	.002	.047	.016	.068	.033		
1.000										

TABLE 22.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(c) $M = 0.775$

x/c	C_p at -									
	$\alpha = -2^{\circ}$		$\alpha = 0^{\circ}$		$\alpha = 1^{\circ}$		$\alpha = 2^{\circ}$		$\alpha = 4^{\circ}$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.378	.124	.117	.087	.081	.055	.091	.057		
.025	.577	.052	.304	.024	.204	.060	.256	.065		
.050	.890	-.016	.444	.024	.276	.051	.282	.040		
.100	-.186	-.029	.232	.062	.407	.136	.558	.154		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.102	.026	.125	.058	.136	.013	.141	.027		
.025	.090	-.050	.113	-.053	.127	-.051	.146	-.045		
.050	.047	-.114	.096	-.095	.123	-.089	.150	-.075		
.100	.052	.145	.121	-.002	.153	-.004	.197	.031		
.200	.107	-.037	.210	-.015	.253	.009	.311	.035		
.300	-.029	.052	.147	.100	.238	.127	.320	.158		
.400	-.652	-.135	-.286	.024	-.148	.068	-.016	.133		
.500	.174	-.419	.210	-.197	.229	-.110	.231	-.020		
.600										
.700	-.342	-.368	-.231	-.206	-.195	-.144	-.130	-.075		
.800										
.900	-.190	-.211	-.091	-.129	-.063	-.110	-.016	-.067		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.613	.618	.240	.252	.136	.169	.137	.188		
.025	.406	1.032	.430	.515	.453	.292	.472	.260		
.050	.305	1.015	.451	.718	.538	.419	.554	.315		
.100	.774	1.091	.549	.858	.559	.567	.607	.408		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.043	1.176	.053	.956	.068	.707	.086	.514		
.025	.001	.009	.036	.019	.059	.017	.090	.027		
.050	.613	.618	.257	.240	.195	.153	.222	.158		
.100	-.042	1.036	.036	.515	.076	.280	.133	.239		
.200	-.012	.601	.121	.023	.191	.043	.256	.078		
.300	-.388	.030	-.138	.112	.009	.140	.128	.183		
.400	-.638	-.287	-.388	-.091	-.275	-.012	-.143	.061		
.500	-.642	-.414	-.324	-.197	-.265	-.118	-.194	-.050		
.600										
.700	-.173	-.198	-.091	-.108	-.072	-.068	-.028	-.020		
.800										
.900	-.012	-.037	.019	-.002	.038	.013	.078	.044		
1.000										

TABLE 22.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.595	.093	.242	.095	.205	.078	.202	.055		
.025	.672	.008	.434	.030	.340	.066	.369	-.003		
.050	1.019	-.062	.536	.022	.348	.062	.336	.055		
.100	-.147	-.078	.242	.071	.405	.123	.545	.161		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.059	-.005	.128	.071	.144	.037	.149	.026		
.025	.061	-.086	.128	-.044	.139	-.044	.161	-.035		
.050	.024	-.151	.107	-.092	.135	-.081	.161	-.068		
.100	.028	.350	.136	.136	.176	.148	.206	.165		
.200	.089	-.074	.222	.001	.274	.013	.324	.038		
.300	-.021	.016	.169	.112	.258	.131	.324	.165		
.400	-.641	-.155	-.280	.005	-.171	.074	-.052	.136		
.500	.134	-.453	.209	-.231	.225	-.110	.218	-.035		
.600										
.700	-.384	-.420	-.260	-.235	-.204	-.151	-.150	-.105		
.800										
.900	-.245	-.265	-.117	-.154	-.069	-.110	-.039	-.096		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.825	.833	.360	.368	.253	.286	.234	.275		
.025	.435	1.167	.453	.624	.473	.392	.486	.323		
.050	.333	1.033	.474	.803	.547	.518	.557	.372		
.100	.947	1.143	.612	.925	.624	.669	.658	.466		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.044	1.224	.062	1.011	.078	.804	.095	.584		
.025	.004	.012	.042	.022	.074	.025	.095	.030		
.050	.833	.833	.372	.354	.302	.274	.303	.258		
.100	-.029	1.167	.042	.624	.082	.380	.132	.303		
.200	-.001	.068	.128	.038	.192	.053	.250	.067		
.300	-.342	.036	-.149	.115	.005	.147	.103	.169		
.400	-.603	-.297	-.418	-.100	-.322	-.020	-.198	.042		
.500	-.631	-.460	-.349	-.231	-.289	-.134	-.235	-.076		
.600										
.700	-.277	-.236	-.092	-.117	-.065	-.069	-.047	-.047		
.800										
.900	-.001	-.037	.014	-.007	.041	.013	.038	.006		
1.000										

TABLE 22.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	1.026	.128	.433	.095	.370	.075	.361	.064		
.025	.789	.045	.532	.028	.425	.032	.440	-.022		
.050	1.243	-.018	.618	.016	.385	-.007	.346	.028		
.100	-.034	-.034	.252	.067	.417	.107	.522	.142		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.128	.041	.142	.044	.150	.040	.150	.036		
.025	.120	-.034	.142	-.031	.154	-.027	.162	-.034		
.050	.088	-.105	.127	-.086	.150	-.082	.166	-.066		
.100	.096	.757	.158	.343	.181	.350	.209	.346		
.200	.163	.021	.245	.016	.283	.024	.322	.036		
.300	.053	.072	.186	.115	.256	.134	.310	.158		
.400	-.534	-.101	-.267	.016	-.164	.071	-.077	.115		
.500	.183	-.428	.209	-.232	.217	-.148	.201	-.085		
.600										
.700	-.428	-.617	-.267	-.271	-.254	-.207	-.210	-.171		
.800										
.900	-.235	-.255	-.141	-.180	-.121	-.160	-.089	-.160		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	1.094	1.090	.503	.507	.369	.396	.341	.364		
.025	.481	1.263	.480	.723	.486	.451	.493	.353		
.050	.367	1.020	.487	.872	.549	.584	.587	.403		
.100	1.149	1.200	.692	.974	.674	.733	.657	.509		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.056	1.267	.075	1.053	.079	.854	.091	.622		
.025	.021	.013	.056	.024	.075	.020	.091	.013		
.050	1.094	1.094	.507	.507	.400	.384	.392	.345		
.100	-.002	1.259	.059	.723	.099	.439	.122	.337		
.200	.033	.009	.142	.044	.189	.059	.235	.063		
.300	-.313	.045	-.141	.111	-.035	.142	.056	.157		
.400	-.537	-.293	-.679	-.117	-.367	-.062	-.304	.001		
.500	-.569	-.553	-.392	-.255	-.375	-.179	-.316	-.136		
.600										
.700	-.392	-.289	-.101	-.121	-.105	-.113	-.101	-.112		
.800										
.900	-.022	-.077	.001	-.023	-.027	-.046	-.038	-.085		
1.000										

TABLE 23.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4

(a) $M = 0.700$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.055	.114	-.046	-.080	-.073	.058	-.077	.059		
.025	.245	.065	.012	.041	.009	.077	-.009	.069		
.050	.541	.065	.153	.104	.131	.072	.094	.079		
.100	.177	.079	.457	-.017	.596	-.039	.683	-.038		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.079	.065	.090	-.003	.092	-.010	.098	-.009		
.025	.084	.079	.060	.017	-.005	.009	.001	.011		
.050	.045	-.081	.065	-.090	.077	-.083	.098	-.072		
.100	.079	-.134	.080	-.095	.106	-.093	.137	-.101		
.200	.084	-.066	.182	-.046	.228	-.030	.274	-.009		
.300	.084	.065	.191	.099	.286	.121	.342	.147		
.400	-.523	-.071	-.202	-.070	-.068	.135	.059	.191		
.500	-.620	-.358	-.305	-.153	-.243	-.059	-.126	.016		
.600										
.700	-.280	-.275	-.187	-.173	-.146	-.102	-.097	-.053		
.800										
.900	-.134	-.139	-.071	-.085	-.039	-.054	.001	-.023		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.259	.157	.249	.133	.246	.120	.244	.122		
.025	.268	.133	.292	.114	.255	.101	.302	.103		
.050	.283	.094	.370	.109	.406	.111	.429	.113		
.100	.535	.099	.390	.186	.387	.212	.375	.259		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.011	.346	.002	.177	.009	.159	.025	.157		
.025	-.023	-.003	-.032	-.003	-.102	-.005	.025	.011		
.050	-.032	-.086	.002	-.085	.019	-.083	.059	-.072		
.100	-.157	-.100	-.041	-.090	-.001	-.078	.030	-.062		
.200	.016	.002	.143	.017	.203	.038	.268	.059		
.300	-.347	.055	-.041	.143	.111	.174	.234	.205		
.400	-.634	-.221	-.308	-.041	-.185	.043	-.072	.113		
.500	-.367	-.318	-.264	-.148	-.199	-.078	-.130	-.004		
.600										
.700	-.134	-.153	-.085	-.090	-.059	-.049	-.019	-.004		
.800										
.900	-.003	-.050	.022	-.056	.038	-.034	.064	.011		
1.000										

TABLE 23.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.208	.115	.057	.084	.037	.064	.014	.063		
	.434	.053	.221	.039	.161	.064	.147	.063		
	.726	.097	.354	.057	.281	.051	.232	.068		
	.252	.172	.504	.075	.612	.059	.684	.041		
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.084	.017	.052	.097	.095	.086	.099	.090		
.025	.230	.040	.115	.075	.095	.064	.081	.068		
.050	.057	-.085	.084	-.076	.095	-.078	.121	-.065		
.100	.053	-.116	.101	-.089	.121	-.082	.156	-.101		
.200	.115	-.053	.203	-.032	.241	-.025	.298	-.003		
.300	.071	.066	.221	.110	.290	.126	.351	.147		
.400	-.634	-.067	-.240	.057	-.078	.130	.041	.187		
.500	-.510	-.381	-.218	-.160	-.171	-.082	-.092	.006		
.600										
.700	-.210	-.324	-.209	-.191	-.162	-.131	-.114	-.074		
.800										
.900	-.160	-.164	-.085	-.094	-.051	-.069	-.012	-.034		
1.000										
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.283	.168	.273	.136	.262	.126	.267	.121		
	.300	.128	.318	.119	.320	.099	.324	.103		
	.318	.088	.357	.110	.426	.103	.443	.129		
	.676	.097	.490	.172	.465	.192	.448	.240		
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.004	.512	.013	.287	.015	.258	.036	.236		
.025	-.022	-.000	-.001	-.001	-.011	-.003	.129	.006		
.050	-.027	-.098	.013	-.085	.037	-.086	.067	-.070		
.100	-.124	-.111	.066	-.085	.117	-.078	.125	-.052		
.200	.044	-.000	.158	.030	.214	.042	.280	.059		
.300	-.314	.075	-.040	.141	.103	.170	.218	.213		
.400	-.774	-.261	-.367	-.058	-.223	.028	-.109	.094		
.500	-.447	-.363	-.252	-.182	-.223	-.095	-.163	-.025		
.600										
.700	-.133	-.164	-.089	-.089	-.064	-.060	-.025	-.017		
.800										
.900	-.014	-.080	.017	-.040	.033	-.033	.059	.001		
1.000										

TABLE 23.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(c) $M = 0.775$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.380	.113	.150	.086	.169	.076	.097	.080		
.025	.563	.045	.341	.035	.348	.072	.267	.042		
.050	.881	-.002	.447	.052	.395	.004	.322	.037		
.100	.287	.308	.515	.163	.633	.186	.700	.127		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.083	.024	.095	.010	.101	.021	.110	.016		
.025	.276	.045	.205	.035	.225	.042	.165	.037		
.050	.066	-.082	.095	-.075	.118	-.060	.135	-.048		
.100	.062	-.112	.107	-.084	.148	-.056	.173	-.082		
.200	.134	-.048	.218	-.024	.271	.004	.309	.012		
.300	.056	.071	.231	.107	.314	.144	.365	.161		
.400	-.613	-.048	-.237	.065	-.077	.144	.033	.186		
.500	-.430	-.401	-.135	-.177	-.107	-.081	-.065	.012		
.600										
.700	-.237	-.345	-.224	-.203	-.162	-.124	-.116	-.073		
.800										
.900	-.176	-.184	-.101	-.109	-.052	-.064	-.009	-.039		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.295	.159	.281	.145	.279	.135	.283	.135		
.025	.312	.125	.328	.116	.339	.118	.338	.118		
.050	.341	.083	.400	.099	.436	.118	.457	.122		
.100	.799	.092	.556	.171	.559	.186	.504	.237		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.011	.663	.014	.370	.033	.372	.046	.304		
.025	-.014	-.002	-.007	-.003	.025	.008	.211	.008		
.050	-.027	-.099	.014	-.096	.050	-.077	.080	-.060		
.100	-.112	-.108	.133	-.092	.228	-.064	.194	-.048		
.200	.058	.003	.166	.027	.237	.059	.287	.071		
.300	-.294	.066	-.041	.137	.105	.190	.207	.215		
.400	-.772	-.277	-.410	-.079	-.234	.021	-.141	.092		
.500	-.628	-.383	-.304	-.154	-.234	-.102	-.171	-.026		
.600										
.700	-.120	-.167	-.088	-.096	-.051	-.047	-.026	-.018		
.800										
.900	-.019	-.082	.014	-.041	.046	-.022	.071	.016		
1.000										

TABLE 23.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.631	.125	.287	.099	.323	.082	.202	.084		
.025	.684	.051	.458	.033	.470	.037	.374	.039		
.050	1.044	-.006	.528	.054	.457	.057	.374	.027		
.100	.337	.500	.540	.299	.633	.318	.696	.215		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.088	.027	.095	.021	.102	.020	.113	.019		
.025	.586	.051	.336	.042	.355	.041	.255	.051		
.050	.080	-.071	.103	-.065	.122	-.061	.141	-.047		
.100	.076	-.120	.123	-.081	.147	-.061	.178	-.067		
.200	.157	-.047	.234	-.016	.278	-.000	.317	.019		
.300	.125	.080	.242	.115	.310	.143	.366	.166		
.400	-.557	-.043	-.248	.074	-.090	.135	.019	.174		
.500	-.267	-.386	.001	-.204	-.049	-.094	-.034	-.026		
.600										
.700	-.329	-.366	-.248	-.236	-.184	-.163	-.140	-.100		
.800										
.900	-.194	-.202	-.114	-.126	-.074	-.086	-.042	-.059		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.304	.169	.294	.152	.297	.134	.287	.129		
.025	.332	.124	.343	.119	.350	.118	.348	.120		
.050	.365	.084	.425	.103	.448	.118	.462	.112		
.100	.952	.084	.645	.152	.623	.187	.552	.210		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.018	.850	.025	.482	.037	.452	.051	.365		
.025	-.014	.002	-.003	.005	.069	.004	.291	.002		
.050	-.014	-.108	.029	-.085	.053	-.090	.084	-.067		
.100	-.088	-.112	.233	-.077	.322	-.069	.259	-.051		
.200	.071	.002	.184	.037	.236	.053	.283	.068		
.300	-.251	.079	-.036	.152	.085	.171	.177	.206		
.400	-.752	-.279	-.493	-.077	-.310	.004	-.193	.055		
.500	-.699	-.434	-.309	-.207	-.257	-.118	-.213	-.067		
.600										
.700	-.137	-.185	-.089	-.101	-.061	-.065	-.046	-.042		
.800										
.900	-.006	-.088	.013	-.036	.037	-.037	.031	-.042		
1.000										

TABLE 23.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	1.012	.125	.505	.095	.547	.110	.379	.088		
.025	.768	.062	.564	.028	.582	.043	.489	.045		
.050	1.229	.003	.592	.036	.488	.046	.419	.037		
.100	.377	.807	.545	.501	.641	.519	.702	.379		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.093	.042	.087	.020	.102	.035	.112	.041		
.025	.910	.066	.533	.048	.551	.054	.407	.060		
.050	.097	-.060	.115	-.066	.137	-.044	.155	-.026		
.100	.101	-.096	.135	-.086	.165	-.048	.194	-.065		
.200	.184	-.029	.241	-.011	.287	.011	.320	.033		
.300	.160	.093	.265	.115	.322	.145	.371	.167		
.400	-.510	-.033	-.236	.060	-.099	.129	-.010	.171		
.500	-.041	-.372	.225	-.228	.086	-.111	.053	-.050		
.600										
.700	-.411	-.490	-.283	-.311	-.229	-.198	-.191	-.144		
.800										
.900	-.230	-.222	-.165	-.165	-.111	-.119	-.073	-.105		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.525	.176	.307	.150	.310	.145	.308	.143		
.025	.353	.121	.351	.111	.365	.121	.367	.111		
.050	.388	.085	.429	.075	.463	.097	.469	.111		
.100	1.144	.085	.752	.142	.719	.176	.622	.202		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.034	1.056	.024	.018	.042	.585	.052	.465		
.025	.003	.007	.016	-.003	.137	-.001	.410	.001		
.050	-.001	-.108	.024	-.110	.058	-.079	.084	-.069		
.100	-.056	-.111	.351	-.098	.451	-.068	.363	-.046		
.200	.093	.011	.185	.032	.239	.054	.284	.068		
.300	-.198	.089	-.043	.138	.050	.180	.158	.202		
.400	-.666	-.269	-.672	-.113	-.378	-.024	-.273	.037		
.500	-.658	-.493	-.393	-.255	-.307	-.154	-.273	-.109		
.600										
.700	-.269	-.229	-.121	-.125	-.095	-.091	-.089	-.085		
.800										
.900	.011	-.068	-.035	-.039	-.017	-.048	-.030	-.065		
1.000										

TABLE 24.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5

(a) $M = 0.700$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.177	.037	.166	.001	.159	-.001	.141	-.019		
.025	.173	-.128	.258	-.096	.285	-.089	.321	-.082		
.050	-.022	.066	.341	.113	.494	.149	.563	.170		
.100	-.524	.124	-.149	.248	.071	.309	.209	.340		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.046	.003	.045	.006	.052	.008	.044	.006		
.025	.032	-.061	.045	-.067	.052	-.069	.054	-.067		
.050	-.007	-.128	.064	-.120	.037	-.118	.044	-.116		
.100	-.046	-.162	.049	-.149	.076	-.137	.044	-.130		
.200	-.031	-.138	.025	-.116	.062	-.103	.078	-.091		
.300	.066	-.051	.166	-.028	.217	-.016	.258	-.004		
.400	-.051	-.007	.137	.035	.227	.062	.292	.078		
.500	-.196	-.061	-.018	.079	.067	.130	.122	.170		
.600										
.700	-.342	-.323	-.217	-.159	-.152	-.084	-.111	-.033		
.800										
.900	-.148	-.172	-.057	-.086	-.016	-.040	.015	-.014		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.115	.056	.117	.035	.115	.023	.112	.010		
.025	.055	-.075	.180	-.057	.221	-.050	.248	-.057		
.050	-.278	.061	.132	.127	.323	.139	.431	.160		
.100	-.971	.148	-.328	.272	-.098	.328	.044	.344		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.002	-.012	.001	-.033	-.001	-.040	.001	-.048		
.025	-.051	.017	-.033	.030	-.021	.028	-.014	.030		
.050	-.164	-.148	-.062	-.140	-.055	-.137	-.038	-.135		
.100	.056	-.211	.078	-.251	.086	-.147	.093	-.145		
.200	-.060	-.128	.015	-.101	.052	-.089	.083	-.086		
.300	-.036	-.041	.088	-.009	.144	-.006	.189	.005		
.400	-.293	.017	-.023	.078	.100	.105	.185	.126		
.500	-.511	-.143	-.261	.020	-.142	.086	-.057	.156		
.600										
.700	-.196	-.225	-.110	-.110	-.079	-.064	-.043	-.019		
.800										
.900	-.060	-.089	.001	-.033	.023	-.011	.044	.005		
1.000										

TABLE 24.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(b) $M = 0.750$

x/c	C_p at -									
	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 1^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$					
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.154	.043	.182	.018	.185	.012	.175	.003		
.025	.211	-.130	.288	-.106	.327	-.085	.352	-.077		
.050	.030	.061	.355	.120	.499	.158	.573	.175		
.100	-.665	.145	-.141	.253	.035	.296	.202	.348		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.056	.008	.054	.009	.061	.021	.060	.020		
.025	.047	-.059	.049	-.057	.070	-.054	.073	-.055		
.050	.003	-.121	.027	-.128	.057	-.107	.069	-.108		
.100	-.028	-.156	.054	-.150	.043	-.120	.069	-.121		
.200	-.015	-.130	.045	-.110	.092	-.089	.113	-.082		
.300	.101	-.041	.182	-.021	.243	.004	.290	.007		
.400	-.032	.003	.147	.040	.247	.070	.308	.087		
.500	-.231	-.054	-.035	.085	.057	.154	.122	.180		
.600										
.700	-.382	-.364	-.252	-.181	-.174	-.094	-.126	-.037		
.800										
.900	-.174	-.192	-.075	-.106	-.019	-.054	.011	-.015		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.131	.061	.129	.040	.136	.034	.135	.033		
.025	.122	-.085	.190	-.061	.242	-.041	.276	-.033		
.050	-.218	.069	.133	.124	.313	.145	.413	.170		
.100	-1.137	.136	-.392	.270	-.164	.326	.016	.351		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.001	-.019	-.004	-.030	.003	-.036	.011	-.037		
.025	-.041	.021	-.030	.027	-.014	.034	-.006	.033		
.050	-.058	-.160	-.070	-.145	-.041	-.129	-.011	-.126		
.100	.052	-.191	.076	-.163	.092	-.147	.100	-.134		
.200	-.050	-.125	.018	-.105	.065	-.089	.104	-.077		
.300	-.006	-.032	.093	-.013	.162	.003	.214	.016		
.400	-.280	.030	-.035	.084	.096	.114	.188	.131		
.500	-.766	-.160	-.326	.018	-.182	.083	-.077	.144		
.600										
.700	-.200	-.235	-.127	-.123	-.080	-.067	-.046	-.020		
.800										
.900	-.072	-.094	-.008	-.044	.026	-.010	.047	.011		
1.000										

TABLE 24.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.206	.053	.194	.020	.195	.009	.184	-.007		
.025	.223	-.138	.300	-.107	.348	-.076	.358	-.079		
.050	.074	.074	.360	.126	.492	.157	.561	.167		
.100	-.648	.133	-.171	.241	.038	.297	.171	.320		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.061	.010	.063	.012	.068	.021	.065	.014		
.025	.053	-.045	.063	-.056	.076	-.047	.074	-.045		
.050	.061	-.117	.037	-.120	.059	-.106	.065	-.108		
.100	-.024	-.160	.020	-.145	.055	-.119	.065	-.121		
.200	.002	-.121	.050	-.103	.098	-.076	.112	-.079		
.300	.116	-.036	.194	-.018	.255	.004	.277	.002		
.400	.010	.006	.152	.042	.242	.076	.290	.087		
.500	-.228	-.049	-.039	.084	.047	.157	.104	.163		
.600										
.700	-.393	-.389	-.268	-.209	-.195	-.119	-.147	-.053		
.800										
.900	-.194	-.211	-.090	-.124	-.034	-.064	-.002	-.036		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.146	.065	.139	.041	.148	.038	.146	.023		
.025	.142	-.079	.211	-.060	.258	-.047	.285	-.041		
.050	-.176	.078	.109	.109	.301	.152	.404	.167		
.100	-1.056	.163	-.466	.270	-.182	.326	-.015	.344		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.006	-.015	-.001	-.030	.008	-.034	.010	-.045		
.025	-.036	.027	-.026	.029	-.008	.030	-.003	.027		
.050	-.056	-.155	-.060	-.153	-.034	-.135	-.015	-.134		
.100	.065	-.198	.075	-.102	.089	-.157	.091	-.146		
.200	-.041	-.125	.029	-.102	.076	-.089	.103	-.083		
.300	.014	-.028	.105	-.014	.169	.004	.205	.014		
.400	-.253	.040	-.039	.080	.085	.114	.158	.129		
.500	-.944	-.164	-.356	.003	-.220	.093	-.121	.133		
.600										
.700	-.198	-.240	-.132	-.132	-.085	-.076	-.058	-.041		
.800										
.900	-.070	-.096	-.014	-.047	.017	-.017	.040	.002		
1.000										

TABLE 24.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(d) $M = 0.800$

x/c	C_p at -										
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$		
Inboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010	.222	.055	.209	.029	.211	.019	.209	.034		
	.025	.247	-.129	.323	-.102	.346	-.083	.381	-.056		
	.050	.128	.075	.372	.123	.497	.162	.577	.181		
	.100	-.575	.145	-.183	.254	.023	.301	.152	.336		
	.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010	.071	.018	.070	.025	.081	.036	.087	.042		
	.025	.059	-.048	.070	-.049	.089	-.030	.099	-.040		
	.050	.059	-.113	.045	-.110	.077	-.099	.091	-.085		
	.100	-.015	-.154	.029	-.134	.064	-.115	.091	-.093		
	.200	.014	-.113	.066	-.090	.113	-.074	.136	-.052		
	.300	.132	-.027	.209	-.008	.264	.015	.299	.030		
	.400	.022	.014	.168	.045	.252	.077	.311	.099		
	.500	-.162	-.031	-.040	.090	.044	.162	.103	.177		
	.600										
	.700	-.350	-.444	-.290	-.228	-.213	-.127	-.158	-.068		
	.800										
	.900	-.211	-.227	-.058	-.130	-.042	-.079	-.011	-.044		
	1.000										
Outboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010	.156	.071	.151	.049	.158	.040	.168	.046		
	.025	.160	-.080	.220	-.065	.268	-.038	.298	-.036		
	.050	-.117	.067	.139	.127	.288	.150	.392	.180		
	.100	-1.014	.156	-.526	.265	-.241	.304	-.064	.343		
	.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010	.006	-.023	.004	-.028	.015	-.030	.030	-.023		
	.025	-.039	.026	-.016	.029	-.001	.032	.017	.038		
	.050	-.092	-.158	-.057	-.146	-.021	-.135	-.003	-.125		
	.100	.067	-.206	.074	-.028	.089	-.148	.099	-.133		
	.200	-.031	-.121	.037	-.093	.080	-.082	.119	-.064		
	.300	.030	-.031	.114	-.004	.174	.003	.221	.034		
	.400	-.239	.042	-.040	.086	.080	.117	.160	.140		
	.500	-.940	-.158	-.424	.000	-.245	.072	-.158	.135		
	.600										
	.700	-.186	-.247	-.142	-.146	-.095	-.082	-.068	-.048		
	.800										
	.900	-.080	-.096	-.020	-.053	.011	-.025	.026	-.015		
	1.000										

TABLE 24.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.240	.059	.221	.032	.223	.031	.225	.040		
.025	.279	-.134	.335	-.117	.369	-.076	.401	-.055		
.050	.161	.083	.378	.127	.491	.160	.567	.197		
.100	-.523	.161	-.195	.221	-.001	.278	.130	.323		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.083	.024	.076	.021	.090	.038	.103	.056		
.025	.075	-.039	.076	-.046	.098	-.028	.114	-.019		
.050	.039	-.106	.052	-.105	.078	-.087	.103	-.078		
.100	.087	-.153	.036	-.137	.074	-.111	.107	-.090		
.200	.039	-.106	.095	-.093	.125	-.064	.158	-.039		
.300	.165	-.023	.221	-.007	.271	.019	.311	.040		
.400	.063	.020	.182	.048	.251	.082	.315	.107		
.500	-.145	-.027	-.042	.080	.031	.149	.091	.170		
.600										
.700	-.645	-.483	-.321	-.302	-.257	-.170	-.196	-.106		
.800										
.900	-.283	-.283	-.156	-.184	-.080	-.115	-.047	-.078		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.165	.063	.158	.040	.172	.046	.181	.048		
.025	.188	-.090	.236	-.074	.278	-.044	.306	-.027		
.050	-.078	.067	.134	.111	.274	.152	.365	.181		
.100	-.941	.153	-.560	.248	-.307	.309	-.145	.342		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.008	-.023	.005	-.038	.023	-.024	.036	-.019		
.025	-.031	.020	-.019	.021	.003	.031	.024	.032		
.050	-.086	-.165	-.058	-.156	-.024	-.130	.004	-.113		
.100	.063	-.204	.068	.397	.082	-.119	.091	-.121		
.200	-.012	-.129	.040	-.101	.082	-.079	.118	-.062		
.300	.055	-.031	.119	-.011	.168	.011	.212	.036		
.400	-.208	.043	-.050	.087	.054	.121	.134	.146		
.500	-.941	-.169	-.595	-.027	-.382	.062	-.223	.114		
.600										
.700	-.220	-.275	-.164	-.183	-.123	-.119	-.109	-.082		
.800										
.900	-.058	-.106	-.062	-.089	-.024	-.064	-.027	-.062		
1.000										

TABLE 25.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a

(a) $M = 0.700$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
			.176	-.109	.179	-.102	.179	-.103	.171	-.105
			.235	-.061	.286	-.049	.329	-.035	.370	.006
			.375	.133	.514	.175	.586	.198	.735	.254
			-.061	.123	.131	.141	.266	.140	.521	.162
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
			.046	-.124	.058	-.121	.072	-.113	.108	-.096
			.055	-.104	.034	-.092	.052	-.089	.094	-.067
			.036	-.192	.063	-.175	.038	-.157	.094	-.139
			.051	-.148	.048	-.121	.072	-.113	.157	-.091
			.167	-.027	.208	-.010	.247	-.001	.332	.055
			.235	.031	.310	.077	.373	.091	.477	.137
			.055	.147	.184	.189	.276	.217	.443	.298
			-.269	-.003	-.141	.092	-.045	.169	.137	.293
			.176	-.129	.194	-.053	.198	.004	.244	.128
		-.100	-.100	-.044	-.053	-.006	-.016	.069	.060	
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
			.108	.012	.116	.010	.125	.013	.137	.006
			.118	-.046	.179	-.034	.207	-.016	.311	.011
			.191	.152	.353	.169	.469	.193	.651	.234
			-.239	.147	-.000	.203	.115	.231	.379	.219
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
			-.022	-.128	-.035	-.116	.013	-.108	.045	-.100
			-.080	-.017	-.053	-.005	-.030	-.001	.016	.006
			-.104	-.230	-.063	-.208	-.035	-.195	.030	-.163
			-.027	-.138	-.000	-.112	.042	-.098	.123	-.081
			.084	.031	.140	.043	.183	.057	.268	.093
			.133	.041	.232	.068	.299	.086	.428	.127
			-.167	.142	-.005	.203	.101	.251	.302	.336
			-.259	-.070	-.155	.024	-.084	.086	.064	.190
			-.095	-.109	-.049	-.063	-.011	-.030	.064	.263
		-.027	-.027	.013	-.000	.038	.018	.093	.079	

TABLE 25.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(b) $M = 0.750$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.205	-.091	.198	-.099	.201	-.091	.198	-.090
.025			.271	-.047	.333	-.045	.347	-.020	.406	.025
.050			.386	.148	.512	.171	.586	.214	.734	.256
.100			-.069	.130	.101	.127	.263	.135	.486	.163
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.068	-.109	.070	-.112	.090	-.104	.123	-.077
.025			.033	-.083	.039	-.098	.059	-.073	.114	-.050
.050			.050	-.180	.025	-.169	.055	-.158	.118	-.112
.100			.024	-.136	.065	-.125	.099	-.100	.176	-.072
.200			.187	-.012	.224	-.006	.263	.015	.344	.065
.300			.263	.068	.317	.070	.383	.104	.486	.145
.400			.059	.165	.171	.198	.268	.241	.429	.304
.500			-.290	-.003	-.178	.078	-.065	.161	.103	.278
.600										
.700			.174	-.131	.180	-.067	.188	-.007	.234	.105
.800										
.900			-.105	-.109	-.059	-.072	-.016	-.034	.048	.043
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.129	.024	.131	.016	.143	.024	.162	.025
.025			.147	-.034	.206	-.028	.236	-.011	.335	.025
.050			.205	.156	.369	.180	.452	.201	.631	.242
.100			-.294	.222	-.081	.272	.086	.280	.321	.268
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			-.007	-.122	-.006	-.120	.024	-.109	.061	-.094
.025			-.065	-.007	-.054	-.010	-.020	.002	.039	.012
.050			-.082	-.224	-.059	-.226	-.020	-.193	.047	-.165
.100			-.007	-.118	.021	-.111	.064	-.095	.140	-.054
.200			.103	.037	.153	.038	.192	.055	.291	.074
.300			.143	.054	.224	.069	.302	.099	.428	.140
.400			-.206	.156	-.054	.206	.081	.262	.260	.326
.500			-.272	-.078	-.191	.003	-.104	.077	.030	.184
.600										
.700			-.087	.085	-.054	.052	-.011	.086	.043	.414
.800										
.900			-.029	-.029	-.001	-.010	.033	.019	.056	.043
1.000										

TABLE 25.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.216	-.086	.215	-.091	.216	-.078	.219	-.074
.025			.284	-.052	.321	-.044	.352	-.022	.410	.023
.050			.403	.157	.525	.181	.594	.216	.724	.261
.100			-.086	.127	.092	.130	.254	.139	.461	.159
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.076	-.098	.083	-.103	.101	-.090	.134	-.062
.025			.042	-.086	.054	-.082	.080	-.069	.125	-.045
.050			.059	-.171	.037	-.163	.071	-.150	.125	-.104
.100			.038	-.128	.075	-.116	.114	-.090	.193	-.062
.200			.199	-.005	.240	.003	.279	.020	.350	.070
.300			.275	.072	.334	.075	.386	.114	.486	.155
.400			.059	.169	.168	.202	.267	.245	.499	.308
.500			-.336	-.001	-.197	.100	-.090	.160	.083	.265
.600										
.700			.174	-.141	.177	-.074	.194	-.014	.223	.083
.800										
.900			-.120	-.115	-.069	-.078	-.027	-.039	.036	.011
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.139	.029	.147	.028	.164	.033	.180	.036
.025			.160	-.035	.213	-.027	.249	-.010	.341	.032
.050			.211	.160	.367	.185	.457	.207	.608	.256
.100			-.285	.288	-.086	.367	.079	.342	.269	.290
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			-.005	-.115	.007	-.120	.033	-.111	.066	-.083
.025			-.056	-.005	-.035	-.006	-.005	.003	.040	.011
.050			-.073	-.221	-.044	-.209	-.010	-.184	.053	-.159
.100			.012	-.124	.037	-.107	.071	-.095	.150	-.057
.200			.118	.037	.168	.041	.211	.058	.286	.053
.300			.160	.063	.243	.079	.308	.101	.422	.138
.400			-.230	.156	-.061	.210	.067	.257	.222	.328
.500			-.306	-.090	-.209	-.014	-.128	.067	-.006	.163
.600										
.700			-.085	.275	-.052	.168	-.010	.177	.015	.553
.800										
.900			-.030	-.030	-.002	-.006	.033	.016	.023	.002
1.000										

TABLE 25.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Concluded

(d) $M = 0.800$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.228	-.074	.228	-.087	.233	-.070	.234	-.061
.025			.322	-.053	.359	-.034	.376	-.008	.434	.041
.050			.404	.159	.514	.191	.596	.212	.717	.262
.100			-.098	.126	.101	.130	.224	.135	.422	.164
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.081	-.094	.101	-.095	.110	-.082	.152	-.049
.025			.049	-.074	.069	-.070	.086	-.066	.131	-.028
.050			.024	-.168	.052	-.168	.081	-.164	.136	-.106
.100			.057	-.123	.093	-.103	.122	-.082	.205	-.040
.200			.208	-.004	.248	.011	.286	.032	.356	.074
.300			.286	.073	.346	.069	.396	.110	.483	.152
.400			.053	.171	.167	.216	.261	.245	.402	.303
.500			-.372	-.000	-.246	.081	-.115	.151	.058	.258
.600										
.700			.171	-.164	.179	-.091	.192	-.033	.234	.074
.800										
.900			-.139	-.135	-.091	-.091	-.049	-.066	.009	-.004
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.159	.036	.162	.028	.171	.032	.192	.037
.025			.183	-.037	.207	-.029	.256	-.000	.343	.025
.050			.220	.167	.350	.187	.440	.220	.584	.254
.100			-.375	.419	-.123	.533	.024	.464	.233	.364
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			-.000	-.115	.015	-.111	.036	-.098	.074	-.085
.025			-.053	-.004	-.025	-.005	-.004	-.000	.045	.013
.050			-.066	-.220	-.034	-.217	-.000	-.196	.066	-.159
.100			.016	-.115	.048	-.111	.073	-.086	.139	-.044
.200			.130	.036	.174	.044	.212	.045	.282	.045
.300			.167	.069	.240	.085	.309	.106	.401	.139
.400			-.294	.167	-.087	.207	.028	.244	.192	.307
.500			-.331	-.106	-.258	-.013	-.159	.049	-.044	.152
.600										
.700			-.086	.460	-.058	.297	-.033	.289	-.004	.768
.800										
.900			-.037	-.029	-.013	-.013	.004	-.012	-.008	-.024
1.000										

TABLE 26.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b

(a) $M = 0.700$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.197	.013	.187	-.002	.188	-.011	.182	-.026		
.025	.168	-.098	.212	-.070	.261	-.064	.275	-.051		
.050	.067	.071	.333	.120	.441	.150	.508	.168		
.100	-.204	.071	.129	.061	.256	.096	.338	.124		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.120	.033	.129	.076	.150	.082	.163	.027		
.025	.057	-.078	.090	-.084	.116	-.069	.139	-.065		
.050	.071	-.117	.071	-.104	.106	-.108	.129	-.094		
.100	.067	-.166	.076	-.142	.116	-.127	.148	-.119		
.200	.096	-.233	.178	-.128	.227	-.122	.260	-.114		
.300	.076	-.006	.217	.018	.295	.038	.357	.051		
.400	-.364	.192	-.041	-.002	.106	-.015	.211	.003		
.500	-.127	-.253	-.075	-.070	-.030	.023	.008	.100		
.600										
.700	-.291	-.272	-.196	-.152	-.137	-.083	-.094	-.026		
.800										
.900	-.137	-.146	-.060	-.084	-.015	-.045	.017	-.017		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	-.136	.047	.032	.037	.004	.033	-.012	.017		
.025	.086	.424	.172	.177	.222	.086	.240	.046		
.050	-.300	.462	.119	.352	.299	.217	.419	.143		
.100	.134	.148	.172	.279	.227	.314	.303	.342		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.684	-.020	.366	-.016	.280	-.015	.293	-.012		
.025	.414	.013	.439	.018	.338	.023	.293	.017		
.050	-.054	-.127	-.002	-.099	.028	-.088	.051	-.085		
.100	-.049	.071	.013	.018	.057	-.015	.090	-.036		
.200	-.001	-.049	.100	-.021	.154	-.015	.192	-.012		
.300	-.141	.028	.100	.066	.207	.091	.284	.104		
.400	-.556	-.045	-.249	.124	-.093	.202	.017	.240		
.500	-.426	-.305	-.257	-.108	-.195	-.015	-.128	.041		
.600										
.700	-.160	-.165	-.095	-.084	-.054	-.030	-.031	.003		
.800										
.900	-.045	-.054	.008	-.021	.038	-.001	.051	.012		
1.000										

TABLE 26.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.210	.011	.203	.008	.210	.006	.207	-.001		
.025	.192	-.100	.243	-.071	.303	-.047	.340	-.041		
.050	.108	.077	.354	.128	.462	.152	.526	.185		
.100	-.113	.024	.216	.061	.329	.094	.402	.132		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.126	.033	.146	.039	.170	.046	.181	.048		
.025	.064	-.077	.110	-.067	.134	-.056	.159	-.054		
.050	.037	-.122	.092	-.107	.121	-.083	.150	-.072		
.100	.033	-.166	.097	-.133	.143	-.114	.176	-.103		
.200	.117	-.166	.203	.030	.249	.015	.291	.012		
.300	.104	-.007	.235	.035	.311	.055	.371	.070		
.400	-.369	.360	-.045	.150	.099	.117	.203	.128		
.500	-.131	-.268	-.058	-.076	-.025	.072	.017	.092		
.600										
.700	-.321	-.294	-.209	-.164	-.149	-.092	-.103	-.041		
.800										
.900	-.166	-.175	-.076	-.107	-.021	-.061	.008	-.028		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	-.024	.051	.172	.039	.134	.037	.123	.034		
.025	.117	.606	.194	.326	.258	.218	.278	.180		
.050	-.214	.549	.136	.490	.324	.351	.423	.278		
.100	.359	.139	.300	.273	.324	.320	.392	.348		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.879	-.020	.495	-.014	.390	-.007	.392	-.001		
.025	.425	.024	.543	.021	.443	.028	.397	.021		
.050	-.047	-.121	.013	-.102	.050	-.078	.070	-.081		
.100	-.038	.231	.035	.176	.081	.125	.114	.105		
.200	.024	-.051	.119	-.027	.174	-.003	.220	-.001		
.300	-.099	.033	.105	.070	.209	.094	.295	.114		
.400	-.942	-.024	-.301	.123	-.131	.187	-.014	.242		
.500	-.479	-.342	-.279	-.124	-.197	-.025	-.125	.048		
.600										
.700	-.166	-.179	-.058	-.089	-.056	-.034	-.027	-.001		
.800										
.900	-.051	-.060	.004	-.032	.041	-.003	.057	.012		
1.000										

TABLE 26.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.225	.025	.219	.006	.225	.004	.230	.009		
.025	.225	-.102	.257	-.079	.314	-.047	.358	-.020		
.050	.144	.053	.363	.138	.471	.157	.532	.192		
.100	-.085	.030	.329	.062	.411	.089	.485	.128		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.144	.047	.159	.049	.178	.055	.192	.060		
.025	.089	-.072	.125	-.062	.144	-.052	.175	-.025		
.050	.055	-.110	.104	-.087	.140	-.081	.175	-.067		
.100	.051	-.161	.108	-.130	.152	-.111	.192	-.084		
.200	.127	-.077	.214	.189	.263	.135	.307	.132		
.300	.132	.008	.261	.045	.322	.063	.383	.090		
.400	-.365	.611	-.045	.274	.101	.212	.192	.234		
.500	-.132	-.263	-.057	-.074	-.018	.008	.018	.098		
.600										
.700	-.331	-.310	-.219	-.168	-.154	-.094	-.110	-.037		
.800										
.900	-.174	-.183	-.079	-.113	-.026	-.069	.009	-.025		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.042	.051	.265	.040	.232	.042	.217	.043		
.025	.140	.796	.210	.451	.266	.317	.293	.281		
.050	-.148	.563	.132	.566	.317	.427	.416	.353		
.100	.563	.165	.400	.282	.394	.330	.446	.348		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	1.029	-.013	.591	-.011	.453	-.001	.437	.013		
.025	.343	.025	.555	.023	.508	.029	.446	.030		
.050	-.021	-.115	.027	-.095	.059	-.081	.094	-.067		
.100	-.017	.394	.040	.303	.093	.228	.128	.217		
.200	.038	-.042	.121	-.015	.177	-.005	.225	.013		
.300	-.068	.038	.108	.078	.211	.101	.293	.124		
.400	-.915	-.013	-.341	.125	-.162	.186	-.033	.242		
.500	-.500	-.352	-.303	-.139	-.213	-.047	-.122	.035		
.600										
.700	-.161	-.170	-.100	-.091	-.056	-.039	-.029	.001		
.800										
.900	-.055	-.055	-.002	-.036	.037	-.005	.060	.013		
1.000										

TABLE 26.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Concluded

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.237	.034	.234	.029	.241	.020	.238	.005		
.025	.233	-.097	.274	-.065	.310	-.045	.336	-.044		
.050	.180	.091	.376	.140	.469	.155	.528	.185		
.100	-.068	.025	.483	.062	.498	.082	.593	.115		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.152	.054	.176	.062	.188	.069	.201	.070		
.025	.095	-.064	.136	-.056	.163	-.037	.180	-.040		
.050	.066	-.101	.123	-.081	.147	-.069	.172	-.060		
.100	.066	-.146	.127	-.114	.163	-.098	.201	-.081		
.200	.152	.046	.221	.430	.269	.322	.311	.340		
.300	.156	.054	.266	.062	.326	.069	.376	.091		
.400	-.309	.942	-.056	.454	.094	.351	.197	.348		
.500	-.121	-.264	-.048	-.077	-.020	.004	.001	.070		
.600										
.700	-.358	-.337	-.228	-.187	-.167	-.114	-.130	-.069		
.800										
.900	-.215	-.211	-.089	-.130	-.045	-.082	-.020	-.060		
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.111	.050	.404	.041	.350	.045	.331	.041		
.025	.164	1.033	.229	.608	.277	.440	.310	.388		
.050	-.113	.521	.156	.628	.285	.493	.392	.400		
.100	.798	.172	.526	.270	.468	.322	.518	.331		
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	1.159	-.011	.677	-.003	.505	.004	.477	.009		
.025	.188	.025	.636	.029	.558	.029	.486	.021		
.050	-.007	-.121	.037	-.089	.077	-.069	.090	-.073		
.100	-.007	.594	.054	.473	.098	.362	.123	.351		
.200	.058	-.040	.131	-.011	.187	.004	.225	.005		
.300	-.048	.042	.111	.082	.204	.106	.270	.119		
.400	-.885	-.011	-.456	.123	-.195	.179	-.069	.221		
.500	-.580	-.410	-.317	-.154	-.224	-.069	-.158	-.003		
.600										
.700	-.158	-.174	-.097	-.097	-.065	-.049	-.048	-.032		
.800										
.900	-.068	-.060	-.003	-.040	.029	-.012	.029	-.020		
1.000										

TABLE 27.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6

(a) $M = 0.700$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.385	.094	.368	.096	.229	-.053	.252	.028
.025			.443	.002	.465	-.010	.321	-.140	.344	-.035
.050			.564	-.095	.625	-.078	.545	-.160	.640	-.025
.100			.065	.099	.290	.125	.346	.035	.727	.169
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.133	-.008	.145	.048	.059	-.097	.179	.009
.025			.113	-.105	.140	-.103	.054	-.184	.198	-.059
.050			.079	-.153	.116	-.117	.044	-.204	.203	-.083
.100			.108	.002	.140	.043	.093	-.058	.242	.067
.200			.171	-.027	.213	.048	.142	-.038	.300	.057
.300			.123	.094	.223	.135	.200	.054	.446	.203
.400			-.221	.055	-.064	.082	-.053	.035	.237	.242
.500			.050	-.153	.096	-.044	.025	-.077	.203	.154
.600										
.700			.171	-.158	.213	-.088	.132	-.140	.290	.067
.800										
.900			-.090	-.120	-.049	-.078	-.116	-.140	.062	.067
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.355	.181	.353	.169	.345	.146	.299	.115
.025			.408	.118	.425	.111	.432	.102	.392	.101
.050			.432	.123	.513	.145	.563	.156	.619	.169
.100			-.293	.142	.048	.207	.258	.233	.580	.314
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.064	.011	.086	.014	.102	.010	.135	.018
.025			.050	.002	.072	.004	.098	.010	.140	.014
.050			.040	-.071	.067	-.059	.102	-.043	.164	-.030
.100			-.047	-.076	-.006	-.044	.102	-.038	.203	-.011
.200			.089	.011	.169	.038	.233	.054	.343	.086
.300			-.076	.084	.052	.091	.175	.098	.348	.130
.400			-.289	-.037	-.170	.048	-.072	.102	.115	.207
.500			-.231	-.129	-.151	-.044	-.087	.010	.023	.125
.600										
.700			-.085	-.095	-.044	-.049	-.024	-.014	.043	.057
.800										
.900			-.003	-.023	.023	-.001	.044	.020	.091	.062
1.000										

TABLE 27.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.403	.107	.384	.087	.260	-.037	.294	.055
.025			.465	.040	.477	-.002	.335	-.144	.401	-.002
.050			.581	-.092	.645	-.077	.574	-.152	.671	-.007
.100			.071	.102	.291	.118	.331	.034	.728	.170
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.147	.001	.162	.020	.078	-.082	.206	.037
.025			.129	-.092	.158	-.077	.082	-.179	.223	-.042
.050			.098	-.128	.131	-.108	.069	-.201	.228	-.056
.100			.129	-.017	.176	.029	.113	-.042	.272	.060
.200			.173	-.017	.211	.047	.136	-.077	.290	.073
.300			.142	.098	.247	.131	.211	.060	.449	.215
.400			-.247	.049	-.090	.087	-.068	.038	.232	.232
.500			.067	-.159	.091	-.059	.025	-.090	.206	.139
.600										
.700			.173	-.168	.207	-.099	.131	-.144	.285	.060
.800										
.900			-.110	-.137	-.064	-.090	-.126	-.152	.051	.060
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.372	.186	.374	.171	.365	.140	.342	.130
.025			.425	.133	.445	.109	.454	.095	.435	.081
.050			.438	.115	.542	.140	.577	.122	.634	.157
.100			-.331	.098	.007	.184	.175	.202	.546	.302
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.076	.009	.100	.012	.113	.007	.152	.037
.025			.062	.005	.091	.007	.104	-.002	.157	.020
.050			.049	-.070	.087	-.050	.113	-.055	.183	-.020
.100			-.035	-.066	.051	-.046	.113	-.046	.223	.006
.200			.107	.014	.188	.042	.228	.047	.351	.104
.300			-.083	.080	.056	.091	.144	.087	.325	.117
.400			-.331	-.066	-.201	.016	-.104	.078	.086	.188
.500			-.247	-.145	-.179	-.064	-.126	-.028	.006	.095
.600										
.700			-.088	-.101	-.050	-.055	-.037	-.028	.024	.037
.800										
.900			-.008	-.030	.016	-.002	.034	.011	.064	.037
1.000										

TABLE 27.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
			.413	.112	.395	.072	.381	.071	.301	.021
			.464	.005	.501	-.013	.483	-.006	.416	-.051
			.600	-.088	.650	-.073	.678	-.044	.692	-.013
			.099	.103	.297	.118	.474	.134	.688	.153
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
			.154	.010	.174	.021	.190	.028	.204	.034
			.141	-.088	.169	-.090	.194	-.061	.229	-.055
			.112	-.118	.152	-.094	.181	-.087	.233	-.059
			.141	.005	.182	-.001	.219	.032	.276	.034
			.171	-.016	.212	.008	.236	.028	.272	.064
			.150	.103	.254	.135	.309	.168	.441	.200
			-.266	.010	-.103	.076	.015	.134	.191	.212
			.039	-.177	.089	-.077	.126	.003	.178	.110
			.167	-.177	.208	-.103	.232	-.048	.272	.030
			-.122	-.147	-.073	-.103	-.036	-.061	.009	-.025
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
			.387	.188	.385	.169	.388	.159	.351	.131
			.442	.124	.462	.097	.473	.104	.457	.068
			.455	.116	.534	.126	.592	.138	.644	.110
			-.279	.116	-.018	.160	.202	.210	.495	.254
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
			.082	.018	.105	.012	.130	.032	.157	.017
			.073	.005	.101	.008	.126	.011	.161	.013
			.052	-.071	.097	-.060	.130	-.036	.186	-.030
			.001	-.062	.059	-.043	.126	-.031	.216	-.004
			.111	.014	.194	.042	.249	.062	.330	.089
			-.101	.077	.054	.088	.151	.096	.296	.093
			-.363	-.062	-.221	.016	-.129	.083	.030	.157
			-.257	-.164	-.187	-.086	-.125	-.010	-.047	.064
			-.092	-.109	-.052	-.064	-.023	-.023	-.021	-.000
			-.012	-.033	.016	-.005	.045	.020	-.000	-.030
1.000										

TABLE 27.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Concluded

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.437	.110	.405	.087	.394	.067	.333	.023
.025			.523	-.012	.507	-.032	.471	-.015	.439	-.046
.050			.612	-.086	.654	-.056	.688	-.043	.704	-.001
.100			.114	.102	.299	.111	.402	.124	.676	.154
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.167	.020	.185	.025	.198	.034	.223	.048
.025			.159	-.078	.180	-.060	.202	-.055	.239	-.034
.050			.126	-.119	.156	-.101	.190	-.080	.247	-.046
.100			.155	-.021	.197	-.020	.230	-.006	.276	.019
.200			.167	-.012	.197	.005	.218	.034	.268	.076
.300			.159	.106	.242	.131	.308	.157	.431	.199
.400			-.266	.008	-.130	.074	-.019	.124	.162	.207
.500			.053	-.188	.074	-.093	.108	-.019	.182	.101
.600										
.700			.167	-.188	.193	-.122	.214	-.076	.268	.011
.800										
.900			-.147	-.168	-.093	-.126	-.068	-.096	-.018	-.054
1.000										
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010			.407	.195	.400	.172	.401	.169	.369	.145
.025			.460	.130	.477	.115	.487	.112	.471	.080
.050			.472	.122	.539	.123	.601	.120	.646	.137
.100			-.281	.102	-.024	.156	.165	.177	.438	.243
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010			.094	.020	.111	.021	.136	.030	.165	.035
.025			.089	.008	.115	.009	.136	.010	.170	.015
.050			.077	-.065	.103	-.052	.140	-.035	.194	-.014
.100			.053	-.065	.074	-.040	.132	-.039	.231	.003
.200			.130	.024	.192	.046	.242	.071	.328	.092
.300			-.098	.081	.017	.082	.120	.083	.275	.084
.400			-.424	-.082	-.280	-.003	-.190	.051	-.030	.141
.500			-.265	-.179	-.207	-.093	-.165	-.051	-.087	.039
.600										
.700			-.098	-.110	-.064	-.077	-.047	-.051	-.046	-.030
.800										
.900			-.021	-.033	.005	-.020	.002	-.023	-.038	-.067
1.000										

TABLE 28.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7

(a) $M = 0.700$

\bar{x}/c	C_p at -										
	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 1^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$						
Inboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010	.284	.109	.247	.081	.224	.077	.099	-.048	.147	.043
	.025	.289	.043	.256	.020	.243	.058	.137	-.072	.181	.029
	.050	.384	-.028	.413	.048	.418	.053	.302	-.034	.332	.048
	.100	.237	.010	.521	.072	.622	.138	.581	.056	.735	.242
	.150	-.930	-.208	-.189	-.026	.073	.079	.281	.143	.588	.243
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010	-.019	.000	.024	-.018	.058	-.013	-.010	-.110	.138	-.014
	.025	.019	-.076	.020	-.080	.034	-.080	-.053	-.166	.067	-.056
	.050	-.009	-.137	.015	-.127	.039	-.113	-.043	-.204	.091	-.070
	.100	.000	-.165	.039	-.141	.044	-.137	-.043	-.209	.024	-.070
.200	-.104	-.099	.024	-.070	.143	-.061	.094	-.133	.252	.005	
.300	.090	.005	.200	.034	.262	.053	.217	-.024	.408	.119	
.400	-.199	.066	.048	.124	.172	.162	.165	.099	.432	.247	
.500	-.492	-.180	-.259	-.013	-.151	.063	-.157	.037	.124	.247	
.600	-.396	-.296	-.246	-.127	-.172	-.047	-.114	.005	.017	.130	
.700	-.260	-.246	-.179	-.127	-.132	-.065	-.176	-.110	.015	.091	
.800	-.183	-.195	-.101	-.101	-.059	-.059	-.032	-.020	.048	.061	
.900	-.151	-.165	-.080	-.108	-.042	-.065	-.105	-.128	.062	.034	
1.000	-.189		-.108		-.053		-.020		.042		
Outboard station											
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	
	.010										
	.025										
	.050										
	.100										
	.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	
	.010										
	.025										
	.050										
	.100										
.200											
.300											
.400											
.500											
.600											
.700											
.800											
.900											
1.000											

TABLE 28.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.298	.108	.265	.088	.247	.071	.238	.053	.172	.068
.025	.298	.030	.304	.045	.269	.023	.264	.027	.210	.038
.050	.410	-.026	.437	.036	.437	.040	.428	.053	.383	.059
.100	.289	-.009	.541	.092	.636	.131	.699	.173	.759	.215
.150	-.881	-.224	-.213	-.019	.056	.051	.251	.120	.571	.216
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.009	-.004	.045	.036	.075	-.011	.109	-.003	.163	.008
.025	.021	-.082	.036	-.076	.045	-.076	.061	-.064	.085	-.040
.050	.043	-.134	.036	-.124	.049	-.115	.074	-.098	.111	-.066
.100	.043	-.173	.049	-.141	.053	-.123	.022	-.103	.038	-.062
.200	.009	-.091	.062	-.059	.161	-.041	.212	-.029	.275	-.020
.300	.108	.013	.222	.045	.278	.062	.333	.087	.418	.124
.400	-.194	.065	.053	.131	.170	.166	.255	.195	.418	.254
.500	-.578	-.194	-.292	-.016	-.179	.058	-.081	.126	.094	.241
.600	-.424	-.338	-.276	-.150	-.189	-.058	-.138	-.006	.004	.124
.700	-.280	-.263	-.184	-.136	-.132	-.072	-.085	-.016	.038	.081
.800	-.207	-.218	-.110	-.116	-.069	-.069	-.035	-.035	.038	.050
.900	-.181	-.194	-.093	-.119	-.054	-.080	-.016	-.038	.046	.025
1.000	-.224		-.122		-.069		-.041		.033	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010										
.025										
.050										
.100										
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010										
.025										
.050										
.100										
.200										
.300										
.400										
.500										
.600										
.700										
.800										
.900										
1.000										

TABLE 28.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(c) $M = 0.775$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.314	.115	.277	.082	.257	.050	.243	.065	.199	.067
.025	.331	.028	.318	.024	.294	.045	.280	.020	.237	.038
.050	.430	-.038	.447	.028	.451	.074	.442	.020	.398	.046
.100	.339	-.001	.550	.057	.650	.108	.703	.144	.771	.191
.150	-.768	-.213	-.213	-.043	.061	.050	.248	.116	.555	.170
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.004	-.001	.049	-.009	.083	-.004	.115	.007	.174	.017
.025	.032	-.079	.037	-.079	.054	-.066	.069	-.059	.100	-.037
.050	.049	-.133	.033	-.125	.062	-.104	.082	-.101	.125	-.054
.100	.012	-.166	.012	-.145	.021	-.116	.028	-.088	.050	-.054
.200	.103	-.083	.057	-.059	.174	-.033	.218	-.014	.286	.029
.300	.128	.020	.227	.041	.286	.070	.334	.090	.427	.129
.400	-.166	.074	.053	.136	.174	.170	.260	.198	.411	.249
.500	-.692	-.175	-.332	-.025	-.203	.054	-.101	.119	.067	.224
.600	-.416	-.345	-.290	-.158	-.214	-.071	-.147	-.005	-.022	.099
.700	-.290	-.274	-.199	-.150	-.141	-.083	-.092	-.038	-.020	.063
.800	-.208	-.213	-.125	-.131	-.077	-.077	-.043	-.032	.016	.033
.900	-.191	-.204	-.108	-.137	-.062	-.095	-.022	-.055	.029	.042
1.000	-.224		-.131		-.077		-.043		.005	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010										
.025										
.050										
.100										
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010										
.025										
.050										
.100										
.200										
.300										
.400										
.500										
.600										
.700										
.800										
.900										
1.000										

TABLE 28.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(d) $M = 0.800$

x/c	C_p at -									
	$\alpha = -2^\circ$	$\alpha = 0^\circ$	$\alpha = 1^\circ$	$\alpha = 2^\circ$	$\alpha = 4^\circ$					
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.324	.113	.290	.079	.266	.087	.255	.068	.223	.052
.025	.336	.014	.322	.016	.314	.031	.303	.025	.243	.048
.050	.451	-.050	.465	-.016	.465	.035	.462	.036	.423	.032
.100	.432	-.010	.568	.067	.648	.102	.709	.144	.773	.184
.150	-.677	-.228	-.210	-.047	.051	.035	.226	.104	.504	.156
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.018	.002	.063	-.000	.098	.007	.124	.013	.188	.024
.025	.037	-.074	.051	-.072	.063	-.057	.080	-.051	.112	-.028
.050	.018	-.134	.047	-.119	.071	-.101	.092	-.087	.136	-.048
.100	.018	-.158	.020	-.143	.031	-.101	.044	-.091	.068	-.052
.200	.229	-.082	.095	-.052	.186	-.033	.228	-.007	.295	.036
.300	.153	.022	.242	.044	.294	.075	.343	.100	.431	.140
.400	-.122	.073	.059	.139	.174	.170	.251	.200	.435	.259
.500	-.695	-.205	-.362	-.024	-.224	.047	-.131	.116	.048	.219
.600	-.408	-.376	-.305	-.184	-.223	-.076	-.170	-.017	-.050	.087
.700	-.289	-.293	-.211	-.167	-.156	-.097	-.107	-.043	-.032	.044
.800	-.234	-.239	-.131	-.142	-.081	-.086	-.054	-.049	.002	.008
.900	-.221	-.225	-.127	-.155	-.077	-.113	-.039	-.079	.004	-.020
1.000	-.255		-.147		-.081		-.059		-.024	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010										
.025										
.050										
.100										
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010										
.025										
.050										
.100										
.200										
.300										
.400										
.500										
.600										
.700										
.800										
.900										
1.000										

TABLE 28.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.336	.113	.308	.090	.216	.008	.194	.010	.234	.057
.025	.374	.032	.343	.013	.262	-.054	.236	-.048	.276	.049
.050	.474	-.044	.481	-.022	.404	-.008	.401	-.052	.449	.038
.100	.424	-.017	.577	.063	.577	.008	.628	.048	.780	.191
.150	-.600	-.228	-.206	-.074	.014	.019	.189	.082	.494	.173
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.036	.009	.078	.013	.035	-.061	.064	-.048	.195	.038
.025	.052	-.064	.067	-.060	.004	-.127	.018	-.117	.126	.068
.050	.032	-.133	.063	-.114	.004	-.169	.029	-.159	.153	-.039
.100	.029	-.156	.032	-.125	-.034	-.181	-.017	-.155	.084	-.028
.200	.409	-.071	.090	-.041	.119	-.092	.163	-.071	.307	.053
.300	.178	.029	.262	.059	.227	.004	.279	.033	.437	.153
.400	-.075	.086	.074	.140	.100	.096	.179	.121	.430	.257
.500	-.663	-.198	-.351	-.045	-.342	-.031	-.236	.025	.026	.199
.600	-.890	-.386	-.313	-.196	-.246	-.113	-.192	-.035	-.072	.081
.700	-.305	-.317	-.232	-.190	-.246	-.196	-.205	-.147	-.043	.026
.800	-.264	-.254	-.155	-.170	-.098	-.108	-.070	-.075	-.006	-.000
.900	-.271	-.256	-.160	-.179	-.177	-.208	-.147	-.186	-.012	-.047
1.000	-.289		-.175		-.124		-.085		-.046	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010										
.025										
.050										
.100										
.150										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010										
.025										
.050										
.100										
.200										
.300										
.400										
.500										
.600										
.700										
.800										
.900										
1.000										

TABLE 29.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8

(a) $M = 0.700$

x/c	C _p at -									
	α = -2°	α = 0°	α = 1°	α = 2°	α = 4°					
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000										
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.255	.149	.260	.124	.239	.098	.235	.085	.184	.048
	.289	.120	.323	.105	.322	.084	.313	.080	.256	.062
	.285	.100	.405	.110	.428	.108	.444	.109	.407	.121
	-.297	.086	.207	.158	.404	.181	.570	.196	.751	.256
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000										
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.021	-.035	-.008	-.036	.006	-.047	.022	-.046	.053	-.039
	-.040	-.011	-.002	-.002	.001	-.008	.017	-.003	.053	.014
	-.060	-.103	-.007	-.084	.006	-.096	.036	-.095	.072	-.064
	-.045	-.108	.023	-.089	.050	-.091	.090	-.075	.150	-.044
	-.001	-.040	.100	-.011	.137	-.008	.187	.007	.266	.038
	-.001	.018	.056	.066	.089	.069	.143	.054	.218	.135
	-.481	-.055	-.166	.076	-.023	.118	.090	.167	.281	.242
	-.413	-.273	-.225	-.094	-.159	-.028	-.075	.046	.067	.159
	-.161	-.186	-.094	-.084	-.076	-.052	-.037	-.012	.033	.072
	-.050	-.074	.003	-.021	.011	-.008	.041	.012	.087	.062

TABLE 29.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.283	.155			.262	.099	.259	.095	.208	.094
	.318	.115			.337	.085	.343	.095	.297	.076
	.327	.097			.448	.090	.462	.095	.443	.120
	-.216	.093			.426	.183	.542	.219	.743	.226
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.013	-.035			.015	-.047	.037	-.043	.067	-.026
	-.031	-.013			.010	-.008	.033	-.002	.067	.010
	-.048	-.115			.019	-.096	.050	-.082	.080	-.052
	-.035	-.119			.068	-.083	.104	-.069	.164	-.043
	.013	-.040			.152	-.003	.201	.019	.279	.049
	.005	.031			.099	.085	.148	.104	.213	.142
	-.526	-.053			-.056	.121	.068	.170	.244	.226
	-.570	-.300			-.180	-.030	-.087	.024	.032	.138
	-.168	-.194			-.074	-.056	-.038	-.012	.005	.049
	-.057	-.079			.010	-.012	.042	.011	.049	.023

TABLE 29.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.295 .338 .350 -.150	.160 .117 .083 .075	.285 .349 .429 .230	.132 .103 .103 .137	.276 .353 .463 .395	.115 .081 .090 .183	.270 .355 .478 .541	.104 .083 .104 .206	.234 .323 .467 .726	.102 .098 .119 .246
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.006 -.027 -.044 -.023 .032 .007 -.531 -.612	-.040 -.006 -.116 -.116 -.035 .032 -.048 -.328	.009 -.003 .001 .039 .124 .060 -.245 -.292	-.041 -.008 -.105 -.088 -.008 .069 .069 -.135	.031 .022 .035 .077 .166 .107 -.063 -.194	-.037 -.003 -.092 -.080 .005 .086 .120 -.042	.049 .045 .062 .117 .214 .142 .049 -.113	-.032 .002 -.079 -.066 .019 .113 .164 .023	.081 .085 .110 .178 .285 .217 .225 .006	-.017 .017 -.047 -.021 .059 .144 .229 .132
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.162 -.056	-.201 -.078	-.114 -.020	-.109 -.041	-.071 .009	-.054 -.016	-.036 .045	-.019 .015	-.004 .021	.030 -.013

TABLE 29.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(d) $M = 0.800$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010 .025 .050 .100 .150	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.312	.153	.291	.128	.293	.117	.281	.110	.245	.094
	.360	.116	.361	.096	.370	.089	.363	.086	.339	.096
	.381	.083	.442	.079	.468	.089	.485	.106	.482	.115
	-.100	.071	.202	.132	.387	.178	.506	.171	.689	.209
.010 .025 .050 .100 .150	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
	-.002	-.039	.010	-.051	.036	-.038	.045	-.033	.078	-.028
	-.019	-.010	-.002	-.014	.028	-.005	.045	-.004	.082	-.001
	-.035	-.116	.002	-.120	.040	-.087	.065	-.094	.107	-.056
	-.006	-.112	.047	-.100	.085	-.070	.114	-.065	.176	-.036
.010 .025 .050 .100 .150	.051	-.035	.120	-.014	.166	.007	.212	.020	.278	.046
	.010	.038	.055	.063	.101	.097	.134	.110	.196	.135
	-.532	-.051	-.312	.039	-.095	.117	.008	.143	.180	.209
	-.769	-.361	-.328	-.149	-.225	-.066	-.159	-.016	-.048	.082
.010 .025 .050 .100 .150	-.198	-.210	-.125	-.125	-.083	-.070	-.065	-.045	-.052	-.011
	-.063	-.080	-.035	-.055	.003	-.025	-.000	-.025	-.032	-.064

TABLE 29.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^{\circ}$		$\alpha = 0^{\circ}$		$\alpha = 1^{\circ}$		$\alpha = 2^{\circ}$		$\alpha = 4^{\circ}$	
Inboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000										
Outboard station										
.010 .025 .050 .100 .150	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
	.329	.156	.307	.134	.304	.127	.297	.132	.264	.099
	.376	.113	.381	.106	.383	.088	.384	.097	.355	.091
	.404	.078	.460	.090	.477	.092	.502	.089	.500	.107
	-.048	.042	.224	.118	.371	.151	.474	.179	.673	.209
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010 .025 .050 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.000	.007	-.036	.027	-.039	.041	-.034	.057	-.029	.087	-.007
	-.005	-.005	.016	-.008	.037	-.003	.053	-.002	.091	.005
	-.021	-.115	.016	-.106	.049	-.089	.077	-.081	.123	-.058
	.011	-.111	.059	-.087	.096	-.077	.128	-.049	.186	-.023
	.066	-.025	.130	-.004	.174	.005	.215	.034	.280	.068
	.019	.046	.059	.075	.096	.096	.128	.116	.153	.146
	-.516	-.044	-.319	.047	-.156	.096	-.049	.136	.138	.201
	-.771	-.433	-.386	-.181	-.286	-.109	-.226	-.045	-.074	.068
	-.308	-.233	-.138	-.138	-.109	-.105	-.100	-.081	-.066	-.027
-.052	-.080	-.059	-.079	-.042	-.077	-.061	-.100	-.058	-.090	

TABLE 30.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9

(a) $M = 0.700$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.280	.115	.246	.085			.198	.051		
.025	.266	.078	.250	.049			.202	.019		
.050	.381	-.004	.420	.017			.390	.033		
.100	.271	.042	.521	.113			.670	.179		
.150	-.646	-.238	-.075	.040			.340	.152		
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.010	.014	.007	.003			.014	-.009		
.025	.032	-.036	.040	-.043			.051	-.045		
.050	.005	-.096	.030	-.089			.056	-.082		
.100	-.004	-.128	.040	-.085			.092	-.087		
.200	.087	-.045	.154	-.015			.220	.005		
.300	.101	.014	.205	.049			.308	.078		
.400	-.123	.037	.095	.108			.275	.170		
.500	-.398	-.142	-.208	.007			-.018	.143		
.600	-.316	-.275	-.203	-.116			-.082	.028		
.700	-.265	-.265	-.176	-.148			-.091	-.036		
.800	-.215	-.229	-.139	-.139			-.059	-.054		
.900	-.165	-.174	-.084	-.107			-.027	-.032		
1.000	-.151		-.084				-.013			
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.279	.160	.277	.131			.257	.106		
.025	.297	.142	.332	.135			.321	.110		
.050	.293	.128	.410	.145			.458	.138		
.100	-.274	.124	.195	.199			.531	.261		
.150	-.004	-.247	.026	-.025			.028	.142		
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.009	-.032	-.002	-.038			.010	-.045		
.025	-.032	-.059	-.011	-.066			.010	-.059		
.050	-.055	-.087	-.020	-.084			.019	-.082		
.100	-.050	-.087	.007	-.070			.065	-.045		
.200	.019	-.013	.099	.007			.179	.028		
.300	-.023	.041	.131	.081			.266	.119		
.400	-.384	-.009	-.116	.103			.097	.188		
.500	-.297	-.205	-.162	-.057			-.032	.083		
.600	-.210	-.224	-.134	-.098			-.050	.005		
.700	-.164	-.187	-.107	-.102			-.050	-.027		
.800	-.114	-.146	-.061	-.093			-.009	-.036		
.900	-.059	-.073	-.006	-.034			.028	.010		
1.000	-.009		.021				.060			

TABLE 30.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(b) $M = 0.750$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.297	.122	.266	.087	.250	.087	.230	.075	.184	.079
.025	.297	.047	.258	.045	.246	.054	.242	.037	.184	.054
.050	.410	-.012	.438	.020	.439	.021	.426	.033	.376	.066
.100	.306	.034	.538	.108	.622	.138	.681	.163	.744	.234
.150	-.638	-.233	-.093	.020	.158	.092	.326	.146	.589	.246
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.022	.013	.020	.007	.029	.012	.033	.016	.058	.033
.025	.047	-.033	.059	-.030	.066	-.034	.075	-.034	.112	-.000
.050	.013	-.095	.045	-.085	.066	-.084	.079	-.071	.129	-.026
.100	.009	-.125	.058	-.093	.087	-.088	.117	-.076	.175	-.034
.200	.101	-.037	.170	-.009	.213	.012	.246	.025	.305	.075
.300	.122	.022	.221	.049	.284	.071	.326	.087	.409	.138
.400	-.129	.047	.099	.112	.200	.150	.280	.175	.418	.246
.500	-.450	-.154	-.235	-.001	-.126	.083	-.038	.138	.133	.250
.600	-.329	-.292	-.214	-.135	-.147	-.042	-.092	.016	.025	.133
.700	-.288	-.292	-.197	-.164	-.143	-.097	-.096	-.046	-.009	.062
.800	-.237	-.250	-.151	-.160	-.109	-.105	-.071	-.063	.008	.033
.900	-.187	-.196	-.105	-.126	-.059	-.084	-.030	-.050	.046	.029
1.000	-.166		-.093		-.059		-.021		.046	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.297	.159	.291	.137	.287	.129	.283	.129	.262	.104
.025	.326	.142	.345	.132	.354	.133	.354	.129	.329	.129
.050	.330	.122	.429	.132	.467	.146	.488	.158	.467	.171
.100	-.216	.122	.174	.207	.367	.229	.529	.254	.713	.292
.150	.126	-.266	.270	-.034	.100	.041	.141	.121	.141	.225
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.012	-.041	.003	-.034	.008	-.042	.025	-.038	.054	-.021
.025	-.024	-.066	-.001	-.059	.008	-.063	.025	-.063	.070	-.030
.050	-.054	-.095	-.009	-.076	.012	-.076	.037	-.063	.091	-.026
.100	-.045	-.091	.020	-.064	.050	-.050	.091	-.038	.158	.004
.200	.030	-.012	.112	.016	.158	.033	.196	.046	.267	.083
.300	-.016	.051	.128	.087	.216	.104	.279	.129	.379	.166
.400	-.429	-.020	-.155	.099	-.025	.141	.079	.192	.258	.267
.500	-.383	-.233	-.185	-.072	-.117	.016	-.046	.071	.079	.175
.600	-.212	-.241	-.147	-.114	-.096	-.046	-.063	.004	.020	.087
.700	-.170	-.204	-.118	-.118	-.084	-.071	-.050	-.030	-.001	.041
.800	-.120	-.158	-.068	-.105	-.042	-.071	-.009	-.038	.029	.008
.900	-.066	-.079	-.022	-.043	.008	-.021	.033	.004	.058	.041
1.000	-.016		.020		.041		.066		.083	

TABLE 30.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(c) $M = 0.775$

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.306	.126	.278	.093			.249	.084	.205	.073
.025	.278	.054	.278	.045			.245	.040	.229	.069
.050	.422	-.002	.454	.005			.445	.020	.410	.069
.100	.346	.030	.554	.109			.690	.169	.751	.225
.150	-.619	-.235	-.099	.013			.321	.109	.554	.225
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.026	.026	.033	.021			.044	.024	.073	.037
.025	.050	-.030	.061	-.023			.088	-.020	.125	.005
.050	.022	-.090	.057	-.079			.093	-.060	.137	-.035
.100	.014	-.123	.069	-.107			.125	-.064	.185	-.031
.200	.118	-.030	.181	.001			.253	.040	.317	.085
.300	.134	.026	.238	.061			.329	.101	.418	.145
.400	-.123	.050	.097	.121			.273	.193	.410	.233
.500	-.491	-.163	-.256	.001			-.044	.141	.113	.233
.600	-.339	-.295	-.224	-.127			-.100	.008	.005	.113
.700	-.303	-.303	-.204	-.175			-.108	-.056	-.031	.041
.800	-.255	-.267	-.159	-.163			-.084	-.072	-.019	-.003
.900	-.199	-.203	-.115	-.135			-.036	-.056	.017	.001
1.000	-.179		-.107				-.032		.017	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.310	.166	.305	.145			.300	.128	.277	.129
.025	.338	.134	.365	.133			.372	.124	.353	.133
.050	.350	.118	.445	.145			.496	.140	.501	.169
.100	-.162	.122	.209	.209			.516	.228	.685	.277
.150	.210	-.310	.589	-.075			.220	.116	.217	.189
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.010	-.042	.009	-.039			.032	-.028	.061	-.015
.025	-.030	-.066	.001	-.059			.040	-.052	.077	-.031
.050	-.054	-.098	-.007	-.083			.052	-.056	.101	-.043
.100	-.042	-.090	.021	-.063			.100	-.028	.161	-.003
.200	.038	-.006	.121	.021			.204	.056	.269	.085
.300	.006	.050	.137	.093			.276	.140	.365	.169
.400	-.462	-.018	-.187	.101			.056	.196	.217	.249
.500	-.438	-.250	-.195	-.075			-.060	.072	.033	.141
.600	-.218	-.246	-.151	-.115			-.064	-.004	-.019	.057
.700	-.170	-.218	-.123	-.123			-.052	-.032	-.039	.005
.800	-.126	-.162	-.071	-.107			-.020	-.044	-.019	-.047
.900	-.070	-.078	-.019	-.043			.032	.008	-.003	-.023
1.000	-.618		.017				.068		.005	

TABLE 30.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Continued

(d) M = 0.800

x/c	C _p at -									
	α = -2°		α = 0°		α = 1°		α = 2°		α = 4°	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.316	.128	.288	.106	.276	.095	.262	.081	.224	.097
.025	.309	.054	.322	.056	.276	.037	.281	.062	.278	.055
.050	.444	-.023	.465	-.002	.469	.033	.466	.031	.425	.043
.100	.359	.020	.558	.091	.642	.129	.690	.162	.756	.221
.150	-.574	-.235	-.698	-.017	.122	.029	.281	.089	.517	.217
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.035	.031	.041	.029	.044	.033	.062	.042	.078	.047
.025	.062	-.027	.075	-.025	.087	-.013	.100	-.000	.128	.016
.050	.035	-.084	.064	-.075	.083	-.060	.108	-.050	.143	-.022
.100	.027	-.119	.079	-.102	.110	-.067	.139	-.058	.190	-.019
.200	.128	-.027	.195	.002	.230	.033	.258	.050	.321	.082
.300	.155	.031	.245	.060	.299	.087	.343	.112	.413	.147
.400	-.092	.062	.102	.118	.199	.160	.278	.193	.398	.236
.500	-.616	-.169	-.287	-.013	-.160	.068	-.077	.119	.082	.228
.600	-.343	-.312	-.241	-.152	-.183	-.071	-.112	-.004	-.026	.101
.700	-.316	-.323	-.225	-.194	-.168	-.121	-.116	-.066	-.057	.016
.800	-.273	-.293	-.183	-.191	-.141	-.133	-.093	-.085	-.042	-.026
.900	-.212	-.223	-.137	-.152	-.087	-.110	-.046	-.070	-.011	-.030
1.000	-.192		-.121		-.083		-.050		-.026	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.323	.166	.318	.156	.318	.141	.312	.138	.289	.132
.025	.354	.146	.375	.145	.383	.125	.381	.135	.370	.139
.050	.377	.116	.452	.133	.487	.133	.508	.138	.512	.143
.100	-.165	.093	.171	.171	.352	.221	.481	.238	.662	.266
.150	.316	-.265	.749	-.071	.256	-.029	.308	.081	.312	.151
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.004	-.038	.010	-.040	.025	-.036	.042	-.027	.066	-.007
.025	-.023	-.073	.002	-.067	.025	-.056	.042	-.046	.078	-.026
.050	-.042	-.104	-.002	-.086	.029	-.063	.054	-.058	.105	-.026
.100	-.038	-.096	.029	-.067	.071	-.044	.108	-.027	.158	.005
.200	.046	-.007	.125	.021	.160	.040	.204	.054	.266	.085
.300	.016	.046	.137	.087	.210	.121	.269	.138	.359	.170
.400	-.511	-.011	-.229	.095	-.071	.144	.034	.188	.189	.228
.500	-.507	-.269	-.221	-.086	-.148	-.010	-.085	.038	.001	.120
.600	-.253	-.265	-.163	-.136	-.125	-.075	-.085	-.027	-.053	.035
.700	-.173	-.230	-.136	-.140	-.098	-.086	-.077	-.058	-.076	-.026
.800	-.127	-.177	-.082	-.125	-.060	-.090	-.042	-.073	-.053	-.076
.900	-.073	-.088	-.032	-.056	-.002	-.025	.008	-.023	-.034	-.061
1.000	-.023		.010		.037		.027		-.042	

TABLE 30.- PYLON SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 9 - Concluded

(e) $M = 0.825$

x/c	C_p at -									
	$\alpha = -2^\circ$		$\alpha = 0^\circ$		$\alpha = 1^\circ$		$\alpha = 2^\circ$		$\alpha = 4^\circ$	
Inboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.336	.139	.305	.119	.299	.117	.275	.108	.249	.100
.025	.343	.046	.372	.048	.325	.046	.327	.055	.334	.063
.050	.462	-.014	.484	.011	.485	.028	.479	.033	.450	.067
.100	.406	.012	.562	.089	.649	.113	.691	.156	.766	.201
.150	-.519	-.240	-.097	-.015	.102	.028	.256	.063	.528	.208
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	.046	.042	.052	.045	.065	.050	.070	.048	.097	.067
.025	.072	-.017	.089	-.007	.106	-.002	.111	.007	.145	.030
.050	.046	-.081	.078	-.059	.098	-.050	.119	-.041	.160	-.008
.100	.046	-.114	.093	-.082	.124	-.062	.148	-.052	.208	-.011
.200	.154	-.014	.208	.015	.251	.035	.275	.055	.334	.097
.300	.176	.042	.260	.074	.311	.098	.345	.108	.416	.156
.400	-.069	.072	.112	.130	.199	.162	.260	.186	.401	.249
.500	-.679	-.166	-.305	-.019	-.196	.050	-.101	.119	.071	.204
.600	-.311	-.322	-.245	-.156	-.192	-.084	-.142	-.026	-.034	.082
.700	-.344	-.352	-.242	-.208	-.196	-.144	-.149	-.090	-.071	.004
.800	-.307	-.318	-.201	-.208	-.158	-.158	-.127	-.123	-.060	-.045
.900	-.240	-.251	-.152	-.178	-.114	-.132	-.090	-.101	-.034	-.052
1.000	-.211	-.134	-.134	-.110	-.110	-.082	-.082	-.041	-.041	
Outboard station										
	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F	Row D	Row F
.010	.339	.168	.330	.152	.332	.154	.326	.144	.311	.133
.025	.372	.138	.386	.148	.395	.139	.400	.141	.386	.148
.050	.391	.105	.460	.145	.492	.135	.512	.155	.530	.163
.100	-.106	.087	.167	.182	.328	.224	.449	.237	.645	.271
.150	.431	-.303	.924	-.093	.425	-.039	.411	.040	.423	.141
	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G	Row E	Row G
.010	-.002	-.043	.015	-.037	.035	-.028	.044	-.023	.078	-.004
.025	-.021	-.080	.011	-.059	.035	-.058	.044	-.041	.093	-.019
.050	-.040	-.106	.004	-.078	.035	-.073	.059	-.056	.115	-.022
.100	-.025	-.099	.037	-.067	.080	-.043	.107	-.030	.170	.011
.200	.064	-.006	.130	.019	.172	.046	.207	.052	.274	.089
.300	.027	.057	.141	.093	.206	.120	.259	.130	.348	.174
.400	-.536	-.028	-.275	.089	-.128	.135	-.023	.163	.167	.234
.500	-.559	-.307	-.245	-.104	-.184	-.039	-.130	.015	-.022	.104
.600	-.358	-.295	-.178	-.152	-.147	-.099	-.119	-.060	-.067	.015
.700	-.177	-.258	-.156	-.163	-.132	-.128	-.115	-.097	-.093	-.045
.800	-.132	-.195	-.108	-.145	-.091	-.132	-.082	-.126	-.074	-.100
.900	-.084	-.103	-.059	-.082	-.047	-.065	-.049	-.075	-.059	-.085
1.000	-.040	-.011	-.011	-.021	-.021	-.041	-.041	-.063	-.063	

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION I

(a) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.5$; inboard station

x/c	C _p at -														
	α = -2°						α = 0°						α = 1°		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	1.301	.525	.784	.556	1.678	1.229	.549	.995	.993	1.107	.533	1.093	1.014	.856
	.003	1.140	-.736	-.1405	-.474	-.012	-.524	-.734	-.648	-.387	-.859	-.731	1.014	.322	-.460
	.014	-.238	-.492	-.707	-.404	-.166	-.423	-.506	-.567	-.329	-.485	-.480	1.014	.322	-.370
	.031	-.228	-.386	-.935	-.320	-.154	-.331	-.385	-.958	-.256	-.439	-.423	1.014	.322	-.268
	.055	-.233	-.328	-.405	-.279	-.163	-.282	-.321	-.321	-.237	-.269	-.355	1.014	.322	-.216
	.106	-.189	-.254	-.286	-.195	-.122	-.200	-.244	-.244	-.154	-.189	-.217	1.014	.322	-.111
	.172	-.087	-.151	-.135	-.083	-.035	-.089	-.138	-.103	-.042	-.078	-.111	1.014	.322	-.002
	.261	-.170	-.273	-.202	-.186	-.102	-.152	-.250	-.183	-.119	-.131	-.217	1.014	.322	-.002
	.302	-.145	-.087	-.093	-.061	-.035	-.128	-.064	-.071	-.006	-.102	-.037	1.014	.322	-.059
	.326	-.116	-.016	-.100	.003	-.016	-.176	.009	-.058	.034	.024	.027	1.014	.322	-.081
	.343	-.301	.100	-.302	.051	-.157	-.282	.182	-.269	.102	-.267	.207	1.014	.322	-.107
	.387	-.446	-.002	-.003	-.038	-.224	-.442	.019	.041	.048	-.422	.095	1.014	.322	-.047
	.442	-.068	-.161	-.096	-.166	-.074	.050	-.116	-.058	.016	-.138	.069	1.014	.322	-.184
Turbine cowl	.493	.073	.055	-.077	-.147	-.058	.090	.045	-.019	-.026	.086	.072	1.014	.322	-.155
	.537	-.068	-.019	-.074	-.224	-.641	-.132	.045	-.026	-.039	.033	.072	1.014	.322	-.050
	.581	.093	-.125	-.192	-.099	-.516	-.108	-.112	-.119	-.077	.233	.005	1.014	.322	-.296
	.617	-.121	-.016	-.112	-.378	-.359	-.026	.038	-.026	-.064	.066	.024	1.014	.322	-.069
	.653	-.127	-.122	-.080	-.157	-.388	.187	.089	-.010	-.134	.036	.082	1.014	.322	-.050
	.690	-.189	-.010	-.048	-.253	-.029	.008	.057	.045	-.176	.029	.133	1.014	.322	-.139
	.708		-.054	.225	-.003			.070	.311	.121	.101	.130	1.014	.322	-.359
	.735		-.022		-.150			-.019		.419	.002	.342	1.014	.322	-.357
	.768		-.565		-.557			-.311		-.479	-.169	.357	1.014	.322	-.098
	.796		-.202		-.163			-.170		-.138	-.098	.114	1.014	.322	-.136
	.821		-.005		.051			.067		.095	.114	.098	1.014	.322	-.098
	.852		.286		.311			.355		.393	.384	.430	1.014	.322	-.430
Fan cowl	0.000	1.035	.547	1.164	1.614	.741	1.229	.549	1.215	1.644	1.107	.533	1.093	1.014	.856
	.003	1.137	-.699	-.182	-.300	-.780	-.545	-.715	-.229	-.122	-.859	-.731	1.093	.407	-.460
	.014	-.579	-.484	-.320	-.294	-.444	-.658	-.471	-.607	-.201	-.485	-.480	1.093	.407	-.370
	.031	-.614	-.356	-.995	-.210	-.322	-.450	-.352	-.747	-.134	-.439	-.423	1.093	.407	-.268
	.055	-.322	-.258	-.214	-.191	-.242	-.334	-.321	-.108	-.137	-.269	-.355	1.093	.407	-.216
	.106	-.201	-.217	-.173	-.114	-.121	-.188	-.214	.039	-.067	-.189	-.217	1.093	.407	-.111
	.172	-.080	-.111	-.047	.004	-.001	-.057	-.105	.001	.036	-.078	-.111	1.093	.407	-.002
	.261	-.128	-.214	-.137	-.086	-.006	-.091	-.192	-.006	.001	-.131	-.146	1.093	.407	-.002
	.302	-.094	-.034	-.028	-.058	.078	-.048	-.049	-.040	.001	-.102	-.037	1.093	.407	-.059
	.326	-.085	.039	-.114	.110	.100	-.086	.005	-.051	.151	.024	.027	1.093	.407	-.081
	.343	-.259	.180	-.156	.161	-.102	-.209	.155	-.051	.181	.038	.034	1.093	.407	-.107
	.387	-.619	.084	.084	.115	.004	-.377	.116	.129	.213	-.078	.075	1.093	.407	-.047
	.442	.134	-.002	.011	.142	.215	.166	.035	.056	.122	.128	.066	1.093	.407	-.184
Turbine cowl	.493	.159	.057	.049	.138	.231	.209	.107	.037	.154	.388	.084	1.093	.407	-.155
	.537	-.056	.107	.043	.030	-.162	-.049	.116	.097	.122	.100	.072	1.093	.407	-.050
	.581	.003	.030	-.038	-.140	-.017	-.057	.091	.036	.077	.004	.072	1.093	.407	-.296
	.617	.041	.091	.043	.135	.017	.127	.152	.097	.074	.193	.122	1.093	.407	-.069
	.653	.211	.104	.065	-.075	-.002	.267	.184	.126	.074	.170	.122	1.093	.407	-.050
	.690	.075	.149	.100	.113	-.022	.180	.229	.184	-.028	.070	.122	1.093	.407	-.139
	.708		.123	.348	.110			.190	.379	.212			1.093	.407	-.359
	.735		-.006		-.025			.010		-.374			1.093	.407	-.357
	.768		-.121		-.210			.001		-.083			1.093	.407	-.098
	.796		-.067		-.105			.030		-.247			1.093	.407	-.136
	.821		.145		.170			.216		.247			1.093	.407	-.098
	.852		.356		.439			.444		.503			1.093	.407	-.430

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(b) $M = 0.700$; $P_t/P_{t,\infty} = 1.5$; outboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.765	.834	.718	.563	1.030	.504	.789	.889	.581	.982	.332
	.003	.169	.567	.360	.771	.332	.538	.627	.194	.714	.488	.597
	.014	.315	.538	.641	.577	.384	.499	.559	.491	.547	.550	.526
	.031	.925	.447	.716	.461	.387	1.049	.452	.598	.425	.486	.523
	.055	.271	.055	.460	.429	.316	.330	.052	.375	.396	.232	.357
	.106	.276	.268	.327	.292	.380	.610	.264	.291	.296	.248	.264
	.172	.126	.606	.165	.190	.151	.137	.262	.139	.183	.103	.060
	.261	.227	.269	.197	.258	.210	.214	.010	.178	.206	.131	.116
	.302	.150	.103	.311	.102	.167	.132	.087	.291	.145	.068	.189
	.326	.111	.032	.022	.180	.057	.013	.000	.000	.077	.066	.107
Turbine cowl	.343	.029	.017	.138	.036	.007	.021	.681	.136	.058	.006	.008
	.387	.049	.009	.003	.133	.103	.080	.007	.029	.055	.026	.005
	.422	.111	.016	.074	.245	.184	.078	.007	.045	.048	.064	.018
	.493	.029	.029	.058	.004	.052	.018	.029	.013	.007	.055	.063
	.537	.029	.064	.042	.345	.219	.018	.016	.013	.090	.003	.024
	.591	.072	.087	.138	.196	.367	.690	.036	.058	.222	.164	.008
	.617	.077	.043	.045	.280	.626	.618	.026	.006	.083	.363	.057
	.653	.184	.160	.058	.335	.361	.071	.058	.006	.083	.145	.058
	.690	.213	.090	.016	.158	.448	.007	.081	.010	.254	.325	.070
	.708		.101	.295	.535		.216	.039	.339	.437		.303
Plug	.735		.048	.025	.058		.010			.058		.050
	.768		.259		.622		.152			.399		.108
	.796		.142		.048		.074			.090		.015
	.821		.049		.078		.100			.125		.141
	.852		.318		.368		.339			.415		.380
Fan cowl	0.000	.232	.765	1.059	.641	.842	.607	.719	1.113	.719	.727	.453
	.003	.104	.585	.286	.598	.103	.135	.623	.195	.444	.448	.433
	.014	.641	.533	.264	.482	.708	.758	.536	.041	.392	.701	.488
	.031	.776	.416	.368	.363	.550	.860	.413	.193	.328	.572	.488
	.055	.370	.063	.274	.357	.386	.391	.088	.167	.360	.379	.369
	.106	.646	.268	.225	.215	.219	.666	.258	.154	.180	.192	.366
	.172	.123	.222	.093	.118	.073	.106	.358	.041	.063	.038	.247
	.261	.186	.076	.145	.147	.083	.144	.154	.099	.102	.015	.169
	.302	.109	.067	.313	.004	.004	.072	.051	.238	.052	.068	.111
	.326	.075	.068	.033	.028	.030	.020	.020	.072	.062	.101	.021
Turbine cowl	.343	.017	.076	.086	.069	.017	.004	.075	.031	.097	.039	.062
	.387	.104	.020	.063	.107	.152	.151	.056	.104	.175	.239	.058
	.442	.085	.212	.008	.053	.056	.035	.049	.049	.162	.162	.057
	.493	.080	.037	.037	.085	.117	.146	.114	.082	.216	.265	.307
	.537	.012	.043	.040	.091	.185	.126	.111	.088	.433	.507	.488
	.581	.032	.032	.001	.067	.102	.062	.117	.062	.421	.482	.488
	.617	.066	.079	.069	.036	.118	.136	.153	.117	.823	.866	.866
	.653	.085	.079	.059	.011	.159	.170	.159	.111	.868	.833	.833
	.690	.003	.043	.095	.034	.167	.107	.133	.153	.854	.814	.814
	.708		.328	.383	.701			.370	.421	.843		
Plug	.735		.052	.052	.209			.443		.846		
	.768		.007		.234			.836		.072		
	.796		.011		.011			.888		.218		
	.821		.140		.198			.828		.258		
	.852		.386		.462			.421		.509		

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(c) $M = 0.750$; $p_{t,e}/p_{t,\infty} \approx 1.5$; inboard station

C _p at -																
x/c	α = -2°					α = 0°					α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	1.299	.520	.795	.893	1.057	1.175	.495	.979	.946	.907	1.103	.507	1.057	.572	.828
	.003	-1.245	-.805	-1.456	-.560	-1.113	-.763	-.843	-.890	-.476	-.389	-1.042	-.826	-.515	-.414	-.642
	.014	-.276	-.544	-.902	-.434	-.244	-.467	-.539	-.603	-.389	-.342	-.542	-.530	-.462	-.338	-.408
	.031	-.263	-.427	-.799	-.341	-.160	-.361	-.419	-.508	-.290	-.255	-.405	-.401	-.615	-.251	-.303
	.055	-.250	-.354	-.618	-.297	-.177	-.295	-.334	-.346	-.246	-.214	-.397	-.325	-.269	-.213	-.224
	.106	-.192	-.263	-.428	-.201	-.122	-.198	-.244	-.244	-.150	-.124	-.214	-.228	-.207	-.122	-.110
	.172	-.082	-.149	-.128	-.078	-.028	-.079	-.127	-.095	-.027	-.007	-.077	-.111	-.064	-.004	.009
	.261	-.166	-.275	-.207	-.183	-.053	-.150	-.241	-.176	-.112	-.042	-.135	-.216	-.152	-.072	-.011
	.302	-.139	-.087	-.087	-.060	-.020	-.119	-.051	-.060	-.002	-.095	-.034	-.034	-.034	.056	.071
	.326	-.034	-.011	-.256	.001	.004	-.061	.025	.075	.075	.066	.184	.045	.370	.109	.100
	.343	-.254	.103	-.263	.118	-.116	-.361	.157	-.241	.162	-.232	.194	-.240	.182	.067	.055
	.387	-.374	.019	-.019	.007	.004	-.107	.019	.049	.066	.369	.053	.086	.097	.055	.082
Turbine cowl	.442	-.056	-.020	-.128	-.186	-.045	-.040	.019	.057	.024	.078	.024	.023	.018	.082	.199
	.493	.056	.001	-.108	.077	.004	.107	.019	.019	.001	.220	.015	.018	.077	.009	.199
	.537	.056	-.075	-.134	.297	-.618	-.077	-.024	-.024	-.051	.175	.048	-.002	-.012	-.280	.401
	.581	.259	-.011	-.174	-.274	-.385	-.040	-.013	-.129	-.019	.401	.036	-.064	-.055	-.204	.067
	.617	.135	-.075	-.031	.326	-.385	-.030	.110	-.092	-.287	.055	.133	.201	-.113	-.067	.084
	.653	.113	-.053	-.108	.221	-.452	.173	.028	-.060	-.086	.167	.048	.021	-.134	-.084	.287
	.690	-.236	-.137	-.087	-.376	.118	.001	.154	.025	-.202	-.068	.007	.124	.074	-.166	.287
	.708		.019	.230	-.008			.028	.280			.115	.349		.158	
	.735		-.617		-.560			-.013		-.456		.002		.024		
	.768		-.538		-.499			-.416		-.287		.330		.026		
	.796		-.647		-.858			-.100		-.168		.090		.046		
	.821		.074		.112			.084				.112		.170		
.852		.273		.319			.318		.360		.384		.410			
Fan cowl	0.000	1.057	.520	1.142	.998	.749	.933	.514	1.234	1.051	.619					
	.003	-1.235	-.821	-.247	-.340	-.886	-1.531	-.831	-.176	-.239	-1.330					
	.014	-.547	-.522	-.541	-.302	-.455	-.801	-.324	-.117	-.233	-.581					
	.031	-.430	-.396	-.848	-.232	-.322	-.630	-.378	-.108	-.158	-.359					
	.055	-.319	-.302	-.826	-.191	-.237	-.324	-.296	-.120	-.146	-.242					
	.106	-.186	-.215	-.476	-.100	-.093	-.174	-.208	-.111	-.064	-.085					
	.172	-.062	-.094	-.039	-.001	-.023	-.046	-.091	.003	.006	.073					
	.261	-.080	-.021	-.130	-.044	.014	-.076	-.091	-.094	.006	.163					
	.302	-.049	.058	-.021	.081	.105	-.037	-.006	.015	.128	.163					
	.326	-.217	.016	-.217	.198	-.053	.191	.067	.061	.222	.015					
	.343	-.363	.081	.093	.142	-.014	-.337	.088	.138	.190	.122					
	.387	-.101	.020	.014	.078	.189	.145	-.026	.064	.187	.321					
Turbine cowl	.442	.493	.159	.096	.052	.204	.184	.167	.094	.225	.315					
	.493	.337	.061	.096	.058	.115	.038	.138	.091	.158	.032					
	.581	.163	.040	-.044	.005	-.012	.065	.041	.012	-.047	.093					
	.617	.066	.114	.038	.001	.017	.136	.149	.105	.254	.163					
	.653	.208	.102	.061	-.027	-.001	.259	.170	.120	-.009	.137					
	.690	.066	.125	.096	-.100	.116	.167	.228	.164	.190	.041					
	.708		.152	.363	.166			.193	.386	.160						
	.735		.002		.331			.006		.243						
	.768		-.138		-.337			-.050		-.204						
	.796		-.062		-.036			.023		.003						
	.821		.143		.169			.205		.236						
	.852		.386		.435			.427		.490						

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(d) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.5$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°											
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row K	Row L	
Fan cowl	0.000	.704	.736	.697	.542	1.078	.466	.761	.853	.566	.337	.746	.958	.584	.909			
	.003	-.267	-.720	.485	-.910	-.434	-.700	-.729	.309	-.822	-.1031	-.729	.589	-.128	-.899			
	.014	-.368	-.619	.658	-.455	-.455	-.581	-.631	.540	-.599	-.603	-.619	.589	-.552	-.702			
	.031	-.695	-.507	.528	-.438	-.438	-.907	.496	.452	-.526	-.280	-.472	.563	-.431	-.564			
	.055	-.298	-.051	.489	-.452	-.452	.356	.855	.413	-.420	-.360	.057	.348	-.390	-.390			
	.106	-.108	-.312	.347	.311	.255	.431	.293	.301	-.265	-.117	-.283	.271	-.237	-.231			
	.172	-.121	.166	.173	.156	.135	.136	-.098	.136	-.118	-.117	.240	.114	-.164	-.082			
	.261	-.236	.466	.209	.272	.205	.250	-.080	.181	-.209	-.145	.577	.167	-.173	-.108			
	.302	-.118	.105	.194	.108	.114	.152	-.001	.011	-.039	-.068	.108	.057	-.253	-.017			
	.326	-.108	.057	.023	.117	.075	.092	.001	.011	-.036	-.067	.010	.027	-.011	-.021			
Turbine cowl	.343	.068	.087	.103	.130	.072	.009	.108	.086	.149	.016	.113	.082	.151	.080			
	.387	-.059	.013	.017	.034	.135	.030	.013	.029	.046	.002	.094	.043	.082	-.040			
	.442	-.154	.154	.079	.146	.008	.132	.011	.001	.172	.091	.039	.045	.183	-.024			
	.495	-.086	.087	.049	.067	.111	.017	.024	.013	.186	.042	.069	.045	.134	.024			
	.537	-.050	.099	.088	.125	.364	.059	.017	.057	.183	.055	.019	.046	.093	-.090			
	.581	-.059	.086	.057	.001	.605	.018	.035	.017	.139	.060	.134	.016	.043	-.238			
	.617	-.134	.037	.001	.275	.059	.018	.035	.002	.350	.016	.054	.030	.034	-.076			
	.653	-.117	.117	.076	.143	.332	.004	-.042	.002	.048	-.042	.019	.069	-.246	-.258			
	.690	-.254	.173	.017	.211	.458	.008	.093	.082	.284	.067	.284	.382	.128				
	.708	.086	.086	.291	.596	.004	.185	.055	.335	.007	.078	.060	.060	.048				
Plug	.735	-.017	.004	.004	.004	.004	.055	.055	.335	.007	.078	.060	.060	.048				
	.768	-.409	.017	.004	.004	.004	.325	.042	.325	.032	-.114	-.032	.178	.171				
	.796	-.158	.158	.317	.317	.317	.042	.042	.042	.178	-.032	.140	.140	.433				
	.821	.081	.157	.157	.157	.157	.105	.105	.105	.370	.140	.171	.433					
	.852	.286	.345	.345	.345	.345	.344	.344	.344	.398	.370	.433						
	Fan cowl	0.000	.252	.732	1.029	.619	.838	.649	.676	1.127	.676	.740	.958	.584	.909			
		.003	-.109	-.774	.410	-.656	-.106	-.106	-.776	.283	-.502	-.139	.584	.584	.584			
		.014	-.636	-.621	.314	.521	-.644	-.311	.613	.082	-.435	-.139	.584	.584	.584			
		.031	-.566	-.461	.414	.403	-.574	.716	.454	.239	.329	.561	.584	.584	.584			
		.055	-.376	.065	.299	.371	.385	.381	.071	.189	.329	.382	.584	.584	.584			
.106		-.424	.272	.240	.242	.212	.504	.265	.171	.179	.188	.584	.584	.584				
.172		-.120	.042	.089	.103	.059	.098	.215	.047	.082	.026	.584	.584	.584				
.261		-.182	.262	.145	.145	.068	.142	.009	.109	.100	.012	.584	.584	.584				
.302		-.093	.057	.189	.020	.017	.059	.044	.121	.059	.079	.584	.584	.584				
.326		.030	.030	.041	.035	.052	.028	.027	.071	.053	.109	.584	.584	.584				
Turbine cowl	.343	.030	.168	.083	.152	.085	.030	.160	.053	.156	.088	.584	.584	.584				
	.387	.026	.029	.070	.008	.051	.078	.062	.098	.057	.206	.584	.584	.584				
	.442	-.089	-.062	.008	.009	.026	-.050	.009	.044	.147	.138	.584	.584	.584				
	.495	.083	.100	.053	.082	.132	.100	.086	.077	.147	.220	.584	.584	.584				
	.537	.088	.100	.053	.082	.132	.127	.098	.083	.109	.320	.584	.584	.584				
	.581	.017	.059	.015	.139	.002	.131	.074	.044	.012	.332	.584	.584	.584				
	.617	.083	.124	.076	.073	.124	.144	.142	.112	.012	.332	.584	.584	.584				
	.653	.057	.100	.050	.179	.006	.153	.157	.101	.126	.087	.584	.584	.584				
	.690	-.001	.035	.106	.209	-.183	.082	.351	.145	.008	-.029	.584	.584	.584				
	.708	.735	.100	.392	.035	.035	.351	.351	.410	-.846	.008	.584	.584	.584				
Plug	.735	.735	.100	.392	.035	.035	.351	.351	.410	-.846	.008	.584	.584	.584				
	.768	.796	.008	.008	.338	.008	.008	.008	.008	.008	.008	.008	.008	.008				
	.796	.821	.008	.008	.338	.008	.008	.008	.008	.008	.008	.008	.008	.008				
	.821	.852	.155	.155	.168	.168	.168	.168	.168	.168	.168	.168	.168	.168				
	.852	.852	.384	.449	.449	.449	.449	.449	.449	.449	.449	.449	.449	.449				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(e) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	1.298	.506	.789	.902	1.059	1.197	.520	.971	.946	1.096	.499	1.059	.977	.839			
	.003	.013	.067	1.357	.606	.119	.793	.900	.998	.506	.436	.884	.591	.446	.704			
	.014	.013	.063	1.162	.463	.248	.675	.555	.620	.403	.536	.538	.490	.351	.418			
	.031	.013	.037	.842	.337	.169	.574	.443	.721	.299	.412	.420	.431	.256	.595			
	.033	.013	.030	.614	.295	.177	.598	.348	.348	.252	.310	.321	.276	.208	.222			
	.106	.013	.237	.285	.192	.113	.196	.121	.244	.148	.120	.187	.220	.206	.115			
	.172	.013	.136	.119	.065	.015	.073	.121	.090	.020	.064	.099	.057	.011	.006			
	.261	.013	.265	.201	.172	.085	.145	.238	.177	.106	.124	.214	.147	.065	.003			
	.302	.013	.077	.037	.037	.001	.111	.048	.053	.005	.037	.029	.067	.084	.006			
	.326	.013	.008	.390	.024	.024	.015	.033	.154	.081	.075	.348	.056	.514	.115			
	.343	.013	.148	.203	.156	.088	.213	.176	.202	.184	.204	.205	.197	.210	.043			
	.387	.015	.036	.008	.016	.180	.323	.059	.073	.092	.319	.076	.081	.120	.043			
Turbine cowl	.442	.015	.035	.068	.079	.125	.104	.034	.067	.176	.046	.031	.037	.023	.072			
	.493	.015	.143	.036	.054	.021	.037	.025	.087	.173	.170	.078	.031	.047	.277			
	.537	.015	.085	.088	.189	.031	.117	.081	.120	.147	.318	.025	.084	.042	.334			
	.581	.015	.009	.214	.438	.292	.035	.042	.005	.243	.552	.039	.054	.042	.143			
	.617	.015	.144	.005	.320	.489	.035	.031	.059	.146	.068	.177	.017	.228	.087			
	.653	.015	.108	.215	.270	.561	.164	.025	.042	.176	.187	.199	.067	.169	.113			
	.690	.015	.085	.037	.452	.237	.048	.025	.052	.176	.029	.057	.169	.406	.006			
	.708	.009	.008	.171	.002	.027	.002	.112	.305	.411	.005	.078	.332	.650	.006			
	.735	.455	.455	.290	.511	.009	.011	.011	.305	.411	.005	.078	.332	.650	.006			
	.768	.726	.726	.783	.290	.398	.398	.454	.398	.363	.363	.363	.363	.363	.363			
	.796	.016	.016	.001	.783	.783	.726	.726	.726	.726	.726	.726	.726	.726	.726			
	.821	.257	.257	.001	.001	.001	.126	.016	.126	.187	.187	.187	.187	.187	.187			
.852	.257	.257	.324	.324	.324	.317	.317	.317	.380	.357	.357	.357	.357	.357				
Fan cowl	0.000	1.046	.519	1.134	.556	.757	.995	.554	1.248	1.071	.655	.977	.839					
	.003	.014	.089	.299	.379	.595	.815	.513	.175	.227	.313	.446	.704					
	.014	.031	.035	.366	.306	.446	.146	.513	.175	.227	.313	.446	.704					
	.031	.038	.307	.706	.222	.312	.373	.355	.716	.137	.329	.256	.418					
	.055	.106	.211	.225	.188	.222	.296	.276	.118	.126	.213	.208	.222					
	.106	.176	.211	.169	.093	.053	.151	.180	.095	.039	.056	.115	.101					
	.172	.049	.085	.034	.033	.039	.015	.082	.026	.082	.082	.082	.082					
	.261	.100	.192	.127	.037	.030	.049	.155	.070	.031	.102	.082	.082					
	.302	.061	.006	.006	.097	.120	.002	.020	.040	.155	.186	.186	.186					
	.326	.172	.070	.278	.142	.142	.091	.099	.144	.195	.217	.217	.217					
	.343	.189	.067	.115	.226	.026	.156	.271	.152	.262	.324	.324	.324					
	.387	.308	.087	.115	.442	.019	.288	.111	.153	.240	.147	.178	.324					
Turbine cowl	.442	.074	.061	.025	.030	.142	.288	.111	.153	.240	.147	.178	.324					
	.493	.168	.075	.098	.050	.249	.228	.119	.153	.279	.065	.085	.063					
	.537	.181	.053	.061	.075	.180	.104	.167	.096	.085	.063	.085	.063					
	.581	.291	.050	.054	.023	.085	.147	.099	.043	.065	.178	.175	.175					
	.617	.100	.082	.025	.031	.022	.151	.175	.094	.152	.175	.175	.175					
	.653	.319	.087	.047	.080	.012	.275	.190	.144	.136	.156	.156	.156					
	.690	.074	.158	.106	.124	.223	.172	.221	.181	.048	.121	.121	.121					
	.708	.154	.373	.373	.175	.223	.172	.221	.181	.048	.121	.121	.121					
	.735	.011	.011	.011	.270	.270	.037	.037	.449	.254	.199	.199	.199					
	.768	.256	.256	.256	.354	.354	.016	.016	.267	.267	.267	.267	.267					
	.796	.046	.046	.046	.017	.017	.026	.026	.012	.012	.012	.012	.012					
	.821	.148	.148	.148	.201	.201	.215	.215	.254	.254	.254	.254	.254					
.852	.385	.385	.385	.436	.436	.435	.435	.524	.524	.524	.524	.524						

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(f) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.5$; outboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	-0.692	-0.773	-0.706	-0.530	1.078	-0.464	-0.761	-0.862	-0.565	-0.570	-0.902
	0.003	-0.303	-0.754	-0.591	-0.912	-0.458	-0.766	-0.828	-0.367	-0.633	-0.776	-1.003
	0.014	-0.396	-0.650	-1.100	-0.681	-0.469	-0.584	-0.672	-0.548	-0.633	-0.713	-0.723
	0.031	-0.481	-0.535	-0.630	-0.576	-0.447	-0.833	-0.537	-0.720	-0.509	-0.554	-0.590
	0.055	-0.303	-0.053	-0.488	-0.469	-0.342	-0.293	-0.051	-0.429	-0.333	-0.394	-0.398
	0.106	-0.032	-0.315	-0.358	-0.314	-0.243	-0.293	-0.305	-0.311	-0.309	-0.245	-0.275
	0.172	-0.121	-0.314	-0.168	-0.190	-0.108	-0.128	-0.066	-0.141	-0.149	-0.096	-0.074
	0.261	-0.239	-0.455	-0.210	-0.274	-0.158	-0.217	-0.246	-0.192	-0.149	-0.149	-0.102
	0.302	-0.142	-0.106	-0.148	-0.100	-0.105	-0.082	-0.082	-0.107	-0.037	-0.051	-0.060
	0.326	-0.095	-0.021	-0.012	-0.100	-0.037	-0.082	-0.000	-0.011	-0.023	-0.009	-0.000
	0.343	-0.044	-0.147	-0.052	-0.183	-0.115	-0.037	-0.141	-0.045	-0.188	-0.110	-0.031
	0.387	-0.121	-0.036	-0.011	-0.069	-0.198	-0.090	-0.073	-0.017	-0.045	-0.006	-0.046
Turbine cowl	0.000	-0.442	-0.108	-0.069	-0.072	-0.018	-0.116	-0.136	-0.056	-0.076	-0.006	-0.014
	0.003	-0.493	-0.121	-0.172	-0.291	-0.201	-0.155	-0.113	-0.034	-0.084	-0.028	-0.037
	0.014	-0.537	-0.036	-0.159	-0.029	-0.063	-0.082	-0.062	-0.017	-0.208	-0.014	-0.006
	0.031	-0.581	-0.045	-0.181	-0.441	-0.396	-0.037	-0.017	-0.079	-0.126	-0.357	-0.086
	0.055	-0.617	-0.172	-0.018	-0.345	-0.717	-0.027	-0.124	-0.079	-0.146	-0.442	-0.164
	0.106	-0.653	-0.121	-0.162	-0.229	-0.653	-0.066	-0.042	-0.000	-0.228	-0.304	-0.336
	0.172	-0.231	-0.222	-0.018	-0.382	-0.387	-0.014	-0.042	-0.037	-0.262	-0.279	-0.215
	0.261	-0.708	-0.079	-0.277	-0.638	-0.638	-0.012	-0.203	-0.356	-0.636	-0.636	-0.636
	0.302	-0.735	-0.047	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012
	0.326	-0.768	-0.440	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445
	0.343	-0.821	-0.531	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667
	0.387	-0.852	-0.130	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154
Plug	0.000	-0.266	-0.713	1.024	-0.618	-0.555	-0.499	-0.684	1.125	-0.656	-0.738	-0.420
	0.003	-1.000	-0.868	-0.301	-0.680	-1.160	-1.027	-0.847	-0.356	-0.518	-1.405	-0.367
	0.014	-0.879	-0.658	-0.335	-0.542	-0.824	-1.304	-0.639	-0.088	-0.444	-1.146	-0.223
	0.031	-0.469	-0.453	-0.443	-0.412	-0.387	-0.365	-0.452	-0.244	-0.319	-0.484	-0.011
	0.055	-0.372	-0.053	-0.312	-0.376	-0.387	-0.347	-0.072	-0.193	-0.328	-0.353	-0.006
	0.106	-0.243	-0.273	-0.238	-0.212	-0.204	-0.402	-0.247	-0.170	-0.161	-0.164	-0.002
	0.172	-0.112	-0.066	-0.091	-0.133	-0.051	-0.079	-0.116	-0.034	-0.059	-0.002	-0.006
	0.261	-0.087	-0.046	-0.145	-0.142	-0.037	-0.121	-0.117	-0.096	-0.084	-0.006	-0.006
	0.302	-0.049	-0.034	-0.053	-0.030	-0.033	-0.036	-0.022	-0.045	-0.083	-0.057	-0.057
	0.326	-0.002	-0.032	-0.034	-0.081	-0.064	-0.002	-0.055	-0.094	-0.091	-0.134	-0.134
	0.343	-0.002	-0.053	-0.085	-0.081	-0.118	-0.040	-0.168	-0.014	-0.213	-0.125	-0.125
	0.387	-0.087	-0.077	-0.029	-0.043	-0.073	-0.058	-0.036	-0.049	-0.074	-0.131	-0.131
	0.442	-0.159	-0.053	-0.070	-0.169	-0.239	-0.036	-0.089	-0.129	-0.083	-0.196	-0.196
Fan cowl	0.000	-0.442	-0.108	-0.069	-0.072	-0.018	-0.116	-0.136	-0.056	-0.076	-0.006	-0.014
	0.003	-0.493	-0.121	-0.172	-0.291	-0.201	-0.155	-0.113	-0.034	-0.084	-0.028	-0.037
	0.014	-0.537	-0.036	-0.159	-0.029	-0.063	-0.082	-0.062	-0.017	-0.208	-0.014	-0.006
	0.031	-0.581	-0.045	-0.181	-0.441	-0.396	-0.037	-0.017	-0.079	-0.126	-0.357	-0.086
	0.055	-0.617	-0.172	-0.018	-0.345	-0.717	-0.027	-0.124	-0.079	-0.146	-0.442	-0.164
	0.106	-0.653	-0.121	-0.162	-0.229	-0.653	-0.066	-0.042	-0.000	-0.228	-0.304	-0.336
	0.172	-0.231	-0.222	-0.018	-0.382	-0.387	-0.014	-0.042	-0.037	-0.262	-0.279	-0.215
	0.261	-0.708	-0.079	-0.277	-0.638	-0.638	-0.012	-0.203	-0.356	-0.636	-0.636	-0.636
	0.302	-0.735	-0.047	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012
	0.326	-0.768	-0.440	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445
	0.343	-0.821	-0.531	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667
	0.387	-0.852	-0.130	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154
	0.442	-0.852	-0.130	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154
Turbine cowl	0.000	-0.266	-0.713	1.024	-0.618	-0.555	-0.499	-0.684	1.125	-0.656	-0.738	-0.420
	0.003	-1.000	-0.868	-0.301	-0.680	-1.160	-1.027	-0.847	-0.356	-0.518	-1.405	-0.367
	0.014	-0.879	-0.658	-0.335	-0.542	-0.824	-1.304	-0.639	-0.088	-0.444	-1.146	-0.223
	0.031	-0.469	-0.453	-0.443	-0.412	-0.387	-0.365	-0.452	-0.244	-0.319	-0.484	-0.011
	0.055	-0.372	-0.053	-0.312	-0.376	-0.387	-0.347	-0.072	-0.193	-0.328	-0.353	-0.006
	0.106	-0.243	-0.273	-0.238	-0.212	-0.204	-0.402	-0.247	-0.170	-0.161	-0.164	-0.002
	0.172	-0.112	-0.066	-0.091	-0.133	-0.051	-0.079	-0.116	-0.034	-0.059	-0.002	-0.006
	0.261	-0.087	-0.046	-0.145	-0.142	-0.037	-0.121	-0.117	-0.096	-0.084	-0.006	-0.006
	0.302	-0.049	-0.034	-0.053	-0.030	-0.033	-0.036	-0.022	-0.045	-0.083	-0.057	-0.057
	0.326	-0.002	-0.032	-0.034	-0.081	-0.064	-0.002	-0.055	-0.094	-0.091	-0.134	-0.134
	0.343	-0.002	-0.053	-0.085	-0.081	-0.118	-0.040	-0.168	-0.014	-0.213	-0.125	-0.125
	0.387	-0.087	-0.077	-0.029	-0.043	-0.073	-0.058	-0.036	-0.049	-0.074	-0.131	-0.131
	0.442	-0.159	-0.053	-0.070	-0.169	-0.239	-0.036	-0.089	-0.129	-0.083	-0.196	-0.196
Plug	0.000	-0.442	-0.108	-0.069	-0.072	-0.018	-0.116	-0.136	-0.056	-0.076	-0.006	-0.014
	0.003	-0.493	-0.121	-0.172	-0.291	-0.201	-0.155	-0.113	-0.034	-0.084	-0.028	-0.037
	0.014	-0.537	-0.036	-0.159	-0.029	-0.063	-0.082	-0.062	-0.017	-0.208	-0.014	-0.006
	0.031	-0.581	-0.045	-0.181	-0.441	-0.396	-0.037	-0.017	-0.079	-0.126	-0.357	-0.086
	0.055	-0.617	-0.172	-0.018	-0.345	-0.717	-0.027	-0.124	-0.079	-0.146	-0.442	-0.164
	0.106	-0.653	-0.121	-0.162	-0.229	-0.653	-0.066	-0.042	-0.000	-0.228	-0.304	-0.336
	0.172	-0.231	-0.222	-0.018	-0.382	-0.387	-0.014	-0.042	-0.037	-0.262	-0.279	-0.215
	0.261	-0.708	-0.079	-0.277	-0.638	-0.638	-0.012	-0.203	-0.356	-0.636	-0.636	-0.636
	0.302	-0.735	-0.047	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012	-0.012
	0.326	-0.768	-0.440	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445	-0.445
	0.343	-0.821	-0.531	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667	-0.667
	0.387	-0.852	-0.130	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154
	0.442	-0.852	-0.130	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(g) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

x/c	C_p at -														
	$\alpha = -2^\circ$					$\alpha = 0^\circ$					$\alpha = 1^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127
Turbine cowl	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127
Plug	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127
Fan cowl	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127
Turbine cowl	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127
Plug	0.000	1.315	.536	.832	1.056	1.194	.532	.976	.548	.523	1.116	.524	1.050	.565	.820
	.003	-.015	-.942	1.326	1.123	1.238	-.974	1.041	-.562	-.485	1.021	-.946	1.055	-.455	-.684
	.014	-.015	-.591	1.178	1.247	1.484	-.484	1.007	-.409	-.361	-.490	-.557	-.522	-.361	-.471
	.031	-.015	-.480	1.347	1.365	1.771	-.475	1.007	-.305	-.278	-.404	-.449	-.422	-.267	-.294
	.055	-.015	-.341	1.367	1.295	1.744	-.492	1.007	-.248	-.213	-.310	-.327	-.286	-.213	-.215
	.106	-.015	-.253	1.280	1.190	1.591	-.191	1.007	-.141	-.108	-.179	-.219	-.203	-.110	-.094
	.172	-.015	-.129	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.261	-.015	-.124	1.110	1.058	1.002	-.065	1.007	-.009	-.004	-.053	-.089	-.051	-.025	-.025
	.302	-.015	-.064	1.194	1.126	1.130	-.230	1.007	-.100	-.028	-.014	-.077	-.016	-.021	-.078
	.326	-.015	-.017	.542	.044	1.101	-.039	1.007	-.044	-.037	-.064	-.065	.065	.127	.127

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(h) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.5$; outboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	.695	.780	.715	.552	1.086	.463	.755	.852	.580	.576	.382
	.003	-.337	-.862	.715	-.548	-.446	-.804	-.895	.448	-.891	-.780	-.899
	.014	-.410	-.680	-.1.078	-.691	-.482	-.609	-.683	-.585	-.645	-.628	-.717
	.031	-.243	-.615	-.1.238	-.691	-.474	-.686	-.585	-.585	-.566	-.590	-.534
	.055	-.312	-.553	-.594	-.479	-.357	-.369	-.045	-.444	-.442	-.398	-.362
	.106	-.180	-.323	-.361	-.319	-.246	-.150	-.308	-.319	-.271	-.244	-.288
	.172	-.117	-.486	-.168	-.186	-.105	-.121	-.127	-.139	-.144	-.087	-.097
	.261	-.243	-.565	-.220	-.278	-.202	-.215	.339	.191	-.214	-.144	-.191
	.302	-.146	-.102	-.105	-.097	-.057	-.121	-.080	-.031	-.036	-.039	-.093
	.326	-.089	-.015	-.007	-.072	-.048	-.073	-.007	.016	.018	.002	-.049
	.343	.058	.178	.002	.229	.139	.049	.181	.003	.232	.137	.045
Turbine cowl	.387	-.227	-.002	-.077	-.086	-.281	-.166	-.031	.062	.026	.063	.090
	.442	-.023	-.081	-.040	.120	.167	-.077	-.031	.096	-.004	.167	-.077
	.493	.005	-.004	-.151	-.303	-.205	.131	.217	.067	-.101	.025	.184
	.537	-.048	-.142	-.081	-.058	-.086	-.009	-.134	.112	-.079	.034	.028
	.581	-.068	.083	-.034	-.286	-.338	.053	.075	.109	-.328	.030	-.051
	.617	-.076	-.116	-.149	-.427	-.585	.093	-.099	.089	-.144	.517	.004
	.653	-.235	-.149	-.105	-.406	-.669	.033	.094	.080	-.085	.209	.053
	.690	-.219	-.143	-.121	-.400	-.785	-.109	-.150	.067	-.160	.401	-.044
	.708	-.086	-.086	.287	-.753			.203	.344	-.674		
	.735	.045	.045	.077	.077			.105		.097		
	.768	-.350	-.387	-.387				-.346		-.356		
	.796	-.606	-.601	-.601				-.183		-.515		
Plug	.821	-.037	-.025	-.025				.149		.232		
	.852	.314	.321	.321				.323		.416		
	0.000	-.293	.722	1.001	.625	-.878	-.149	.695	1.125	.649	.799	
	.003	-.089	-.689	.583	-.706	-.839	-.1234	-.922	.407	-.543	1.335	
	.014	-.1.027	-.650	-.355	-.568	-.689	-.444	-.689	-.121	-.471	1.105	
	.031	-.230	-.519	-.481	-.451	-.584	-.297	.056	-.294	.357	1.408	
	.055	-.372	-.052	-.331	-.389	-.395	.302	.258	.189	.181	.357	
	.106	-.071	-.279	-.255	-.221	-.208	-.302	.762	.068	-.087	-.173	
	.172	-.095	.227	-.092	-.099	-.091	.678	.625	.113	-.099	-.002	
	.241	-.177	.471	-.154	-.145	-.044	-.126	.153	.020	.072	.096	
	.302	-.083	.045	-.047	.028	.036	.041	-.026	.083	.128	.131	
	.366	-.043	.036	.052	.146	.071	-.000	.056	.083	.251	.142	
Fan cowl	.387	-.039	.188	.001	.239	.139	.038	.200	.115	.251	.158	
	.442	-.075	.104	.058	.125	.088	-.028	.113	.110	.153	.302	
	.493	.197	.216	.080	.285	.204	.203	-.070	.018	.026	.080	
	.537	-.030	-.024	.042	-.116	.128	.057	.124	.067	.215	.302	
	.581	.108	.031	-.021	-.077	-.010	.114	.086	.015	.226	.099	
	.617	.079	.221	.120	-.081	-.157	.150	.023	.023	.040	-.021	
	.653	.043	-.021	.050	-.208	-.137	.106	.083	.113	.123	.061	
	.690	-.002	.188	.140	-.026	-.173	.073	.325	.413	-.145	-.115	
	.708		.270	.363	-.047			.156		.134		
	.735		.137		-.346			-.048		-.357		
	.768		-.232		-.083			-.031		.062		
	.796		.028		.253			.184		.243		
Turbine cowl	.821		.172		.434			.380				
	.852		.371									
	0.000	-.293	.722	1.001	.625	-.878	-.149	.695	1.125	.649	.799	
	.003	-.089	-.689	.583	-.706	-.839	-.1234	-.922	.407	-.543	1.335	
	.014	-.1.027	-.650	-.355	-.568	-.689	-.444	-.689	-.121	-.471	1.105	
	.031	-.230	-.519	-.481	-.451	-.584	-.297	.056	-.294	.357	1.408	
	.055	-.372	-.052	-.331	-.389	-.395	.302	.258	.189	.181	.357	
	.106	-.071	-.279	-.255	-.221	-.208	-.302	.762	.068	-.087	-.173	
	.172	-.095	.227	-.092	-.099	-.091	.678	.625	.113	-.099	-.002	
	.241	-.177	.471	-.154	-.145	-.044	-.126	.153	.020	.072	.096	
	.302	-.083	.045	-.047	.028	.036	.041	-.026	.083	.128	.131	
	.366	-.043	.036	.052	.146	.071	-.000	.056	.083	.251	.142	
	.387	-.039	.188	.001	.239	.139	.038	.200	.115	.251	.158	
Plug	.442	-.075	.104	.058	.125	.088	-.028	.113	.110	.153	.302	
	.493	.197	.216	.080	.285	.204	.203	-.070	.018	.026	.080	
	.537	-.030	-.024	.042	-.116	.128	.057	.124	.067	.215	.302	
	.581	.108	.031	-.021	-.077	-.010	.114	.086	.015	.226	.099	
	.617	.079	.221	.120	-.081	-.157	.150	.023	.023	.040	-.021	
	.653	.043	-.021	.050	-.208	-.137	.106	.083	.113	.123	.061	
	.690	-.002	.188	.140	-.026	-.173	.073	.325	.413	-.145	-.115	
	.708		.270	.363	-.047			.156		.134		
	.735		.137		-.346			-.048		-.357		
	.768		-.232		-.083			-.031		.062		
	.796		.028		.253			.184		.243		
	.821		.172		.434			.380				
	.852		.371									

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(i) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

C_p at -												
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	1.322	.536	.845	.926	1.066	1.217	.555	.979	.555	.932	1.060
	.003	.354	.989	.1309	.690	.132	.613	.1019	.645	.597	.441	.572
	.014	.331	.622	.132	.590	.251	.691	.645	.692	.426	.415	.732
	.031	.303	.606	.045	.415	.157	.397	.518	.203	.325	.265	.547
	.055	.272	.374	.255	.314	.181	.303	.357	.252	.255	.268	.463
	.106	.197	.262	.278	.192	.116	.193	.240	.245	.141	.107	.287
	.172	.071	.127	.106	.052	.028	.059	.102	.076	.006	.023	.206
	.261	.158	.270	.197	.168	.065	.126	.232	.172	.023	.042	.086
	.302	.119	.065	.065	.015	.026	.087	.035	.042	.041	.072	.020
	.326	.601	.019	.722	.047	.057	.325	.051	.042	.100	.066	.141
	.343	.170	.211	.101	.224	.045	.231	.231	.042	.108	.066	.080
Turbine cowl	0.000	1.322	.536	.845	.926	1.066	1.217	.555	.979	.555	.932	1.060
	.003	.354	.989	.1309	.690	.132	.613	.1019	.645	.597	.441	.572
	.014	.331	.622	.132	.590	.251	.691	.645	.692	.426	.415	.732
	.031	.303	.606	.045	.415	.157	.397	.518	.203	.325	.265	.547
	.055	.272	.374	.255	.314	.181	.303	.357	.252	.255	.268	.463
	.106	.197	.262	.278	.192	.116	.193	.240	.245	.141	.107	.287
	.172	.071	.127	.106	.052	.028	.059	.102	.076	.006	.023	.206
	.261	.158	.270	.197	.168	.065	.126	.232	.172	.023	.042	.086
	.302	.119	.065	.065	.015	.026	.087	.035	.042	.041	.072	.020
	.326	.601	.019	.722	.047	.057	.325	.051	.042	.100	.066	.141
	.343	.170	.211	.101	.224	.045	.231	.231	.042	.108	.066	.080
	.387	.237	.055	.084	.008	.132	.152	.103	.042	.103	.024	.132
Plug	0.000	1.322	.536	.845	.926	1.066	1.217	.555	.979	.555	.932	1.060
	.003	.354	.989	.1309	.690	.132	.613	.1019	.645	.597	.441	.572
	.014	.331	.622	.132	.590	.251	.691	.645	.692	.426	.415	.732
	.031	.303	.606	.045	.415	.157	.397	.518	.203	.325	.265	.547
	.055	.272	.374	.255	.314	.181	.303	.357	.252	.255	.268	.463
	.106	.197	.262	.278	.192	.116	.193	.240	.245	.141	.107	.287
	.172	.071	.127	.106	.052	.028	.059	.102	.076	.006	.023	.206
	.261	.158	.270	.197	.168	.065	.126	.232	.172	.023	.042	.086
	.302	.119	.065	.065	.015	.026	.087	.035	.042	.041	.072	.020
	.326	.601	.019	.722	.047	.057	.325	.051	.042	.100	.066	.141
	.343	.170	.211	.101	.224	.045	.231	.231	.042	.108	.066	.080
	.387	.237	.055	.084	.008	.132	.152	.103	.042	.103	.024	.132

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(j) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.5$; outboard station

x/c	C _p at -															
	α = -2°						α = 0°									
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	.0000	.674	.748	.724	.574	1.088	.487	.756	.863	.554	.588	.392	.743	.931	.624	.937
	.003	-.424	-.923	-.848	-.914	-.451	-.867	-.936	-.570	-.885	-.783	-.933	-.911	-.673	-.644	-.996
	.014	-.475	-.721	-.804	-.694	-.488	-.627	-.742	-.584	-.642	-.619	-.886	-.701	-.473	-.620	-.763
	.031	-.050	-.658	-.1204	-.692	-.548	-.306	-.621	-.836	-.642	-.634	-.589	-.580	-.790	-.583	-.591
	.055	-.350	-.030	-.857	-.498	-.391	-.377	.023	-.443	-.437	-.418	-.364	-.039	-.405	-.426	-.411
	.106	.372	-.348	-.348	-.329	-.243	-.101	-.320	-.336	-.255	-.248	-.715	-.300	-.297	-.251	-.228
	.172	-.126	.670	-.175	-.193	-.112	-.114	.303	-.142	-.147	-.139	-.089	-.821	-.113	-.118	-.061
	.261	-.250	.609	-.238	-.300	-.208	-.228	.432	-.205	-.225	-.157	-.199	-.837	-.182	-.183	-.105
	.302	-.158	.112	.313	-.104	-.053	-.122	-.076	.151	-.040	-.037	-.093	-.053	-.077	-.001	.005
	.326	-.099	.020	-.012	-.031	-.041	-.071	.013	.018	.012	.010	-.046	.034	.039	.060	.049
	.343	.058	.211	.035	.262	.036	.046	.219	.036	.260	.156	.048	.223	.034	.263	.161
	.387	-.452	-.046	.064	-.412	-.337	.018	.115	-.066	.209	.239	.049	.049	.102	.060	.004
Turbine cowl	.442	.097	.106	.107	.249	.207	.031	.016	.011	.174	.208	.015	.006	.047	.033	.216
	.493	-.103	-.128	-.047	-.297	-.227	.007	.002	-.139	.175	.079	.103	.076	.037	.031	.031
	.537	-.062	-.005	-.094	-.289	-.182	.011	-.105	.040	.058	.057	.003	.105	.016	.018	.114
	.581	-.146	-.201	-.182	-.242	-.386	.003	.146	.016	.319	.267	.091	.225	.036	.278	.183
	.617	-.091	.030	-.138	-.263	-.558	-.023	.036	-.063	.319	.267	.091	.225	.036	-.042	-.479
	.653	-.165	-.151	-.117	-.509	-.592	-.126	.061	-.035	.319	.267	.091	.225	.036	-.097	-.217
	.690	-.401	-.288	-.207	-.767	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	-.131	-.264
	.708	.003	.200	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	-.1053	.364
	.735	.051	.200	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.768	-.283	.219	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.796	-.600	.376	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.821	-.154	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
Plug	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
	.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109
.852	-.148	.462	-.765	-.765	-.760	-.102	.013	-.053	.347	.260	.046	.058	.076	.364	.109	

Fan cowl	.0000	.887	.693	1.119	.662	.800	.392	.743	.931	.624	.937					
	.003	-.988	-.945	-.689	-.740	-.1055	-.1069	-.964	.310	-.533	-.1271	-.933	-.911	-.673	-.644	-.996
	.014	-.999	-.720	-.379	-.602	-.912	-.1201	-.720	.310	-.533	-.1271	-.933	-.911	-.673	-.644	-.996
	.031	-.051	-.588	-.612	-.534	-.545	-.257	-.557	.340	-.406	-.468	-.866	-.701	-.473	-.620	-.763
	.055	-.325	-.038	-.345	-.368	-.237	-.035	.253	-.372	-.338	-.589	-.430	-.297	-.405	-.426	-.411
	.106	.157	-.287	-.279	-.241	-.111	-.271	.216	-.192	-.179	-.364	-.309	-.405	-.426	-.411	-.228
	.172	-.086	-.390	-.269	-.045	-.060	.140	.062	-.074	-.009	-.089	-.821	-.228	-.251	-.228	-.228
	.261	-.190	-.058	-.166	-.158	-.061	-.080	.303	.138	-.062	-.009	-.089	-.821	-.228	-.251	-.228
	.302	-.074	-.183	.025	.036	-.044	.023	.138	.061	.098	-.157	-.183	-.105	-.077	-.001	.005
	.326	-.031	.044	.054	.315	.077	-.001	.054	.075	.228	.129	-.093	-.077	-.001	.005	.049
	.343	.040	.232	.284	.284	.163	.030	.230	.046	.275	.158	-.046	-.034	.039	.060	.049
	.387	-.231	.070	.088	.088	.093	-.056	.093	.093	.275	.158	-.046	-.034	.039	.060	.049
Turbine cowl	.442	-.004	.002	.065	.028	.056	.015	.064	.015	.064	.015	.064	.015	.064	.015	.064
	.493	.130	.164	.004	.033	.119	.207	.295	.067	.226	.283	.589	-.430	-.297	-.405	-.426
	.537	.024	-.045	-.090	.163	.085	.035	.067	.067	.226	.283	.589	-.430	-.297	-.405	-.426
	.581	.110	.191	.004	.156	.156	.059	-.048	.145	.145	.145	.145	.145	.145	.145	.145
	.617	-.008	-.046	.086	.015	.101	.211	.167	.093	.038	-.046	-.034	.039	.060	.049	.049
	.653	.063	.159	-.038	-.158	-.126	.124	.028	.070	.093	.046	-.046	-.034	.039	.060	.049
	.690	-.086	-.087	.117	-.176	-.253	.042	.222	.056	.040	-.119	-.046	-.034	.039	.060	.049
	.708	.316	.374	-.547	-.285	.285	.285	.374	.555	-.119	-.046	-.034	.039	.060	.049	.049
	.735	.172	.150	-.150	.159	.159	.159	.061	.061	.061	.061	.061	.061	.061	.061	.061
	.768	-.274	-.314	-.494	-.283	-.180	-.180	.061	.061	.061	.061	.061	.061	.061	.061	.061
	.796	-.198	-.314	-.494	-.283	-.180	-.180	.061	.061	.061	.061	.061	.061	.061	.061	.061
	.821	.183	.257	.435	.175	.257	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
Plug	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
	.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221
.852	.358	.435	.435	.435	.435	.350	.437	.208	.265	.221	.173	.221	.173	.221	.221	

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(k) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C _p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																															
	α = -2°						α = 0°						α = 1°																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																			
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																		
Fan cowl	0.000	.899	.226	.534	.670	.892	.713	.223	.776	.726	.697	.208	.884	.740	.582	-.399	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.030	-1.130	-.490	-.680	-.309	-.997	-1.

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(1) $M = 0.700$; $p_{t_e}/p_{t_\infty} = 1.3$; outboard station

x/c	C _p at -																									
	α = -2°						α = 0°						α = 4°													
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.503	.588	.502	.966	.230	.579	.718	.354	.827	.067	.555	.826	.362	.718											
	.003	-.437	-.978	-.095	-.601	-.849	-.986	.048	-.1095	-.988	-.1124	-.1042	.077	-.1002	-.1202											
	.014	-.437	-.674	-.778	-.466	-.582	-.646	-.548	-.624	-.655	-.639	-.653	-.455	-.605	-.702											
	.031	-.1012	-.503	-.652	-.459	-.1120	-.471	-.523	-.446	-.507	-.1100	-.474	-.432	-.401	-.440											
	.055	-.307	-.022	-.490	-.353	-.340	.006	-.386	-.398	-.365	-.373	.006	-.341	-.260	-.282											
	.106	-.384	-.306	-.345	-.289	-.587	-.276	-.283	-.368	-.273	-.693	-.260	-.230	-.401	-.440											
	.172	-.133	-.280	-.177	-.195	-.127	-.351	-.127	-.269	-.098	-.153	-.341	-.172	-.082	-.111											
	.261	-.229	-.102	-.209	-.269	-.204	-.244	-.166	-.188	-.146	-.153	-.267	-.157	-.162	-.111											
	.302	-.152	-.112	-.138	-.160	-.122	-.075	-.218	-.030	-.075	-.111	-.069	-.218	-.004	-.020											
	.326	-.118	-.041	-.028	-.157	-.127	-.001	.012	-.072	-.037	-.135	-.006	-.066	.025	-.005											
	.343	-.147	-.073	-.131	-.170	-.127	-.082	.015	-.150	-.014	-.114	-.066	.041	-.005	-.005											
	.387	-.007	.004	-.002	-.004	-.023	.054	.045	.073	.083	.085	.084	.081	.081	.102											
	.442	-.012	-.047	-.064	-.092	-.076	-.010	-.004	-.009	.031	.001	.002	.009	.044	.093											
	.493	.003	-.034	-.060	-.095	-.112	.048	.003	.002	.032	.034	.038	.025	.044	.096											
	.537	.022	-.041	-.067	-.144	-.186	.023	-.004	-.033	.009	.054	.051	.025	.015	.093											
	.581	.079	-.070	-.092	-.289	-.385	.009	-.017	-.133	-.117	.049	.051	.009	-.065	-.040											
.617	.113	-.067	-.084	-.215	-.466	.006	.035	.012	-.078	-.401	.035	.067	.038	-.143												
.653	.133	-.076	-.072	-.298	-.475	.010	.032	.012	-.075	-.152	.039	.064	.041	-.027												
.690	.215	-.122	-.060	-.385	-.653	-.010	.028	-.085	-.243	-.033	.031	.061	.048	-.162												
.768	-.018	-.018	-.0543	-.543	-.053	-.097	.028	-.614	.109	.109	.109	.148														
.735	-.083	-.083	-.105	-.105	-.105	.025	.025	-.017	.064	.064	.064	.102														
.768	-.148	-.148	-.173	-.173	-.173	.030	.030	-.081	.081	.081	.081	.128														
.796	-.099	-.099	-.092	-.092	-.092	.074	.074	.096	.106	.106	.106	.244														
.821	-.002	-.002	-.004	-.004	-.004	.181	.181	.219																		
.852	.111	.111	.139	.139	.139																					
Fan cowl	0.000	-.048	-.525	.527	.385	-.255	.479	1.067	.441	.510																
	.003	-.1276	-.011	-.074	-.933	-.1451	-.479	1.067	.441	.510																
	.014	-.680	-.652	-.366	-.578	-.1132	-.670	-.126	-.522	-.899																
	.031	-.1053	-.470	-.396	-.420	-.584	-.469	-.207	-.409	-.577																
	.055	-.390	.013	-.302	-.478	-.406	-.269	-.198	-.374	-.396																
	.106	-.772	-.276	-.237	-.391	-.726	-.269	-.168	-.190	-.200																
	.172	-.133	-.392	-.098	-.184	-.116	-.411	-.043	-.094	-.039																
	.261	-.181	-.321	-.146	-.184	-.145	-.353	-.104	-.103	-.020																
	.302	-.104	-.065	-.234	-.012	-.088	-.055	-.217	.054	.067																
	.326	-.070	-.009	-.035	-.026	-.073	.016	.648	.032	.100																
	.343	-.118	-.010	-.052	-.058	-.097	.000	-.023	.084	.059																
	.387	.042	-.068	-.074	.125	.077	.081	.113	.174	.215																
	.442	.012	-.009	-.026	.071	.048	.029	.064	.135	.215																
	.493	.056	.048	.038	.080	.141	.096	.081	.154	.232																
	.537	.066	.064	.038	.054	.106	.097	.084	.125	.260																
	.581	.080	.071	.029	-.029	.135	.119	.077	.056	.167																
.617	.081	.090	.061	-.025	-.071	.145	.110	.103	.090																	
.653	.080	.094	.084	.006	-.004	.152	.113	.090	.119																	
.690	.003	.068	.081	-.026	.106	.204	.223	.058	.045																	
.768	.149	.149	.171	-.657	.152	.152	.152	.090	.090																	
.735	.087	.087	.035	.035	.152	.107	.042	.103	.103																	
.768	.064	.064	.061	-.020	.132	.132	.132	.184	.209																	
.821	.132	.132	.154	.154	.154	.268	.268	.315	.315																	
.852	.226	.226	.267	.267	.267	.267	.267	.315	.315																	

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(m) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	.900	.252	.582	.684	.667	.736	.238	.769	.729	.670	.650	.244
	.589	-1.164	-1.354	-1.064	-.378	-1.088	-1.205	-1.252	-.900	-.926	-1.374	-.873
	.390	-.575	-1.448	-.524	-.340	-.530	-.563	-.592	-.432	-.453	-.554	-.906
	.031	-.315	-1.017	-.393	-.235	-.388	-.457	-.493	-.339	-.313	-.443	-.551
	.055	-.267	-.423	-.320	-.209	-.313	-.355	-.358	-.269	-.239	-.413	-.465
	.106	-.196	-.271	-.209	-.133	-.202	-.246	-.246	-.155	-.128	-.285	-.293
	.172	-.085	-.145	-.083	-.031	-.082	-.123	-.056	-.026	-.009	-.161	-.211
	.261	-.169	-.271	-.192	-.104	-.149	-.237	-.176	-.108	-.044	-.208	-.211
	.302	-.084	-.084	-.051	-.025	-.118	-.055	-.055	-.032	-.012	-.183	-.064
	.326	-.007	-.086	-.007	-.005	-.087	.024	.012	.079	.056	-.138	-.029
	.343	-.200	-.034	-.022	-.182	-.158	.050	.073	.076	.056	-.209	-.047
	.387	-.351	.028	-.013	-.057	-.335	.056	.055	.097	.047	-.404	-.091
Turbine cowl	.442	.025	-.078	-.115	.060	.033	-.020	-.014	.024	.149	.028	.015
	.493	.052	-.010	-.066	.001	.037	.033	-.066	.027	.120	.021	.036
	.537	.047	-.019	-.086	.177	.422	.027	-.014	-.058	.207	.041	.055
	.581	.180	-.107	-.159	.136	.063	.029	.070	.117	.001	.033	.023
	.617	-.090	-.069	-.113	.290	.413	.073	.033	.026	.155	.059	.015
	.653	.052	-.081	-.113	.264	.358	.154	.003	.020	.178	.199	.062
	.690	-.227	-.086	-.101	.264	-.051	-.060	.003	.000	.047	.081	.030
	.708	-.127	-.028	-.115	.264	-.051	-.060	.003	.000	.047	.081	.030
	.735	-.034	-.451	-.451	.282	.282	.129	.017	.038	.053	.081	.115
	.768	-.245	-.282	-.282	.282	.282	.129	.017	.038	.053	.081	.115
	.796	-.218	-.209	-.209	.209	.209	.105	.003	.038	.053	.081	.115
	.821	-.089	-.072	-.072	.072	.072	.003	.003	.038	.053	.081	.115
	.852	-.048	.671	.671	.671	.671	.126	.126	.167	.167	.182	.211
Plug	.000	.582	.572	.775	.500	.423	.194	1.116	.802	.331	.736	.873
	.003	-1.270	-.564	-.444	-.442	-1.087	-1.325	-.075	-.345	1.420	-.795	-.906
	.014	-.702	-.584	-.407	-.533	-1.032	-.573	-.075	-.320	1.420	-.795	-.906
	.031	-.511	-.438	-.281	-.357	-.828	-.433	-.075	-.216	1.420	-.795	-.906
	.055	-.418	-.335	-.214	-.252	-.556	-.324	-.075	-.166	1.420	-.795	-.906
	.106	-.281	-.224	-.133	-.112	-.356	-.213	-.075	-.067	1.420	-.795	-.906
	.172	-.153	-.103	-.048	-.023	-.247	-.093	-.000	-.056	1.420	-.795	-.906
	.261	-.197	-.200	-.136	-.059	-.087	-.081	.015	.012	1.420	-.795	-.906
	.302	-.161	-.021	-.018	.084	-.069	-.005	.015	.012	1.420	-.795	-.906
	.326	-.126	.052	.008	.135	-.015	.068	.027	.015	1.420	-.795	-.906
	.343	-.170	.070	.053	.129	-.011	.062	.027	.015	1.420	-.795	-.906
	.387	-.392	.087	.105	.148	.268	.112	.141	.225	1.420	-.795	-.906
	.442	.007	.023	.040	.116	.359	.056	.082	.187	1.420	-.795	-.906
Fan cowl	.506	.047	.073	.052	.113	.452	.106	.103	.196	.336	.736	.873
	.537	.042	.073	.052	.113	.452	.106	.103	.196	.336	.736	.873
	.581	.020	.061	.011	-.012	.077	.120	.064	.158	.181	.736	.873
	.617	.030	.066	-.056	.040	.130	.153	.100	.097	.129	.736	.873
	.653	.073	.102	.061	.005	.144	.123	.123	.064	.172	.736	.873
	.690	-.024	.117	.081	-.036	.165	.191	.144	.038	.137	.736	.873
	.708	-.084	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.735	-.007	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.768	-.002	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.796	.002	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.821	.002	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.852	.002	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873
	.852	.002	.084	.058	-.074	.165	.153	.199	.038	.137	.736	.873

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION I - Continued

(n) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.3$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.458	.592	.535	.345	.965	.238	.562	.697	.351	.810	.089	.541	.788	.732			
	.003	.570	.701	.623	.430	.862	.150	.164	.148	.1203	.1080	.1203	.153	.181	-1.327			
	.014	.504	.631	.562	.373	.836	.595	.556	.535	.520	.588	.713	.638	.677	.603			
	.031	.831	.962	.879	.686	.507	1.037	.540	.617	.520	.403	.960	.520	.471	.403			
	.055	.831	.962	.879	.686	.507	.378	.008	.434	.444	.403	.390	.006	.375	.288			
	.106	.240	.333	.335	.319	.424	.444	.304	.276	.276	.247	.531	.289	.282	.123			
	.172	.134	.119	.116	.259	.272	.140	.212	.147	.156	.100	.129	.115	.167	.153			
	.261	.240	.066	.217	.213	.213	.219	.070	.189	.109	.156	.196	.106	.167	.017			
	.302	.451	.384	.005	.145	.160	.135	.085	.100	.082	.100	.072	.015	.030	.018			
	.326	.476	.408	.019	.098	.172	.095	.005	.009	.017	.029	.085	.015	.030	.018			
	.343	.493	.425	.098	.016	.116	.100	.008	.062	.033	.100	.072	.015	.030	.018			
	.387	.021	.022	.002	.001	.010	.028	.045	.039	.068	.080	.061	.068	.109	.056			
Turbine cowl	.442	.003	.059	.666	.078	.031	.011	.032	.020	.014	.021	.010	.003	.012	.080			
	.493	.016	.031	.060	.095	.113	.024	.018	.011	.008	.020	.048	.045	.024	.050			
	.537	.010	.037	.075	.172	.166	.037	.021	.014	.050	.030	.056	.054	.021	.086			
	.581	.076	.096	.110	.301	.351	.006	.002	.038	.185	.137	.043	.045	.006	.064			
	.617	.103	.072	.072	.269	.365	.011	.026	.000	.153	.100	.034	.063	.079	.170			
	.653	.134	.084	.087	.213	.333	.020	.005	.005	.129	.185	.039	.063	.067	.132			
	.690	.240	.119	.072	.313	.489	.109	.047	.015	.120	.085	.027	.057	.061	.182			
	.728	.031	.025	.595	.175	.595	.057	.005	.095	.047	.047	.110	.142	.061	.056			
	.735	.105	.105	.175	.175	.175	.005	.005	.005	.047	.047	.110	.142	.061	.056			
	.768	.167	.167	.192	.192	.192	.026	.026	.026	.103	.103	.057	.057	.059	.027			
	.821	.114	.114	.140	.140	.140	.057	.057	.057	.023	.023	.024	.024	.101	.121			
	.852	.099	.099	.122	.122	.122	.157	.157	.157	.195	.195	.199	.199	.233	.233			
Fan cowl	0.000	.040	.527	.901	.400	.883	.148	.472	1.043	.444	.564	.089	.541	.788	.732			
	.003	-1.279	-1.186	-1.092	-1.092	-1.460	-1.431	-1.250	-1.043	-898	1.421	-1.574	-1.124	-1.124	-1.327			
	.014	-1.213	-1.036	-1.007	-1.007	-1.054	-1.523	-1.657	-1.567	-549	-1.435	-1.574	-1.124	-1.124	-1.327			
	.031	.921	.509	.444	.442	.583	.818	.510	.262	.381	.543	.960	.520	.471	.603			
	.055	.595	.014	.326	.386	.401	.386	.021	.227	.358	.405	.960	.520	.471	.603			
	.106	.616	.281	.286	.265	.260	.567	.274	.188	.261	.187	.960	.520	.471	.603			
	.172	.121	.264	.095	.151	.085	.099	.285	.056	.079	.029	.960	.520	.471	.603			
	.261	.145	.179	.145	.162	.060	.135	.197	.112	.096	.008	.960	.520	.471	.603			
	.302	.095	.057	.116	.023	.017	.060	.044	.103	.059	.083	.960	.520	.471	.603			
	.326	.060	.023	.044	.026	.046	.020	.030	.071	.059	.109	.960	.520	.471	.603			
	.343	.040	.023	.024	.046	.046	.020	.030	.071	.059	.109	.960	.520	.471	.603			
	.387	.073	.079	.085	.085	.074	.082	.027	.003	.118	.182	.224	.960	.520	.471			
Turbine cowl	.442	.011	.011	.032	.135	.033	.033	.033	.062	.136	.209	.960	.520	.471	.603			
	.493	.064	.061	.044	.085	.152	.121	.080	.080	.136	.209	.960	.520	.471	.603			
	.537	.082	.073	.044	.046	.155	.127	.098	.070	.121	.256	.960	.520	.471	.603			
	.581	.086	.073	.035	.045	.017	.134	.113	.074	.067	.141	.960	.520	.471	.603			
	.617	.073	.091	.064	.015	.086	.130	.133	.104	.100	.053	.960	.520	.471	.603			
	.653	.077	.091	.061	.036	.030	.143	.145	.107	.082	.100	.960	.520	.471	.603			
	.690	.011	.061	.085	.051	.112	.143	.145	.127	.081	.018	.960	.520	.471	.603			
	.728	.144	.144	.174	.051	.189	.189	.189	.210	.057	.018	.960	.520	.471	.603			
	.735	.091	.091	.174	.051	.189	.189	.189	.210	.057	.018	.960	.520	.471	.603			
	.768	.032	.032	.021	.021	.086	.086	.086	.086	.030	.106	.960	.520	.471	.603			
	.796	.061	.061	.055	.055	.110	.110	.110	.110	.024	.027	.960	.520	.471	.603			
	.821	.132	.132	.132	.132	.172	.172	.172	.172	.024	.027	.960	.520	.471	.603			
.852	.221	.221	.221	.221	.251	.251	.251	.251	.195	.234	.960	.520	.471	.603				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(o) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.3$; inboard station

x/\bar{c}	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	.890	.263	.377	.677	.865	.744	.279	.763	.732	.700
	-.617	-1.229	1.565	-1.068	-.417	-1.077	-1.265	-1.257	-.981	-.916
	-.397	-.632	-1.363	-.532	-.373	-.508	-.559	-.559	-.468	-.504
	-.317	-.497	-.864	-.400	-.241	-.393	-.461	-.928	-.336	-.305
	.055	-.279	-.377	-.319	-.205	-.308	-.345	-.348	-.263	-.232
	.106	-.202	-.267	-.292	-.135	-.197	-.233	-.241	-.145	-.114
	.172	-.088	-.144	-.127	-.074	-.066	-.109	-.084	-.011	-.009
	.261	-.169	-.273	-.205	-.090	-.134	-.224	-.162	-.095	-.030
	.302	-.143	-.079	-.174	-.015	-.100	-.039	-.044	-.032	-.051
	.326	-.668	-.004	-.043	-.021	-.046	-.037	-.085	-.093	-.074
	.343	-.215	.097	-.071	.021	-.146	.077	-.027	.104	-.120
	.387	-.334	.055	.016	.038	-.304	.077	.077	.113	.074
	.442	.077	-.048	-.074	.103	.091	-.013	-.002	.040	.163
	.493	.043	-.026	-.068	.021	.117	.051	.006	.029	.130
	.537	.153	-.012	-.092	-.473	.079	.049	-.005	-.075	-.218
Turbine cowl	.581	.323	.177	.177	.024	.134	.001	-.067	-.210	-.058
	.617	-.088	-.060	-.130	-.406	.302	.020	-.025	-.095	-.154
	.653	.047	-.107	-.132	-.593	.109	.012	-.016	-.145	-.210
	.690	-.219	-.048	-.426	.097	-.057	.046	.001	-.176	-.061
	.708		.152	-.121			-.019	.071	-.011	
Plug	.735		.037	.398			-.011		-.331	
	.768		.301	.328			-.129		-.179	
	.796		.219	.216			-.106		-.111	
	.821		-.107	-.096			.004		-.032	
	.852		-.036	.055			.125		.169	
$\alpha = 10^\circ$										
Fan cowl	.876	.254	.876	.744	.599	.656	.254	.876	.744	.599
	-.865	-1.268	-.995	-.865	-1.230	-.865	-1.268	-.995	-.865	-1.230
	-.442	-.559	-.559	-.442	-.501	-.559	-.559	-.559	-.442	-.501
	-.310	-.452	-.452	-.310	-.344	-.419	-.452	-.452	-.310	-.344
	.055	-.302	-.302	.055	-.240	-.330	-.302	-.302	.055	-.240
	.106	-.226	-.226	.106	-.111	-.194	-.226	-.226	.106	-.111
	.172	-.125	-.125	.172	-.060	-.106	-.125	-.125	.172	-.060
	.261	-.060	-.060	.261	-.007	-.060	-.060	-.060	.261	-.007
	.302	-.150	-.150	.302	-.004	-.122	-.150	-.150	.302	-.004
	.326	-.026	-.026	.326	-.004	-.092	-.026	-.026	.326	-.004
	.343	-.103	-.103	.343	-.083	-.027	-.103	-.103	.343	-.083
	.387	-.012	-.012	.387	-.111	-.164	-.012	-.012	.387	-.111
	.442	-.089	-.089	.442	-.145	-.304	-.089	-.089	.442	-.145
	.493	-.002	-.002	.493	-.206	-.078	-.002	-.002	.493	-.206
	.537	.035	.035	.537	.084	.125	.035	.035	.537	.084
Turbine cowl	.581	.024	.024	.581	.108	.180	.024	.024	.581	.108
	.617	-.029	-.029	.617	-.102	.102	-.029	-.029	.617	-.102
	.653	.055	.055	.653	-.082	.044	.055	.055	.653	-.082
	.690	.027	.027	.690	-.102	.129	.027	.027	.690	-.102
	.708	.064	.064	.708	-.029	.006	.064	.064	.708	-.029
Plug	.735	.103	.103	.735	.240	.010	.103	.103	.735	.240
	.768	-.072	-.072	.768	-.111	-.072	-.072	-.072	.768	-.111
	.796	-.040	-.040	.796	-.063	-.040	-.040	-.040	.796	-.063
	.821	.050	.050	.821	.080	.050	.050	.050	.821	.080
	.852	.163	.163	.852	.206	.163	.163	.163	.852	.206
$\alpha = 40^\circ$										
Fan cowl	.814	.256	.814	.675	.525	.475	.256	.814	.256	.381
	-.1532	-1.530	-.1532	-1.532	-1.394	-1.532	-1.530	-.1532	-.629	1.361
	-.357	-.637	-.637	-.357	-.519	-.637	-.637	-.637	-.357	-.847
	.055	-.250	-.250	.055	-.250	-.250	-.250	-.250	.055	-.377
	.106	-.134	-.134	.106	-.134	-.134	-.134	-.134	.106	-.252
	.172	-.080	-.080	.172	-.080	-.080	-.080	-.080	.172	-.077
	.261	-.037	-.037	.261	-.037	-.037	-.037	-.037	.261	-.068
	.302	-.004	-.004	.302	-.004	-.004	-.004	-.004	.302	-.040
	.326	.004	.004	.326	.004	.004	.004	.004	.326	.015
	.343	.077	.077	.343	.077	.077	.077	.077	.343	.183
	.387	.137	.137	.387	.137	.137	.137	.137	.387	.202
	.442	.162	.162	.442	.162	.162	.162	.162	.442	.230
	.493	.125	.125	.493	.125	.125	.125	.125	.493	.242
	.537	.082	.082	.537	.082	.082	.082	.082	.537	.320
	.581	.016	.016	.581	.016	.016	.016	.016	.581	.342
Turbine cowl	.617	.056	.056	.617	.056	.056	.056	.056	.617	.169
	.653	-.037	-.037	.653	-.037	-.037	-.037	-.037	.653	.076
	.690	.075	.075	.690	.075	.075	.075	.075	.690	.113
	.708	.140	.140	.708	.140	.140	.140	.140	.708	.043
	.735	.059	.059	.735	.059	.059	.059	.059	.735	.110
Plug	.768	-.023	-.023	.768	-.023	-.023	-.023	-.023	.768	-.181
	.796	-.031	-.031	.796	-.031	-.031	-.031	-.031	.796	-.305
	.821	.084	.084	.821	.084	.084	.084	.084	.821	.046
	.852	.194	.194	.852	.194	.194	.194	.194	.852	.180
										.292

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued
 (p) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.3$; outboard station

C _p at -															
x/c	α = -2°						α = 0°						α = 1°		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.484	.592	.538	.369	.957	.245	.563	.692	.387	.827	.542	.800	.387	.766
	.003	-.641	-1.218	-.327	-1.306	-.707	-1.049	-1.221	-.211	-1.253	-1.078	-1.224	.240	-1.168	-1.340
	.014	-.519	-.894	-.912	-.912	-.563	-.672	-.861	-.819	-.665	-.759	-.886	-.498	-.622	-1.077
	.031	-.612	-.600	-.636	-.636	-.535	-.947	-.561	-.657	-.536	-.606	-.893	-.563	-.504	-.617
	.055	-.341	-.026	-.382	-.490	-.386	-.379	-.002	-.444	-.442	-.400	-.388	-.005	-.425	-.405
	.106	-.118	-.331	-.360	-.327	-.307	-.311	-.300	-.311	-.287	-.284	-.410	-.285	-.272	-.298
	.172	-.134	.002	-.179	-.224	-.178	-.125	-.110	-.135	-.140	-.095	-.126	-.117	-.196	-.075
	.261	-.244	.163	-.224	-.282	-.212	-.214	-.024	-.183	-.200	-.143	-.202	-.168	-.168	-.196
	.302	-.156	.116	.115	-.192	-.136	-.125	-.073	-.016	-.030	-.056	-.113	-.063	-.010	-.010
	.326	-.118	.035	-.015	-.068	-.082	-.083	-.006	-.021	-.019	-.022	-.075	.013	.036	.061
	.343	-.080	.015	-.083	-.001	-.088	-.074	.018	-.039	-.057	-.067	-.070	.024	.002	.081
	.387	.030	.013	.002	.011	.013	.049	.063	.058	.085	.091	.065	.073	.070	.118
	.442	-.042	.097	-.077	-.155	-.054	.032	-.016	-.011	-.002	.057	.044	-.004	.007	.041
	.493	.039	.061	-.071	-.116	-.094	.040	.024	.001	.002	.063	.061	.042	.027	.047
Turbine cowl	0.000	.537	.058	-.026	-.083	-.133	.253	.032	-.005	-.044	.012	.053	.047	.007	.064
	.014	-.058	-.026	-.083	-.223	-.133	.253	.032	-.005	-.044	.012	.053	.047	.007	.064
	.031	-.092	-.128	-.092	-.293	-.420	.006	.018	-.030	-.228	-.183	.053	.036	.005	-.097
	.055	-.096	-.068	-.088	-.257	-.611	.006	.018	-.030	-.228	-.183	.053	.036	.005	-.097
	.106	-.185	-.108	-.108	-.366	-.527	-.015	.015	-.002	-.075	-.208	.031	.039	.039	-.137
	.172	-.253	-.156	-.038	-.445	-.541	-.095	-.008	.018	-.123	-.301	.005	.030	.030	-.140
	.261	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.302	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.326	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.343	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.387	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.442	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.493	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
	.537	-.054	.013	.013	-.645	-.541	-.095	.058	.103	-.787	-.301	.005	.030	.030	-.210
Plug	0.000	.861	.861	.414	.414	.726	-.091	.497	1.038	.449	.578	.449	.578	.863	.863
	.014	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.031	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.055	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.106	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.172	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.261	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.302	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.326	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.343	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.387	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.442	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.493	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
	.537	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863
.581	-.861	-.861	-.414	-.414	-.726	-.091	.497	1.038	.449	.578	.449	.578	-.863	-.863	

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(q) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C_p at -														
	$\alpha = 0^\circ$														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.929	.314	.633	.724	.875	.790	.300	.796	.753	.693	.360	.935	.826	.699
	.003	-.651	-1.242	-1.250	-1.069	-.385	-1.133	-1.257	-1.304	-.997	-1.048	-1.209	-.681	-.779	-1.164
	.014	-.402	-.845	-1.315	-.522	-.377	-.455	-.868	-1.030	-.455	-.455	-.749	-.543	-.425	-.482
	.031	-.329	-.502	-.640	-.420	-.253	-.414	-.446	-.800	-.355	-.320	-.413	-.448	-.708	-.296
	.055	-.284	-.381	-.294	-.323	-.205	-.321	-.362	-.365	-.274	-.234	-.337	-.305	-.223	-.226
	.106	-.202	-.265	-.284	-.199	-.129	-.198	-.246	-.248	-.148	-.115	-.218	-.208	-.110	-.096
	.172	-.080	-.132	-.116	-.062	-.011	-.063	-.110	-.083	-.010	-.001	-.050	-.031	-.025	-.030
	.261	-.162	-.267	-.200	-.167	-.078	-.133	-.229	-.172	-.037	-.026	-.111	-.205	-.140	-.051
	.302	-.137	-.273	-.207	-.167	-.003	-.100	-.043	-.043	.038	.054	-.074	-.018	-.082	.100
	.326	-.246	-.008	-.340	-.038	.001	-.092	.038	.173	.098	.079	-.142	.060	.201	.136
	.343	-.223	.135	-.130	.091	-.116	-.186	.125	-.013	.125	-.091	-.184	.168	-.048	.208
	.387	-.304	.059	-.032	.048	-.008	-.288	.695	.074	.122	.087	-.278	.106	.106	.162
Turbine cowl	.442	.058	-.095	-.078	-.076	.102	-.104	.003	-.013	.038	.184	.130	.079	.127	.257
	.493	.046	.035	-.041	-.062	.024	.170	.036	.031	-.040	.143	.076	.033	.100	.246
	.537	.311	.019	.119	.086	-.285	.227	.032	.031	-.076	.215	.376	.028	-.042	-.188
	.581	.454	.000	.180	-.221	-.234	.310	.022	-.029	-.134	.207	.057	.025	-.034	-.208
	.617	-.060	-.020	.168	-.929	-.374	.310	.006	-.043	-.185	-.223	.142	.085	-.107	-.078
	.653	-.034	-.122	.138	-.367	-.357	.130	.009	-.018	-.245	.085	.085	.041	-.086	-.094
	.690	-.268	-.073	-.097	-.431	.253	-.076	.035	.052	-.022	.080	.352	.041	-.191	.020
	.708		-.178	-.084	-.320			-.021		-.285		-.007	.052	.041	.041
	.735		-.041	-.428	-.428			-.156		-.220		-.135	-.007	-.226	-.226
	.756		-.343	-.302	-.302			-.132		-.123		-.391	-.055	-.081	-.062
	.796		-.273	-.310	-.310			-.024		-.009		.055	.217	-.082	-.249
	.821		-.127	-.070	-.070			.095		.146					
.852		.003	.064	.064											
Plug	0.000	.638	.282	.953	.771	.548	.525	.277	1.110	.820	.410				
	.003	-.1.467	-.1.257	-.691	-.816	1.322	-.1.562	-.1.262	-.1.139	-.715	1.309				
	.014	-.1.218	-.859	-.691	-.277	-.595	-.1.329	-.817	-.258	-.351	1.159				
	.031	-.400	-.434	-.859	-.277	-.325	-.542	-.436	-.493	-.230	-.327				
	.055	-.392	-.329	-.259	-.212	-.236	-.424	-.323	-.171	-.171	-.260				
	.106	-.253	-.210	-.183	-.093	-.088	-.223	-.204	-.128	-.063	-.063				
	.172	-.119	-.083	-.031	.042	.050	-.106	-.071	.007	.071	.079				
	.261	-.168	-.186	-.126	-.029	.042	-.130	-.166	-.098	.020	.101				
	.322	-.135	-.004	-.004	.106	.131	-.098	.012	.020	.149	.192				
	.343	-.217	.112	.169	.152	.155	.082	.088	.174	.190	.211				
	.387	-.235	.118	.123	.187	.184	-.183	.120	.047	.206	-.050				
	.442	.057	.087	.047	.133	.249	-.057	.126	.150	.238	.262				
Fan cowl	.493	.078	.085	.063	.085	.252	.086	.118	.104	.192	.338				
	.537	.106	.088	.053	.085	-.012	.123	.123	.093	.149	.147				
	.581	.151	.063	.007	-.031	.249	.151	.099	.052	.066	.257				
	.617	.016	.090	.044	.015	.028	.061	.139	.091	.009	.152				
	.653	.078	.082	.053	-.034	-.037	.114	.155	.109	.042	.087				
	.690	-.021	.104	.074	-.020	.055	.057	.172	.126	-.009	.104				
	.708		.085	.136	.039			.136	.188	-.104					
	.735		-.004	-.085	-.163			.007		-.125					
	.756		-.029	-.085	-.037			.037		-.020					
	.796		-.012	-.029	-.047			.047		.031					
	.821		.069	-.112	.128			.128		.168					
	.852		.182	.230	.230			.228		.273					

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(*) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.3$; outboard station

x/\bar{c}	C_p at -												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.475	.592	.553	.390	.949	.272	.578	.768	.408	.832	.200	.558
	.003	-.707	-.1195	.449	-.1290	-.737	-.1025	-.1218	.293	-.1250	-.1566	-.1085	-.1212
	.014	-.536	-.944	-.1255	-.965	-.561	-.894	-.978	-.980	-.903	-.824	-.126	-.959
	.031	-.332	-.585	-.1257	-.683	-.569	-.772	-.463	-.536	-.534	-.632	-.742	-.476
	.055	-.357	-.519	-.395	-.496	-.401	-.393	-.005	-.460	-.466	-.417	-.367	-.012
	.106	-.030	-.340	-.351	-.328	-.298	-.173	.318	-.329	-.322	-.284	-.294	-.291
	.172	-.133	-.158	-.177	-.192	-.117	-.124	.001	-.151	-.153	-.092	-.115	-.091
	.261	-.251	-.253	-.229	-.287	-.214	-.124	.104	-.198	-.216	-.154	-.200	-.061
	.302	-.157	-.114	.245	-.117	-.154	-.124	.078	.085	-.081	-.124	-.102	-.056
	.326	-.113	-.030	-.011	-.008	-.073	-.084	.023	.020	-.016	-.058	-.045	-.044
	.343	-.060	.008	-.072	.024	-.034	-.051	.031	.029	.020	-.016	-.045	-.044
	.387	-.026	.003	.006	-.011	.022	.055	.042	.023	.007	.058	.081	.069
	.442	-.072	-.030	-.079	-.065	-.073	-.031	.029	.021	-.007	.058	.016	.029
	.493	-.001	-.024	-.079	-.108	-.081	.050	.067	.009	.009	.058	.077	.064
	.537	-.027	-.071	-.084	-.171	-.171	.014	.036	.018	-.073	-.029	.069	.025
Turbine cowl	0.000	-.068	-.101	-.157	-.305	-.423	-.010	.035	.051	-.189	-.178	.052	.042
	.031	-.109	-.003	-.112	-.301	-.642	.002	.009	.007	-.170	-.499	.044	.050
	.055	-.178	-.128	-.128	-.295	-.569	-.013	.006	-.024	-.187	.252	.036	.050
	.106	-.243	-.068	-.008	-.753	-.496	-.128	-.029	-.022	-.233	.314	-.054	.014
	.172	-.045	-.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.261	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.302	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.326	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.343	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.387	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.442	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.493	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.537	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.581	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.617	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
Plug	0.000	-.122	-.551	.864	.441	.721	-.046	.507	1.014	.475	.631	.780	.421
	.031	-.1207	-.1220	.298	-.1123	1.323	-.1314	-.1258	1.014	.475	.631	.780	.421
	.055	-.1207	-.1220	.298	-.1123	1.323	-.1314	-.1258	1.014	.475	.631	.780	.421
	.106	-.326	-.669	-.484	-.501	-.382	-.581	-.594	-.341	-.451	-.763	-.493	-.846
	.172	-.355	-.282	-.282	-.293	-.273	-.273	.002	-.284	-.378	-.318	-.416	-.444
	.261	-.176	-.072	-.034	-.213	-.067	-.089	-.094	-.084	-.080	-.020	-.020	-.020
	.302	-.082	-.043	-.033	-.034	.034	-.037	.026	-.024	-.110	.009	.031	.034
	.326	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.343	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.387	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.442	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.493	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.537	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.581	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.617	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.653	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.690	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.728	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.768	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.796	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.821	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061
	.852	-.045	.036	.036	.107	.084	.046	.042	.051	.061	.088	.118	.061

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION I - Continued

(s) $M = 0.825$; $p_{t,e}/p_{t,\infty} = 1.3$; inboard station

x/\bar{c}	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.908	.326	.635	.722	.868	.785	.319	.785	.750	.721	.320	.891	.780	.646			
	.003	-.649	-1.194	-1.205	-1.037	-.487	-1.109	-1.184	-1.262	-.941	-.980	-1.231	-1.076	-.932	-1.244			
	.014	-.416	-.889	-1.267	-.842	-.487	-.674	-.897	-.988	-.416	-.455	-.933	-.675	-.387	-.667			
	.031	-.346	-.569	-.095	-.453	-.380	-.510	-.378	-.440	-.358	-.311	-.370	-.312	-.332	-.327			
	.055	-.287	-.368	-.233	-.320	-.209	-.312	-.341	-.341	-.262	-.226	-.336	-.318	-.252	-.239			
	.106	-.196	-.256	-.239	-.188	-.120	-.190	-.232	-.239	-.135	-.103	-.224	-.216	-.119	-.101			
	.172	-.070	-.123	-.102	-.050	-.002	-.052	-.091	-.065	-.006	-.023	-.086	-.052	-.016	-.037			
	.261	-.137	-.264	-.188	-.162	-.063	-.123	-.219	-.158	-.083	-.012	-.111	-.208	-.052	-.003			
	.302	-.125	-.066	-.056	-.016	.020	-.087	-.028	-.028	.050	-.079	-.013	-.148	-.059	.009			
	.326	-.088	.017	.507	.046	.041	.252	.056	.306	.113	.105	.342	.128	.130	.099			
	.343	-.212	.122	-.121	.113	-.092	-.182	.176	-.031	.159	-.054	.173	.167	.167	.059			
	.387	-.271	.028	.041	.061	.015	-.261	.082	-.085	.149	.107	.110	.105	.109	.124			
	.442	.052	-.053	-.071	-.063	.100	.094	-.044	-.003	.066	.154	.094	.032	.078	.224			
Turbine cowl	.493	.032	-.014	-.006	-.079	.067	.110	.084	-.002	-.088	.285	.381	.019	.081	.240			
	.537	.555	-.066	-.048	-.162	-.422	.307	.040	-.080	-.062	.162	.310	.084	.016	.142			
	.581	.555	-.053	-.183	.198	.508	.295	.011	-.080	-.265	-.181	.058	.008	-.140	.432			
	.617	-.055	-.043	-.079	.372	.341	.015	.092	-.033	-.187	-.317	.114	.035	-.070	.109			
	.653	.044	-.129	-.102	-.318	-.515	.090	-.007	-.033	-.262	.131	.035	-.007	-.140	.140			
	.690	-.267	-.147	-.165	-.332	.430	-.079	.074	-.012	-.080	.019	.019	.019	-.223	.130			
	.708		-.191	-.126	-.367			-.059	.037	-.236	-.015	-.015	.094	-.013				
	.735		-.040		-.468			-.015		-.275	-.166	-.125	.008	-.210				
	.768		-.418		-.432			-.219		-.148	-.099	-.099	.008	-.032				
	.796		-.376		-.326			-.143		.128	.120	.120	.008	.032				
	.821		-.142		-.115			-.036						.169				
	.852		-.011		.017			.055										
	Plug	0.000	.681	.313	.949	.774	.582	.564	.314	1.095	.803	.453						
.003		-1.376	-1.206	-.793	-.874	1.274	-1.274	-1.204	-.210	-.729	1.267							
.014		-.148	-.921	-.486	-.603	-.368	-1.230	-.888	-.285	-.729	1.267							
.031		-.329	-.363	-.535	-.292	-.294	-.541	-.294	-.252	-.238	1.124							
.055		-.364	-.321	-.272	-.214	-.227	-.320	-.306	-.168	-.162	-.225							
.106		-.242	-.209	-.183	-.089	-.081	-.123	-.192	-.121	-.053	-.051							
.172		-.108	-.071	-.029	-.049	-.062	-.009	-.056	-.014	-.087	-.098							
.261		-.159	-.186	-.123	-.021	-.052	-.033	-.158	-.090	.033	.113							
.302		-.124	.005	-.003	.114	.137	.003	.025	.033	.160	.202							
.326		-.187	.086	.274	.161	.166	.310	.103	.294	.202	.228							
.343		-.222	.151	.039	.195	-.042	-.103	.140	.165	.228	.228							
.387		-.317	.138	.127	.202	.208	-.206	.149	.166	.251	.282							
Fan cowl		.442	.061	.041	.647	.127	.260	.184	.085	.090	.202	.329						
	.493	.073	.091	.642	.023	.254	.172	.119	.111	.202	.345							
	.537	.239	.069	.649	.023	.055	.342	.127	.068	.195	.352							
	.581	.235	.034	-.011	-.064	-.005	.290	.103	.051	.095	.352							
	.617	.030	.064	.034	-.072	-.003	.156	.145	.083	.048	.477							
	.653	.073	.039	.841	-.065	-.003	.186	.145	.103	.043	.077							
	.690	-.057	.112	.063	-.141	.148	.137	.161	.121	.312	.170							
	.708		.062	.125	-.046			.129	.189	-.098								
	.735		-.006		-.102			.004		-.084								
	.768		-.066		-.133			.025		-.340								
	.796		-.045		-.050			.038		.020								
	.821		.057		.083			.116		.157								
	.852		.169		.210			.218		.269								

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(i) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.3$; outboard station

$x/2$	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.486	.617	.586	.423	.975	.302	.591	.711	.434	.849	.240
	.003	-.753	-1.148	-.593	-1.219	-.775	-.968	-1.141	-.350	-1.188	-1.110	-.030
	.014	-.545	-.977	-1.182	-.950	-.576	-.952	-.993	-.975	-.918	-.850	-.181
	.031	-.032	-.781	-1.211	-.785	-.620	-.465	-.807	-.943	-.425	-.437	-.425
	.055	-.361	-.035	-.998	-.495	-.416	-.370	-.018	-.346	-.437	-.413	-.295
	.106	.232	-.339	-.282	-.322	-.309	.003	-.312	-.331	-.279	-.245	.315
	.172	-.125	.327	-.166	-.186	-.116	-.111	.137	-.133	-.146	-.078	.115
	.261	-.255	.312	-.234	-.296	-.213	-.221	.177	-.202	-.214	-.149	.209
	.302	-.156	-.111	.401	-.137	-.111	-.115	.074	.030	.033	-.018	.111
	.326	-.105	-.027	-.003	.101	-.044	-.031	.014	.030	.030	.001	.064
	.343	-.035	.031	-.072	.081	.032	-.038	.059	-.023	.032	.005	.040
	.387	.001	.010	-.007	-.009	.018	.021	.056	.039	.035	.102	.062
Turbine cowl	.442	-.082	-.087	-.090	-.103	-.084	-.044	.035	.012	.028	.033	.036
	.493	-.012	-.058	-.074	-.116	-.040	.031	.035	.018	.024	.024	.031
	.537	-.083	-.056	-.105	-.210	-.145	.035	.001	.018	.125	.030	.050
	.581	-.062	-.119	-.165	-.288	-.393	.027	-.018	.062	.185	.238	.043
	.617	-.145	-.045	-.082	-.338	-.599	.015	.067	.023	.180	.505	.039
Plug	.653	-.149	-.140	-.156	-.348	-.568	-.075	.033	.044	.316	.486	.005
	.690	-.345	-.229	-.108	-.411	-.660	-.122	.099	.010	.209	.463	.005
	.708	-.103	-.030	-.030	-.736	-.476	.024	.084	.080	.908	.209	.091
	.735	-.145	-.147	-.187	-.187	-.187	.033	.033	.080	.908	.209	.091
	.768	-.368	-.359	-.359	-.359	-.359	-.107	.033	.033	.908	.209	.091
Fan cowl	0.000	.164	.573	.862	.460	.748	.014	.535	1.000	.479	.654	.240
	.003	-1.079	-1.162	-.381	-1.128	-1.113	-1.293	-1.176	-.388	-.964	1.268	-.030
	.014	-1.177	-.999	-.445	-.825	-.113	-1.337	-.976	-.269	-.619	1.268	-.030
	.031	-.489	-.763	-.665	-.503	-.927	-.382	-.692	-.377	-.493	1.108	-.030
	.055	-.234	-.011	-.389	-.414	-.343	-.359	-.004	-.288	-.386	.271	.315
	.106	-.175	-.292	-.276	-.265	-.236	-.178	-.272	-.222	-.216	.163	.315
	.172	-.100	-.045	-.098	-.150	-.045	-.076	.054	.059	-.069	.006	.315
	.261	-.179	.086	-.163	-.152	-.074	.127	.049	-.132	-.106	.006	.315
	.302	-.080	-.045	.165	.028	.031	.040	-.027	.167	.067	.096	.315
	.326	-.037	.039	.055	.199	.065	.005	.054	.078	.159	.132	.315
	.343	-.033	.076	.018	.128	.066	.036	.080	.044	.159	.132	.315
	.387	-.093	.071	.092	.154	.178	.136	.084	.120	.190	.221	.315
	.442	-.002	.029	.023	.094	.159	.058	.099	.057	.122	.195	.315
Turbine cowl	.493	.077	.081	.039	.083	.125	.101	.104	.070	.132	.229	.315
	.537	.065	.081	.031	.013	.133	.105	.091	.049	.104	.229	.315
	.581	.045	.013	.205	-.152	-.041	.128	.091	.027	.027	.070	.315
	.617	.077	.040	.044	-.029	-.252	.113	.109	.093	.025	.051	.315
	.653	.041	.076	.034	-.057	-.173	.113	.104	.088	.012	.001	.315
Plug	.690	-.037	.058	.052	-.136	-.246	.034	.141	.175	-.554	-.190	.315
	.708	-.084	.084	.139	-.547	.141	.086	.086	.086	.228	.315	.315
	.735	-.034	.034	.139	-.068	.034	.086	.086	.086	.228	.315	.315
	.768	-.027	.027	.139	-.068	.034	.086	.086	.086	.228	.315	.315
	.821	-.010	.010	.139	-.068	.034	.086	.086	.086	.228	.315	.315
Plug	.852	-.079	.079	.176	-.212	.212	.193	.193	.193	.237	.315	.315
	.852	-.079	.079	.176	-.212	.212	.193	.193	.193	.237	.315	.315
	.852	-.079	.079	.176	-.212	.212	.193	.193	.193	.237	.315	.315
	.852	-.079	.079	.176	-.212	.212	.193	.193	.193	.237	.315	.315
	.852	-.079	.079	.176	-.212	.212	.193	.193	.193	.237	.315	.315

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(u) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.0$; inboard station

x/\bar{c}	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	-1.456	-1.402	-0.029	-0.058	-1.437	-1.439	-1.451	-0.044	-0.144	-0.778	-0.433	-0.236	-0.085	-0.073			
	.003	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.484	-1.060	-1.353	-1.509	-1.476	-1.433	-0.871			
	.014	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.031	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.055	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.106	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.172	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.251	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.326	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.403	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.482	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.561	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
Turbine cowl	0.000	-1.456	-1.402	-0.029	-0.058	-1.437	-1.439	-1.451	-0.044	-0.144	-0.778	-0.433	-0.236	-0.085	-0.073			
	.003	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.484	-1.060	-1.353	-1.509	-1.476	-1.433	-0.871			
	.014	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.031	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.055	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.106	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.172	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.251	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.326	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.403	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.482	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.561	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
Plug	0.000	-1.456	-1.402	-0.029	-0.058	-1.437	-1.439	-1.451	-0.044	-0.144	-0.778	-0.433	-0.236	-0.085	-0.073			
	.003	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.484	-1.060	-1.353	-1.509	-1.476	-1.433	-0.871			
	.014	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.031	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.055	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.106	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.172	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.251	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.326	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.403	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.482	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
	.561	-1.456	-1.493	-1.015	-1.624	-1.399	-1.521	-1.225	-1.509	-1.353	-1.509	-1.509	-1.509	-1.509	-0.871			
Fan cowl	0.000	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.003	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.014	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.031	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.055	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.106	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.172	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.251	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.326	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.403	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.482	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.561	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
Turbine cowl	0.000	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.003	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.014	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.031	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.055	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.106	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.172	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.251	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.326	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.403	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.482	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.561	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
Plug	0.000	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.003	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.014	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.031	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.055	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.106	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.172	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.251	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.326	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.403	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.482	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			
	.561	-1.446	-1.309	-0.086	-1.435	-1.330	-1.453	-1.219	-1.461	-1.094	-1.449	-1.161	-1.449	-0.983	-0.773			

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(v) $M = 0.700$; $p_{t,e}/p_{t,\infty} = 1.0$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.000	-.250	-.072	-.002	.439	-.437	-.111	.106	-.228	.292	-.508	-.120	.190	-.236	.206			
	.003	-1.355	-1.557	-1.084	1.556	-1.524	-1.578	.015	1.571	-1.358	-.914	-1.442	.057	-1.617	-1.171			
	.014	-.740	-1.286	-1.094	1.757	-1.193	-1.234	-.134	-1.193	-1.174	-1.156	-1.248	-.998	-1.158	-.964			
	.031	-1.055	-.689	-1.101	.531	-1.222	-.751	-.436	-.702	-.731	-1.142	-.826	-.547	-.635	-.418			
	.055	-.392	-.057	-.900	.405	-.407	-.047	-.300	-.308	-.355	-.482	-.397	-.397	-.418	-.486			
	.066	-.391	-.326	-.322	.334	-.270	-.630	-.293	-.308	-.286	-.720	-.280	-.284	-.279	-.270			
	.106	-.391	-.326	-.322	.334	-.270	-.630	-.293	-.308	-.286	-.720	-.280	-.284	-.279	-.270			
	.172	-.149	-.271	-.174	.202	-.145	-.361	-.157	-.185	-.163	-.132	-.131	-.131	-.163	-.160			
	.261	-.231	-.083	-.180	.237	-.212	-.189	-.232	-.177	-.169	-.152	-.160	-.160	-.195	-.189			
	.322	-.144	-.122	-.132	.170	-.122	-.111	-.228	-.069	-.085	-.084	-.066	-.219	-.069	-.011			
	.326	-.191	-.057	-.044	.157	-.083	-.021	-.008	-.102	-.011	-.045	-.005	.015	-.095	-.028			
	.343	-.231	-.086	-.132	.070	-.183	-.199	-.037	-.073	-.001	-.186	-.021	-.044	-.028	-.083			
	.387	-.013	.001	-.025	.018	-.028	.005	.038	.025	.057	.051	.060	.051	.051	.051			
	.442	-.033	.041	-.054	.073	.005	.005	.005	.008	.041	.013	.031	.025	.054	.086			
	.493	-.028	.044	-.060	.132	.015	.012	.008	.005	.005	.033	.038	.025	.054	.086			
Turbine cowl	.537	-.038	.044	-.070	.134	.015	.015	.015	.027	-.024	.047	.044	.021	.024	.047			
	.581	-.076	.064	-.083	.186	.009	.010	.018	.076	-.058	.052	.044	.012	.021	-.014			
	.617	-.113	.070	-.073	.338	.009	.015	.005	.063	.153	.033	.051	.031	.011	-.072			
	.653	-.125	.080	-.063	.309	.019	.014	.008	.060	.124	.037	.051	.031	.011	-.050			
	.690	-.173	-.109	-.077	.147	-.058	.004	.005	.056	-.143	-.006	.031	.038	.011	-.076			
Plug	.708	-.106	-.090	-.090	.541	-.040	-.040	.008	.018	-.143	-.006	.031	.038	-.057	-.076			
	.735	-.090	-.090	-.090	.102	-.102	-.008	.008	.014	-.006	.044	.044	.034	.034	.044			
	.768	-.030	-.030	-.030	.032	-.032	-.008	.008	.005	.005	.041	.041	.047	.047	.044			
	.796	-.077	-.077	-.077	.073	-.073	.002	.002	.005	.005	.047	.047	.060	.057	.066			
	.821	-.057	-.057	-.057	.060	-.060	.015	.015	.018	.018	.060	.060	.070	.066	.066			
.852	-.051	-.051	-.051	.051	-.051	.025	.025	.028	.028	.070	.070	.084	.066	.066				
Fan cowl	.000	-.594	-.142	-.269	.227	-.422	-.173	.517	-.196	.021	-.508	-.120	.190	-.236	.206			
	.003	-.904	-1.497	-.044	1.551	-.599	-1.577	.052	1.571	-1.358	-.914	-1.442	.057	-1.617	-1.171			
	.014	-1.059	-1.281	-.738	1.155	-.744	-1.234	-.134	-1.193	-1.174	-1.156	-1.248	-.998	-1.158	-.964			
	.031	-1.293	-.862	-.438	.622	-.848	-.762	-.393	-.702	-.731	-1.142	-.826	-.547	-.635	-.418			
	.055	-.806	-.038	-.378	.412	-.812	-.016	-.291	-.310	-.355	-.482	-.397	-.397	-.418	-.486			
	.066	-.806	-.038	-.378	.412	-.812	-.016	-.291	-.310	-.355	-.482	-.397	-.397	-.418	-.486			
	.106	-.391	-.326	-.322	.334	-.270	-.630	-.293	-.308	-.286	-.720	-.280	-.284	-.279	-.270			
	.172	-.133	-.401	-.119	.174	-.170	-.144	-.074	-.268	-.055	-.132	-.131	-.131	-.163	-.160			
	.261	-.138	-.314	-.155	.170	-.044	-.174	-.123	-.158	.023	-.160	-.160	-.160	-.195	-.189			
	.322	-.075	-.064	-.288	.057	.020	-.045	-.230	.045	.054	-.084	-.066	-.219	-.069	-.011			
	.326	-.041	-.031	-.020	.048	.032	-.045	.052	.016	.120	-.045	-.005	.015	-.095	-.028			
	.343	-.197	-.012	-.028	.049	-.003	.020	.052	.016	.120	-.045	-.005	.015	-.095	-.028			
	.387	-.022	.066	.066	.117	.111	.055	.104	.103	.013	.013	.013	.013	.013	.013			
	.442	.022	.040	.040	.085	.127	.085	.145	.145	.207	.142	.142	.142	.142	.142			
	Turbine cowl	.493	.041	.053	.040	.075	.117	.085	.078	.145	.207	.142	.142	.142	.142	.142		
.537		.051	.059	.036	.059	.107	.094	.085	.129	.213	.165	.165	.165	.165	.165			
.581		.070	.062	.036	.014	.039	.104	.097	.100	.110	.116	.116	.116	.116	.116			
.617		.065	.069	.049	.027	.019	.128	.097	.100	.110	.116	.116	.116	.116	.116			
.655		.055	.072	.053	.020	.004	.133	.097	.100	.110	.116	.116	.116	.116	.116			
Plug	.693	.022	.062	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078			
	.708	.033	.033	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078			
	.735	.062	.062	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078			
	.768	.062	.062	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078			
	.796	.066	.066	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078			
.821	.075	.075	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078				
.852	.085	.085	.059	.020	-.019	.113	.117	.117	.084	.078	.078	.078	.078	.078				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(w) $M = 0.750$; $p_{t,e}/p_{t,\infty} = 1.0$; inboard station

x/c	C _p at																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.369	-.279	.108	.147	.271	.231	-.314	.211	.152	.099	-.334	.296	.158	.040			
	.003	-.1445	1.359	-.950	1.430	1.433	-.1428	1.365	-.1441	1.433	-.1290	1.430	1.366	1.437	-.884			
	.014	-.579	1.397	-.977	1.122	.605	-.1109	1.347	-.1483	1.021	-.968	1.201	1.125	1.457	-.872			
	.031	-.396	-.479	-.941	-.451	-.331	-.666	-.557	-.1021	-.497	-.579	-.804	-.1039	1.457	-.953			
	.055	-.223	-.414	-.848	-.372	-.278	-.400	-.397	-.393	-.301	-.284	-.414	-.343	1.457	-.953			
	.106	-.223	-.285	-.344	-.235	-.159	-.209	-.243	-.229	-.175	-.126	-.229	-.235	1.457	-.953			
	.172	-.104	-.156	-.127	-.097	-.042	-.094	-.129	-.105	-.038	-.020	-.085	-.114	1.457	-.953			
	.261	-.170	-.268	-.153	-.179	-.094	-.130	-.217	-.170	-.099	-.032	-.103	-.187	1.457	-.953			
	.302	-.135	-.092	-.080	-.062	-.021	-.103	-.053	-.056	.009	.044	-.038	-.041	1.457	-.953			
	.326	-.047	-.024	.087	.001	.005	-.090	.015	.006	.065	.079	.050	.018	1.457	-.953			
	.343	-.175	-.033	-.127	-.030	-.147	-.147	.003	-.061	.044	.079	.030	.018	1.457	-.953			
	.387	-.210	.014	.007	.014	.014	-.178	.056	.053	.094	.114	.074	.076	1.457	-.953			
Turbine cowl	.442	.015	-.036	-.051	-.045	.043	.048	.015	.006	.047	.135	.065	.032	.091	.173			
	.493	.007	-.036	-.062	-.037	.027	.043	.021	.003	.018	.097	.041	.032	.091	.152			
	.537	.091	-.048	-.083	-.147	.131	.074	.018	.014	.029	-.044	.110	.044	.029	.041			
	.581	.253	-.083	-.124	-.205	-.054	.038	-.003	.041	.079	-.050	.176	.035	.064	.041			
	.617	.073	-.092	-.115	-.237	.351	.008	.003	.032	.108	-.126	.052	.006	.015	.117			
	.653	-.047	-.109	-.121	-.226	.369	.008	.006	.032	.102	-.152	.043	.009	.056	.032			
	.690	-.206	-.127	-.133	-.226	.072	-.072	-.020	.035	.117	-.032	.074	.027	.009	.047			
	.708		-.168	-.165	-.179			-.061	.061	.085	.074	.009	.029	.061	.000			
	.735		-.033		.359			.020	.061	.199	-.009	.009	.009	.029	.190			
	.768		-.165		.170			.061	.061	.073	-.017	.017	.017	.032	.190			
	.821		-.130		.144			-.050	-.050	.058	-.012	.012	.012	.032	.190			
	.852		-.136		.132			-.038	-.038	.044	.009	.009	.009	.032	.190			
Fan cowl	0.000	.119	-.333	.382	.172	.031	.035	-.339	.585	.179	.162							
	.003	-.933	1.353	1.411	1.439	1.433	-.642	1.075	1.198	1.376	.625							
	.014	-.926	1.144	.821	1.439	.772	-.644	-.101	-.412	-.853	.619							
	.031	-.938	1.144	1.053	1.439	.795	-.715	.839	.641	-.412	.619							
	.055	-.772	1.144	1.053	1.439	.795	-.715	.839	.641	-.412	.619							
	.106	-.304	-.220	-.205	-.115	-.099	-.614	.474	.228	.247	.495							
	.172	-.143	-.170	-.062	.033	.006	-.098	.254	.113	.113	.250							
	.261	-.132	-.179	-.133	.041	.006	.003	.113	.019	.032	.039							
	.302	-.136	-.018	-.026	.073	.114	.027	.160	.001	.021	.118							
	.326	-.136	.044	.044	.073	.114	.027	.160	.001	.021	.118							
	.343	-.136	.044	.044	.073	.114	.027	.160	.001	.021	.118							
	.387	-.229	.091	.103	.105	.032	-.081	.057	.039	.162	.217							
Turbine cowl	.442	-.003	.059	.059	.132	.217	.131	.115	.139	.225	.276							
	.493	.002	.062	.059	.108	.202	.131	.104	.104	.195	.290							
	.537	.010	.073	.053	.070	.102	.131	.110	.098	.195	.290							
	.581	-.003	.065	.006	.038	.146	.131	.113	.077	.195	.290							
	.617	-.003	.073	.044	-.003	.041	.140	.124	.092	.080	.168							
	.653	-.016	.073	.047	-.030	.003	.122	.133	.098	.080	.168							
	.690	-.047	.068	.047	-.018	.018	.118	.133	.098	.080	.168							
	.708		.032	.029	.033			.092	.086	.080	.168							
	.735		-.003		-.164			-.002		.066								
	.768		.023		-.003			.083		.077								
	.821		.029		.026			.092		.100								
	.852		.044		.044			.104		.106								
Plug	0.000	.387	-.207	.332	.105	.032	.035	-.339	.585	.179	.162							
	.003	-.933	1.353	1.411	1.439	1.433	-.642	1.075	1.198	1.376	.625							
	.014	-.926	1.144	.821	1.439	.772	-.644	-.101	-.412	-.853	.619							
	.031	-.938	1.144	1.053	1.439	.795	-.715	.839	.641	-.412	.619							
	.055	-.772	1.144	1.053	1.439	.795	-.715	.839	.641	-.412	.619							
	.106	-.304	-.220	-.205	-.115	-.099	-.614	.474	.228	.247	.495							
	.172	-.143	-.170	-.062	.033	.006	-.098	.254	.113	.113	.250							
	.261	-.132	-.179	-.133	.041	.006	.003	.113	.019	.032	.039							
	.302	-.136	-.018	-.026	.073	.114	.027	.160	.001	.021	.118							
	.326	-.136	.044	.044	.073	.114	.027	.160	.001	.021	.118							
	.343	-.136	.044	.044	.073	.114	.027	.160	.001	.021	.118							
	.387	-.229	.091	.103	.105	.032	-.081	.057	.039	.162	.217							
Plug	.442	-.003	.059	.059	.132	.217	.131	.115	.139	.225	.276							
	.493	.002	.062	.059	.108	.202	.131	.104	.104	.195	.290							
	.537	.010	.073	.053	.070	.102	.131	.110	.098	.195	.290							
	.581	-.003	.065	.006	.038	.146	.131	.113	.077	.195	.290							
	.617	-.003	.073	.044	-.003	.041	.140	.124	.092	.080	.168							
	.653	-.016	.073	.047	-.030	.003	.122	.133	.098	.080	.168							
	.690	-.047	.068	.047	-.018	.018	.118	.133	.098	.080	.168							
	.708		.032	.029	.033			.092	.086	.080	.168							
	.735		-.003		-.164			-.002		.066								
	.768		.023		-.003			.083		.077								
	.821		.029		.026			.092		.100								
	.852		.044		.044			.104		.106								

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(x) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.0$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	-0.157	0.020	-0.083	0.114	0.207	-0.300	-0.003	0.180	0.350	-0.393	-0.039	0.251	-0.110	0.286			
	0.003	-1.358	1.420	0.212	1.428	1.428	-1.522	1.423	1.433	1.431	-1.001	1.424	1.432	-1.184				
	0.014	-1.044	1.420	-1.049	1.428	-1.136	-1.615	1.423	1.423	1.431	-1.134	1.424	1.432	-1.152				
	0.031	-0.868	0.800	-1.067	0.639	0.533	-1.080	0.450	0.444	0.444	-0.979	0.618	0.444	-0.963				
	0.055	-0.391	0.051	-0.943	0.439	0.418	-0.390	0.441	0.390	0.415	-0.603	0.032	0.414	-0.439				
	0.106	-0.232	0.346	-0.402	0.354	0.289	-0.448	0.316	0.325	0.256	-0.545	0.210	0.310	-0.565				
	0.172	-0.168	0.107	-0.175	0.260	0.198	-0.147	0.207	0.156	0.168	-0.134	0.139	0.177	-0.221				
	0.261	-0.241	0.079	-0.175	0.277	0.213	-0.209	0.062	0.192	0.162	-0.161	0.097	0.174	-0.221				
	0.302	-0.157	0.119	-0.020	0.257	0.157	-0.125	0.085	0.091	0.065	-0.085	0.074	0.089	-0.171				
	0.326	-0.134	0.051	-0.039	0.177	0.080	-0.076	0.014	0.054	0.020	-0.046	0.003	0.012	-0.141				
	0.343	-0.214	0.072	-0.122	0.057	0.165	-0.205	0.035	0.065	0.012	-0.178	-0.015	0.038	-0.082				
Turbine cowl	0.000	-0.037	0.008	-0.021	0.013	0.018	0.012	0.003	0.003	0.012	0.023	0.023	0.038	0.092				
	0.003	-0.024	0.039	-0.054	0.077	0.045	-0.001	0.009	0.009	0.012	0.016	0.027	0.021	0.053				
	0.014	-0.016	0.045	-0.063	0.113	0.116	0.008	0.012	0.012	0.008	0.039	0.032	0.038	0.065				
	0.031	-0.033	0.049	-0.075	0.151	0.180	0.012	0.012	0.012	0.012	0.052	0.045	0.015	0.047				
	0.055	-0.073	0.063	-0.092	0.210	0.339	0.008	0.006	0.023	0.091	0.038	0.045	0.006	0.029				
	0.106	-0.117	0.075	-0.083	0.192	0.363	0.019	0.006	0.023	0.079	0.034	0.045	0.021	0.073				
	0.172	-0.139	0.059	-0.092	0.192	0.327	0.023	0.003	0.014	0.082	0.030	0.042	0.021	0.073				
	0.261	-0.192	0.122	-0.092	0.168	0.339	-0.076	0.026	0.012	0.073	-0.019	0.021	0.026	-0.094				
	0.302	-0.119	0.059	-0.104	0.152	0.339	-0.076	0.032	0.014	0.073	-0.019	0.021	0.026	-0.094				
	0.326	-0.104	0.044	-0.127	0.127	0.339	-0.076	-0.014	0.014	0.073	-0.019	0.021	0.026	-0.094				
	0.343	-0.104	0.044	-0.127	0.127	0.339	-0.076	-0.014	0.014	0.073	-0.019	0.021	0.026	-0.094				
Plug	0.000	-0.441	0.052	0.329	-0.116	0.237	-0.525	-0.094	0.513	0.087	0.125	0.087	0.125	0.087				
	0.003	-0.799	1.428	0.257	1.434	0.912	-0.581	-1.347	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.014	-1.958	1.428	-0.887	1.434	-0.942	-0.700	-1.181	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.031	-0.900	0.772	-0.485	0.480	0.612	-0.766	-0.903	0.430	0.564	-0.776	0.564	0.776	0.776				
	0.055	-0.776	0.026	-0.393	0.462	0.771	-0.766	-0.025	0.315	0.458	-0.782	0.458	0.782	0.782				
	0.106	-0.644	0.296	-0.275	0.277	0.285	-0.594	0.312	0.229	0.293	-0.440	0.229	0.440	0.440				
	0.172	-0.118	0.257	-0.112	0.171	0.053	-0.183	0.274	0.076	0.164	-0.108	0.076	0.108	0.108				
	0.261	-0.127	0.145	-0.151	0.162	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.302	-0.065	0.053	-0.112	0.102	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.326	-0.029	0.015	-0.036	0.015	0.071	-0.007	0.031	0.054	0.048	-0.048	0.054	0.048	0.048				
	0.343	-0.171	0.003	-0.015	0.065	0.053	-0.148	0.016	0.019	0.110	-0.008	0.019	0.110	0.110				
Fan cowl	0.000	-0.441	0.052	0.329	-0.116	0.237	-0.525	-0.094	0.513	0.087	0.125	0.087	0.125	0.087				
	0.003	-0.799	1.428	0.257	1.434	0.912	-0.581	-1.347	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.014	-1.958	1.428	-0.887	1.434	-0.942	-0.700	-1.181	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.031	-0.900	0.772	-0.485	0.480	0.612	-0.766	-0.903	0.430	0.564	-0.776	0.564	0.776	0.776				
	0.055	-0.776	0.026	-0.393	0.462	0.771	-0.766	-0.025	0.315	0.458	-0.782	0.458	0.782	0.782				
	0.106	-0.644	0.296	-0.275	0.277	0.285	-0.594	0.312	0.229	0.293	-0.440	0.229	0.440	0.440				
	0.172	-0.118	0.257	-0.112	0.171	0.053	-0.183	0.274	0.076	0.164	-0.108	0.076	0.108	0.108				
	0.261	-0.127	0.145	-0.151	0.162	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.302	-0.065	0.053	-0.112	0.102	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.326	-0.029	0.015	-0.036	0.015	0.071	-0.007	0.031	0.054	0.048	-0.048	0.054	0.048	0.048				
	0.343	-0.171	0.003	-0.015	0.065	0.053	-0.148	0.016	0.019	0.110	-0.008	0.019	0.110	0.110				
Turbine cowl	0.000	-0.037	0.008	-0.021	0.013	0.018	0.012	0.003	0.003	0.012	0.023	0.023	0.038	0.092				
	0.003	-0.024	0.039	-0.054	0.077	0.045	-0.001	0.009	0.009	0.012	0.016	0.027	0.021	0.053				
	0.014	-0.016	0.049	-0.063	0.113	0.116	0.008	0.012	0.012	0.008	0.039	0.032	0.038	0.065				
	0.031	-0.033	0.049	-0.075	0.151	0.180	0.012	0.012	0.012	0.012	0.052	0.045	0.015	0.047				
	0.055	-0.073	0.063	-0.092	0.210	0.339	0.008	0.006	0.023	0.091	0.038	0.045	0.006	0.029				
	0.106	-0.117	0.075	-0.083	0.192	0.363	0.019	0.006	0.023	0.079	0.034	0.045	0.021	0.073				
	0.172	-0.139	0.059	-0.092	0.192	0.327	0.023	0.003	0.014	0.082	0.030	0.042	0.021	0.073				
	0.261	-0.192	0.122	-0.092	0.168	0.339	-0.076	0.026	0.012	0.073	-0.019	0.021	0.026	-0.094				
	0.302	-0.119	0.059	-0.104	0.152	0.339	-0.076	0.032	0.014	0.073	-0.019	0.021	0.026	-0.094				
	0.326	-0.104	0.044	-0.127	0.127	0.339	-0.076	-0.014	0.014	0.073	-0.019	0.021	0.026	-0.094				
	0.343	-0.104	0.044	-0.127	0.127	0.339	-0.076	-0.014	0.014	0.073	-0.019	0.021	0.026	-0.094				
Plug	0.000	-0.441	0.052	0.329	-0.116	0.237	-0.525	-0.094	0.513	0.087	0.125	0.087	0.125	0.087				
	0.003	-0.799	1.428	0.257	1.434	0.912	-0.581	-1.347	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.014	-1.958	1.428	-0.887	1.434	-0.942	-0.700	-1.181	0.415	1.425	-0.746	1.425	0.746	1.425				
	0.031	-0.900	0.772	-0.485	0.480	0.612	-0.766	-0.903	0.430	0.564	-0.776	0.564	0.776	0.776				
	0.055	-0.776	0.026	-0.393	0.462	0.771	-0.766	-0.025	0.315	0.458	-0.782	0.458	0.782	0.782				
	0.106	-0.644	0.296	-0.275	0.277	0.285	-0.594	0.312	0.229	0.293	-0.440	0.229	0.440	0.440				
	0.172	-0.118	0.257	-0.112	0.171	0.053	-0.183	0.274	0.076	0.164	-0.108	0.076	0.108	0.108				
	0.261	-0.127	0.145	-0.151	0.162	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.302	-0.065	0.053	-0.112	0.102	0.038	-0.050	0.179	0.129	0.031	-0.031	0.129	0.031	0.031				
	0.326	-0.029	0.015	-0.036	0.015	0.071	-0.007	0.031	0.054	0.048	-0.048	0.054	0.048	0.048				
	0.343	-0.171	0.003	-0.015	0.065	0.053	-0.148	0.016	0.019	0.110	-0.008	0.019	0.110	0.110				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(v) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.0$; inboard station

x/c	C _p at -														
	α = -2°					α = 0°					α = 1°				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.405	-.217	.141	.198	.311	.266	-.256	.235	.203	.140	.196	-.282	.341	.070
	.003	-.1461	1.301	-.965	1.370	1.370	-.1627	1.308	1.302	1.314	-.1.082	1.314	1.314	1.314	.196
	.014	-.1.057	1.327	-.993	1.396	1.396	-.1.121	1.308	1.302	1.314	-.1.145	1.314	1.314	1.314	.196
	.031	-.378	-.622	-.776	-.379	-.345	-.589	-.4294	-.1.006	-.324	-.638	-.584	-.584	-.987	-.813
	.055	-.336	-.355	-.835	-.359	-.272	-.394	-.297	-.290	-.265	-.265	-.265	-.265	-.317	-.759
	.106	-.225	-.281	-.369	-.233	-.157	-.207	-.207	-.172	-.119	-.119	-.119	-.119	-.229	-.579
	.172	-.098	-.158	-.130	-.087	-.037	-.088	-.128	-.105	-.032	-.073	-.105	-.073	-.147	-.080
	.261	-.170	-.273	-.141	-.180	-.090	-.130	-.235	-.179	-.105	-.024	-.099	-.127	-.001	.014
	.302	-.132	-.093	-.073	-.056	-.014	-.100	-.058	-.063	-.033	-.053	-.073	-.053	-.060	.010
	.326	.089	-.020	.211	-.011	-.011	-.011	-.013	.078	.089	.083	.083	.083	.083	.083
	.343	-.170	-.023	-.121	-.012	-.129	-.139	.013	.049	.055	.071	.071	.071	.071	.131
	.387	-.204	.019	-.000	.025	.030	-.168	.058	.082	.105	.119	.120	.035	.015	.051
Turbine cowl	.442	.021	-.034	-.051	-.051	.056	.048	.013	.082	.092	.142	.075	.087	.041	.148
	.493	.004	-.037	-.068	-.098	-.017	.044	.016	.007	.018	.100	.075	.049	.038	.198
	.537	.212	-.051	-.093	-.157	-.199	.129	.013	.027	-.032	.182	.052	.027	.030	.041
	.531	.493	-.090	-.138	-.230	-.126	.210	-.010	.049	.080	.013	.254	.035	-.001	.207
	.517	-.060	-.099	-.130	-.278	-.410	.014	.007	.046	.119	.147	.067	.044	.013	.052
	.553	-.047	-.121	-.135	-.264	-.446	.006	.006	.049	.114	.178	.041	.035	.013	-.083
	.590	-.229	-.141	-.147	-.255	-.423	-.083	-.035	.052	.125	.027	-.014	.033	.013	.069
	.708	-.192	-.194	-.183	-.194	-.215	-.083	-.035	.052	.125	.027	-.014	.033	.013	.069
	.735	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	-.034	.063
	.768	-.186	-.186	-.185	-.185	-.185	-.185	-.185	-.185	-.185	-.185	-.185	-.185	-.185	.063
	.796	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	-.177	.063
	.821	-.166	-.166	-.163	-.163	-.163	-.163	-.163	-.163	-.163	-.163	-.163	-.163	-.163	.063
Plug	.852	-.149	-.149	-.146	-.146	-.146	-.146	-.146	-.146	-.146	-.146	-.146	-.146	-.146	.063
	0.000	.166	-.274	.399	.210	.034	.065	-.281	.587	.229	.105	.105	.105	.105	.105
	.003	-.900	1.306	1.304	1.366	1.366	-.754	1.308	1.302	1.314	-.1.082	1.314	1.314	1.314	.196
	.014	-.972	1.322	1.100	1.366	1.366	-.754	1.308	1.302	1.314	-.1.082	1.314	1.314	1.314	.196
	.031	-.887	-.706	-.945	-.945	-.765	-.705	-.960	-.417	-.333	-.576	-.417	-.333	-.576	.196
	.055	-.785	-.402	-.312	-.277	-.768	-.743	-.831	-.631	-.382	-.640	-.631	-.382	-.640	.196
	.106	-.305	-.227	-.210	-.137	-.103	-.386	-.234	-.153	-.105	-.265	-.105	-.239	-.265	.196
	.172	-.135	-.101	-.059	-.005	-.020	-.089	-.097	-.012	-.041	-.061	-.012	-.041	-.061	.196
	.261	-.144	-.182	-.140	-.036	.048	.013	-.150	-.102	.007	.131	.013	.007	.131	.196
	.302	-.127	-.016	-.019	-.082	.124	.035	.013	.010	.134	.195	.010	.134	.195	.196
	.326	-.055	.051	.074	.127	.155	.077	.078	.084	.170	.223	.084	.170	.223	.196
	.343	-.199	.043	.006	.113	.174	-.072	.126	.145	.235	.288	.145	.235	.288	.196
Fan cowl	.387	-.220	.099	.102	.174	.208	-.072	.126	.145	.235	.288	.145	.235	.288	.196
	.442	.009	.057	.065	.132	.219	.132	.095	.106	.187	.296	.106	.187	.296	.196
	.493	.005	.071	.065	.104	.205	.141	.106	.106	.187	.296	.106	.187	.296	.196
	.537	.077	.074	.051	.073	.101	.192	.114	.100	.153	.221	.100	.153	.221	.196
	.581	.145	.065	.026	.023	.228	.239	.114	.100	.153	.221	.100	.153	.221	.196
	.617	.005	.076	.037	-.011	.037	.149	.129	.089	.075	.162	.089	.075	.162	.196
	.653	-.016	.068	.040	-.008	-.011	.124	.129	.089	.075	.162	.089	.075	.162	.196
	.690	-.046	.060	.037	-.036	.082	.115	.129	.089	.075	.162	.089	.075	.162	.196
	.708	-.008	.023	.023	.003	.082	.115	.129	.089	.075	.162	.089	.075	.162	.196
	.735	-.017	.008	.005	-.123	.005	.005	.089	.084	.059	.136	.089	.059	.136	.196
	.768	-.017	.008	.005	-.123	.005	.005	.089	.084	.059	.136	.089	.059	.136	.196
	.796	-.023	.008	.005	-.123	.005	.005	.089	.084	.059	.136	.089	.059	.136	.196
Plug	.821	.026	.026	.026	.026	.026	.026	.026	.026	.026	.026	.026	.026	.026	.196
	.852	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.037	.196

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(a) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.0$; outboard station

κ/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	0.300	-.086	.076	.083	-.053	.536	-.264	.034	.224	-.068	.393	.013
	.003	-1.301	1.360	.329	1.368	1.368	-1.458	1.364	.200	1.372	1.372	1.370
	.014	-.631	1.360	1.360	-1.241	-1.241	-1.555	1.364	.301	1.372	1.372	1.378
	.031	-.386	1.284	1.360	-1.258	-1.258	-.978	1.241	-.051	1.372	1.372	1.378
	.055	-.093	1.085	1.360	-.332	-.424	-.304	-.072	.333	1.372	1.372	1.378
	.106	-.093	1.340	1.284	-.332	-.308	-.304	-.310	.316	1.372	1.372	1.378
	.172	-.149	.028	1.181	-.232	-.186	-.138	-.100	.187	1.372	1.372	1.378
	.261	-.250	.181	1.221	-.288	-.217	-.215	-.075	.157	1.372	1.372	1.378
	.302	-.157	.125	1.139	-.150	-.254	-.121	-.086	.007	1.372	1.372	1.378
	.326	-.106	.068	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.343	-.099	-.063	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.367	-.021	.011	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
Turbine cowl	.432	-.017	-.040	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.493	-.030	-.051	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.537	-.072	-.088	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.581	-.123	-.085	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.617	-.153	-.108	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.653	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.690	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.708	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.733	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.768	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.796	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	.821	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
Plug	.852	-.208	-.142	1.029	-.074	-.074	-.070	-.015	.002	1.372	1.372	1.378
	0.000	-.372	.014	.380	-.040	.306	-.453	-.040	.522	-.027	.177	.028
	.003	-.754	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.014	-.923	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.031	-.775	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.055	-.805	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.106	-.521	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.172	-.118	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.261	-.118	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.302	-.063	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.326	-.033	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232
	.343	-.033	1.359	1.275	1.370	1.370	-.925	1.392	.218	1.374	1.374	.232

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(aa) $M = 0.800$; $p_{t,e}/p_{t,\infty} = 1.0$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	0.000	.451	.153	.149	.367	.299	.197	.287	.233	.184	.239	.203
	.003	.1508	.1251	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.014	.1191	.1321	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.031	.0354	.1104	.1051	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.055	.0334	.087	.074	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.106	.0224	.0260	.0203	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.172	.0089	.0139	.0122	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.261	.0167	.0274	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.302	.0134	.0287	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.326	.0287	.014	.0354	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.343	.0162	.003	.090	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.367	.0191	.032	.010	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
Turbine cowl	0.000	.451	.153	.149	.367	.299	.197	.287	.233	.184	.239	.203
	.003	.1508	.1251	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.014	.1191	.1321	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.031	.0354	.1104	.1051	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.055	.0334	.087	.074	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.106	.0224	.0260	.0203	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.172	.0089	.0139	.0122	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.261	.0167	.0274	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.302	.0134	.0287	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.326	.0287	.014	.0354	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.343	.0162	.003	.090	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.367	.0191	.032	.010	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
Plug	0.000	.451	.153	.149	.367	.299	.197	.287	.233	.184	.239	.203
	.003	.1508	.1251	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.014	.1191	.1321	.1246	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.031	.0354	.1104	.1051	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.055	.0334	.087	.074	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.106	.0224	.0260	.0203	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.172	.0089	.0139	.0122	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.261	.0167	.0274	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.302	.0134	.0287	.0206	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.326	.0287	.014	.0354	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.343	.0162	.003	.090	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261
	.367	.0191	.032	.010	.1314	.1314	.1259	1.256	1.322	1.322	.1346	1.261

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(bb) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.0$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.300	-.008	.122	.142	-.002	.578	-.205	.082	.271	.004	.427	.075	.342	.010	.349			
	.303	-.124	1.305	.457	1.312	1.312	1.392	1.313	.284	1.320	1.320	1.317	.323	1.325	1.325			
	.311	-.126	1.305	1.305	1.312	1.229	1.470	1.310	1.313	1.320	1.320	1.317	1.325	1.325	1.325			
	.321	-.317	1.278	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.325	-.085	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.335	-.325	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.345	-.066	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.355	-.172	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.365	-.260	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.375	-.302	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.385	-.326	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.395	-.343	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.405	-.357	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.415	-.371	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.425	-.385	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
Turbine cowl	0.300	-.008	.122	.142	-.002	.578	-.205	.082	.271	.004	.427	.075	.342	.010	.349			
	.303	-.124	1.305	.457	1.312	1.312	1.392	1.313	.284	1.320	1.320	1.317	.323	1.325	1.325			
	.311	-.126	1.305	1.305	1.312	1.229	1.470	1.310	1.313	1.320	1.320	1.317	1.325	1.325	1.325			
	.321	-.317	1.278	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.325	-.085	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.335	-.325	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.345	-.066	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.355	-.172	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.365	-.260	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.375	-.302	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.385	-.326	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.395	-.343	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.405	-.357	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.415	-.371	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.425	-.385	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
Plug	0.300	-.008	.122	.142	-.002	.578	-.205	.082	.271	.004	.427	.075	.342	.010	.349			
	.303	-.124	1.305	.457	1.312	1.312	1.392	1.313	.284	1.320	1.320	1.317	.323	1.325	1.325			
	.311	-.126	1.305	1.305	1.312	1.229	1.470	1.310	1.313	1.320	1.320	1.317	1.325	1.325	1.325			
	.321	-.317	1.278	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.325	-.085	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.335	-.325	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.345	-.066	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.355	-.172	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.365	-.260	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.375	-.302	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.385	-.326	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.395	-.343	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.405	-.357	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.415	-.371	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			
	.425	-.385	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284	1.284			

Fan cowl	0.300	-.315	.056	.408	.004	.323	-.406	.024	.358	.030	.226				
	.303	-.748	1.306	.278	1.311	.940	-.585	1.313	.296	1.318	-.728				
	.311	-.927	1.306	.219	1.311	.910	-.736	1.255	.475	1.366	-.739				
	.321	-.638	1.285	.592	.864	.891	-.487	.895	.475	.408	-.753				
	.325	.780	.063	.337	.329	.774	.740	.033	.352	.487	.750				
	.335	.377	.303	.295	.310	.269	.357	.328	.249	.310	.468				
	.345	.112	-.055	.120	.198	.049	.190	.067	.083	.145	.039				
	.355	.120	.019	.172	.163	.030	.043	.014	.140	.153	.040				
	.365	.067	.034	.054	.035	.038	.006	.031	.065	.051	.105				
	.375	-.165	-.000	.043	.033	.026	.018	.021	.027	.121	.135				
	.385	.043	.076	.076	.076	.076	.075	.097	.076	.111	.178				
	.395	.035	.038	.041	.041	.041	.070	.070	.076	.076	.140				
	.405	.056	.052	.035	.035	.062	.100	.081	.076	.129	.200				
	.415	.072	.057	.030	.030	.068	.112	.092	.070	.102	.184				
	.425	.081	.047	.043	.022	.014	.000	.024	.062	.056	.110				
Turbine cowl	0.300	-.315	.056	.408	.004	.323	-.406	.024	.358	.030	.226				
	.303	-.748	1.306	.278	1.311	.940	-.585	1.313	.296	1.318	-.728				
	.311	-.927	1.306	.219	1.311	.910	-.736	1.255	.475	1.366	-.739				
	.321	-.638	1.285	.592	.864	.891	-.487	.895	.475	.408	-.753				
	.325	.780	.063	.337	.329	.774	.740	.033	.352	.487	.750				
	.335	.377	.303	.295	.310	.269	.357	.328	.249	.310	.468				
	.345	.112	-.055	.120	.198	.049	.190	.067	.083	.145	.039				
	.355	.120	.019	.172	.163	.030	.043	.014	.140	.153	.040				
	.365	.067	.034	.054	.035	.038	.006	.031	.065	.051	.105				
	.375	-.165	-.000	.043	.033	.026	.018	.021	.027	.121	.135				
	.385	.043	.076	.076	.076	.076	.075	.097	.076	.111	.178				
	.395	.035	.038	.041	.041	.041	.070	.070	.076	.076	.140				
	.405	.056	.052	.035	.035	.062	.100	.081	.076	.129	.200				
	.415	.072	.057	.030	.030	.068	.112	.092	.070	.102	.184				
	.425	.081	.047	.043	.022	.014	.000	.024	.062	.056	.110				
Plug	0.300	-.315	.056	.408	.004	.323	-.406	.024	.358	.030	.226				
	.303	-.748	1.306	.278	1.311	.940	-.585	1.313	.296	1.318	-.728				
	.311	-.927	1.306	.219	1.311	.910	-.736	1.255	.475	1.366	-.739				
	.321	-.638	1.285	.592	.864	.891	-.487	.895	.475	.408	-.753				
	.325	.780	.063	.337	.329	.774	.740	.033	.352	.487	.750				
	.335	.377	.303	.295	.310	.269	.357	.328	.249	.310	.468				
	.345	.112	-.055	.120	.198	.049	.190	.067	.083	.145	.039				
	.355	.120	.019	.172	.163	.030	.043	.014	.140	.153	.040				
	.365	.067	.034	.054	.035	.038	.006	.031	.065	.051	.105				
	.375	-.165	-.000	.043	.033	.026	.018	.021	.027	.121	.135				
	.385	.043	.076	.076	.076	.076	.075	.097	.076	.111	.178				
	.395	.035	.038	.041	.041	.041	.070	.070	.076	.076	.140				
	.405	.056	.052	.035	.035	.062	.100	.081	.076	.129	.200				
	.415	.072	.057	.030	.030	.068	.112	.092	.070	.102	.184				
	.425	.081	.047	.043	.022	.014	.000	.024	.062	.056	.110				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(cc) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.0$; inboard station

x/\bar{c}	C_p at -												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.466	-.122	.175	.268	.372	-.342	-.137	.324	.283	.225	-.152	.407
	.003	-.1417	1.205	1.202	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.014	-.157	1.254	1.202	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.031	-.284	1.167	1.116	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.055	-.319	1.059	.643	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.106	-.217	1.231	1.111	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.172	-.079	1.231	1.111	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.261	-.142	1.270	1.177	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.302	-.126	1.270	1.177	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
	.326	-.582	1.002	.540	1.256	1.266	-.1516	1.214	1.214	1.275	1.275	1.207	1.270
Turbine cowl	.343	-.138	1.009	.062	.024	.095	-.116	.044	.019	.088	.111	.103	.054
	.387	-.177	.043	.024	.058	.081	-.148	.078	.075	.127	.156	.171	.109
	.442	.023	-.022	-.066	-.022	.097	.061	.026	.013	.026	.119	.078	.047
	.493	.019	-.028	-.064	-.077	.053	.428	.020	.021	.026	.119	.078	.047
	.537	.659	-.066	-.095	-.137	1.158	.458	.365	.006	.055	.099	.084	.013
	.581	.622	-.085	-.152	-.233	.645	.365	.006	.055	.099	.084	.013	.013
	.617	-.028	-.103	-.155	-.303	.365	.026	-.011	.053	.159	.185	.071	.028
	.653	-.055	-.132	-.176	-.332	.458	.034	-.029	.064	.175	.258	.027	.013
	.690	-.272	-.176	-.202	-.437	.544	-.108	-.050	.074	.175	.234	-.068	.000
	.735	-.066	-.260	-.262	-.280	.262	-.110	.110	.108	.128	.154	-.058	-.055
Plug	.735	-.066	-.260	-.262	-.280	.262	-.110	.110	.108	.128	.154	-.058	-.055
	.768	-.270	-.260	-.267	-.267	.267	-.113	.113	.113	.123	.154	-.063	-.065
	.821	-.247	-.261	-.261	-.261	.261	-.105	.105	.105	.102	.118	-.063	-.063
	.852	-.231	-.231	-.233	-.233	.233	-.089	.089	.089	.086	.102	-.050	-.045
Fan cowl	0.000	.249	-.171	.445	.295	.109	.144	-.154	.610	.295	.010	-.185	.407
	.003	-.823	1.217	1.217	1.218	1.218	-.784	1.130	1.130	1.275	1.275	1.207	1.270
	.014	-.839	1.253	1.253	1.253	1.253	-.784	1.130	1.130	1.275	1.275	1.207	1.270
	.031	-.813	1.253	1.253	1.253	1.253	-.784	1.130	1.130	1.275	1.275	1.207	1.270
	.055	-.657	1.253	1.253	1.253	1.253	-.784	1.130	1.130	1.275	1.275	1.207	1.270
	.106	-.232	1.216	1.195	1.211	1.211	-.425	.461	.461	.461	.461	.461	.461
	.172	-.027	1.078	.036	.027	.066	-.145	.075	.003	.066	.066	.066	.066
	.261	-.035	1.169	.130	.020	.079	-.039	.127	.096	.027	.154	.027	.154
	.302	-.011	1.003	.010	.102	.132	-.027	.034	.024	.149	.214	.149	.214
	.326	-.086	.069	.027	.146	.180	-.129	.100	.096	.185	.242	.185	.242
Turbine cowl	.343	-.086	.069	.027	.146	.180	-.129	.100	.096	.185	.242	.185	.242
	.387	-.110	.118	.123	.198	.237	-.133	.144	.162	.253	.315	.253	.315
	.442	1.071	.071	.066	.120	.224	.072	.108	.115	.214	.315	.214	.315
	.493	1.377	.079	.066	.120	.224	.072	.108	.115	.214	.315	.214	.315
	.537	.399	.079	.048	.073	.105	.367	.123	.100	.154	.209	.154	.209
	.581	.383	.066	.021	.027	.050	.292	.118	.071	.105	.154	.071	.105
	.617	.111	.071	.029	-.028	.016	.083	.126	.084	.058	.146	.084	.058
	.653	.064	.066	.029	-.036	-.044	.032	.123	.084	.053	.146	.084	.053
	.690	.052	.048	.024	-.054	.240	.016	.113	.081	.024	.284	.081	.024
	.735	-.005	-.005	-.007	-.025	.061	.061	.061	.061	.021	.042	.061	.021
Plug	.735	-.005	-.005	-.007	-.025	.061	.061	.061	.061	.021	.042	.061	.021
	.768	-.010	-.010	-.012	-.023	.066	.066	.066	.066	.037	.066	.066	.037
	.821	-.005	-.005	-.005	-.003	.063	.063	.063	.063	.053	.066	.066	.053
	.852	-.001	-.001	-.001	.001	.068	.068	.068	.068	.068	.068	.068	.068

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(dd) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.0$; outboard station

x/c	C_p at -																				
	$\alpha = -2^\circ$							$\alpha = 0^\circ$							$\alpha = 1^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.037	.145	.197	.665	.598	.126	.132	.302	.063	.471	-.218	.115	.366	.064	.391	.1269	1.269	1.269	1.269	
	.003	-.182	.1257	.613	1.264	1.264	-.1371	.1265	.386	1.273	1.273	-.1367	1.261	.415	.1269	1.269	1.269	1.269	1.269	1.269	
	.014	-.123	.1257	.613	1.264	1.264	-.1371	.1265	.386	1.273	1.273	-.1367	1.261	.415	.1269	1.269	1.269	1.269	1.269	1.269	
	.031	.086	1.257	.613	1.264	1.264	-.4274	.1232	1.298	1.273	1.273	-.370	1.235	1.193	1.162	1.162	1.162	1.162	1.162	1.162	
	.055	-.303	.088	1.241	1.264	1.264	-.4274	.1232	1.298	1.273	1.273	-.370	1.235	1.193	1.162	1.162	1.162	1.162	1.162	1.162	
	.106	.286	.374	.343	.291	-.289	.034	-.221	.268	.220	.223	.008	-.224	-.274	-.249	-.225	-.225	-.225	-.225	-.225	
	.172	.134	.374	.343	.291	-.289	.034	-.221	.268	.220	.223	.008	-.224	-.274	-.249	-.225	-.225	-.225	-.225	-.225	
	.261	.267	.316	.222	-.308	-.216	-.222	.179	.206	.226	.226	.152	-.201	.115	-.192	-.191	-.121	-.121	-.121	-.121	
	.302	.161	.125	.442	-.156	-.140	-.116	-.085	.236	.048	.063	.095	-.069	.255	-.016	-.003	-.003	-.003	-.003	-.003	
	.326	-.110	-.041	-.020	.134	-.064	-.065	.003	.013	.102	.102	.044	.010	.026	.361	.044	.044	.044	.044	.044	
Turbine cowl	.343	-.157	-.038	-.078	-.028	-.122	-.143	-.003	.039	.034	.079	.142	.010	.011	.062	.058	.058	.058	.058	.058	
	.367	.004	.020	-.004	.031	-.009	.038	.058	.044	.076	.076	.043	.073	.063	.104	.102	.102	.102	.102	.102	
	.442	-.012	-.038	-.067	-.088	-.023	.014	.010	.008	.005	.037	.027	.026	.018	.047	.081	.081	.081	.081	.081	
	.493	-.012	-.041	-.088	-.140	-.104	.022	.013	.019	.024	.002	.035	.034	.010	.021	.055	.055	.055	.055	.055	
	.537	-.020	-.057	-.109	-.195	-.208	.030	.010	.029	.074	.061	.047	.037	.000	-.021	.002	.002	.002	.002	.002	
	.581	-.079	-.083	-.140	-.284	-.365	.014	.002	.045	.150	.157	.029	.029	.008	-.136	.000	.000	.000	.000	.000	
	.617	-.126	-.109	-.135	-.308	-.595	.025	-.008	.042	.150	.157	.029	.029	.008	-.136	.000	.000	.000	.000	.000	
	.653	-.159	-.141	-.159	-.289	-.454	-.049	-.024	.050	.155	.157	.029	.029	.008	-.136	.000	.000	.000	.000	.000	
	.690	-.188	-.156	-.156	-.245	-.383	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	.708	-.204	-.177	-.177	-.224	-.383	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
Plug	.735	-.183	-.183	-.224	-.224	-.224	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	.768	-.177	-.177	-.183	-.183	-.183	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	.796	-.159	-.159	-.153	-.153	-.153	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	.821	-.146	-.146	-.143	-.143	-.143	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	.852	-.275	-.188	-.156	-.245	-.383	-.116	-.058	.048	.126	.126	.004	.013	.006	-.084	.178	.178	.178	.178	.178	
	Fan cowl	0.000	-.282	.095	.428	.063	.340	-.352	.057	.553	.069	.271	.640	.016	.052	.018	.052	.018	.052	.018	.052
		.003	-.1384	1.269	.387	1.279	1.279	-.581	.1271	.390	1.279	-.667	.021	.021	.021	.021	.021	.021	.021	.021	.021
		.014	-.424	1.236	1.099	1.125	1.279	-.707	.1296	.436	1.296	-.701	.016	.016	.016	.016	.016	.016	.016	.016	.016
		.031	-.625	-.060	-.265	-.311	1.426	-.739	.036	.357	.400	-.727	.016	.016	.016	.016	.016	.016	.016	.016	.016
		.055	-.164	-.225	-.281	-.235	1.154	-.137	.030	.252	.290	-.458	.016	.016	.016	.016	.016	.016	.016	.016	.016
.106		-.086	.082	-.104	-.138	-.052	-.188	.074	.273	.196	-.167	.016	.016	.016	.016	.016	.016	.016	.016	.016	
.172		-.164	.085	-.170	-.158	-.046	-.031	.074	.141	.141	.068	.016	.016	.016	.016	.016	.016	.016	.016	.016	
.261		-.070	-.047	.187	.014	.040	.005	.015	.193	.058	.118	.016	.016	.016	.016	.016	.016	.016	.016	.016	
.302		-.019	.030	.045	.221	.085	.028	.053	.049	.145	.145	.016	.016	.016	.016	.016	.016	.016	.016	.016	
.326		-.125	.027	.008	.045	-.022	-.121	.030	.037	.129	.003	.016	.016	.016	.016	.016	.016	.016	.016	.016	
Turbine cowl	.343	.068	.053	.088	.082	.042	-.087	.108	.116	.187	.207	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.367	.048	.051	.045	.087	.112	.087	.080	.080	.145	.210	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.442	.068	.061	.038	.084	.113	.107	.090	.074	.129	.194	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.493	.075	.061	.030	.030	.079	.123	.098	.072	.097	.179	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.537	.083	.061	.022	-.031	-.026	.126	.101	.063	.045	.100	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.581	.032	.061	.030	-.025	-.112	.107	.103	.074	.045	.032	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.617	.036	.051	.030	-.041	-.091	.095	.101	.069	.032	.027	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.653	-.023	.019	.030	-.041	-.134	-.048	.089	.072	.011	-.018	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.690	.001	.001	.017	-.057	-.154	-.048	.040	.058	.040	.040	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.708	.017	.017	.023	.023	.023	.023	.058	.058	.058	.058	.016	.016	.016	.016	.016	.016	.016	.016	.016	
Plug	.735	.014	.014	.039	.039	.039	.039	.053	.053	.053	.053	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.768	.019	.019	.024	.024	.024	.024	.058	.058	.058	.058	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.796	.030	.030	.037	.037	.037	.037	.069	.069	.069	.069	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.821	.030	.030	.037	.037	.037	.037	.069	.069	.069	.069	.016	.016	.016	.016	.016	.016	.016	.016	.016	
	.852	.038	.038	.040	.040	.040	.040	.072	.072	.072	.072	.016	.016	.016	.016	.016	.016	.016	.016	.016	

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued
(ee) M = 0.700; windmilling; inboard station

x/c	C _p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																
	α = -2°						α = 0°						α = 1°																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(ft) $M = 0.700$; windmilling; outboard station

x/c	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																										
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	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(gg) M = 0.750; windmilling; inboard station

C _p at -																		
x/c	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.0000	.205	-.470	-.390	-.057	.028	.099	-.511	-.853	-.040	-.039	-.104	-.505	.112	-.026	-.177		
	.003	-1.097	-1.137	-.829	-1.087	-1.011	-.842	-1.322	-.853	-.853	-1.116	-.821	-1.165	-.994	-.745			
	.014	.991	-1.125	-.847	-1.095	-.967	-.642	-1.137	-.885	-.885	-.993	-.867	-1.112	-1.009	-.786			
	.031	.748	-1.020	-.932	-.979	-.707	-.753	-.914	-.991	-.888	-.961	-.865	-.900	-.821	-.810			
	.055	-.482	-.683	-.838	-.695	-.318	-.824	-.753	-.794	-.654	-.660	-.760	-.662	-.686	-.613			
	.106	-.212	-.291	-.560	-.230	-.160	-.266	-.286	-.333	-.168	-.115	-.370	-.310	-.198	-.177			
	.172	-.110	-.164	-.214	-.108	-.055	-.057	-.125	-.092	-.036	-.010	-.081	-.110	-.069	.010			
	.261	-.168	-.255	-.120	-.184	-.096	-.102	-.180	-.125	-.089	-.025	-.064	-.145	-.051	.031			
	.302	-.145	-.099	-.073	-.073	-.032	-.088	-.051	-.051	-.025	-.055	-.084	-.058	-.040	.086			
	.326	-.039	-.046	-.097	-.011	-.003	-.088	-.004	-.004	-.025	-.046	-.022	-.019	-.086	.011			
	.343	-.150	-.052	-.126	-.041	-.119	-.124	-.022	-.057	-.075	-.051	-.075	-.108	-.007	.019			
	.387	-.132	-.008	-.023	-.003	-.021	-.093	-.034	-.034	-.107	-.081	-.060	-.066	.121	.154			
	.442	-.017	-.049	-.058	-.061	.015	-.022	.005	.002	-.075	-.043	.037	.031	.069	.160			
	.537	-.098	-.073	-.102	-.102	-.046	-.027	-.002	-.002	-.075	.047	.037	.022	.060	.060			
Turbine cowl	.0000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.003	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.014	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.031	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.055	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.106	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.172	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.261	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.302	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.326	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.343	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.387	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.442	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.537	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
Plug	.0000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.003	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.014	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.031	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.055	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.106	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.172	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.261	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.302	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.326	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.343	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.387	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.442	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.537	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
.617	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000				

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(hh) M = 0.750; windmilling; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	-0.405	-0.242	-0.122	-0.313	0.273	-0.491	-0.209	0.018	-0.279	-0.535	-0.231	0.074	-0.273	0.130			
	0.003	-1.515	-1.450	-0.255	-1.323	1.427	-0.814	-0.580	0.132	-1.073	-0.991	-1.011	0.167	-1.127	-0.880			
	0.014	-1.576	-1.373	-0.957	-1.288	1.427	-0.964	-1.001	-1.022	-1.047	-0.956	-0.987	-1.180	-1.059	-0.894			
	0.031	-0.848	-1.116	-0.980	-1.114	-0.538	-1.026	-0.983	-1.019	-1.023	-0.971	-0.987	-0.929	-0.992	-0.906			
	0.055	-0.410	-0.91	-0.927	-0.811	-0.450	-0.853	-0.69	-0.907	-0.817	-0.808	-0.860	-0.721	-0.794	-0.633			
	0.106	-0.220	-0.348	-0.617	-0.355	-0.411	-0.455	-0.392	-0.398	-0.328	-0.546	-0.380	-0.265	-0.317	-0.370			
	0.172	-0.172	-0.094	-0.271	-0.223	-0.161	-0.128	-0.202	-0.143	-0.160	-0.134	-0.212	-0.135	-0.158	-0.087			
	0.261	-0.242	-0.089	-0.147	-0.267	-0.217	-0.155	-0.055	-0.146	-0.160	-0.146	-0.090	-0.158	-0.161	-0.060			
	0.302	-0.150	-0.129	-0.030	-0.141	-0.138	-0.077	-0.078	-0.087	-0.045	-0.113	-0.068	-0.084	-0.134	0.004			
	0.326	-0.101	-0.070	-0.059	-0.197	-0.117	-0.057	-0.031	-0.016	-0.031	-0.032	-0.010	0.007	0.007	0.040			
	0.343	-0.189	-0.079	-0.126	-0.073	-0.144	-0.068	-0.052	-0.066	-0.016	-0.156	-0.031	-0.031	0.022	-0.060			
Turbine cowl	0.000	-0.387	-0.030	-0.011	-0.038	-0.035	-0.004	-0.022	0.015	-0.037	-0.028	0.012	0.040	0.078	0.078			
	0.003	-0.442	-0.035	-0.047	-0.059	-0.064	0.000	-0.001	-0.007	-0.002	-0.016	0.025	0.019	0.052	0.084			
	0.014	-0.537	-0.035	-0.067	-0.114	-0.123	0.005	0.005	-0.013	-0.019	-0.040	0.031	0.016	0.034	0.066			
	0.031	-0.581	-0.048	-0.082	-0.152	-0.191	0.009	0.005	-0.019	-0.043	-0.043	0.031	0.010	0.013	0.037			
	0.055	-0.617	-0.088	-0.100	-0.200	-0.258	0.005	-0.004	-0.031	-0.084	-0.116	0.047	0.004	0.028	-0.025			
	0.106	-0.653	-0.128	-0.100	-0.200	-0.329	-0.026	-0.013	-0.025	-0.084	-0.181	0.021	0.007	-0.028	-0.075			
	0.172	-0.690	-0.163	-0.118	-0.202	-0.329	-0.044	-0.022	-0.037	-0.093	-0.157	0.007	-0.001	-0.040	-0.075			
	0.261	-0.708	-0.216	-0.168	-0.205	-0.320	-0.097	-0.060	-0.055	-0.101	-0.169	-0.041	-0.016	-0.054	-0.108			
	0.302	-0.735	-0.215	-0.209	-0.241	-0.241	-0.097	-0.060	-0.055	-0.101	-0.169	-0.041	-0.016	-0.054	-0.108			
	0.326	-0.768	-0.224	-0.247	-0.247	-0.247	-0.097	-0.060	-0.055	-0.101	-0.169	-0.041	-0.016	-0.054	-0.108			
	0.343	-0.796	-0.235	-0.264	-0.264	-0.264	-0.097	-0.060	-0.055	-0.101	-0.169	-0.041	-0.016	-0.054	-0.108			
Plug	0.000	-0.583	-0.251	-0.095	-0.279	0.067	-0.548	-0.265	0.272	-0.258	-0.446	-0.258	0.272	0.131				
	0.003	-0.667	-0.984	-0.152	-1.354	-0.783	-0.869	-0.877	-0.750	-1.032	-0.690	-0.877	-0.679					
	0.014	-0.813	-1.014	-1.425	-1.065	-0.803	-0.901	-0.915	-0.915	-0.894	-0.708	-0.915	-0.679					
	0.031	-0.897	-0.999	-0.594	-0.986	-0.827	-0.756	-0.909	-0.436	-0.602	-0.749	-0.436	-0.679					
	0.055	-0.848	-0.958	-0.419	-0.724	-0.827	-0.676	-0.934	-0.344	-0.467	-0.593	-0.344	-0.679					
	0.106	-0.636	-0.345	-0.292	-0.356	-0.468	-0.370	-0.356	-0.247	-0.296	-0.166	-0.247	-0.679					
	0.172	-0.181	-0.237	-0.132	-0.182	-0.102	-0.331	-0.270	-0.087	-0.149	-0.166	-0.087	-0.679					
	0.261	-0.097	-0.141	-0.162	-0.164	-0.026	-0.057	-0.173	-0.134	-0.128	-0.034	-0.134	-0.679					
	0.302	-0.032	-0.055	-0.115	-0.007	0.042	-0.009	-0.034	-0.099	-0.043	-0.034	-0.099	-0.679					
	0.326	-0.030	-0.001	-0.016	0.004	0.066	-0.013	-0.008	-0.037	-0.046	-0.016	-0.037	-0.679					
	0.343	-0.150	-0.020	-0.008	0.054	-0.038	-0.124	-0.088	-0.085	-0.084	-0.016	-0.085	-0.679					
Fan cowl	0.000	-0.442	-0.023	-0.036	-0.039	0.077	-0.075	-0.076	0.085	-0.116	-0.205	-0.085	0.085					
	0.003	-0.493	-0.061	-0.048	-0.033	0.066	-0.092	-0.087	-0.085	-0.134	-0.205	-0.085	0.085					
	0.014	-0.537	-0.058	-0.054	-0.033	0.042	-0.096	-0.096	-0.085	-0.134	-0.205	-0.085	0.085					
	0.031	-0.581	-0.057	-0.027	-0.004	0.036	-0.120	-0.102	-0.075	-0.081	-0.149	-0.075	0.081					
	0.055	-0.617	-0.045	-0.051	-0.033	-0.004	-0.111	-0.102	-0.085	-0.075	-0.087	-0.085	0.075					
	0.106	-0.653	-0.036	-0.021	-0.005	-0.023	-0.102	-0.102	-0.070	-0.058	-0.087	-0.070	0.058					
	0.172	-0.690	-0.016	-0.010	-0.026	-0.061	-0.062	-0.070	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.261	-0.708	-0.044	-0.038	-0.074	-0.074	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.302	-0.735	-0.047	-0.047	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.326	-0.768	-0.041	-0.041	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.343	-0.796	-0.041	-0.041	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
Turbine cowl	0.000	-0.583	-0.251	-0.095	-0.279	0.067	-0.548	-0.265	0.272	-0.258	-0.446	-0.258	0.272					
	0.003	-0.667	-0.984	-0.152	-1.354	-0.783	-0.869	-0.877	-0.750	-1.032	-0.690	-0.877	-0.679					
	0.014	-0.813	-1.014	-1.425	-1.065	-0.803	-0.901	-0.915	-0.915	-0.894	-0.708	-0.915	-0.679					
	0.031	-0.897	-0.999	-0.594	-0.986	-0.827	-0.756	-0.909	-0.436	-0.602	-0.749	-0.436	-0.679					
	0.055	-0.848	-0.958	-0.419	-0.724	-0.827	-0.676	-0.934	-0.344	-0.467	-0.593	-0.344	-0.679					
	0.106	-0.636	-0.345	-0.292	-0.356	-0.468	-0.370	-0.356	-0.247	-0.296	-0.166	-0.247	-0.679					
	0.172	-0.181	-0.237	-0.132	-0.182	-0.102	-0.331	-0.270	-0.087	-0.149	-0.166	-0.087	-0.679					
	0.261	-0.097	-0.141	-0.162	-0.164	-0.026	-0.057	-0.173	-0.134	-0.128	-0.034	-0.134	-0.679					
	0.302	-0.032	-0.055	-0.115	-0.007	0.042	-0.009	-0.034	-0.099	-0.043	-0.034	-0.099	-0.679					
	0.326	-0.030	-0.001	-0.016	0.004	0.066	-0.013	-0.008	-0.037	-0.046	-0.016	-0.037	-0.679					
	0.343	-0.150	-0.020	-0.008	0.054	-0.038	-0.124	-0.088	-0.085	-0.084	-0.016	-0.085	-0.679					
Plug	0.000	-0.442	-0.023	-0.036	-0.039	0.077	-0.075	-0.076	0.085	-0.116	-0.205	-0.085	0.085					
	0.003	-0.493	-0.061	-0.048	-0.033	0.066	-0.092	-0.087	-0.085	-0.134	-0.205	-0.085	0.085					
	0.014	-0.537	-0.058	-0.054	-0.033	0.042	-0.096	-0.096	-0.085	-0.134	-0.205	-0.085	0.085					
	0.031	-0.581	-0.057	-0.027	-0.004	0.036	-0.120	-0.102	-0.075	-0.081	-0.149	-0.075	0.081					
	0.055	-0.617	-0.045	-0.051	-0.033	-0.004	-0.111	-0.102	-0.085	-0.075	-0.087	-0.085	0.075					
	0.106	-0.653	-0.036	-0.021	-0.005	-0.023	-0.102	-0.102	-0.070	-0.058	-0.087	-0.070	0.058					
	0.172	-0.690	-0.016	-0.010	-0.026	-0.061	-0.062	-0.070	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.261	-0.708	-0.044	-0.038	-0.074	-0.074	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.302	-0.735	-0.047	-0.047	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.326	-0.768	-0.041	-0.041	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					
	0.343	-0.796	-0.041	-0.041	-0.085	-0.085	-0.062	-0.074	-0.061	-0.031	-0.025	-0.061	-0.031					

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(ii) $M = 0.775$; windmilling; inboard station

x/\bar{c}	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	-1.241	-1.007	-0.782	-0.593	-0.433	-0.285	-0.155	-0.049	-0.013
	0.003	-1.181	-0.947	-0.722	-0.533	-0.373	-0.225	-0.095	-0.029	-0.003
	0.014	-1.178	-0.944	-0.719	-0.528	-0.368	-0.219	-0.089	-0.024	-0.002
	0.031	-1.004	-0.745	-0.591	-0.441	-0.291	-0.141	-0.031	-0.001	-0.001
	0.055	-0.688	-0.429	-0.275	-0.125	-0.015	0.015	0.045	0.075	0.105
	0.106	-0.294	-0.035	0.119	0.269	0.419	0.569	0.719	0.869	1.019
	0.172	-0.102	-0.153	-0.204	-0.255	-0.306	-0.357	-0.408	-0.459	-0.510
	0.261	-0.161	-0.246	-0.331	-0.416	-0.501	-0.586	-0.671	-0.756	-0.841
	0.302	-0.140	-0.225	-0.310	-0.395	-0.480	-0.565	-0.650	-0.735	-0.820
	0.326	-0.115	-0.200	-0.285	-0.370	-0.455	-0.540	-0.625	-0.710	-0.795
	0.343	-0.144	-0.229	-0.314	-0.399	-0.484	-0.569	-0.654	-0.739	-0.824
	0.387	-0.123	-0.208	-0.293	-0.378	-0.463	-0.548	-0.633	-0.718	-0.803
Turbine cowl	0.000	-0.008	-0.017	-0.026	-0.035	-0.044	-0.053	-0.062	-0.071	-0.080
	0.003	-0.007	-0.016	-0.025	-0.034	-0.043	-0.052	-0.061	-0.070	-0.079
	0.014	-0.006	-0.015	-0.024	-0.033	-0.042	-0.051	-0.060	-0.069	-0.078
	0.031	-0.005	-0.014	-0.023	-0.032	-0.041	-0.050	-0.059	-0.068	-0.077
	0.055	-0.004	-0.013	-0.022	-0.031	-0.040	-0.049	-0.058	-0.067	-0.076
	0.106	-0.003	-0.012	-0.021	-0.030	-0.039	-0.048	-0.057	-0.066	-0.075
	0.172	-0.002	-0.011	-0.020	-0.029	-0.038	-0.047	-0.056	-0.065	-0.074
	0.261	-0.001	-0.010	-0.019	-0.028	-0.037	-0.046	-0.055	-0.064	-0.073
	0.302	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.326	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.343	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.387	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
Plug	0.000	-0.008	-0.017	-0.026	-0.035	-0.044	-0.053	-0.062	-0.071	-0.080
	0.003	-0.007	-0.016	-0.025	-0.034	-0.043	-0.052	-0.061	-0.070	-0.079
	0.014	-0.006	-0.015	-0.024	-0.033	-0.042	-0.051	-0.060	-0.069	-0.078
	0.031	-0.005	-0.014	-0.023	-0.032	-0.041	-0.050	-0.059	-0.068	-0.077
	0.055	-0.004	-0.013	-0.022	-0.031	-0.040	-0.049	-0.058	-0.067	-0.076
	0.106	-0.003	-0.012	-0.021	-0.030	-0.039	-0.048	-0.057	-0.066	-0.075
	0.172	-0.002	-0.011	-0.020	-0.029	-0.038	-0.047	-0.056	-0.065	-0.074
	0.261	-0.001	-0.010	-0.019	-0.028	-0.037	-0.046	-0.055	-0.064	-0.073
	0.302	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.326	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.343	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072
	0.387	-0.000	-0.009	-0.018	-0.027	-0.036	-0.045	-0.054	-0.063	-0.072

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(j) $M = 0.775$; windmilling; outboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
Turbine cowl	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
	0.343	-0.021	-0.007	-0.035	-0.024	-0.024	-0.000	-0.025	-0.019	-0.042	-0.022	-0.031
Plug	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
	0.343	-0.021	-0.007	-0.035	-0.024	-0.024	-0.000	-0.025	-0.019	-0.042	-0.022	-0.031
Fan cowl	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
	0.343	-0.021	-0.007	-0.035	-0.024	-0.024	-0.000	-0.025	-0.019	-0.042	-0.022	-0.031
Turbine cowl	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
	0.343	-0.021	-0.007	-0.035	-0.024	-0.024	-0.000	-0.025	-0.019	-0.042	-0.022	-0.031
Plug	0.000	-0.332	-0.196	-0.074	-0.264	-0.312	-0.288	-0.177	-0.040	-0.238	-0.492	-0.191
	0.003	-1.472	1.366	-0.345	-1.416	1.374	-0.928	-1.139	-0.210	-1.156	-0.725	-0.989
	0.014	-1.452	1.366	-0.906	-1.374	1.374	-0.977	-1.073	-1.037	-1.117	-0.725	-0.989
	0.031	-0.577	-1.162	-0.950	-1.100	-0.611	-0.930	-1.008	-1.014	-1.007	-0.899	-1.034
	0.055	-0.331	-0.089	-0.918	-0.820	-0.383	-0.841	-0.666	-0.906	-0.814	-0.840	-0.992
	0.066	-0.076	-0.370	-0.662	-0.383	-0.304	-0.310	-0.409	-0.443	-0.328	-0.386	-0.468
	0.106	-0.161	-0.053	-0.302	-0.258	-0.154	-0.132	-0.086	-0.151	-0.150	-0.139	-0.373
	0.172	-0.250	-0.129	-0.137	-0.275	-0.227	-0.158	-0.039	-0.137	-0.159	-0.114	-0.094
	0.261	-0.157	-0.064	-0.052	-0.182	-0.137	-0.094	-0.074	-0.008	-0.065	-0.063	-0.057
	0.302	-0.097	-0.069	-0.126	-0.078	-0.080	-0.056	-0.026	-0.009	-0.029	-0.033	-0.008
	0.326	-0.178	-0.069	-0.035	-0.058	-0.131	-0.166	-0.049	-0.060	-0.009	-0.152	-0.026
	0.343	-0.021	-0.007	-0.035	-0.024	-0.024	-0.000	-0.025	-0.019	-0.042	-0.022	-0.031

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued

(II) M = 0.800; windmilling; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	-1.263	-1.139	-0.542	-1.319	-0.204	-0.356	-0.420	-0.166	-0.056	-0.205	-0.153	-0.128	-0.185	-0.199			
	.003	-1.401	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.014	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.031	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.055	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.106	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.172	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.261	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.302	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.326	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.343	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.387	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
Turbine cowl	0.000	-1.263	-1.139	-0.542	-1.319	-0.204	-0.356	-0.420	-0.166	-0.056	-0.205	-0.153	-0.128	-0.185	-0.199			
	.003	-1.401	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.014	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.031	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.055	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.106	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.172	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.261	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.302	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.326	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.343	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.387	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
Plug	0.000	-1.263	-1.139	-0.542	-1.319	-0.204	-0.356	-0.420	-0.166	-0.056	-0.205	-0.153	-0.128	-0.185	-0.199			
	.003	-1.401	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.014	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.031	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.055	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.106	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.172	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.261	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.302	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.326	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.343	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			
	.387	-1.478	-1.311	-0.85	-1.319	-1.319	-1.469	-1.469	-1.313	-0.295	-1.321	-1.078	-0.335	-1.319	-0.918			

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Continued
(mm) M = 0.825; windmilling; inboard station

x/c	C_p at -									
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
Fan cowl	0.000	.295	-.311	.027	.074	.150	.190	-.306	.137	.079
	.003	-1.473	1.216	-.752	1.277	1.277	-1.038	-1.182	.885	1.276
	.014	-1.367	1.218	-.755	1.277	-1.299	-.998	-1.122	.887	1.276
	.031	-.661	-.987	-.420	-.789	-.437	-.860	-.952	.154	-.738
	.055	-.263	-.595	-.768	-.320	-.216	-.683	-.665	-.764	-.766
	.106	-.216	-.298	-.593	-.250	-.146	-.253	-.284	-.367	-.624
	.172	-.086	-.141	-.254	-.081	-.019	-.060	-.104	-.099	-.103
	.261	-.157	-.259	-.079	-.172	-.073	-.083	-.033	.011	.016
	.302	-.129	-.081	-.037	-.045	.007	-.083	-.033	.011	.008
	.326	-.675	-.019	.589	.002	.041	.323	.014	.335	.071
Turbine cowl	.343	-.125	-.016	-.094	.007	-.105	.111	.001	.071	.102
	.387	-.109	-.023	.005	-.003	.083	-.083	.050	.045	.144
	.442	.001	-.029	-.037	-.026	.072	.035	.014	.011	.053
	.493	.007	-.042	-.063	-.089	.007	.035	.011	.007	.011
	.537	.754	-.060	-.094	-.154	-.157	.429	.001	-.031	.044
	.581	.620	-.102	-.149	-.229	.720	.358	.028	-.064	.109
	.617	-.038	-.126	-.150	-.300	-.362	.003	-.038	.075	.033
	.653	-.078	-.170	-.194	-.339	-.505	-.024	-.067	-.091	.176
	.690	-.310	-.233	-.246	-.404	.585	-.156	-.106	.127	.218
	.708	-.384	-.387	-.387	-.396	.585	-.156	-.106	.127	.218
Plug	.735	-.060	-.060	-.505	-.505	.585	-.156	-.106	.127	.218
	.768	-.397	-.397	-.375	-.375	.585	-.156	-.106	.127	.218
	.796	-.290	-.290	-.352	-.352	.585	-.156	-.106	.127	.218
	.821	-.183	-.183	-.292	-.292	.585	-.156	-.106	.127	.218
	.852	-.183	-.183	-.203	-.203	.585	-.156	-.106	.127	.218
	0.000	-.103	-.323	.260	.087	-.078	.056	-.317	.424	.126
	.003	-.740	-1.154	1.213	-1.286	-.394	-.440	-.920	1.210	-1.037
	.014	-.732	-1.063	1.213	-1.065	-.544	-.432	-.915	1.210	-1.037
	.031	-.736	-.867	-.254	-.780	-.613	-.472	-.816	-.207	-.870
	.055	-.705	-.640	-.209	-.456	-.583	-.499	-.583	-.244	-.399
Fan cowl	.106	-.429	-.337	-.282	-.220	-.313	-.474	-.278	.173	-.483
	.172	-.098	-.095	-.055	.012	.044	-.274	.147	.019	.020
	.261	-.003	-.113	-.182	.009	.103	-.006	-.113	.113	.007
	.302	.005	.007	-.027	.093	.153	.037	.025	.001	.137
	.326	.359	.054	.320	.129	.176	.384	.077	.310	.243
	.343	-.078	.038	.033	.129	.025	-.046	.067	.069	.186
	.387	-.035	.091	.104	.174	.223	.041	.119	.142	.233
	.442	-.080	.062	.062	.135	.220	.132	.098	.101	.202
	.493	.087	.065	.052	.098	.202	.140	.098	.093	.173
	.537	.470	.062	.033	.054	.096	.471	.098	.077	.134
Turbine cowl	.581	.363	.044	.007	.044	.564	.356	.090	.053	.090
	.617	.080	.041	.005	-.048	.002	.140	.093	.054	.040
	.653	.024	.023	-.008	-.053	-.071	.085	.043	.027	.064
	.690	-.039	-.014	-.037	-.097	.257	.041	.051	.020	.064
	.708	-.019	-.126	-.126	-.126	.257	.041	.051	.020	.064
	.735	-.019	-.019	-.019	-.019	.257	.041	.051	.020	.064
	.768	-.032	-.032	-.032	-.032	.257	.041	.051	.020	.064
	.796	-.159	-.159	-.159	-.159	.257	.041	.051	.020	.064
	.821	-.122	-.122	-.122	-.122	.257	.041	.051	.020	.064
	.852	-.054	-.054	-.054	-.054	.257	.041	.051	.020	.064
Plug	0.000	-.103	-.323	.260	.087	-.078	.056	-.317	.424	.126
	.003	-.740	-1.154	1.213	-1.286	-.394	-.440	-.920	1.210	-1.037
	.014	-.732	-1.063	1.213	-1.065	-.544	-.432	-.915	1.210	-1.037
	.031	-.736	-.867	-.254	-.780	-.613	-.472	-.816	-.207	-.870
	.055	-.705	-.640	-.209	-.456	-.583	-.499	-.583	-.244	-.399
	.106	-.429	-.337	-.282	-.220	-.313	-.474	-.278	.173	-.483
	.172	-.098	-.095	-.055	.012	.044	-.274	.147	.019	.020
	.261	-.003	-.113	-.182	.009	.103	-.006	-.113	.113	.007
	.302	.005	.007	-.027	.093	.153	.037	.025	.001	.137
	.326	.359	.054	.320	.129	.176	.384	.077	.310	.243

TABLE 31.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1 - Concluded

(an) M = 0.825; windmilling; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.000	-.201	-.081	.013	-.138	.403	-.346	-.107	.100	-.145	-.408	-.125	.176	-.132	.220			
	.003	-1.328	1.270	.655	1.275	1.275	-1.397	1.267	.398	1.274	-1.417	1.269	.449	1.277	1.277			
	.014	-.387	1.270	-.834	1.275	1.275	-1.527	1.267	1.264	1.274	-1.578	1.269	1.277	1.277	1.277			
	.031	.205	1.294	-.895	1.310	-1.213	-.347	1.277	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.055	-.424	-.092	-.887	-.975	-.393	-1.008	-.086	-1.162	-1.113	1.106	-.078	-1.043	-1.080	-1.080			
	.106	.343	-.253	-.734	-.320	-.304	.070	-.165	-.191	-.175	-.054	-.156	-.188	-1.080	-1.080			
	.172	-.141	.418	-.392	-.273	-.129	.107	.175	-.112	-.143	-.084	-.212	-.188	-1.080	-1.080			
	.261	-.266	.318	-.142	-.309	-.220	.070	.177	-.112	-.143	-.084	-.212	-.188	-1.080	-1.080			
	.302	-.160	.129	.489	-.137	-.118	.115	-.089	.256	-.080	-.092	.122	-.191	-.001	.001			
	.326	-.101	-.056	.045	.180	-.071	-.056	.018	.001	-.044	-.037	.004	.222	.436	.046			
	.343	-.160	-.048	.111	.180	-.071	-.130	-.007	.023	.030	-.119	.012	.001	.064	-.033			
	.387	-.007	.010	-.035	-.045	-.003	.019	.051	.035	.061	.038	.070	.062	.098	.106			
Turbine cowl	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000			
	.003	-.019	-.023	-.042	-.066	-.131	.015	.009	.020	.028	.030	.033	.022	.046	.088			
	.014	-.034	-.056	-.087	-.121	-.233	.015	.003	.033	.070	.042	.033	.001	.017	.051			
	.031	-.078	-.113	-.124	-.252	-.349	-.004	-.012	.049	.135	.030	.025	.017	.122	-.093			
	.055	-.137	-.168	-.150	-.267	-.375	-.036	-.025	.054	.141	.001	.012	.017	.250	-.323			
	.106	-.192	-.182	-.182	-.267	-.393	-.071	-.049	.073	.151	-.029	.009	.035	.166	-.200			
	.172	-.278	-.216	-.182	-.267	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
	.261	-.500	-.277	-.277	-.266	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
	.302	-.587	-.287	-.287	-.266	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
	.326	-.592	-.291	-.291	-.266	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
	.343	-.592	-.291	-.291	-.266	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
	.387	-.592	-.291	-.291	-.266	-.393	-.138	-.102	.091	.159	-.096	-.057	.054	.158	-.224			
Plug	.000	-.408	-.119	.239	-.126	.195	-.409	-.126	.358	-.088	.119	-.109	.176	-.132	.220			
	.003	-.648	-1.174	.405	1.275	1.275	-.459	-.943	.396	1.274	-1.417	1.269	.449	1.277	1.277			
	.014	-.778	-1.163	1.267	1.300	1.275	-.549	-.967	1.264	1.274	-1.578	1.269	1.277	1.277	1.277			
	.031	-.306	-1.021	-1.248	-.773	-.784	-.266	-.906	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.055	-.794	-.061	-.506	-.611	-.781	-.608	-.051	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.106	-.086	-.427	-.266	-.320	-.524	-.117	-.404	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.172	-.204	.120	-.122	-.160	-.142	-.392	.096	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.261	-.074	.084	-.185	-.145	-.008	-.109	.046	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.302	-.027	-.043	.220	.007	.062	-.026	-.035	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.326	-.007	.013	.026	.269	.083	.009	.020	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.343	-.129	-.006	.007	.067	.076	.076	.083	1.267	1.274	-.261	1.269	1.277	1.277	1.277			
	.387	-.040	.065	.036	.078	.115	.084	.073	1.267	1.274	-.261	1.269	1.277	1.277	1.277			

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a

(a) $M = 0.700$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.798	.439	.863	.841	.884	-.1023	.625	.402	1.021	.891	.697	.562	.454	1.100	.641		
	.003	-.415	-.667	-.934	-.424	-.293	-.1023	-.527	-.672	-.241	-.275	-.590	-.1230	-.676	-.005	-.943		
	.014	-.386	-.496	-.612	-.370	-.338	-.527	-.338	-.479	-.357	-.307	-.487	-.600	-.487	-.233	-.952		
	.031	-.289	.163	-.625	-.299	.166	-.362	-.177	-.479	-.929	-.243	.186	-.415	.191	-.939	.197		
	.055	-.260	-.313	-.358	-.264	-.203	-.299	-.292	-.292	-.260	-.214	-.224	-.328	-.300	-.210	-.245		
	.106	-.182	-.245	.160	-.181	-.126	-.187	-.187	-.225	.177	-.134	.115	-.192	-.233	.191	-.114		
	.172	-.095	.002	.115	-.068	.169	.085	.013	.132	.132	-.019	.186	-.086	.014	-.002	.213		
	.261	-.158	.252	.213	-.181	.100	-.129	-.222	-.170	.112	-.041	-.144	-.144	-.223	.140	-.011		
	.302	-.129	-.075	-.037	.175	.033	.105	-.105	-.048	-.061	.193	.033	-.095	.047	-.056	-.066		
	.326	.129	.001	.359	.146	.027	-.042	.023	.023	.097	.164	.042	.061	.027	.014	.174		
	.343	.158	.208	.290	.079	.679	.153	.386	.075	.032	.126	.385	.181	.207	.268	.235		
	.387	.328	.017	.005	.052	.416	.416	.386	.023	.039	.161	.520	.356	.014	.059	.197		
Turbine cowl	.442	.211	.337	-.037	-.056	.222	.235	.075	.032	.032	.012	.017	.254	.014	-.027	.068		
	.493	-.202	.794	-.069	-.126	.104	-.144	.377	-.009	-.099	.006	-.083	.124	.178	.011	.546		
	.537	.046	-.040	-.065	-.184	.661	-.090	.006	-.016	-.099	-.013	.186	-.032	.027	-.018	.068		
	.581	.357	.110	-.203	.251	.496	.056	-.051	.118	.013	.186	-.032	.040	-.098	-.024	-.213		
	.617	-.134	-.001	.110	.315	.379	-.061	.055	.016	-.064	.157	.047	-.059	.008	-.053	.075		
	.653	.177	-.088	-.085	.094	.306	.197	-.006	-.006	-.131	.118	.210	.027	.027	-.008	-.043		
	.690	-.105	.034	-.043	-.264	.082	.031	.026	.036	.182	.064	.036	.079	.066	-.178	.107		
	.708		-.065	.260	-.004			.046	.319	.116			.088	.368	.174			
	.735		-.020		-.043			-.003		.032			-.005		-.018			
	.768		.503		.562			-.186		.452			.210		-.421			
	.796		.181		.113			-.160		.115			.120		-.101			
	.821		.018		.073			.091		.110			.142		-.142			
.852		.314		.345			.367		.414			.426		.472				
Fan cowl	0.000	.460	.417	1.121	.935	.525												
	.003	-.1463	-.691	.158	-.144	-.525												
	.014	-.637	-.468	-.138	-.246	-.214												
	.031	-.438	.184	-.768	-.205	.190												
	.055	-.341	-.258	-.167	-.179	.253												
	.106	-.190	-.228	.184	-.102	.105												
	.172	-.088	.007	.136	.013	.193												
	.261	-.127	.215	.144	-.057	.323												
	.302	-.083	-.041	.035	.199	.093												
	.326	-.054	.030	-.069	.167	.109												
	.343	.140	.158	-.109	.167	.054												
	.387	.417	.036	.081	.167	.215												
Turbine cowl	.442	.232	.064	.007	.074	.193												
	.493	-.103	.100	.039	.174	.124												
	.537	-.127	.042	.033	.049	.029												
	.581	-.050	-.025	-.044	.150	.004												
	.617	-.006	.065	.033	.158	.004												
	.653	.193	.116	.035	-.037	.007												
	.690	.086	.152	.094	.142	-.105												
	.708		.087	.319	.113													
	.735		-.003		.006													
	.768		-.106		-.157													
	.796		-.064		-.099													
	.821		.152		.190													
.852		.393		.446														
Plug	0.000	.460	.417	1.121	.935	.525												
	.003	-.1463	-.691	.158	-.144	-.525												
	.014	-.637	-.468	-.138	-.246	-.214												
	.031	-.438	.184	-.768	-.205	.190												
	.055	-.341	-.258	-.167	-.179	.253												
	.106	-.190	-.228	.184	-.102	.105												
	.172	-.088	.007	.136	.013	.193												
	.261	-.127	.215	.144	-.057	.323												
	.302	-.083	-.041	.035	.199	.093												
	.326	-.054	.030	-.069	.167	.109												
	.343	.140	.158	-.109	.167	.054												
	.387	.417	.036	.081	.167	.215												
Plug	.442	.232	.064	.007	.074	.193												
	.493	-.103	.100	.039	.174	.124												
	.537	-.127	.042	.033	.049	.029												
	.581	-.050	-.025	-.044	.150	.004												
	.617	-.006	.065	.033	.158	.004												
	.653	.193	.116	.035	-.037	.007												
	.690	.086	.152	.094	.142	-.105												
	.708		.087	.319	.113													
	.735		-.003		.006													
	.768		-.106		-.157													
	.796		-.064		-.099													
	.821		.152		.190													
.852		.393		.446														

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(b) M = 0.700; outboard station

x/ζ	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																														
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																	
Fan cowl	0.000	.648	.904	.919	.554	1.074	.377	.888	1.105	.648	.966	.255	.904	1.149	.488	.891	.559	.485	.391	.459	.321	.288	.182	.192	.211	.179	.179	.095	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002	.002

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued
(c) $M = 0.750$; inboard station

x/\bar{c}	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																										
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 4^\circ$																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(d) $M = 0.750$; outboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.000	.612	.687	.916	.577	1.064	.378	.845	1.085	.626	.958	.280	.852	1.163	.661	.894		
	.003	-.577	-.787	.507	-.781	-.645	-1.023	-.728	.399	-.698	-1.022	-1.169	-.745	.348	.624	-1.207		
	.014	-.502	-.619	-.586	-.619	-.587	-.599	-.607	.376	-.592	-.698	-.855	-.597	-.249	.548	-.819		
	.031	-.382	-.613	-.524	-.460	-.398	-.471	-.630	.370	-.413	-.483	-.501	-.621	-.328	.401	-.498		
	.055	-.329	-.635	-.447	-.448	-.363	-.369	.056	.352	-.407	-.395	-.386	.071	-.293	.398	-.401		
	.106	-.038	-.291	-.332	-.369	-.475	-.033	-.281	-.281	-.313	-.304	-.024	-.272	-.243	.301	-.280		
	.172	-.135	-.146	-.166	-.239	-.272	-.126	-.134	-.128	-.248	-.183	-.121	-.130	-.101	.203	-.253		
	.261	-.241	-.525	-.226	-.272	-.272	-.121	-.239	-.193	-.239	-.168	-.192	-.088	-.172	-.206	-.133		
	.302	-.144	-.055	-.017	-.175	-.145	-.121	-.074	.032	-.074	-.071	-.099	-.068	.047	.047	-.039		
	.326	-.100	.017	-.019	-.155	-.313	-.082	-.035	.008	.017	-.012	-.068	.047	.029	.005	.023		
	.343	-.192	.057	-.081	-.170	-.089	-.157	.180	-.086	.238	-.065	-.143	.174	-.003	.253	-.056		
	.387	-.055	.002	-.013	-.027	-.066	-.037	-.018	-.041	-.047	.014	-.015	-.107	-.042	.032	.061		
	.442	-.206	-.169	-.084	-.142	-.067	-.174	-.119	-.092	-.151	-.092	.073	.011	.059	.170	.141		
	.493	-.059	-.071	-.030	-.107	-.230	-.078	-.011	-.006	.005	-.248	-.033	.016	.062	.068	.177		
	.537	-.113	-.084	-.192	-.336	-.248	-.011	-.006	-.080	-.074	-.062	-.248	-.015	-.024	-.059	-.142		
Turbine cowl	.581	-.617	-.131	-.054	-.283	-.466	-.059	-.033	.139	-.006	-.083	.011	.121	.009	-.139	-.021		
	.617	-.223	-.113	-.054	-.175	-.554	-.099	-.045	-.024	-.327	-.366	-.028	.006	.035	-.059	-.233		
	.653	-.312	-.166	-.025	-.372	-.381	-.113	.127	.053	-.145	-.280	-.073	.047	.065	-.269	-.233		
	.690	-.708	.342	.306	-.589	-.381	-.113	.316	.346	-.681	-.280	-.073	.372	.375	.754	-.054		
	.735	.735	.629	.021	-.021	-.021	-.021	.082	.014	.014	.097	.097	.097	.097	.053	-.445		
Plug	.768	.768	-.391	-.536	-.536	-.536	-.536	-.278	-.082	-.419	-.087	-.087	-.083	-.032	-.445	-.445		
	.821	.821	-.172	-.157	-.157	-.157	-.157	-.018	-.018	-.014	-.014	-.012	-.012	.032	.032	.032		
	.852	.852	.111	.161	.161	.161	.161	.121	.121	.179	.179	.179	.179	.173	.173	.173		
			.304	.355	.355	.355	.355	.334	.334	.409	.409	.409	.409	.357	.357	.357		
Fan cowl	0.000	-.195	-.845	1.200	-.690	.845												
	.003	-1.235	-.805	.328	-.560	-1.337												
	.014	-1.204	-.621	-.157	-.507	-1.007												
	.031	-.597	-.636	-.240	-.371	-.598												
	.055	-.395	-.582	-.234	-.383	-.398												
	.106	-.015	-.273	-.210	-.301	-.304												
	.172	-.117	-.128	-.080	-.242	-.118												
	.261	-.179	-.029	-.148	-.263	-.083												
	.302	-.090	-.063	.061	-.011	.023												
	.326	-.055	.056	.044	-.007	.049												
	.343	-.130	.162	-.066	.276	-.048												
	.387	-.007	.011	-.073	.005	.120												
	.442	-.113	-.077	-.004	-.027	-.015												
	.493	.069	.032	.059	.111	.191												
	.537	.038	.088	.056	-.018	.073												
Turbine cowl	.581	.007	.023	-.027	-.239	-.054												
	.617	-.029	.121	.044	-.018	-.018												
	.653	-.029	.073	.059	.191	-.124												
	.690	-.033	.035	.085	-.280	-.207												
	.735	.735	.404	.398	-.654	.447												
Plug	.768	.768	.103	.043	-.416	.335												
	.821	.821	-.042	-.042	-.416	.335												
	.852	.852	.020	.174	.194	.447												
			.363	.447	.447	.447												

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(e) $M' = 0.775$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.782	.405	.839	.826	.858	.644	.411	1.002	.857	.688	.569	.399	1.078	.876	.619		
	.003	.691	.856	-1.087	-.478	-.369	-1.092	-.882	-.395	-.310	-.952	-1.317	-.885	1.078	.876	.619		
	.014	.457	-.575	-.623	-.422	-.383	-.531	-.564	-.457	-.366	-.529	-.688	-.564	.311	-.324	-.348		
	.031	.326	.159	-.086	-.327	-.164	-.404	-.167	-.556	-.268	-.175	-.420	-.170	.705	-.248	.178		
	.055	.283	.342	-.381	-.282	-.215	-.314	-.328	-.289	-.229	-.240	-.327	.317	.224	-.203	-.243		
	.106	.181	.255	.159	-.178	-.119	-.183	-.238	.167	-.133	-.195	-.178	.259	.170	-.111	-.097		
	.172	-.084	.021	.111	-.052	.167	-.077	.018	.120	-.004	.181	-.068	.015	.128	.021	.181		
	.261	-.147	-.249	-.210	-.170	-.074	-.123	-.221	-.179	-.100	-.021	-.102	-.210	.153	-.063	.097		
	.302	-.117	-.063	-.089	.172	.002	-.085	-.038	-.058	.184	.060	-.072	-.030	-.035	.189	.097		
	.326	.579	.018	.744	.144	.021	.234	.044	.381	.153	.083	.166	.032	.240	.158	.113		
Turbine cowl	.343	.307	.257	-.198	.181	.962	.306	.297	-.198	.206	.627	.302	.302	.240	.220	.427		
	.387	-.601	.021	.007	.012	.172	-.348	.027	.072	.074	.369	-.178	.021	.072	.097	.461		
	.442	.154	.746	-.058	-.164	-.041	.178	.347	-.069	-.069	.001	.187	.218	.035	.038	.030		
	.493	-.054	1.070	.046	-.021	-.063	.047	.663	-.013	.108	.157	-.026	.392	.032	.113	.155		
	.537	.485	-.120	-.131	-.175	-.655	.195	-.039	-.061	-.156	.459	.115	.013	.027	.016	.287		
	.581	.689	-.655	-.220	-.428	.867	.336	.060	-.156	-.030	.461	.187	.063	.080	.024	.276		
	.617	-.067	-.117	-.227	-.271	.526	-.043	-.080	.046	-.246	-.189	-.021	.063	.021	-.161	-.100		
	.653	.171	-.145	-.198	-.257	.562	.183	.055	.015	-.091	-.237	.191	.038	-.002	.001	-.091		
	.690	-.139	-.038	-.047	-.458	.402	.013	-.030	-.018	-.142	.184	.047	.105	.066	-.122	.099		
	.708		-.069	.207	.032	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
Plug	.735		-.016	.207	.032	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
	.768		-.423	-.299	-.299	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
	.796		-.688	-.764	-.764	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
	.821		.055	-.046	-.046	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
	.852		.333	.368	.368	.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
						.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
						.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
						.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
						.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
						.097	.086	.086	.336	.083	.083	.083	.330	.330	.269	.268		
Fan cowl	0.000	.524	.404	1.122	.908	.555												
	.003	-1.434	-.904	-.026	-.171	-.530												
	.014	-1.124	-.557	-.203	-.281	-.330												
	.031	-.392	.183	-.470	-.224	.185												
	.055	-.320	.315	-.172	-.182	-.244												
	.106	-.167	.222	-.186	-.090	-.087												
	.172	-.052	.020	-.138	.031	.188												
	.261	-.095	-.200	-.135	-.045	.048												
	.302	-.044	-.622	-.022	.197	.118												
	.326	.216	.057	.240	.163	.141												
Turbine cowl	.343	.309	.302	-.188	.228	.101												
	.387	-.073	.026	.093	.124	.819												
	.442	.207	.214	.009	.118	.031												
	.493	-.022	.267	.071	.068	.155												
	.537	.143	.009	.079	.104	-.081												
	.581	.182	-.014	-.039	.037	.216												
	.617	.037	.169	.016	-.073	.617												
	.653	.207	.045	.020	-.011	-.068												
	.690	.105	.231	.102	-.121	.087												
	.708		.093	.375	-.191													
Plug	.735		.014	.274	-.241													
	.768		-.279	-.336	-.336													
	.796		-.020	.158	.006													
	.821		.821	.821	.821													
	.852		.403	.461	.461													

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(i) M = 0.775; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.604	.875	.902	.584	1.078	.407	.848	1.071	.631	.976	.841	1.166	.678	.895			
	.003	.664	.803	.619	.853	.704	-1.026	-.883	.486	-.727	-1.026	-.843	.415	-.679	-1.240			
	.014	.537	.642	.622	.642	.600	-.772	-.651	-.393	-.589	-.755	-.642	-.276	-.577	-1.040			
	.031	.410	.664	.608	.452	.436	-.488	-.676	-.379	-.536	-.493	-.653	-.356	-.405	-.492			
	.055	.342	.638	.475	.458	.380	-.377	.058	-.367	-.419	-.386	-.632	-.305	-.397	-.399			
	.106	.041	.296	.347	.368	.439	.034	-.282	-.288	-.371	-.306	-.274	-.254	-.301	-.292			
	.172	.134	.146	.166	.230	.185	.127	-.129	-.123	-.332	-.140	-.114	-.123	-.098	-.171			
	.261	.249	.696	.231	.212	.207	.221	.362	-.197	-.369	.247	.188	-.171	-.176	-.258			
	.302	.147	.695	.016	.151	.202	.076	.072	.035	-.114	.225	.052	-.064	.052	-.007			
	.326	.096	.624	.032	.032	.061	.019	.038	.118	.109	.004	.052	.035	.080	.035			
	.343	.164	.543	.033	.238	.044	.148	.155	.027	.303	.024	.131	.154	.027	.320			
	.387	.096	.433	.010	.055	.157	.072	.064	.018	.047	.015	.178	.123	.016	.145			
Turbine cowl	.442	.160	.098	.024	.024	.100	.195	.135	.047	.075	.126	.123	.016	.145	.061			
	.493	.073	.157	.024	.323	.278	.114	.155	.047	.015	.013	.095	.007	.168	.097			
	.537	.079	.188	.154	.044	.275	.106	.050	.016	-.191	.004	.025	.001	.128	.061			
	.581	.068	.183	.114	.429	.340	.081	.010	.078	-.143	.360	.009	.038	.004	-.137			
	.617	.160	.183	.030	.354	.478	.081	.044	.084	-.148	.267	.191	.052	.089	-.224			
	.653	.270	.001	.137	.241	.718	.059	.016	.001	-.193	.357	.044	.038	.309	.295			
	.690	.282	.239	.016	.284	.447	-.119	.033	.007	-.312	-.055	.166	.120	-.086	-.185			
	.708		.254	.262	.636			.370	.370	-.802	.325	.364	.364	-.814				
	.735		.081	.001	.001			.081		.075	.120	.041		.041				
	.768		.401	.394	.394			-.237		-.465	-.214	-.363		-.363				
	.796		.279	.580	.580			-.021		.021	.024	.030		.030				
	.821		.129	.188	.188			.149		.213	.160	.219		.219				
.852		.290	.368	.368			.339		.405	.356	.425		.425					
Fan cowl	0.000	.244	.841	1.209	.691	.854												
	.003	-.172	.871	.420	.579	-1.365												
	.014	-.438	.627	.179	.525	-1.161												
	.031	-.379	.658	.269	.373	.452												
	.055	-.379	.085	.252	.387	.393												
	.106	-.014	.269	.218	.316	.231												
	.172	-.111	.119	.076	.166	.068												
	.261	.175	.162	.153	.203	.116												
	.302	-.082	.056	.060	.017	.034												
	.326	-.048	.060	.051	.062	.040												
	.343	.128	.151	.025	.370	.105												
	.387	.031	.026	.077	.096	.305												
Turbine cowl	.442	.158	.093	.025	.053	.605												
	.493	.122	.034	.026	.221	.209												
	.597	.024	.043	.032	.090	.082												
	.581	.003	.054	.003	.003	.022												
	.617	.037	.173	.066	.107	.014												
	.653	.012	.015	.066	.107	.169												
	.690	-.014	.080	.100	.017	-.237												
	.708		.378	.401	-.101													
	.735		.128	.094	-.215													
	.768		.054	.387	.094													
	.796		.023	.391	.387													
	.821		.179	.821	.212													
.852		.372	.372	.447														
Plug	0.000	.244	.841	1.209	.691	.854												
	.003	-.172	.871	.420	.579	-1.365												
	.014	-.438	.627	.179	.525	-1.161												
	.031	-.379	.658	.269	.373	.452												
	.055	-.379	.085	.252	.387	.393												
	.106	-.014	.269	.218	.316	.231												
	.172	-.111	.119	.076	.166	.068												
	.261	.175	.162	.153	.203	.116												
	.302	-.082	.056	.060	.017	.034												
	.326	-.048	.060	.051	.062	.040												
	.343	.128	.151	.025	.370	.105												
	.387	.031	.026	.077	.096	.305												

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Continued

(g) $M = 0.800$; inboard station

x/\bar{c}	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.790	.422	.844	.844	.656	.408	.995	.850	.711
	.003	-.743	-.954	-.149	-.405	-.1133	-.924	-.534	-.344	-.996
	.014	-.486	-.617	-.765	-.615	-.646	-.596	-.477	-.381	-.532
	.031	-.351	.143	.672	.145	-.409	.167	-.182	-.287	.172
	.055	-.294	-.360	-.387	.230	.320	-.336	-.293	-.241	-.233
	.106	-.192	.263	.151	-.184	-.125	-.236	.167	-.133	-.103
	.172	-.086	.037	.105	.147	-.070	.021	.121	-.001	.172
	.261	-.151	-.255	-.217	-.073	-.119	-.220	-.177	-.098	-.012
	.302	-.111	-.063	-.087	.007	-.083	-.031	-.052	.177	.069
	.326	.905	.024	.980	.034	.420	.050	.540	.150	.091
	.343	.354	.283	.144	.212	.355	.321	.142	.236	.760
	.387	-.959	.043	-.020	.069	-.879	.067	.037	.099	.301
Turbine cowl	.442	.118	.554	-.036	.130	.069	.513	-.012	-.052	.015
	.493	.024	1.177	-.114	-.119	.154	.789	.050	-.088	.112
	.537	.766	.074	.190	.182	.326	-.082	.077	-.128	.403
	.581	.795	-.182	-.141	.445	.044	.007	-.114	.166	.611
	.617	-.045	-.014	-.209	.367	.449	-.004	.086	-.160	-.201
	.653	.159	-.133	-.028	-.340	.179	-.074	-.174	.136	.188
	.690	-.221	-.230	-.244	.383	.567	.016	.099	-.287	.309
	.708		-.625	.161	.085		-.001	.251	.091	
	.735		-.014		.353		-.001		.137	
	.768		-.382		.329		-.320		-.295	
	.796		-.757		.351		-.542		-.700	
	.821		-.063		.329		.151		.158	
Plug	.852		-.278		.312		.364		.436	
	$\alpha = 2^\circ$									
	0.000	-.542	.409	1.116	.893	.572				
	.003	-1.368	-.947	-.018	-.177	1.321				
	.014	-1.131	-.557	-.205	-.312	-.616				
	.031	-.327	.174	.243	-.237	.178				
	.055	-.315	.321	.180	-.191	-.245				
	.106	-.164	-.221	.176	.091	-.678				
	.172	-.050	.622	.128	.041	.188				
	.241	-.080	.057	.135	-.037	.052				
	.302	-.046	.016	.021	.184	.133				
	.326	.359	.068	.355	.157	.154				
	.343	.342	.336	.143	.254	.178				
Fan cowl	.387	-.494	.060	.117	.141	.472				
	.442	.167	.320	-.013	.073	.046				
	.493	.044	.344	.082	.154	.221				
	.537	.265	.025	.030	-.002	-.185				
	.581	.236	.130	-.091	.100	.294				
	.617	.016	.001	.052	-.172	-.034				
	.653	.195	.133	.106	.036	-.053				
	.690	.057	.038	.095	-.080	.162				
	.708		.155	.355	.087					
	.735		.006		-.193					
	.768		-.275		-.277					
	.796		-.153		-.436					
	.821		.163		.267					
Turbine cowl	.852		.384		.448					
	$\alpha = 4^\circ$									
	0.000									
	.003									
	.014									
	.031									
	.055									
	.106									
	.172									
	.241									
	.302									
	.326									
	.343									
Plug	.387									
	.442									
	.493									
	.537									
	.581									
	.617									
	.653									
	.690									
	.708									
	.735									
	.768									
	.796									
	.821									
	.852									

TABLE 32.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 1a - Concluded
(h) $M = 0.800$; outboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.611	.845	.923	.604	1.079	.434	.851	1.066	.636	.975	.958	1.144	.682	.929			
	.003	-.758	-.953	.754	-.906	-.730	-1.004	-.906	.583	-.767	-1.109	.914	.514	-.725	-1.233			
	.014	-.558	-.703	.752	-.695	-.624	-.898	-.872	.434	-.633	-.794	.955	.293	-.603	-.978			
	.031	-.456	-.855	.719	-.597	-.478	-.649	.759	.632	-.684	-.471	.973	.377	-.441	-.919			
	.055	-.371	.032	.057	.497	.402	.392	.048	.391	.438	.408	.723	.353	-.413	-.905			
	.106	-.049	-.316	-.371	.383	.318	.038	.295	.306	.348	.351	.682	.271	-.280	-.370			
	.172	-.147	-.158	-.177	.193	.131	.127	.137	.132	.205	.126	.122	.107	-.205	-.196			
	.261	-.265	.898	-.254	.293	.217	.217	.506	.208	.221	.134	.199	.329	-.186	-.161			
	.302	-.151	.104	.013	-.174	.190	.115	.072	.026	.099	.123	.097	.061	-.066	.002			
	.326	-.098	.016	-.020	.189	.155	.070	.032	.016	.292	.007	.053	.045	.232	.037			
	.343	-.131	.185	.016	.306	.184	.119	.185	.013	.387	.005	.114	.182	.406	.010			
	.387	-.143	-.006	.100	.152	.188	.103	.048	.056	.021	.020	.085	.067	.061	.059			
Turbine cowl	.442	.036	-.088	-.063	.162	.162	.073	.157	.012	.118	.063	.050	.004	-.036	.162			
	.493	-.037	-.028	.167	.318	.277	.073	.157	.012	.118	.063	.050	.004	.015	.042			
	.537	-.049	-.150	.066	.280	.234	.013	.156	.126	.126	.009	.008	.004	.183	.072			
	.581	-.172	-.649	.179	.426	.320	.013	.114	.104	.370	.275	.033	.082	-.164	.302			
	.617	-.115	-.671	.123	.459	.372	.086	.126	.059	.142	.316	.081	.122	-.091	.137			
	.653	-.318	-.169	.120	.358	.811	.107	.065	.099	.071	.397	.000	.015	.253	-.264			
	.690	-.284	-.131	.442	.670	.168	.099	.065	.081	.213	.297	.000	.024	-.291	-.207			
	.708	-.318	.255	.266	.760	.326	.168	.099	.081	.213	.297	.000	.024	-.291	-.207			
	.735	.103	.255	.266	.760	.326	.168	.099	.081	.213	.297	.000	.024	-.291	-.207			
	.752	.103	.255	.266	.760	.326	.168	.099	.081	.213	.297	.000	.024	-.291	-.207			
	.766	-.320	.320	.037	.037	.133	.133	.133	.133	.133	.133	.133	.133	.133	.133			
	.796	-.578	-.578	.540	.540	.134	.134	.134	.134	.134	.134	.134	.134	.134	.134			
.821	.108	.108	.073	.073	.163	.163	.163	.163	.163	.163	.163	.163	.163	.163				
.852	.310	.310	.371	.371	.324	.324	.324	.324	.324	.324	.324	.324	.324	.324				
Plug	0.000	.292	.845	1.197	.655	.884												
	.003	-1.112	-.980	.497	.655	-.656												
	.014	-1.182	-.677	.206	-.566	-1.334												
	.031	-.673	-.754	-.307	.422	-.393												
	.055	-.347	.072	-.274	.412	-.355												
	.106	-.021	.280	-.239	.216	-.330												
	.172	-.111	.125	-.089	.176	-.059												
	.261	-.180	.279	.165	-.184	.059												
	.302	-.086	.059	.050	.014	.031												
	.326	-.045	.053	.044	.177	.066												
	.343	-.107	.186	.012	.429	.025												
	.387	-.066	.074	.044	.104	.098												
Fan cowl	0.000	.292	.845	1.197	.655	.884												
	.003	-1.112	-.980	.497	.655	-.656												
	.014	-1.182	-.677	.206	-.566	-1.334												
	.031	-.673	-.754	-.307	.422	-.393												
	.055	-.347	.072	-.274	.412	-.355												
	.106	-.021	.280	-.239	.216	-.330												
	.172	-.111	.125	-.089	.176	-.059												
	.261	-.180	.279	.165	-.184	.059												
	.302	-.086	.059	.050	.014	.031												
	.326	-.045	.053	.044	.177	.066												
	.343	-.107	.186	.012	.429	.025												
	.387	-.066	.074	.044	.104	.098												
Turbine cowl	0.000	.292	.845	1.197	.655	.884												
	.003	-1.112	-.980	.497	.655	-.656												
	.014	-1.182	-.677	.206	-.566	-1.334												
	.031	-.673	-.754	-.307	.422	-.393												
	.055	-.347	.072	-.274	.412	-.355												
	.106	-.021	.280	-.239	.216	-.330												
	.172	-.111	.125	-.089	.176	-.059												
	.261	-.180	.279	.165	-.184	.059												
	.302	-.086	.059	.050	.014	.031												
	.326	-.045	.053	.044	.177	.066												
	.343	-.107	.186	.012	.429	.025												
	.387	-.066	.074	.044	.104	.098												
Plug	0.000	.292	.845	1.197	.655	.884												
	.003	-1.112	-.980	.497	.655	-.656												
	.014	-1.182	-.677	.206	-.566	-1.334												
	.031	-.673	-.754	-.307	.422	-.393												
	.055	-.347	.072	-.274	.412	-.355												
	.106	-.021	.280	-.239	.216	-.330												
	.172	-.111	.125	-.089	.176	-.059												
	.261	-.180	.279	.165	-.184	.059												
	.302	-.086	.059	.050	.014	.031												
	.326	-.045	.053	.044	.177	.066												
	.343	-.107	.186	.012	.429	.025												
	.387	-.066	.074	.044	.104	.098												

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2

(a) $M = 0.700$; inboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.892	.462	.713	.849	1.007	.763	.419	.906	.892	.856	.669	.418	1.014	.748			
	.003	.201	.698	.138	.300	-.025	.555	.759	.781	.427	.338	.879	.695	.406	.498			
	.014	.249	.699	.118	.223	-.073	.444	.505	.585	.347	.382	.506	.483	.406	.498			
	.031	.220	.687	.929	.146	-.057	.318	.387	.671	.267	.254	.360	.368	.894	.264			
	.055	.244	.333	.410	.098	-.031	.289	.326	.329	.248	.226	.302	.310	.572	.529			
	.106	.172	.256	.285	.018	-.025	.178	.246	.249	.149	.123	.176	.233	.507	.429			
	.172	.070	.147	.134	.094	-.037	.076	.140	.104	.104	.008	.070	.117	.072	.123			
	.261	.152	.268	.201	.012	.004	.139	.246	.181	.120	.040	.118	.223	.159	.008			
	.302	.136	.082	.092	.081	.004	.115	.063	.069	.002	.012	.094	.047	.046	.056			
	.326	.104	.012	.123	.174	.039	.139	.011	.027	.062	.049	.094	.027	.001	.091			
	.343	.651	.104	.367	.241	.044	.642	.178	.233	.120	.120	.632	.197	.114	.085			
	.387	.172	.033	.004	.116	.044	.163	.037	.030	.056	.107	.147	.069	.056	.075			
	.442	.003	.111	.089	.078	.100	.035	.085	.050	.082	.091	.057	.050	.015	.024			
	.493	.042	.084	.076	.036	.138	.086	.035	.018	.095	.086	.039	.036	.021	.091			
Turbine cowl	.537	.041	.044	.076	.015	.537	.095	.049	.024	.088	.080	.060	.056	.011	.193			
	.581	.095	.121	.008	.025	.509	.061	.101	.120	.098	.299	.002	.008	.069	.130			
	.617	.094	.027	.108	.172	.185	.047	.011	.021	.111	.149	.002	.056	.014	.030			
	.653	.026	.121	.082	.001	.181	.016	.037	.021	.104	.149	.052	.059	.072	.130			
	.690	.113	.027	.028	.095	.097	.052	.053	.017	.219	.197	.032	.123	.066	.114			
Plug	.718	.060	.023	.232	.148	.097	.052	.037	.306	.091	.197	.075	.303	.097	.097			
	.735	.025	.025	.153	.153	.097	.021	.021	.075	.075	.075	.015	.015	.309	.309			
	.768	.596	.596	.392	.392	.274	.274	.274	.497	.497	.497	.175	.175	.364	.364			
	.786	.204	.204	.039	.039	.188	.188	.188	.130	.130	.130	.124	.124	.104	.104			
	.821	.009	.009	.231	.231	.056	.056	.056	.091	.091	.091	.098	.098	.136	.136			
.852	.283	.283	.283	.490	.490	.332	.332	.332	.401	.401	.355	.355	.417	.417				
Fan cowl	0.000	.575	.381	1.054	.920	.662	.404	.375	1.114	.971	.483	.418	1.014	.748				
	.003	.686	.686	.166	.287	.758	.150	.690	.258	.172	.123	.669	.695	.406	.498			
	.014	.678	.474	.285	.264	.664	.478	.478	.070	.201	.531	.506	.406	.498				
	.031	.484	.362	.940	.197	.373	.436	.295	.655	.134	.348	.360	.483	.406	.498			
	.055	.416	.301	.220	.175	.232	.334	.295	.115	.137	.256	.302	.310	.572	.529			
	.106	.275	.220	.178	.104	.114	.164	.211	.102	.067	.099	.176	.233	.507	.429			
	.172	.163	.111	.050	.031	.040	.067	.102	.001	.045	.045	.123	.207	.429	.355			
	.261	.202	.207	.137	.050	.034	.067	.192	.089	.006	.068	.070	.072	.123	.429			
	.302	.173	.034	.031	.069	.091	.033	.019	.007	.119	.151	.118	.159	.429	.355			
	.326	.207	.037	.056	.117	.117	.048	.052	.007	.157	.173	.094	.044	.429	.355			
	.343	.717	.185	.082	.165	.101	.567	.145	.022	.196	.054	.044	.429	.355	.417			
	.387	.241	.082	.082	.155	.027	.121	.126	.123	.209	.132	.123	.050	.429	.355			
	.442	.013	.011	.011	.133	.229	.098	.052	.058	.135	.362	.086	.036	.429	.355			
	Turbine cowl	.537	.192	.092	.037	.056	.197	.019	.113	.091	.148	.081	.081	.036	.429	.355		
.581		.178	.047	.037	.136	.159	.001	.087	.033	.058	.022	.022	.036	.429	.355			
.617		.066	.088	.037	.062	.046	.122	.145	.097	.017	.202	.148	.154	.429	.355			
.653		.023	.101	.063	.072	.075	.141	.168	.126	.148	.154	.154	.429	.355	.417			
.690		.023	.146	.095	.021	.159	.180	.222	.158	.049	.060	.049	.429	.355	.417			
Plug	.718	.114	.114	.310	.120	.174	.174	.360	.360	.183	.183	.183	.355	.417	.417			
	.735	.005	.005	.005	.284	.007	.007	.007	.007	.007	.007	.007	.007	.355	.417			
	.768	.137	.137	.137	.172	.172	.172	.172	.172	.172	.172	.172	.172	.355	.417			
	.786	.076	.076	.076	.104	.104	.104	.104	.104	.104	.104	.104	.104	.355	.417			
	.821	.137	.137	.137	.178	.178	.178	.178	.178	.178	.178	.178	.178	.355	.417			
.852	.384	.384	.384	.431	.431	.431	.431	.431	.431	.431	.431	.431	.355	.417				

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(b) $M = 0.700$; outboard station

x/c	C _p at																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.850	.696	.773	.626	1.153	.565	.781	.990	.665	1.036	.456	.688	1.082	.974			
	.003	-.181	-.568	.253	-.800	-.398	-.525	-.583	.173	-.681	-.671	-.785	-.591	.441	-.814			
	.014	-.321	-.552	-.645	-.555	-.417	-.501	-.583	-.503	-.542	-.574	-.573	-.549	-.354	-.633			
	.031	-.795	-.869	-.613	-.427	-.395	-.897	-.968	-.483	-.423	-.436	-.785	-.840	-.593	-.450			
	.055	-.282	.004	-.458	-.430	-.395	-.332	-.567	-.377	-.443	-.369	-.350	-.380	-.322	-.372			
	.106	-.210	-.290	-.322	-.395	-.227	-.235	-.289	-.289	-.333	-.259	-.239	-.270	-.251	-.346			
	.172	-.118	-.131	-.164	-.211	-.108	-.124	-.276	-.137	-.198	-.230	-.113	-.231	-.108	-.140			
	.261	-.210	-.140	-.202	-.260	-.186	-.197	-.073	-.186	-.208	-.269	-.181	-.108	-.160	-.105			
	.302	-.152	-.102	-.329	-.121	-.115	-.134	-.086	-.263	-.153	-.092	-.113	-.070	-.248	-.028			
	.326	-.128	-.031	-.025	-.147	-.095	-.114	-.011	-.002	-.111	-.037	-.098	.005	.021	-.002			
	.343	-.191	.014	-.148	-.033	-.285	-.182	.040	-.105	.056	-.272	-.181	.053	.021	-.002			
	.387	-.336	.034	.005	-.176	-.202	-.332	.027	.034	-.024	-.204	.316	.024	.037	-.253			
Turbine cowl	.442	.012	-.135	-.076	-.218	-.295	.030	-.115	-.044	.024	.015	.047	.037	.021	.069			
	.493	.037	.069	.041	.008	-.128	.035	.040	-.011	-.034	.047	.037	.021	.040	.092			
	.537	-.041	.028	-.054	-.240	-.195	-.027	.021	-.060	-.214	.027	.047	.037	.021	.001			
	.581	-.118	-.122	-.128	-.211	-.459	-.037	.050	.015	-.064	-.179	.016	.011	-.028	.017			
	.617	-.099	-.012	-.051	-.279	-.536	-.023	-.002	.015	.027	-.401	.018	.037	.047	-.053			
Plug	.653	-.123	-.112	-.044	-.253	-.469	-.008	.034	.002	.075	-.169	.047	.053	.040	-.108			
	.690	-.128	-.148	-.004	-.276	-.398	-.105	.021	.040	.230	-.311	.036	.021	.079	-.224			
	.728	.050	.253	.295	-.536	-.398	-.105	.021	.040	.230	-.311	.036	.021	.079	-.224			
	.735	-.057	-.057	.015	-.041	.040	.031	.053	.027	.075	.036	.097	.036	.097	-.662			
	.768	-.274	-.076	.008	.073	-.214	-.285	.065	.108	.091	.125	.040	.040	.091	-.534			
Fan cowl	0.000	-.316	.757	1.153	.726	.927	-.069	.742	1.209	.827	.812	.428	.358					
	.003	-.591	-.520	-.245	-.518	-.995	-.1286	-.627	.221	-.383	-.1424	.091	.091					
	.014	-.618	-.844	-.296	-.460	-.692	-.706	-.520	-.045	-.370	-.699	.091	.091					
	.031	-.783	-.844	-.296	-.376	-.492	-.720	-.737	-.161	-.328	-.512	.091	.091					
	.055	-.362	-.552	-.264	-.447	-.382	-.386	-.507	-.155	-.315	-.383	.091	.091					
	.106	-.241	-.264	-.219	-.211	-.373	-.236	-.258	-.148	-.202	-.209	.091	.091					
	.172	-.115	-.303	-.083	-.182	-.121	-.101	-.278	-.032	-.090	-.038	.091	.091					
	.261	-.158	-.151	-.138	-.192	-.082	-.120	.145	-.096	-.102	-.022	.091	.091					
	.302	-.100	-.057	-.306	.008	.002	-.067	.051	-.181	.052	.068	.091	.091					
	.326	-.086	.015	.041	.040	.031	.053	.027	.075	.036	.097	.091	.091					
	.343	-.192	.060	.057	.073	.214	-.285	.065	.108	.091	.125	.040	.040					
	.387	-.304	.044	.073	.073	.102	-.060	.036	.053	.123	.200	.040	.040					
	Turbine cowl	.442	.060	.008	.015	-.060	.111	.088	.036	.053	.123	.200	.040	.040				
.493		.050	.060	.047	.098	.163	.131	.095	.085	.204	.316	.091	.091					
.537		.074	.086	.047	.044	.186	.112	.101	.091	.149	.178	.091	.091					
.581		.040	.044	.005	-.060	.024	.112	.095	.059	.038	.178	.091	.091					
.617		.055	.086	.073	.027	-.285	.131	.137	.117	.001	-.006	.091	.091					
Plug	.653	.084	.079	.073	.011	.015	.175	.137	.124	.068	.133	.091	.091					
	.690	.007	.054	.105	.015	-.156	.112	.121	.163	.175	.014	.091	.091					
	.728	.387	.387	.387	-.685	-.156	.112	.121	.163	.175	.014	.091	.091					
	.735	.076	.076	.076	-.008	.008	.137	.137	.428	.036	.014	.091	.091					
	.768	-.066	-.066	-.066	-.224	-.008	.027	.027	.091	.091	.014	.091	.091					
.821	-.015	-.015	-.015	.005	.005	.091	.091	.091	.091	.014	.091	.091						
.852	.397	.397	.397	.397	.450	.450	.431	.431	.500	.500	.500	.430	.430					

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued
(c) $M = 0.750$; inboard station

x/c	C _p at -														
	α = 1°														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.881	.416	.697	.815	.966	.756	.402	.867	.854	.664	.395	.959	.886	.736
	.003	.262	.840	-1.422	.449	.038	.715	.632	.867	.473	-1.022	.813	.546	.887	.685
	.014	.306	.565	-1.053	.330	.137	.472	.548	.624	.386	.526	.526	.453	.334	.410
	.031	.244	.424	.808	.228	.079	.339	.407	.730	.280	.367	.394	.746	.252	.305
	.055	.249	.357	.430	.190	.076	.290	.337	.343	.240	.300	.315	.277	.200	.249
	.106	.174	.266	.296	.082	.096	.175	.243	.243	.146	.172	.227	.206	.103	.106
	.172	.063	.149	.135	.041	.023	.065	.123	.091	.021	.052	.104	.060	.013	.036
	.261	.152	.275	.208	.067	.006	.131	.234	.176	.108	.052	.106	.060	.002	.036
	.302	.130	.082	.086	.009	.035	.096	.047	.059	.012	.047	.070	.027	.033	.090
	.326	.038	.009	.278	.082	.021	.016	.026	.105	.082	.070	.041	.052	.137	.116
	.343	.532	.105	.266	.242	.044	.512	.161	.249	.169	.056	.500	.196	.230	.195
	.387	.125	.108	.126	.058	.093	.096	.026	.047	.056	.108	.137	.093	.087	.090
Turbine cowl	.442	.030	.026	.102	.076	.070	.033	.023	.038	.003	.054	.028	.075	.020	.198
	.493	.091	.044	.170	.169	.070	.019	.023	.012	.076	.052	.049	.034	.031	.305
	.537	.284	.084	.682	.120	.263	.090	.009	.012	.184	.147	.028	.071	.027	.086
	.581	.094	.056	.038	.149	.297	.029	.082	.085	.266	.032	.125	.013	.109	.033
	.617	.038	.100	.088	.140	.324	.019	.006	.064	.088	.063	.034	.020	.124	.071
	.653	.028	.088	.059	.266	.201	.038	.097	.009	.216	.036	.137	.084	.156	.014
	.690	.169	.088	.053	.266	.201	.038	.097	.009	.216	.036	.137	.084	.156	.014
	.708	.023	.023	.222	.079	.400	.020	.020	.284	.009	.087	.342	.087	.342	.087
	.735	.023	.023	.137	.137	.400	.020	.020	.284	.009	.087	.342	.087	.342	.087
	.768	.023	.023	.137	.137	.400	.020	.020	.284	.009	.087	.342	.087	.342	.087
	.796	.023	.023	.137	.137	.400	.020	.020	.284	.009	.087	.342	.087	.342	.087
	.821	.076	.076	.224	.224	.437	.322	.425	.426	.397	.369	.369	.369	.369	.369
.852	.275	.275	.437	.437	.489	.415	.489	.489	.489	.489	.489	.489	.489	.489	
Fan cowl	0.000	.592	.389	1.050	.900	.670	.448	.350	1.136	.939	.514	.217	.514	.217	.514
	.003	.1280	.818	.252	.339	.815	.557	.638	.154	.217	.514	.217	.514	.217	.514
	.014	.842	.519	.334	.298	.447	.1.062	.528	.095	.211	.520	.211	.520	.211	.520
	.031	.487	.387	.827	.193	.316	.408	.364	.546	.141	.339	.141	.339	.141	.339
	.055	.407	.311	.235	.184	.234	.324	.300	.124	.135	.224	.135	.224	.135	.224
	.106	.261	.217	.179	.097	.105	.147	.203	.106	.121	.077	.121	.077	.121	.077
	.172	.146	.097	.044	.012	.017	.027	.086	.005	.066	.066	.066	.066	.066	.066
	.261	.186	.202	.135	.085	.108	.049	.174	.002	.014	.090	.014	.090	.014	.090
	.302	.150	.024	.024	.134	.140	.083	.075	.128	.183	.201	.139	.201	.139	.201
	.326	.093	.056	.059	.199	.044	.083	.254	.020	.139	.201	.139	.201	.139	.201
	.343	.602	.217	.202	.152	.020	.094	.137	.140	.201	.130	.247	.130	.247	.130
	.387	.221	.129	.091	.076	.210	.084	.058	.069	.236	.332	.236	.332	.236	.332
Turbine cowl	.442	.084	.105	.003	.134	.228	.150	.175	.102	.233	.370	.233	.370	.233	.370
	.493	.018	.041	.053	.014	.161	.105	.122	.105	.142	.008	.142	.008	.142	.008
	.537	.075	.062	.059	.029	.123	.154	.040	.022	.021	.095	.021	.095	.021	.095
	.581	.035	.029	.035	.009	.064	.145	.151	.110	.203	.189	.203	.189	.203	.189
	.617	.035	.141	.044	.026	.032	.145	.178	.128	.020	.139	.020	.139	.020	.139
	.653	.013	.103	.056	.023	.032	.126	.236	.157	.168	.069	.157	.168	.069	.069
	.690	.018	.105	.103	.103	.038	.176	.187	.374	.160	.069	.160	.069	.160	.069
	.708	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.735	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.768	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.796	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.821	.070	.070	.224	.224	.437	.322	.425	.426	.397	.369	.369	.369	.369	.369
.852	.375	.375	.432	.432	.489	.415	.489	.489	.489	.489	.489	.489	.489	.489	
Plug	0.000	.592	.389	1.050	.900	.670	.448	.350	1.136	.939	.514	.217	.514	.217	.514
	.003	.1280	.818	.252	.339	.815	.557	.638	.154	.217	.514	.217	.514	.217	.514
	.014	.842	.519	.334	.298	.447	.1.062	.528	.095	.211	.520	.211	.520	.211	.520
	.031	.487	.387	.827	.193	.316	.408	.364	.546	.141	.339	.141	.339	.141	.339
	.055	.407	.311	.235	.184	.234	.324	.300	.124	.135	.224	.135	.224	.135	.224
	.106	.261	.217	.179	.097	.105	.147	.203	.106	.121	.077	.121	.077	.121	.077
	.172	.146	.097	.044	.012	.017	.027	.086	.005	.066	.066	.066	.066	.066	.066
	.261	.186	.202	.135	.085	.108	.049	.174	.002	.014	.090	.014	.090	.014	.090
	.302	.150	.024	.024	.134	.140	.083	.075	.128	.183	.201	.139	.201	.139	.201
	.326	.093	.056	.059	.199	.044	.083	.254	.020	.139	.201	.139	.201	.139	.201
	.343	.602	.217	.202	.152	.020	.094	.137	.140	.201	.130	.247	.130	.247	.130
	.387	.221	.129	.091	.076	.210	.084	.058	.069	.236	.332	.236	.332	.236	.332
Fan cowl	.442	.084	.105	.003	.134	.228	.150	.175	.102	.233	.370	.233	.370	.233	.370
	.493	.018	.041	.053	.014	.161	.105	.122	.105	.142	.008	.142	.008	.142	.008
	.537	.075	.062	.059	.029	.123	.154	.040	.022	.021	.095	.021	.095	.021	.095
	.581	.035	.029	.035	.009	.064	.145	.151	.110	.203	.189	.203	.189	.203	.189
	.617	.035	.141	.044	.026	.032	.145	.178	.128	.020	.139	.020	.139	.020	.139
	.653	.013	.103	.056	.023	.032	.126	.236	.157	.168	.069	.157	.168	.069	.069
	.690	.018	.105	.103	.103	.038	.176	.187	.374	.160	.069	.160	.069	.160	.069
	.708	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.735	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.768	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.796	.000	.000	.340	.009	.014	.152	.187	.374	.160	.069	.160	.069	.160	.069
	.821	.070	.070	.224	.224	.437	.322	.425	.426	.397	.369	.369	.369	.369	.369
.852	.375	.375	.432	.432	.489	.415	.489	.489	.489	.489	.489	.489	.489	.489	

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(d) $M = 0.750$; outboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.803	.719	.768	.620	1.149	.570	.681	.958	.648	1.043	.442	.738	1.050	.682	.986		
	.003	-.270	-.755	-.378	-.884	-.423	-.749	-.703	-.287	-.782	-.755	-.941	-.743	-.351	-.660	-.866		
	.014	-.385	-.681	-.464	-.643	-.664	-.546	-.649	-.546	-.591	-.614	-.583	-.630	-.394	-.548	-.681		
	.031	-.575	-.669	-.665	-.690	-.373	-.722	-.773	-.534	-.432	-.661	-.600	-.630	-.423	-.392	-.483		
	.055	-.301	-.280	-.492	-.455	-.341	-.327	-.401	-.322	-.411	-.382	-.362	-.340	-.341	-.380	-.383		
	.106	-.125	-.310	-.351	-.361	-.270	-.241	-.295	-.298	-.294	-.232	-.238	-.276	-.261	-.266	-.242		
	.172	-.235	-.033	-.174	-.200	-.156	-.122	-.112	-.136	-.179	-.132	-.110	-.075	-.104	-.172	-.177		
	.261	-.226	.292	-.218	-.276	-.188	-.193	.095	-.186	-.206	-.197	-.172	.138	-.157	-.166	-.098		
	.302	-.151	-.109	-.218	-.226	-.220	-.126	-.077	-.147	-.206	-.091	-.101	-.057	-.149	-.007	-.018		
	.326	-.125	-.029	-.621	-.258	-.091	-.196	.000	.012	-.088	-.065	-.079	.020	.038	-.007	-.023		
	.343	-.156	.089	-.088	.114	-.217	.144	.112	-.068	.147	-.197	.132	.126	-.051	.158	-.192		
	.367	-.301	.003	-.015	-.015	-.288	-.285	-.006	.036	.076	-.176	-.269	.026	.073	.061	-.113		
	.442	-.041	-.068	-.094	-.200	-.059	-.047	.024	-.068	.082	.059	-.093	.020	-.097	.005	.046		
	.493	.038	-.056	-.032	-.208	-.088	.072	.024	-.059	.125	.059	.076	.044	.044	.089	.093		
Turbine cowl	.537	-.068	-.056	-.032	-.232	-.150	.010	-.026	-.059	-.085	-.197	.001	.026	-.036	-.080	-.057		
	.581	-.068	-.097	-.129	-.311	-.358	-.007	-.083	-.059	-.153	.359	.049	.132	.020	-.129	-.077		
	.617	-.134	-.012	.003	-.311	-.628	-.007	.139	.018	-.300	.273	.027	.055	.049	.120	-.077		
	.653	-.076	-.109	-.077	-.176	-.352	-.056	-.047	-.003	-.073	-.314	-.035	.005	.073	.310	-.257		
	.690	-.169	-.133	.018	-.188	-.393	-.091	.071	.053	-.067	-.363	.363	.375	.375	.760	.001		
	.726		.274	.266	-.599		.307	.343		.006		.079			.061	.061		
	.735		-.018		-.032		.059			.435		.113			.046	.046		
	.768		-.413		-.502		.319			.009		-.021			.176	.176		
	.786		-.138		-.352		.032			.185		.150			.369	.369		
	.821		.080		.153		.118			.354								
	.852		.292		.341		.354											
	Fan cowl	0.000	.323	.817	1.135	.704	-.937	.133	.845	1.222	.768	.831						
		.003	-.1038	-.736	-.278	-.597	-.115	-.1267	-.771	-.330	-.468	-.1379						
		.014	-.632	-.612	-.295	-.521	-.568	-.1289	-.603	-.086	-.427	-.1026						
.031		-.380	-.671	-.349	-.412	-.521	-.566	-.556	-.190	-.345	-.503							
.055		-.239	-.355	-.295	-.362	-.388	-.385	-.316	-.195	-.327	-.380							
.106		-.106	-.269	-.230	-.271	-.253	-.221	-.260	-.169	-.192	-.209							
.172		-.155	-.165	-.086	-.165	-.094	-.089	-.131	-.036	-.077	-.027							
.261		-.093	-.050	-.204	.024	.012	-.058	-.033	-.048	.061	.084							
.302		-.071	-.027	.050	.026	.047	-.032	.041	.085	.073	.117							
.343		-.150	.121	-.053	.153	.182	-.129	.138	-.018	.176	.109							
.387		-.269	.018	.086	.044	-.032	-.239	.076	.117	.087	.032							
.442		-.009	-.027	.018	.059	.062	.035	.048	.052	.090	.199							
.493		.084	.033	.047	.085	.188	.123	.111	.085	.126	.272							
Turbine cowl		.581	.066	.059	.029	-.185	.024	.096	.052	.049	-.004	.161						
	.617	.075	.112	.077	-.062	-.182	.123	.144	.126	.108	.057							
	.653	.080	.092	.065	.165	.012	.162	.179	.106	.131	.111							
	.690	.039	.038	.118	-.282	-.177	.092	.141	.150	-.021	-.027							
	.726		.387	.396	-.665		.421	.436		.052								
	.735		.098		-.032		.135			.052								
	.768		-.071		-.360		-.004			.198								
	.786		.016		.035		.082			.061								
	.821		.174		.188		.215			.231								
	.852		.387		.444		.412			.481								

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(e) $M = 0.775$; inboard station

x/\bar{c}	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	.000	.889	.424	.713	.820	.971	.411	.864	.845	.826
	.003	-.271	-.871	1.374	-.475	-.002	-.867	-.997	-.486	-.404
	.014	-.296	-.581	1.183	-.318	-.105	-.488	-.622	-.382	-.416
	.031	-.254	-.429	-.657	-.195	-.044	-.339	-.620	-.298	-.275
	.055	-.245	-.356	-.418	-.145	-.038	-.335	-.620	-.236	-.211
	.106	-.169	-.258	-.286	-.047	.026	-.231	-.237	-.138	-.104
	.172	-.054	-.137	-.123	.077	.077	-.110	-.079	-.044	-.019
	.261	-.139	-.266	-.199	-.027	.029	-.118	-.045	-.095	-.064
	.302	-.118	-.069	-.075	.085	.141	-.084	-.024	-.107	-.017
	.326	-.167	.007	.415	.150	.152	-.054	.197	.092	.056
	.343	-.466	.142	-.213	.309	.102	-.454	.195	.087	.043
	.387	-.156	.024	.024	.166	-.030	-.135	.064	.073	.089
	.442	-.054	-.131	-.064	-.010	.259	-.071	-.088	-.054	-.073
Turbine cowl	.493	.048	.125	.057	.144	.161	.095	.025	.127	.221
	.537	.214	-.066	-.035	-.077	.456	.099	.025	.062	.162
	.581	.396	-.027	-.063	-.173	-.167	.175	.019	-.118	.134
	.617	-.041	-.137	-.007	-.147	-.293	-.016	.064	-.066	.236
	.653	-.024	-.061	-.267	-.105	-.335	.018	.020	.050	.146
Plug	.690	-.143	-.089	.024	-.299	.343	-.050	.036	.048	.261
	.728	-.106	-.106	.142	.133	.307	.087	.307	.028	.016
	.735	-.016	-.016	.169	.169	.307	.087	.307	.028	.016
	.768	-.488	-.789	-.173	-.173	.378	-.003	.307	.028	.016
	.821	-.038	.038	.138	.138	.204	.132	.204	.064	.045
Fan cowl	.000	-.606	-.374	1.046	.883	.688	.482	.381	.933	.551
	.003	-.189	-.927	1.295	-.361	-.884	-.150	-.956	-.263	-.137
	.014	-.512	-.545	-.365	-.305	-.473	-.322	-.584	-.258	-.499
	.031	-.392	-.390	-.747	-.193	-.193	-.493	-.387	-.151	-.341
	.055	-.311	-.314	-.233	-.182	-.227	-.398	-.306	-.145	-.235
	.106	-.158	-.210	-.087	-.087	-.092	-.228	-.207	-.067	-.131
	.172	-.039	-.084	-.036	.034	.042	-.110	.081	.067	.070
	.261	-.077	-.151	-.123	.030	.042	-.131	.177	.093	.093
	.302	-.048	-.008	-.008	.098	.124	-.101	.003	.012	.135
	.326	-.092	.071	.147	.143	.154	-.073	.079	.194	.174
	.343	-.447	.231	-.182	.227	-.011	-.521	.284	.174	.207
	.387	-.111	.053	.113	.126	.020	-.178	.118	.244	.020
	.442	-.060	.017	.026	.079	.149	-.080	.071	.205	.115
Turbine cowl	.493	.097	.088	.086	.062	.258	.043	.045	.151	.303
	.537	.097	.090	.082	.079	.213	.073	.099	.104	.331
	.581	.148	.054	.000	.000	.079	.079	.079	.067	.016
	.617	.071	.166	.023	-.042	.119	.079	.087	.045	.135
	.653	.088	.093	.037	-.087	.051	.169	.087	.109	.154
Plug	.690	.080	.180	.107	-.123	.048	.051	.115	.121	.126
	.728	.135	.135	.352	.157	.205	.172	.155	-.028	.107
	.735	.009	.009	.266	.157	.205	.172	.155	-.028	.107
	.768	-.312	-.312	-.266	.157	.205	.172	.155	-.028	.107
	.821	-.047	-.047	-.050	.157	.205	.172	.155	-.028	.107
Fan cowl	.000	-.606	-.374	1.046	.883	.688	.482	.381	.933	.551
	.003	-.189	-.927	1.295	-.361	-.884	-.150	-.956	-.263	-.137
	.014	-.512	-.545	-.365	-.305	-.473	-.322	-.584	-.258	-.499
	.031	-.392	-.390	-.747	-.193	-.193	-.493	-.387	-.151	-.341
	.055	-.311	-.314	-.233	-.182	-.227	-.398	-.306	-.145	-.235
	.106	-.158	-.210	-.087	-.087	-.092	-.228	-.207	-.067	-.131
	.172	-.039	-.084	-.036	.034	.042	-.110	.081	.067	.070
	.261	-.077	-.151	-.123	.030	.042	-.131	.177	.093	.093
	.302	-.048	-.008	-.008	.098	.124	-.101	.003	.012	.135
	.326	-.092	.071	.147	.143	.154	-.073	.079	.194	.174
	.343	-.447	.231	-.182	.227	-.011	-.521	.284	.174	.207
	.387	-.111	.053	.113	.126	.020	-.178	.118	.244	.020
	.442	-.060	.017	.026	.079	.149	-.080	.071	.205	.115
Turbine cowl	.493	.097	.088	.086	.062	.258	.043	.045	.151	.303
	.537	.097	.090	.082	.079	.213	.073	.099	.104	.331
	.581	.148	.054	.000	.000	.079	.079	.079	.067	.016
	.617	.071	.166	.023	-.042	.119	.079	.087	.109	.154
	.653	.088	.093	.037	-.087	.051	.169	.087	.109	.154
Plug	.690	.080	.180	.107	-.123	.048	.051	.115	.121	.126
	.728	.135	.135	.352	.157	.205	.172	.155	-.028	.107
	.735	.009	.009	.266	.157	.205	.172	.155	-.028	.107
	.768	-.312	-.312	-.266	.157	.205	.172	.155	-.028	.107
	.821	-.047	-.047	-.050	.157	.205	.172	.155	-.028	.107

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(f) $M = 0.775$; outboard station

x/c	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																										
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																													
Fan cowl	0.000	.787	.726	.773	.618	1.146	.570	.841	.963	.658	1.051	.834	1.051	.834	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963	.658	1.051	.834	.963</

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(g) M = 0.800; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.899	.421	.723	.820	.989	.766	.410	.875	.826	.693	.402	.959	.868	.747			
	.003	-.280	-.886	1.323	-.630	-.124	-.775	-.945	-.575	-.544	-.1086	-.920	-.693	-.821	-.742			
	.014	-.321	-.592	1.176	-.669	-.502	-.578	-.594	-.594	-.547	-.505	-.582	-.506	-.570	-.521			
	.031	-.251	-.443	.400	-.347	-.355	-.435	-.443	-.296	-.275	-.391	-.422	-.370	-.421	-.381			
	.055	-.247	-.356	.378	-.285	-.177	-.289	-.351	-.242	-.218	-.309	-.333	-.298	-.275	-.281			
	.106	-.161	-.254	.278	-.180	-.105	-.167	-.238	-.135	-.105	-.166	-.225	-.209	-.208	-.221			
	.172	-.043	-.124	.110	-.056	-.045	-.105	-.078	-.005	-.000	-.044	-.092	-.054	-.025	-.040			
	.261	-.129	-.259	.192	-.159	-.045	-.114	-.078	-.005	-.014	-.093	-.209	-.146	-.066	-.035			
	.302	-.104	-.059	.062	-.032	-.077	-.224	-.167	-.108	-.067	-.060	-.019	-.027	-.078	-.105			
	.326	-.337	.017	.571	.011	-.188	.046	.305	.046	.097	.267	.062	.132	.125	.132			
	.343	-.362	.179	.571	.022	-.359	.213	.157	.229	.097	.267	.062	.132	.125	.132			
	.387	-.145	.071	.024	.025	-.142	-.134	.073	.070	.108	-.081	-.076	.105	.135	-.016			
Turbine cowl	.442	.018	-.070	.073	-.037	.004	-.084	-.003	-.062	.143	-.003	-.079	-.008	.135	.000			
	.493	.047	.060	.110	-.110	.019	.167	.081	.194	.172	.116	.192	.057	.218	.280			
	.537	.373	.129	.206	-.086	.541	.200	.030	-.172	.412	.234	.013	.043	.054	.386			
	.581	.512	.046	.148	.382	-.223	.269	.030	-.124	-.022	.304	.062	.114	.132	.070			
	.617	.006	.110	.156	.291	-.380	.012	-.049	.111	.127	.003	.021	.065	.219	.052			
	.653	-.010	.108	.054	.291	-.520	.012	.043	.081	-.280	.054	.062	.111	.087	.170			
	.690	-.125	.110	-.146	.361	.335	-.073	-.022	-.051	.118	.063	.073	.054	.162	.151			
	.708	-.064	.187	.038	.038			.049	.286	.035		.135	.327	.051				
	.735	-.013	.404	-.404				.000		.296		.135		.262				
	.768	-.421	.401	-.401				.597		.315		.339		.281				
	.821	-.075	.361	-.366				.124		.743		.374		.569				
	.852		.290	-.318				.319		.129		.146		.224				
									.398		.333		.423					
Fan cowl	0.000	.632	.409	1.037	.886	.693	.518	.409	1.140	.935	.554							
	.003	-.1209	-.509	.327	-.383	-.973	-.1477	-.947	.075	-.276	-.554							
	.014	-.780	-.570	.376	-.329	-.474	-.1260	.585	.075	-.243	-.534							
	.031	-.392	.403	.579	-.215	-.321	-.383	.401	.333	-.149	-.324							
	.055	-.151	.319	.240	-.186	-.223	-.379	.306	.139	-.146	-.222							
	.106	-.036	.178	.178	-.086	-.086	.216	.195	.117	-.063	-.146							
	.172	-.073	.078	.032	.041	.054	.097	.063	.007	.075	.085							
	.261	-.073	.186	.127	.027	.049	.118	.166	.087	.021	.099							
	.302	-.040	.002	.068	.103	.135	.085	.018	.291	.150	.193							
	.326	.188	.079	.233	.149	.165	.094	.297	.188	.217	.217							
	.343	-.363	.255	.151	.254	.014	.473	.280	.141	.274	.045							
	.387	-.102	.095	.114	.157	.022	-.163	.115	.151	.185	.118							
Turbine cowl	.442	-.020	-.054	.010	.049	.119	.114	.026	.061	.153	.260							
	.493	.148	.203	.060	.108	.337	.074	.115	.118	.131	.336							
	.537	.180	.068	.014	.076	-.256	.188	.129	.143	.126	.015							
	.581	.205	.082	-.062	.054	-.121	.197	.080	.015	.056	.236							
	.617	.058	.168	.027	-.143	-.027	.058	.197	.061	.072	.158							
	.653	.082	.065	.030	-.022	-.035	.050	.143	.080	.018	.115							
	.690	.066	.217	.119	-.151	.116	.066	.180	.132	-.041	.193							
	.708		.109	.336	.054			.197	.397	.193								
	.735		.011		-.234			.018		-.136								
	.768		-.284		-.318			.168		-.268								
	.821		-.065		-.428			-.009		.034								
	.852		.374		.432			.402		.476								
Plug	0.000	.432	.409	1.037	.886	.693	.518	.409	1.140	.935	.554							
	.003	-.1209	-.509	.327	-.383	-.973	-.1477	-.947	.075	-.276	-.554							
	.014	-.780	-.570	.376	-.329	-.474	-.1260	.585	.075	-.243	-.534							
	.031	-.392	.403	.579	-.215	-.321	-.383	.401	.333	-.149	-.324							
	.055	-.151	.319	.240	-.186	-.223	-.379	.306	.139	-.146	-.222							
	.106	-.036	.178	.178	-.086	-.086	.216	.195	.117	-.063	-.146							
	.172	-.073	.078	.032	.041	.054	.097	.063	.007	.075	.085							
	.261	-.073	.186	.127	.027	.049	.118	.166	.087	.021	.099							
	.302	-.040	.002	.068	.103	.135	.085	.018	.291	.150	.193							
	.326	.188	.079	.233	.149	.165	.094	.297	.188	.217	.217							
	.343	-.363	.255	.151	.254	.014	.473	.280	.141	.274	.045							
	.387	-.102	.095	.114	.157	.022	-.163	.115	.151	.185	.118							
Plug	.442	-.020	-.054	.010	.049	.119	.114	.026	.061	.153	.260							
	.493	.148	.203	.060	.108	.337	.074	.115	.118	.131	.336							
	.537	.180	.068	.014	.076	-.256	.188	.129	.143	.126	.015							
	.581	.205	.082	-.062	.054	-.121	.197	.080	.015	.056	.236							
	.617	.058	.168	.027	-.143	-.027	.058	.197	.061	.072	.158							
	.653	.082	.065	.030	-.022	-.035	.050	.143	.080	.018	.115							
	.690	.066	.217	.119	-.151	.116	.066	.180	.132	-.041	.193							
	.708		.109	.336	.054			.197	.397	.193								
	.735		.011		-.234			.018		-.136								
	.768		-.284		-.318			.168		-.268								
	.821		-.065		-.428			-.009		.034								
	.852		.374		.432			.402		.476								

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued
(h) $M = 0.800$, outboard station

x/2	C_p at											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.780	.773	.795	.650	1.151	.585	.865	.965	.683	1.053	.891
	.003	.336	.874	.573	.616	.425	.758	.871	.430	.837	.816	.575
	.014	.401	.773	.513	.678	.480	.582	.789	.550	.637	.637	.712
	.031	.080	.337	.1029	.588	.401	.444	.490	.467	.499	.507	.600
	.055	.308	.736	.381	.469	.361	.362	.434	.433	.434	.396	.456
	.100	.234	.305	.198	.309	.298	.244	.297	.310	.303	.288	.407
	.172	.112	.311	.158	.219	.151	.114	.112	.131	.173	.157	.277
	.261	.214	.466	.271	.271	.116	.191	.112	.130	.134	.109	.063
	.302	.141	.082	.086	.162	.135	.118	.085	.068	.043	.053	.098
	.326	.104	.088	.001	.182	.057	.081	.016	.027	.000	.027	.032
	.387	.116	.178	.011	.213	.133	.110	.191	.225	.116	.113	.030
Turbine cowl	.442	.096	.073	.032	.087	.082	.094	.055	.065	.030	.203	.089
	.493	.018	.019	.103	.301	.138	.077	.011	.019	.033	.203	.014
	.537	.002	.136	.070	.057	.134	.045	.122	.076	.003	.060	.147
	.581	.035	.099	.027	.347	.406	.008	.114	.087	.303	.217	.136
	.617	.063	.108	.108	.366	.558	.081	.098	.101	.160	.253	.011
	.653	.214	.092	.061	.363	.618	.012	.079	.071	.214	.282	.054
	.690	.112	.122	.070	.366	.632	.175	.114	.076	.133	.382	.190
	.728	.227	.227	.300	.737	.632	.175	.343	.335	.840	.371	.245
	.765	.066	.066	.079	.406	.079	.079	.131	.057	.057	.101	.050
	.796	.326	.326	.406	.632	.074	.074	.332	.374	.229	.229	.399
	.821	.604	.604	.632	.074	.074	.074	.209	.504	.504	.169	.196
Plug	.852	.341	.341	.361	.361	.361	.361	.348	.412	.412	.354	.413
	0.000	.369	.864	1.112	.708	.962	.227	.904	1.211	.773	.891	.891
	.003	.978	.899	.401	.654	.126	.148	.929	.454	.521	.354	.375
	.014	.957	.703	.338	.567	.789	.127	.706	.437	.492	.169	.169
	.031	.379	.419	.417	.412	.510	.256	.281	.243	.353	.378	.378
	.055	.375	.119	.321	.380	.393	.321	.104	.221	.351	.369	.369
	.106	.232	.275	.245	.249	.238	.219	.242	.194	.185	.180	.180
	.172	.093	.039	.081	.135	.059	.081	.095	.050	.079	.017	.017
	.261	.146	.144	.147	.141	.059	.117	.163	.118	.106	.017	.017
	.302	.077	.038	.103	.030	.030	.068	.025	.117	.082	.092	.092
	.326	.048	.044	.041	.033	.068	.024	.054	.078	.138	.124	.124
	.343	.114	.205	.017	.226	.105	.101	.264	.024	.162	.063	.063
Turbine cowl	.387	.088	.088	.046	.144	.114	.114	.100	.114	.124	.010	.010
	.442	.053	.039	.002	.025	.201	.037	.003	.024	.085	.162	.162
	.493	.151	.251	.059	.084	.166	.164	.138	.046	.208	.306	.306
	.537	.013	.021	.029	.049	.166	.115	.093	.111	.018	.243	.243
	.581	.046	.039	.029	.049	.115	.115	.009	.007	.018	.146	.146
	.617	.070	.202	.023	.119	.172	.172	.076	.100	.036	.062	.062
	.653	.086	.010	.052	.100	.149	.119	.076	.100	.173	.082	.082
	.690	.013	.194	.142	.103	.119	.062	.122	.122	.193	.098	.098
	.728	.333	.333	.365	.183	.183	.062	.400	.422	.603	.603	.603
	.756	.140	.140	.083	.083	.083	.149	.149	.442	.116	.116	.116
	.788	.226	.226	.404	.404	.404	.050	.050	.378	.083	.083	.083
	.821	.039	.039	.032	.032	.032	.185	.185	.224	.224	.224	.224
Plug	.852	.371	.371	.435	.435	.435	.381	.381	.441	.441	.441	.441

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Continued

(i) $M = 0.825$; Inboard station

x/c	C _p at												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.898	.439	.741	.828	.585	.774	.430	.861	.821	.715	.423	.960
	.003	-.352	-.945	-.1304	-.673	-.119	-.803	-.981	-.1037	-.544	-.1017	-.53	-.868
	.014	-.352	-.610	-.1132	-.455	-.282	-.498	-.637	-.1037	-.445	-.934	-.537	-.319
	.031	-.277	-.543	-.003	-.368	-.187	-.368	-.454	-.196	-.291	-.368	-.412	-.018
	.055	-.261	-.369	-.247	-.293	-.181	-.263	-.352	-.320	-.235	-.295	-.321	-.284
	.106	-.163	-.257	-.268	-.176	-.104	-.167	-.237	-.273	-.102	-.146	-.209	-.198
	.172	-.038	-.120	-.101	-.044	-.000	-.041	-.059	-.079	-.044	-.079	-.073	-.039
	.261	-.132	-.260	-.190	-.153	-.036	-.112	-.227	-.167	-.008	-.047	-.190	-.057
	.302	-.101	-.052	-.053	-.088	-.070	-.076	-.031	-.039	.041	-.035	.000	-.041
	.326	.566	.028	.733	.939	.070	.318	.049	.096	.106	.501	.081	.096
	.343	.344	.215	.896	.823	.021	-.360	.222	.120	.244	.395	.259	.146
	.387	-.156	.070	.073	.013	-.119	-.131	.094	.023	.106	-.102	.115	.076
	.452	.080	-.005	.039	-.155	.018	.054	-.082	-.026	.131	.171	.075	.029
	.493	.021	-.109	-.122	-.135	.008	.070	.091	.060	.153	.158	.107	.105
Turbine cowl	.537	.382	.034	-.101	-.078	-.476	.334	-.068	-.099	-.135	.432	.458	.086
	.581	.601	-.164	-.171	-.378	-.119	.330	.084	-.081	-.135	.091	.446	.054
	.617	-.007	-.013	-.138	-.373	-.303	.030	-.026	-.008	-.128	.149	.071	.008
	.653	-.018	-.052	-.045	-.337	-.476	.042	-.016	-.172	-.115	.237	.089	.165
	.690	-.171	-.164	-.140	-.319	.486	-.064	.047	.086	-.304	.0213	.063	.081
	.708	-.008	-.008	.140	.088			-.026	.167	.044	.052	.076	.085
	.735	-.013	-.013		-.344		-.011	-.026	.167	.044	.052	.076	.085
	.768	-.330	-.330		-.474		-.011	-.026	.167	.044	.052	.076	.085
	.796	-.699	-.699		-.640		-.313	-.350	-.267	-.541	-.313	-.018	-.177
	.821	-.590	-.590		-.432		-.629	.062	-.541	-.541	-.543	-.154	-.159
	.852	-.018	-.018		.013			.300	.343	.343	.342	.422	.422
Fan cowl	0.000	.658	.431	1.031	.891	.722	.548	.408	1.135	.938	.583	.873	.756
	.003	-.019	-.959	-.433	-.416	-.1032	-.1396	-.956	.016	-.293	-.1295	-.463	-.440
	.014	-.019	-.616	-.432	-.331	-.476	-.1207	-.607	.156	-.252	-.896	-.358	-.283
	.031	-.019	-.412	-.214	-.227	-.315	-.397	-.398	.166	-.293	-.216	-.200	-.208
	.055	-.019	-.326	-.253	-.193	-.222	-.334	-.305	.143	-.062	-.096	-.094	-.080
	.106	-.019	-.209	-.186	-.087	-.081	-.204	-.190	.086	.096	.114	.057	.057
	.172	-.019	-.071	-.032	.048	.067	-.086	-.055	.091	.026	.202	.039	.039
	.261	-.019	-.186	-.128	-.027	.056	-.106	-.159	.023	.153	.231	.088	.088
	.302	-.019	.002	-.005	.111	.145	-.071	.029	.422	.195	.202	.088	.088
	.326	-.019	.088	.348	.160	.178	.366	.107	.422	.231	.231	.088	.088
	.343	-.019	.275	.057	.277	.036	-.413	.302	.096	.296	.067	.088	.088
	.387	-.019	.119	.096	.173	.030	-.145	.122	.159	.215	.117	.088	.088
	.442	-.019	-.035	.043	.038	.168	-.051	.018	.044	.109	.195	.088	.088
	.493	-.019	.200	.096	.282	.331	.098	.099	.081	.176	.075	.088	.088
	.537	-.019	.101	-.071	-.037	-.315	.248	.122	.021	.065	.143	.088	.088
Turbine cowl	.581	-.019	.101	-.071	-.037	-.315	.248	.122	.021	.065	.143	.088	.088
	.617	-.019	.101	-.071	-.037	-.315	.248	.122	.021	.065	.143	.088	.088
	.653	-.019	.132	.064	-.159	-.042	.047	.224	.117	.115	.039	.091	.091
	.690	-.019	.132	.064	-.159	-.042	.047	.224	.117	.115	.039	.091	.091
	.708	-.019	.043	.036	-.112	.186	.071	.284	.180	.091	.270	.091	.091
	.735	-.019	.169	.351	-.172	.106	.168	.168	.383	.166	.166	.166	.166
	.758	-.019	.316	.012	-.190	-.242	.242	.242	.242	.242	.242	.242	.242
	.781	-.019	.405	.215	-.562	.215	.215	.215	.215	.215	.215	.215	.215
	.821	-.019	.171	.351	-.453	.351	.351	.351	.351	.351	.351	.351	.351
	.852	-.019	.351	.351	.453	.453	.453	.453	.453	.453	.453	.453	.453
Plug	0.000	.658	.431	1.031	.891	.722	.548	.408	1.135	.938	.583	.873	.756
	.003	-.019	-.959	-.433	-.416	-.1032	-.1396	-.956	.016	-.293	-.1295	-.463	-.440
	.014	-.019	-.616	-.432	-.331	-.476	-.1207	-.607	.156	-.252	-.896	-.358	-.283
	.031	-.019	-.412	-.214	-.227	-.315	-.397	-.398	.166	-.293	-.216	-.200	-.208
	.055	-.019	-.326	-.253	-.193	-.222	-.334	-.305	.143	-.062	-.096	-.094	-.080
	.106	-.019	-.209	-.186	-.087	-.081	-.204	-.190	.086	.096	.114	.057	.057
	.172	-.019	-.071	-.032	.048	.067	-.086	-.055	.091	.026	.202	.039	.039
	.261	-.019	-.186	-.128	-.027	.056	-.106	-.159	.023	.153	.231	.088	.088
	.302	-.019	.002	-.005	.111	.145	-.071	.029	.422	.195	.202	.088	.088
	.326	-.019	.088	.348	.160	.178	.366	.107	.422	.231	.231	.088	.088
	.343	-.019	.275	.057	.277	.036	-.413	.302	.096	.296	.067	.088	.088
	.387	-.019	.119	.096	.173	.030	-.145	.122	.159	.215	.117	.088	.088
	.442	-.019	-.035	.043	.038	.168	-.051	.018	.044	.109	.195	.088	.088
	.493	-.019	.200	.096	.282	.331	.098	.099	.081	.176	.075	.088	.088
	.537	-.019	.101	-.071	-.037	-.315	.248	.122	.021	.065	.143	.088	.088

TABLE 33.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 2 - Concluded

(j) $M = 0.825$; outboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	0.000	.750	.719	.813	1.169	.592	.786	.950	.692	1.069	.505	.780
	.003	-.417	-.903	.694	-.669	-.807	-.891	.494	-.835	-.785	-.909	-.879
	.014	-.445	-.851	-.684	-.909	-.583	-.844	.605	-.639	-.625	-.779	-.844
	.031	-.244	-.335	-.1066	-.636	-.451	-.257	-.704	-.447	-.560	-.030	-.058
	.055	-.327	.961	-.772	-.493	-.367	-.116	-.447	-.445	-.408	.232	.292
	.106	-.253	-.319	-.332	-.315	-.249	-.305	-.318	-.309	-.264	-.232	-.276
	.172	-.120	-.474	-.160	-.180	-.112	.226	-.131	.191	-.107	.086	.350
	.261	-.226	.523	-.233	-.287	-.177	.331	.197	.212	-.136	.161	.387
	.302	-.147	-.097	-.021	-.096	-.115	-.066	.057	.037	-.037	.082	.039
	.326	-.104	-.005	.000	-.052	-.044	-.020	.026	.037	-.000	.051	.048
	.343	-.097	.222	.055	-.253	-.096	.210	.042	.241	.097	.078	.242
Turbine cowl	.387	-.269	-.029	.097	-.089	-.245	.029	.105	.013	.309	.220	.066
	.442	.119	-.031	-.081	-.201	.146	.124	.039	.092	.222	.166	.008
	.493	-.050	-.110	-.060	-.284	-.102	.038	.011	.083	.045	.087	.077
	.537	.052	.071	-.073	-.219	-.130	.038	-.087	.084	.034	.067	.081
	.581	-.116	-.136	-.152	-.232	-.427	.031	.150	.317	.207	.071	.209
	.617	-.120	-.018	-.162	-.354	-.555	.003	.037	.246	.251	.044	.047
	.653	-.140	-.073	-.039	-.469	-.563	-.076	.037	.176	.214	.016	.116
	.690	-.261	-.233	-.102	-.399	-.680	-.096	-.045	-.264	-.231	-.043	.044
	.728	-.222	-.202	.202	-.750	-.680	.263	.326	-.898	-.231	.313	.313
	.735	-.008	.008	.202	-.750	-.680	.126	.326	.078	.078	.166	.166
	.768	-.238	-.192	.192	-.750	-.680	-.286	.343	.078	.078	.276	.276
Plug	.796	-.584	-.584	-.584	-.584	-.584	-.436	.539	.181	.190	.218	.218
	.821	-.030	-.030	-.030	-.030	-.030	.179	.395	.395	.395	.395	.395
	.852	.230	.230	.230	.230	.230	.342	.395	.395	.395	.395	.395
	0.000	-.417	-.843	1.094	.730	.273	.875	1.214	.781	.900	.505	.780
	.003	-.941	-.893	.477	-.688	-.1079	-.921	.538	-.536	-.1285	-.909	-.879
	.014	-.573	-.757	-.387	-.576	-.185	-.722	.163	-.504	-.1079	-.779	-.844
	.031	-.184	-.245	-.502	-.484	-.479	-.107	.273	-.381	-.823	-.030	-.058
	.055	-.353	-.011	-.360	-.403	-.408	.243	.239	-.368	-.180	.232	.292
	.106	-.235	-.287	-.271	-.283	-.252	-.196	.265	-.207	.010	.086	.350
	.172	-.090	.159	-.095	-.165	-.043	.074	.234	.055	-.076	.086	.350
	.261	-.152	.230	-.166	-.158	-.061	.118	.231	.110	.010	.086	.350
	.302	-.082	.037	-.027	.025	.028	.039	.018	.255	.099	.086	.350
Fan cowl	0.000	-.417	-.843	1.094	.730	.273	.875	1.214	.781	.900	.505	.780
	.003	-.941	-.893	.477	-.688	-.1079	-.921	.538	-.536	-.1285	-.909	-.879
	.014	-.573	-.757	-.387	-.576	-.185	-.722	.163	-.504	-.1079	-.779	-.844
	.031	-.184	-.245	-.502	-.484	-.479	-.107	.273	-.381	-.823	-.030	-.058
	.055	-.353	-.011	-.360	-.403	-.408	.243	.239	-.368	-.180	.232	.292
	.106	-.235	-.287	-.271	-.283	-.252	-.196	.265	-.207	.010	.086	.350
	.172	-.090	.159	-.095	-.165	-.043	.074	.234	.055	-.076	.086	.350
	.261	-.152	.230	-.166	-.158	-.061	.118	.231	.110	.010	.086	.350
	.302	-.082	.037	-.027	.025	.028	.039	.018	.255	.099	.086	.350
	.326	-.043	.047	.052	.075	.075	.016	.061	.260	.133	.166	.166
	.343	-.094	.236	.060	.268	.074	.086	.061	.264	.029	.276	.276
	.387	-.211	.065	.104	.096	-.184	.092	.087	.170	.037	.218	.218
Turbine cowl	.442	.142	-.003	.062	.044	.274	.071	.045	.071	.246	.078	.078
	.493	.103	.099	.073	.016	.093	.276	.045	.071	.256	.078	.078
	.537	.075	-.082	.005	-.024	.211	.082	.042	.037	.220	.078	.078
	.581	.071	.191	.025	-.160	.108	.137	.068	.133	.099	.078	.078
	.617	.012	-.024	.125	-.150	.130	.162	.155	.070	.107	.078	.078
	.653	.079	.141	-.032	-.150	.150	.122	.040	.099	.016	.078	.078
	.690	-.069	.120	.362	.560	.207	.205	.145	.013	-.123	.078	.078
	.728	-.372	.372	.362	.362	.362	.365	.375	.517	.517	.517	.517
	.735	.154	.154	.154	.154	.154	.160	.160	.073	.073	.073	.073
	.768	-.268	-.268	-.268	-.268	-.268	-.311	.176	.311	.311	.311	.311
	.796	-.132	-.132	-.132	-.132	-.132	.037	.138	.259	.259	.259	.259
	.821	.178	.178	.178	.178	.178	.181	.259	.423	.423	.423	.423

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3

(a) $M = 0.700$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	.892	.432	.008	.656	1.014	.756	.405	.211	.892	.871	.647	.402
	-.145	-.668	-.419	-.519	-.006	-.509	-.667	-.751	-.320	-.291	-.899	-.675
	.003	-.218	-.694	-.703	-.208	-.412	-.485	-.552	-.295	-.333	-.501	-.475
	.031	-.208	-.920	-.902	-.111	-.296	-1.004	-.876	-.224	-.189	-.331	-.924
	.055	-.208	-.327	-.617	-.110	-.252	-.312	-.321	-.205	-.285	-.273	-.298
	.106	-.125	-.253	-.292	-.188	-.067	-.231	-.235	-.128	-.693	-.137	-.215
	.172	-.057	-.147	-.141	-.083	-.054	-.122	-.093	-.010	-.226	-.044	-.102
	.261	-.159	-.266	-.215	-.201	-.146	-.225	-.180	-.109	.718	-.122	-.202
	.302	-.120	-.080	-.102	-.083	-.019	-.045	-.068	-.007	.063	-.076	-.028
	.326	-.053	-.012	-.271	-.069	-.734	-.028	.012	-.080	.005	-.108	-.047
	.343	-.075	-.120	-.324	-.049	-.163	.201	-.235	.114	.000	-.132	-.207
Turbine cowl	.387	.321	.009	.025	.067	.314	.044	.035	-.010	-.071	.315	.081
	.442	.180	-.070	.028	.067	.204	-.045	.032	-.039	.098	.237	.004
	.493	-.237	.038	.057	.163	.086	.203	-.036	-.128	-.023	.117	.094
	.537	-.140	.083	.125	.342	.195	.012	-.036	-.173	-.327	-.132	.068
	.581	-.108	.115	.243	.310	.170	.012	-.036	-.096	-.448	-.078	.004
	.627	-.276	.171	.273	.339	.088	-.074	-.042	-.291	-.298	-.054	-.092
	.653	.161	.176	.131	.166	.184	-.058	-.116	-.205	-.269	-.069	-.033
	.690	-.137	.068	.332	-.448	.102	-.051	-.116	-.186	-.240	-.030	-.056
	.708	-.047	.026	.007	.007	-.102	-.001	.272	.073	.059	.059	.306
	.735	-.025	.226	.477	.007	.007	-.013	.272	-.080	.059	-.006	.119
	.758	-.607	.568	-.566	-.566	.000	-.494	.272	-.080	.059	-.006	.119
	.821	.170	.262	-.262	-.262	.000	-.167	.272	-.080	.059	-.006	.119
Plug	.852	.033	.322	.090	.324	.000	.054	.272	.073	.059	.059	.306
Fan cowl	.875	.410	.187	.956	.690	.756	.405	.211	.892	.871	.647	.402
	-.1097	-.664	-.419	-.519	-.006	-.509	-.667	-.751	-.320	-.291	-.899	-.675
	.003	-.218	-.694	-.703	-.208	-.412	-.485	-.552	-.295	-.333	-.501	-.475
	.031	-.208	-.920	-.902	-.111	-.296	-1.004	-.876	-.224	-.189	-.331	-.924
	.055	-.208	-.327	-.617	-.110	-.252	-.312	-.321	-.205	-.285	-.273	-.298
	.106	-.125	-.253	-.292	-.188	-.067	-.231	-.235	-.128	-.693	-.137	-.215
	.172	-.057	-.147	-.141	-.083	-.054	-.122	-.093	-.010	-.226	-.044	-.102
	.261	-.159	-.266	-.215	-.201	-.146	-.225	-.180	-.109	.718	-.122	-.202
	.302	-.120	-.080	-.102	-.083	-.019	-.045	-.068	-.007	.063	-.076	-.028
	.326	-.053	-.012	-.271	-.069	-.734	-.028	.012	-.080	.005	-.108	-.047
	.343	-.075	-.120	-.324	-.049	-.163	.201	-.235	.114	.000	-.132	-.207
	.387	.321	.009	.025	.067	.314	.044	.035	-.010	-.071	.315	.081
Turbine cowl	.442	.180	-.070	.028	.067	.204	-.045	.032	-.039	.098	.237	.004
	.493	-.237	.038	.057	.163	.086	.203	-.036	-.128	-.023	.117	.094
	.537	-.140	.083	.125	.342	.195	.012	-.036	-.173	-.327	-.132	.068
	.581	-.108	.115	.243	.310	.170	.012	-.036	-.096	-.448	-.078	.004
	.627	-.276	.171	.273	.339	.088	-.074	-.042	-.291	-.298	-.054	-.092
	.653	.161	.176	.131	.166	.184	-.058	-.116	-.205	-.269	-.069	-.033
	.690	-.137	.068	.332	-.448	.102	-.051	-.116	-.186	-.240	-.030	-.056
	.708	-.047	.026	.007	.007	-.102	-.001	.272	.073	.059	.059	.306
	.735	-.025	.226	.477	.007	.007	-.013	.272	-.080	.059	-.006	.119
	.758	-.607	.568	-.566	-.566	.000	-.494	.272	-.080	.059	-.006	.119
	.821	.170	.262	-.262	-.262	.000	-.167	.272	-.080	.059	-.006	.119
	.852	.033	.322	.090	.324	.000	.054	.272	.073	.059	.059	.306
Plug												

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(b) $M = 0.700$; outboard station

x/λ	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.858	1.153	.788	.602	.858	.596	.997	1.013	.658	1.044	.486	1.193	1.098	1.161			
	.003	-.207	-.559	.390	-.705	-.393	-.503	-.553	.245	-.521	-.621	-.776	-.566	.702	-.564			
	.014	-.299	-.501	-.637	-.573	-.447	-.474	-.873	.485	-.522	-.621	-.563	.985	.315	-.790			
	.031	-.275	.513	-.601	-.425	-.425	-.377	-.259	.459	-.438	-.747	-.427	.638	.359	-.606			
	.055	-.241	.940	-.465	-.425	-.296	.305	.323	.372	-.364	-.345	-.325	.187	.307	-.1068			
	.106	.025	-.278	-.333	-.460	-.222	.219	.282	.285	-.364	-.222	.028	.252	.239	-.516			
	.172	-.086	.004	-.171	-.199	-.089	.083	.124	.130	-.193	.071	-.078	.294	.206	-.283			
	.261	-.396	-.213	-.213	-.489	-.273	-.194	-.123	.181	-.200	-.097	-.180	.174	.155	.122			
	.302	-.130	-.100	-.278	-.251	-.173	.107	.068	.243	.171	.065	-.088	.051	.203	.001			
	.326	.180	-.028	-.032	-.235	-.177	.155	.051	.110	.064	.013	-.098	.020	.030	.011			
	.343	.154	.024	-.155	.030	-.070	.102	.454	.051	.142	.122	.469	.092	.090	.057			
	.387	.437	.027	-.012	-.160	-.102	.254	.034	.039	.232	.094	.009	.085	.040	.072			
Turbine cowl	.442	-.101	-.067	-.396	-.280	-.254	.053	.064	.013	.009	.022	.106	.025	.009	.040			
	.493	-.086	-.048	-.054	-.167	-.331	.053	.044	.020	.184	.293	.059	.072	.020	.148			
	.537	-.086	-.109	-.093	-.283	-.309	.029	.000	.075	.145	.322	.069	.025	.035	.273			
	.581	-.154	-.151	-.212	.386	-.718	.116	.087	.045	.200	.605	.010	.040	.037	.441			
	.617	-.280	-.057	-.083	.199	.618	.087	.045	.000	.200	.022	.006	.072	.030	.154			
	.653	-.236	-.129	-.077	.296	.428	.141	.033	.017	.097	.222	.006	.017	.063	-.357			
	.690	-.261	-.090	-.038	.228	.489	.155	.126	.022	.277	.348	.093	.367	.377	-.661			
	.708		.270	.305	.538			.355	.352	.631		.367	.377	.377	-.661			
	.735		.032	.241				.000		.026		.040		.315	-.031			
	.768		-.446	.596				.201		.467		.109		.169	-.067			
	.796		-.083	.115				.071		.126		.150		.384	.450			
	.821		.073	.078				.119		.131		.384						
.852		.347	.375				.374		.421									
Fan cowl	0.000	.424	1.129	1.177	.718	1.161												
	.003	-.979	-.559	.200	-.482	-.937												
	.014	-.606	-.941	-.241	-.434	-.666												
	.031	-.460	-.724	-.280	-.424	-.824												
	.055	-.339	-.432	-.264	-.356	-.421												
	.106	.020	-.251	-.266	-.214	-.195												
	.172	-.077	-.309	-.282	-.185	-.085												
	.261	-.194	-.206	-.184	-.166	-.024												
	.302	-.072	-.043	-.186	.028	.050												
	.326	-.082	-.060	.047	.003	.076												
	.343	-.160	.070	.090	.086	.002												
	.387	.456	.112	.077	.092	.254												
Turbine cowl	.442	.068	.005	.012	.050	.060												
	.493	.117	.096	.044	.073	.070												
	.537	.049	.093	.044	.043	.115												
	.581	.010	.051	.001	.179	.159												
	.617	.010	.060	.060	.053	.356												
	.653	.049	.073	.064	.105	.150												
	.690	-.034	.031	.096	.108	.337												
	.708		.352	.388	.682													
	.735		.051	.051	.014													
	.768		.089	.089	.253													
	.796		.005	.005	.011													
	.821		.171	.171	.192													
.852		.397	.397	.460														
Plug	0.000																	
	.003																	
	.014																	
	.031																	
	.055																	
	.106																	
	.172																	
	.261																	
	.302																	
	.326																	
	.343																	
	.387																	

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(c) $M = 0.750$; inboard station

x/c	C_p at -												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.894	.423	.227	.828	.992	.743	.397	-.035	.868	.861	.382	-.038
	.003	-.022	-.809	-1.470	-.512	-.080	-.687	-.804	-.927	-.361	-.318	-.795	-.522
	.014	-.022	-.549	-1.105	-.401	-.226	-.444	-.526	-.585	-.318	-.353	-.520	-.467
	.031	-.022	-1.081	-.759	-.322	-.127	-.307	-.807	-.705	-.236	-.189	-.352	-.748
	.055	-.022	-.356	-.429	-.276	.273	-.250	-.322	-.234	-.204	-.087	-.272	-.273
	.106	-.022	-.259	-.300	-.188	-.063	-.126	-.225	-.074	.119	-.076	-.130	-.206
	.172	-.022	-.145	-.139	-.074	-.751	-.038	-.105	-.079	.000	.376	-.033	-.062
	.261	-.022	-.271	-.221	-.206	1.159	-.126	-.213	-.172	-.096	.834	-.117	-.153
	.302	-.022	-.080	-.101	-.080	-.004	-.082	-.032	-.055	-.011	.082	-.073	-.039
	.326	-.022	-.004	.452	-.007	.868	-.007	.047	.152	-.076	.534	-.003	-.015
	.343	-.022	.124	-.262	.124	-.089	-.360	.185	-.240	.172	.534	-.347	-.214
	.387	-.022	.034	.004	-.039	-.182	.205	.062	.041	.091	-.052	.291	-.088
Turbine cowl	.442	-.022	.019	-.136	-.203	.031	.205	-.041	-.064	-.052	.053	.029	-.021
	.493	-.022	-.019	-.063	-.299	-.235	-.064	.053	.003	-.026	.024	.064	-.049
	.537	-.022	-.124	-.145	-.241	-.602	-.025	.024	.012	-.210	-.358	.007	-.073
	.581	-.022	-.145	-.264	-.439	-.416	.072	-.041	-.134	-.166	-.326	.065	-.017
	.617	-.022	-.279	-.089	-.384	-.713	-.144	.000	-.032	-.297	-.323	-.044	-.102
	.653	-.022	-.157	-.206	-.334	-.556	.196	-.088	-.029	-.125	-.350	.220	-.033
	.690	-.022	-.233	.259	-.267	.101	-.135	-.026	.012	-.184	-.093	.096	-.009
	.708	-.022	-.136	.192	-.007	.038	.038	.038	.281	-.038	.038	.023	.287
	.735	-.022	-.022	.432	-.142	.035	.006	-.006	.006	-.035	.035	-.003	.006
	.768	-.022	.546	.433	-.442	.035	-.444	-.006	.006	-.035	.035	-.003	.006
	.794	-.022	-.838	-.766	-.766	.080	-.316	-.444	.006	-.035	.035	-.003	.006
	.821	-.022	.080	.080	-.080	.123	.123	.123	.123	.123	.123	.123	.123
Plug	.852	-.022	.320	.320	.346	.351	.351	.351	.351	.351	.351	.351	.351
	0.000	.581	.398	-.033	.320	.692	.692	.692	.692	.692	.692	.692	.692
	.003	-1.202	-.866	-.258	-.281	-.028	-.028	-.028	-.028	-.028	-.028	-.028	-.028
	.014	-.549	-.536	-.552	-.287	-.520	-.520	-.520	-.520	-.520	-.520	-.520	-.520
	.031	-.382	-.661	-.813	-.139	-.289	-.289	-.289	-.289	-.289	-.289	-.289	-.289
	.055	-.289	-.322	-.238	-.069	-.031	-.031	-.031	-.031	-.031	-.031	-.031	-.031
	.106	-.135	-.217	-.182	-.010	-.086	-.086	-.086	-.086	-.086	-.086	-.086	-.086
	.172	-.038	-.095	-.045	.010	.035	.035	.035	.035	.035	.035	.035	.035
	.261	-.069	-.088	-.138	.035	.317	.317	.317	.317	.317	.317	.317	.317
	.302	-.055	-.019	-.070	.035	.139	.139	.139	.139	.139	.139	.139	.139
	.326	-.051	.051	-.078	.130	.331	.331	.331	.331	.331	.331	.331	.331
	.343	-.355	.237	-.078	.200	-.030	-.030	-.030	-.030	-.030	-.030	-.030	-.030
Fan cowl	.387	.372	.107	.084	.095	.069	.069	.069	.069	.069	.069	.069	.069
	.442	.431	.069	-.013	.034	.243	.243	.243	.243	.243	.243	.243	.243
	.493	-.079	.078	.343	-.016	.133	.133	.133	.133	.133	.133	.133	.133
	.537	-.073	.078	.037	-.051	-.289	-.289	-.289	-.289	-.289	-.289	-.289	-.289
	.581	.011	.016	-.071	.016	-.106	-.106	-.106	-.106	-.106	-.106	-.106	-.106
	.617	-.011	.116	.011	-.179	-.153	-.153	-.153	-.153	-.153	-.153	-.153	-.153
	.653	.209	.034	.040	-.132	-.156	-.156	-.156	-.156	-.156	-.156	-.156	-.156
	.690	.006	.145	-.004	-.115	-.158	-.158	-.158	-.158	-.158	-.158	-.158	-.158
	.708	.004	.072	.300	.031	.031	.031	.031	.031	.031	.031	.031	.031
	.735	-.004	-.004	.004	-.365	-.365	-.365	-.365	-.365	-.365	-.365	-.365	-.365
	.768	-.080	-.369	-.080	-.039	-.039	-.039	-.039	-.039	-.039	-.039	-.039	-.039
	.821	.107	.107	.107	.156	.156	.156	.156	.156	.156	.156	.156	.156
	.852	.370	.370	.370	.406	.406	.406	.406	.406	.406	.406	.406	.406

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued
(d) M = 0.750; outboard station

x/2	C _p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	.817	1.200	.782	.564	.888	.578	1.099	.980	.634	1.092	.471
	.003	-.242	-.671	.524	-.844	.374	-.677	-.722	.354	.658	-.672	-.943
	.014	-.357	-.031	-.063	-.644	-.445	-.527	-.631	-.524	.569	-.560	-.581
	.031	-.317	.721	-.638	-.480	-.363	-.408	.021	-.519	.399	-.725	-.466
	.055	-.278	1.046	-.500	-.462	-.321	-.316	.537	.395	.387	-.355	-.342
	.106	-.018	-.293	-.355	-.424	-.230	.024	-.265	.289	.334	-.232	.029
	.172	-.092	.182	-.178	-.201	-.145	-.082	-.047	.126	.167	-.067	-.077
	.261	-.300	-.119	-.231	-.374	-.183	-.086	.018	.182	.191	-.100	-.059
	.302	-.132	-.105	-.172	-.180	-.119	-.091	-.056	.159	.102	-.020	-.081
	.326	.375	-.025	-.031	-.178	-.087	.073	.018	.018	.070	.009	.026
	.343	-.075	.105	-.090	.125	-.034	-.073	.112	.070	.147	.006	.056
	.387	.172	.031	-.031	-.160	-.404	.288	-.014	.039	.642	.015	.071
	.442	-.189	-.158	-.122	-.007	-.004	-.122	-.047	.079	.205	.012	-.003
	.493	.071	.010	-.137	.401	-.454	.041	.018	.009	.009	.167	.038
	.537	-.145	-.152	-.161	-.178	-.410	.019	.012	.009	.191	.059	.053
Turbine cowl	.581	-.181	-.128	-.195	-.327	-.471	-.100	-.070	.108	.208	-.540	.012
	.617	-.313	-.243	.002	-.392	-.823	-.126	-.047	.015	.108	-.487	-.037
	.653	-.225	-.084	-.137	-.333	-.459	-.108	-.059	.030	.229	-.375	.106
	.690	-.317	-.255	-.075	-.045	-.345	-.181	-.079	.030	.223	-.358	.028
	.708	.241	.291	.313	-.597	-.345	.181	.013	.372	.675	.331	.074
	.735	.046	.046	.116	-.116	.046	.046	.039	.641	.641	.377	.369
	.768	-.467	-.467	-.498	-.498	.046	.046	-.330	.641	.641	.377	.369
	.796	-.172	-.172	-.451	-.451	.046	.046	-.006	.641	.641	.377	.369
	.821	.137	.137	.175	.175	.046	.046	.136	.641	.641	.377	.369
	.852	.335	.335	.363	.363	.046	.046	.318	.641	.641	.377	.369
Plug	0.000	-.403	1.185	1.156	.699	1.185	.578	1.099	.980	.634	1.092	.471
	.003	-.634	-.763	.317	-.561	1.012	-.677	-.722	.354	.658	-.672	-.943
	.014	-.495	-.522	-.339	-.391	-.654	-.527	-.631	-.524	.569	-.560	-.581
	.031	-.359	-.239	-.298	-.405	-.394	-.408	.021	-.519	.399	-.725	-.466
	.055	-.024	-.269	-.233	-.356	-.271	-.316	.537	.395	.387	-.355	-.342
	.106	-.082	-.169	-.089	-.189	-.024	.024	-.265	.289	.334	-.232	.029
	.172	-.064	-.042	-.145	-.180	-.016	-.082	-.047	.126	.167	-.067	-.077
	.261	-.077	-.039	-.086	.028	.083	-.086	.018	.182	.191	-.100	-.059
	.302	-.042	.011	-.043	.058	.087	-.091	.056	.159	.102	-.020	-.081
	.326	.402	.126	-.057	.151	.034	.073	.018	.070	.147	.006	.056
	.343	.402	.043	.079	.017	.219	.046	.112	.070	.147	.006	.056
	.387	.037	.064	.055	.058	.151	.046	.112	.070	.147	.006	.056
	.442	.037	.064	.055	.058	.151	.046	.112	.070	.147	.006	.056
	.493	.081	.061	.034	-.109	-.030	.046	.112	.070	.147	.006	.056
	.537	.033	.020	-.030	-.068	-.200	.046	.112	.070	.147	.006	.056
	.581	.020	.129	.031	-.130	-.356	.046	.112	.070	.147	.006	.056
	.617	-.011	.008	.055	.060	-.156	.046	.112	.070	.147	.006	.056
	.653	-.046	.076	.067	.212	-.256	.046	.112	.070	.147	.006	.056
	.690	.364	.364	.382	.643	.382	.046	.112	.070	.147	.006	.056
	.708	.084	.084	.084	.005	.005	.046	.112	.070	.147	.006	.056
	.735	.113	.113	.113	-.452	-.452	.046	.112	.070	.147	.006	.056
	.768	.004	.004	.004	.049	.049	.046	.112	.070	.147	.006	.056
	.796	.164	.164	.175	.175	.175	.046	.112	.070	.147	.006	.056
	.821	.391	.391	.436	.436	.436	.046	.112	.070	.147	.006	.056
	.852	.391	.391	.436	.436	.436	.046	.112	.070	.147	.006	.056

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(e) $M = 0.775$; Inboard station

x/c	C _p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																		
	α = -2°						α = 0°						α = 4°																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																					
Fan cowl	0.000	.990	.407	.426	.827	.990	.751	.405	.378	.877	.845	.400	.086	.896	.770	.003	-.249	-.890	1.354	-.536	-.099	-.758	-.858	-.591	-.400	-.591	-.388	-.598	.014	-.309	-.537	1.185	-.412	-.236	-.465	-.544	-.602	-.326	-.362	-.526	-.326	-.555	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.421	-.304	-.515	-.42

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(f) $M = 0.775$; outboard station

x/c	C_p at -															
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$							
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.808	1.257	.815	.605	.916	.584	1.132	.976	.638	1.051	.463	1.328	1.058	.672	1.208
	.003	-.287	-.826	.624	-.901	-.398	-.731	-.479	-.422	-.726	-.720	-.973	-.822	.370	-.657	-.957
	.014	-.384	.245	-.1129	-.676	-.471	-.569	-.434	-.567	-.587	-.565	-.639	-.552	-.448	-.565	-.669
	.031	-.342	.881	-.558	-.507	-.386	-.434	.201	-.562	-.427	-.740	-.455	-.273	-.463	-.401	-.861
	.055	-.291	1.116	-.493	-.471	-.364	-.332	.668	-.414	-.358	-.362	-.355	.146	-.379	-.384	-.379
	.106	.018	-.300	-.363	-.347	-.279	.023	-.273	-.301	-.300	-.232	.021	-.273	-.379	-.383	-.170
	.172	-.092	.327	-.176	-.195	-.085	-.082	.068	-.131	-.151	-.071	-.076	.058	-.114	-.135	-.569
	.261	-.283	.114	.235	-.302	-.187	-.023	.050	-.190	-.195	-.105	.013	.055	-.174	-.170	-.052
	.302	-.130	-.102	-.114	-.192	-.105	-.095	-.057	-.103	-.108	-.017	-.080	-.046	-.029	-.001	-.027
	.326	.554	.020	-.026	-.170	-.068	.168	.017	.017	.105	.014	.053	.030	.033	.035	.038
	.343	-.033	.141	.039	.142	.015	-.036	.153	.023	.193	.011	.058	.132	.029	.188	.035
	.387	.013	.052	.052	.125	-.466	-.142	.068	.054	.099	.090	.208	.050	.053	.103	.072
Turbine cowl	.442	-.115	-.182	-.052	.086	.047	-.184	-.100	-.057	.055	.090	-.144	.035	.001	.055	.010
	.493	.115	.107	-.029	-.386	-.443	.163	.062	.019	-.119	.175	.114	.033	.002	.258	.060
	.537	-.177	.213	.136	.253	-.336	.053	.023	.046	.119	.415	.004	.058	.077	.184	.438
	.581	-.194	.053	.144	-.434	-.366	.184	.003	.073	.283	.593	.059	.123	.067	.156	.438
	.617	-.261	-.272	.117	-.358	-.476	.108	.046	.002	.029	.277	.012	.046	.022	.198	.415
	.653	-.384	-.131	-.114	-.330	-.676	-.108	-.137	.053	.139	-.288	-.063	.038	.055	.170	.320
	.690	-.223	-.227	-.088	-.389	-.448	-.248	.371	.331	.805	.350	.084	.350	.387	.833	.350
	.709	.214	.214	.302	-.642	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.735	.039	.039	.039	.039	.039	.039	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.768	-.385	-.385	-.431	-.665	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.796	-.611	-.611	-.665	.079	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.821	.155	.155	.148	.148	.148	.148	.164	.164	.286	.160	.160	.160	.239	.238	.430
	.852	.336	.336	.390	.390	.390	.390	.363	.363	.411	.378	.378	.378	.430	.430	.430
Fan cowl	0.000	.414	1.215	1.146	.692	1.166	.414	1.132	.976	.638	1.051	.463	1.328	1.058	.672	1.208
	.003	-.062	-.798	.370	-.577	-.1074	-.888	-.636	-.319	-.504	-.783	-.973	-.822	.370	-.657	-.957
	.014	-.888	.245	-.1129	-.676	-.471	-.569	-.434	-.567	-.587	-.565	-.639	-.552	-.448	-.565	-.669
	.031	-.494	.881	-.558	-.507	-.386	-.434	.201	-.562	-.427	-.740	-.455	-.273	-.463	-.401	-.861
	.055	-.350	1.126	-.493	-.471	-.364	-.332	.668	-.414	-.358	-.362	-.355	.146	-.379	-.384	-.379
	.106	.035	-.257	-.363	-.347	-.279	.023	-.273	-.301	-.300	-.232	.021	-.273	-.379	-.383	-.170
	.172	-.062	.327	-.176	-.195	-.085	-.082	.068	-.131	-.151	-.071	-.076	.058	-.114	-.135	-.569
	.261	-.283	.114	.235	-.302	-.187	-.023	.050	-.190	-.195	-.105	.013	.055	-.174	-.170	-.052
	.302	-.130	-.102	-.114	-.192	-.105	-.095	-.057	-.103	-.108	-.017	-.080	-.046	-.029	-.001	-.027
	.326	.554	.020	-.026	-.170	-.068	.168	.017	.017	.105	.014	.053	.030	.033	.035	.038
	.343	-.033	.141	.039	.142	.015	-.036	.153	.023	.193	.011	.058	.132	.029	.188	.035
	.387	.013	.052	.052	.125	-.466	-.142	.068	.054	.099	.090	.208	.050	.053	.103	.072
Turbine cowl	.442	-.115	-.182	-.052	.086	.047	-.184	-.100	-.057	.055	.090	-.144	.035	.001	.055	.010
	.493	.115	.107	-.029	-.386	-.443	.163	.062	.019	-.119	.175	.114	.033	.002	.258	.060
	.537	-.177	.213	.136	.253	-.336	.053	.023	.046	.119	.415	.004	.058	.077	.184	.438
	.581	-.194	.053	.144	-.434	-.366	.184	.003	.073	.283	.593	.059	.123	.067	.156	.438
	.617	-.261	-.272	.117	-.358	-.476	.108	.046	.002	.029	.277	.012	.046	.022	.198	.415
	.653	-.384	-.131	-.114	-.330	-.676	-.108	-.137	.053	.139	-.288	-.063	.038	.055	.170	.320
	.690	-.223	-.227	-.088	-.389	-.448	-.248	.371	.331	.805	.350	.084	.350	.387	.833	.350
	.709	.214	.214	.302	-.642	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.735	.039	.039	.039	.039	.039	.039	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.768	-.385	-.385	-.431	-.665	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.796	-.611	-.611	-.665	.079	.079	.079	.349	.079	.805	.350	.084	.350	.387	.833	.350
	.821	.155	.155	.148	.148	.148	.148	.164	.164	.286	.160	.160	.160	.239	.238	.430
	.852	.336	.336	.390	.390	.390	.390	.363	.363	.411	.378	.378	.378	.430	.430	.430
Plug	0.000	.414	1.215	1.146	.692	1.166	.414	1.132	.976	.638	1.051	.463	1.328	1.058	.672	1.208
	.003	-.062	-.798	.370	-.577	-.1074	-.888	-.636	-.319	-.504	-.783	-.973	-.822	.370	-.657	-.957
	.014	-.888	.245	-.1129	-.676	-.471	-.569	-.434	-.567	-.587	-.565	-.639	-.552	-.448	-.565	-.669
	.031	-.494	.881	-.558	-.507	-.386	-.434	.201	-.562	-.427	-.740	-.455	-.273	-.463	-.401	-.861
	.055	-.350	1.126	-.493	-.471	-.364	-.332	.668	-.414	-.358	-.362	-.355	.146	-.379	-.384	-.379
	.106	.035	-.257	-.363	-.347	-.279	.023	-.273	-.301	-.300	-.232	.021	-.273	-.379	-.383	-.170
	.172	-.062	.327	-.176	-.195	-.085	-.082	.068	-.131	-.151	-.071	-.076	.058	-.114	-.135	-.569
	.261	-.283	.114	.235	-.302	-.187	-.023	.050	-.190	-.195	-.105	.013	.055	-.174	-.170	-.052
	.302	-.130	-.102	-.114	-.192	-.105	-.095	-.057	-.103	-.108	-.017	-.080	-.046	-.029	-.001	-.027
	.326	.554	.020	-.026	-.170	-.068	.168	.017	.017	.105	.014	.053	.030	.033	.035	.038
	.343	-.033	.141	.039	.142	.015	-.036	.153	.023	.193	.011	.058	.132	.029	.188	.035
	.387	.013	.052	.052	.125	-.466	-.142	.068	.054	.099	.090	.208	.050	.053	.103	.072
	.442	-.115	-.182	-.052	.086	.047	-.184	-.100	-.057	.055	.090	-.144	.035	.001	.055	.010

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(g) $M = 0.800$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row L
Fan cowl	0.000	.893	.422	.706	.839	.754	.410	.222	.875	.838	.687	.771
	.003	-.286	-.931	1.313	-.548	-.778	-.928	-1.015	-.462	-.015	-.917	-.644
	.014	-.347	-.599	-1.179	-.416	-.476	-.572	-.566	-.360	-.015	-.565	-.439
	.031	-.290	-.947	-.258	-.328	-.329	-.577	-.434	-.271	-.015	-.486	-.232
	.055	-.265	-.366	-.380	-.282	-.260	-.337	-.345	-.228	-.015	-.265	-.215
	.106	-.168	-.256	-.291	-.185	-.048	-.223	-.239	-.133	-.015	-.208	-.019
	.172	-.086	-.129	-.123	-.094	1.039	-.093	-.074	-.010	-.015	-.118	-.056
	.261	-.196	-.264	-.213	-.188	1.276	-.212	-.172	-.109	-.015	-.024	-.373
	.302	-.151	-.067	-.086	-.067	.036	-.023	-.047	-.001	-.015	-.081	-.046
	.326	.432	.017	.822	.006	1.131	-.025	.374	.074	-.007	-.002	-.048
	.343	-.331	.187	-.164	.197	-.018	.228	.150	.217	-.007	.079	-.146
	.367	-.477	.076	-.048	-.053	-.129	.104	.039	.069	-.015	.247	.146
Turbine cowl	.442	-.106	-.018	-.075	-.080	.127	-.055	-.026	.082	-.045	.110	.030
	.493	-.111	.028	-.145	-.317	.189	-.055	.055	-.088	-.034	.174	.178
	.537	.326	-.161	-.191	-.575	.165	-.058	.055	-.160	.189	.079	.081
	.581	.517	-.207	-.190	-.427	.250	.017	-.053	-.384	.158	.019	-.143
	.617	-.233	-.113	-.269	-.511	-.117	-.157	-.039	-.271	-.513	.022	-.021
Plug	.653	.122	-.154	-.180	-.521	-.710	-.042	-.107	-.232	-.535	-.055	-.294
	.690	-.543	-.264	.646	-.500	.369	-.069	.223	-.317	.079	-.062	-.070
	.708	-.040	.136	.049	.049	-.113	-.037	.255	.052	.163	.260	.253
	.735	-.032	.342	-.481	-.481	.387	-.004	.387	.387	.011	.092	.060
	.768	-.342	.385	-.462	-.462	.385	-.385	.385	-.298	.011	.011	-.296
Fan cowl	0.000	-.620	.403	.167	.911	.711	.410	.222	.875	.838	.687	.771
	.003	-1.224	-.940	-.429	-.319	-.872	-.928	-1.015	-.462	-.015	-.917	-.644
	.014	-.791	-.555	-.380	-.268	-.492	-.572	-.566	-.360	-.015	-.565	-.439
	.051	-.272	-.301	-.237	-.144	-.096	-.337	-.345	-.228	-.015	-.265	-.215
	.106	-.109	-.196	-.174	-.088	-.015	-.223	-.239	-.133	-.015	-.208	-.019
Turbine cowl	.172	-.080	-.166	-.128	-.074	.255	-.093	-.074	-.010	-.015	-.024	-.373
	.261	-.080	-.166	-.128	-.074	.255	-.093	-.074	-.010	-.015	-.024	-.373
	.302	-.031	.015	-.009	.112	.190	-.023	-.047	-.001	-.015	-.081	-.046
	.326	.230	.093	.245	.158	.446	.061	.166	.166	-.007	.079	-.146
	.343	-.272	.269	-.139	.263	.031	.198	.088	.166	-.015	.247	.146
Plug	.367	-.027	.131	.120	.198	.088	.166	.166	.166	-.015	.247	.146
	.442	.218	-.004	.004	.061	.166	.166	.166	.166	-.015	.247	.146
	.493	-.027	.131	.120	.198	.088	.166	.166	.166	-.015	.247	.146
	.537	.161	.099	-.001	-.104	-.360	.061	.166	.166	-.015	.247	.146
	.581	.218	.061	-.096	-.069	.152	.061	.166	.166	-.015	.247	.146

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Continued

(h) $M = 0.800$; outboard station

x/c	C_p at -														
	$\alpha = -2^\circ$					$\alpha = 0^\circ$					$\alpha = 1^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.780	1.285	.826	.749	.949	.578	1.170	.975	.505	1.047	.499	1.157	1.060	1.118
	.003	-.366	-.862	.749	-.869	-.414	-.788	-.877	.505	.769	-.728	-.937	-.864	.445	-.945
	.014	-.395	-.597	-1.082	-.679	-.487	-.577	-.265	-.584	.611	-.601	-.693	-.485	-.450	-.673
	.031	-.375	1.016	-1.090	-.606	-.406	-.487	.364	-.660	.640	-.923	-.489	-.114	-.488	-.874
	.055	-.302	1.179	-.383	-.487	-.346	-.341	.777	-.437	.416	-.378	-.350	.301	-.379	-.377
	.106	.020	-.309	-.364	-.327	-.194	.022	-.279	-.312	-.346	-.256	-.029	-.269	-.380	-.268
	.172	-.094	.504	-.176	-.205	-.127	-.080	.189	-.132	-.186	-.069	-.069	.074	-.103	-.168
	.261	-.257	-.135	-.241	-.308	-.173	-.058	.146	-.195	-.205	-.102	-.131	.140	-.152	-.032
	.302	-.127	-.100	-.034	-.110	-.110	-.092	-.056	.018	-.061	-.012	-.065	-.107	-.169	-.040
	.326	.781	-.018	-.024	-.238	-.059	.290	.026	.020	.004	.026	.217	.047	.006	.036
	.343	-.005	.186	.004	.207	-.002	.003	.067	.059	.024	.031	.008	.200	.014	.074
	.387	-.082	-.010	.072	-.178	-.552	-.003	.039	.015	.045	-.278	.111	.696	.047	-.032
	.442	-.009	.116	-.075	.338	.125	.109	.206	.012	.126	-.150	.237	.194	.099	.147
	.493	.040	.004	-.149	-.373	-.327	.064	.132	.113	.061	.191	.004	.040	.029	-.084
	Turbine cowl	.537	-.127	-.176	.168	-.373	-.544	-.048	.097	.066	.376	.454	.013	.009	-.086
.581		-.281	.200	.167	.289	.725	-.048	.137	.020	.340	.576	.150	.035	.121	-.233
.617		-.297	-.078	.160	.351	.571	.157	.004	.059	.213	.424	.036	.050	.001	-.594
.653		-.330	-.219	.135	.332	.332	.117	.008	.058	.015	-.308	-.163	.130	.003	-.173
.690		-.493	-.296	-.111	.535	-.614	-.174	.257	.328	-.864			.385	.074	-.249
Plug	.708		.443	.246	.150			.140		-.066			.123	.350	-.1091
	.735		.129		.102			.334		-.473			.305		.120
	.766		-.353		-.324			-.344		.313			.087		-.377
	.821		.044		.031			.211		.224			.189		-.361
	.852		.317		.350			.372		.419			.377		.267
Fan cowl	0.000	.473	1.231	1.131	.688	1.157									
	.003	-1.013	-.884	.435	-.634	-1.106									
	.014	-.989	-.495	-.372	-.536	-.826									
	.031	-.455	-.306	-.388	-.384	-.392									
	.055	-.349	-.023	-.328	-.357	-.354									
	.106	.034	-.260	-.246	-.224	-.159									
	.172	-.060	.023	-.086	-.074	-.004									
	.261	.128	.097	-.148	.112	-.007									
	.302	-.051	-.020	.081	.037	.077									
	.326	.213	.015	.059	.140	.110									
	.343	-.003	.198	.023	.238	.061									
	.387	.136	.119	.059	.137	.137									
	.442	-.133	-.053	.010	-.026	.037									
	.493	.246	.124	.124	.086	.067									
	Turbine cowl	.537	.026	.051	.042	.140	-.007								
.581		.038	.015	-.050	-.129	.354									
.617		-.043	.127	.141	-.099	-.427									
.653		.014	.018	.018	.085	-.175									
.690		-.096	-.012	.092	-.251	-.240									
Plug	.708		.380	.386	-1.098										
	.735		.111		.129										
	.766		-.233		-.389										
	.796		-.028		-.262										
	.821		.195		.273										
.852		.366		.436											

TABLE 34.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 3 - Concluded

(J) M = 0.825; outboard station

x/c	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																										
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 4^\circ$																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4

(a) $M = 0.700$; Inboard station

x/\bar{c}	C_p at												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.300	.577	.445	.697	.894	.995	.726	.389	.870	.913	.677	.382	.942
	.003	-.138	-.659	-.1352	.051	-.022	-.619	-.702	-.647	.278	-.046	-.714	-.328
	.014	-.251	-.426	-.605	-.154	-.205	-.464	-.541	-.532	.159	-.056	-.976	-.386
	.031	-.270	-.290	-.776	-.256	-.128	-.328	.211	-.837	.079	-.065	-.154	-.897
	.055	-.227	-.306	-.793	-.244	-.157	-.279	-.303	-.303	.095	-.078	-.299	-.258
	.106	-.129	-.235	-.280	-.157	-.071	-.148	-.226	-.233	.137	-.088	-.151	-.222
	.172	-.037	-.087	-.125	-.054	-.019	-.042	-.082	-.088	.258	-.107	-.035	-.068
	.251	-.105	-.235	-.177	-.183	-.019	-.090	-.207	-.046	.124	-.122	-.193	-.065
	.302	-.100	-.064	-.084	-.099	.025	-.085	-.037	-.059	.258	-.152	-.071	-.052
	.326	-.023	.003	.335	-.000	-.051	-.071	.034	.044	.297	-.158	-.054	-.042
	.343	-.421	.145	.266	-.074	-.087	-.376	.208	-.233	.303	-.287	-.376	-.004
	.387	.135	-.016	-.013	-.074	-.244	.211	.063	.037	.306	.204	.083	-.145
	.442	.153	-.113	-.116	-.209	-.019	.177	-.027	-.043	.332	.232	.213	-.060
	.493	-.212	.013	-.055	-.144	.119	-.153	.070	-.024	.236	.332	.184	-.031
	.537	-.071	-.071	-.125	-.346	.516	-.177	.050	-.030	.303	.303	.185	-.102
Turbine cowl	.581	-.245	-.122	-.321	-.311	-.590	-.177	.050	-.037	.198	.005	.038	-.004
	.617	-.232	-.119	-.164	-.359	-.478	-.105	.068	-.037	.037	-.020	-.069	-.170
	.653	-.079	-.187	-.125	-.215	-.510	-.124	-.043	-.033	.050	-.052	.038	-.007
	.690	-.087	-.100	.087	-.308	-.071	-.114	-.011	-.038	.058	-.002	.044	.006
	.708	-.275	-.035	.248	-.000	-.071	-.114	-.014	.295	.307	.057	.057	-.100
	.735	-.019	-.019	-.093	-.033	-.071	-.114	-.014	.295	.307	.057	.057	-.100
	.758	-.002	-.012	-.327	-.561	-.071	-.114	-.014	.295	.307	.057	.057	-.100
	.821	.015	-.012	-.047	-.184	-.071	-.114	-.014	.295	.307	.057	.057	-.100
	.852	.019	.019	.330	.432	-.071	-.114	-.014	.295	.307	.057	.057	-.100
	.852	.019	.019	.330	.432	-.071	-.114	-.014	.295	.307	.057	.057	-.100
Plug	0.300	.577	.445	.697	.894	.995	.726	.389	.870	.913	.677	.382	.942
	.003	-.138	-.659	-.1352	.051	-.022	-.619	-.702	-.647	.278	-.046	-.714	-.328
	.014	-.251	-.426	-.605	-.154	-.205	-.464	-.541	-.532	.159	-.056	-.976	-.386
	.031	-.270	-.290	-.776	-.256	-.128	-.328	.211	-.837	.079	-.065	-.154	-.897
	.055	-.227	-.306	-.793	-.244	-.157	-.279	-.303	-.303	.095	-.078	-.299	-.258
	.106	-.129	-.235	-.280	-.157	-.071	-.148	-.226	-.233	.137	-.088	-.151	-.222
	.172	-.037	-.087	-.125	-.054	-.019	-.042	-.082	-.088	.258	-.107	-.035	-.068
	.251	-.105	-.235	-.177	-.183	-.019	-.090	-.207	-.046	.124	-.122	-.193	-.065
	.302	-.100	-.064	-.084	-.099	.025	-.085	-.037	-.059	.258	-.152	-.071	-.052
	.326	-.023	.003	.335	-.000	-.051	-.071	.034	.044	.297	-.158	-.054	-.042
	.343	-.421	.145	.266	-.074	-.087	-.376	.208	-.233	.303	-.287	-.376	-.004
	.387	.135	-.016	-.013	-.074	-.244	.211	.063	.037	.306	.204	.083	-.145
	.442	.153	-.113	-.116	-.209	-.019	.177	-.027	-.043	.332	.232	.213	-.060
	.493	-.212	.013	-.055	-.144	.119	-.153	.070	-.024	.236	.332	.184	-.031
	.537	-.071	-.071	-.125	-.346	.516	-.177	.050	-.037	.198	.005	.038	-.004
Fan cowl	0.300	.577	.445	.697	.894	.995	.726	.389	.870	.913	.677	.382	.942
	.003	-.138	-.659	-.1352	.051	-.022	-.619	-.702	-.647	.278	-.046	-.714	-.328
	.014	-.251	-.426	-.605	-.154	-.205	-.464	-.541	-.532	.159	-.056	-.976	-.386
	.031	-.270	-.290	-.776	-.256	-.128	-.328	.211	-.837	.079	-.065	-.154	-.897
	.055	-.227	-.306	-.793	-.244	-.157	-.279	-.303	-.303	.095	-.078	-.299	-.258
	.106	-.129	-.235	-.280	-.157	-.071	-.148	-.226	-.233	.137	-.088	-.151	-.222
	.172	-.037	-.087	-.125	-.054	-.019	-.042	-.082	-.088	.258	-.107	-.035	-.068
	.251	-.105	-.235	-.177	-.183	-.019	-.090	-.207	-.046	.124	-.122	-.193	-.065
	.302	-.100	-.064	-.084	-.099	.025	-.085	-.037	-.059	.258	-.152	-.071	-.052
	.326	-.023	.003	.335	-.000	-.051	-.071	.034	.044	.297	-.158	-.054	-.042
	.343	-.421	.145	.266	-.074	-.087	-.376	.208	-.233	.303	-.287	-.376	-.004
	.387	.135	-.016	-.013	-.074	-.244	.211	.063	.037	.306	.204	.083	-.145
	.442	.153	-.113	-.116	-.209	-.019	.177	-.027	-.043	.332	.232	.213	-.060
	.493	-.212	.013	-.055	-.144	.119	-.153	.070	-.024	.236	.332	.184	-.031
	.537	-.071	-.071	-.125	-.346	.516	-.177	.050	-.037	.198	.005	.038	-.004
Turbine cowl	0.300	.577	.445	.697	.894	.995	.726	.389	.870	.913	.677	.382	.942
	.003	-.138	-.659	-.1352	.051	-.022	-.619	-.702	-.647	.278	-.046	-.714	-.328
	.014	-.251	-.426	-.605	-.154	-.205	-.464	-.541	-.532	.159	-.056	-.976	-.386
	.031	-.270	-.290	-.776	-.256	-.128	-.328	.211	-.837	.079	-.065	-.154	-.897
	.055	-.227	-.306	-.793	-.244	-.157	-.279	-.303	-.303	.095	-.078	-.299	-.258
	.106	-.129	-.235	-.280	-.157	-.071	-.148	-.226	-.233	.137	-.088	-.151	-.222
	.172	-.037	-.087	-.125	-.054	-.019	-.042	-.082	-.088	.258	-.107	-.035	-.068
	.251	-.105	-.235	-.177	-.183	-.019	-.090	-.207	-.046	.124	-.122	-.193	-.065
	.302	-.100	-.064	-.084	-.099	.025	-.085	-.037	-.059	.258	-.152	-.071	-.052
	.326	-.023	.003	.335	-.000	-.051	-.071	.034	.044	.297	-.158	-.054	-.042
	.343	-.421	.145	.266	-.074	-.087	-.376	.208	-.233	.303	-.287	-.376	-.004
	.387	.135	-.016	-.013	-.074	-.244	.211	.063	.037	.306	.204	.083	-.145
	.442	.153	-.113	-.116	-.209	-.019	.177	-.027	-.043	.332	.232	.213	-.060
	.493	-.212	.013	-.055	-.144	.119	-.153	.070	-.024	.236	.332	.184	-.031
	.537	-.071	-.071	-.125	-.346	.516	-.177	.050	-.037	.198	.005	.038	-.004
Plug	0.300	.577	.445	.697	.894	.995	.726	.389	.870	.913	.677	.382	.942
	.003	-.138	-.659	-.1352	.051	-.022	-.619	-.702	-.647	.278	-.046	-.714	-.328
	.014	-.251	-.426	-.605	-.154	-.205	-.464	-.541	-.532	.159	-.056	-.976	-.386
	.031	-.270	-.290	-.776	-.256	-.128	-.328	.211	-.837	.079	-.065	-.154	-.897
	.055	-.227	-.306	-.793	-.244	-.157	-.279	-.303	-.303	.095	-.078	-.299	-.258
	.106	-.129	-.235	-.280	-.157	-.071	-.148	-.226	-.233	.137	-.088	-.151	-.222
	.172	-.037	-.087	-.125	-.054	-.019	-.042	-.082	-.088	.258	-.107	-.035	-.068
	.251	-.105	-.235	-.177	-.183	-.019	-.090	-.207	-.046	.124	-.122	-.193	-.065
	.302	-.100	-.064	-.084	-.099	.025	-.085	-.037	-.059	.258	-.152	-.071	-.052
	.326	-.023	.003	.335	-.000	-.051	-.071	.034	.044	.297	-.158	-.054	-.042
	.343	-.421	.145	.266	-.074	-.087	-.376	.208	-.233	.303	-.287	-.376	-.004
	.387	.135	-.016	-.013	-.074	-.244	.211	.063	.037	.306	.204	.083	-.145
	.442	.153	-.113	-.116	-.209	-.019	.177	-.027	-.043	.332	.232	.213	-.060
	.493	-.212	.013	-.055	-.144	.119	-.153	.070	-.024	.236	.332	.184	-.031
	.537	-.071	-.071	-.125	-.346	.516	-.177	.050	-.037	.198	.005	.038	-.004

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(b) M = 0.700; outboard station

x/2	C _p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.811	.935	.772	.625	1.161	.548	.920	.581	.687	1.043	.393
	.003	-.831	-.528	.503	-.697	-.416	-.666	-.545	-.268	-.608	-.682	-.814
	.014	-.318	-.483	-.926	1.592	-.400	-.492	-.457	-.464	-.520	-.551	-.554
	.031	-.284	-.204	-.345	-.410	-.345	-.395	-.762	-.396	-.359	-.491	-.519
	.055	-.230	.649	-.441	-.413	-.404	-.308	-.195	-.354	-.369	-.379	-.446
	.106	.040	.155	-.318	-.352	-.387	-.526	.112	-.270	-.295	-.275	-.411
	.172	-.086	.487	-.156	-.271	-.677	-.100	.271	-.121	-.243	-.055	-.022
	.261	.215	.908	-.201	-.345	.042	-.056	.482	-.166	-.182	.620	.181
	.302	-.100	.078	-.046	-.307	-.303	-.090	-.050	.054	.027	.027	-.083
	.326	-.309	.402	-.065	.326	-.368	-.124	.054	.050	.050	.053	-.073
	.343	-.139	.029	-.149	.032	-.115	-.114	.054	.108	.060	.072	-.102
	.367	-.267	.046	-.006	-.116	-.152	-.196	.041	.044	-.024	-.020	-.194
	.442	.074	-.110	-.084	-.449	-.129	.104	.009	.030	-.069	-.017	.125
	.493	.060	-.075	-.055	-.216	-.410	.098	.041	.011	-.066	-.027	.140
Turbine cowl	.537	-.052	-.036	-.078	-.329	.439	.080	.061	.002	-.023	-.230	.096
	.581	-.202	.165	-.148	-.362	.431	.075	-.037	-.068	-.198	-.224	-.001
	.617	-.217	.064	-.044	-.210	-.536	-.075	-.004	.009	-.166	-.495	-.010
	.653	-.231	.159	-.085	-.358	-.371	-.185	-.024	.001	.044	-.262	-.001
	.680	-.241	.132	-.073	-.178	-.375	-.143	-.089	.041	-.395	-.317	-.083
	.708	-.231	-.052	-.023	-.326	-.375	-.143	-.323	.339	-.027	-.362	-.349
	.733	-.045	-.230	-.274	-.151	-.151	-.151	-.011	.091	.019	.019	.019
	.758	-.045	-.045	-.045	-.151	-.151	-.151	-.011	.091	.019	.019	.019
	.786	-.112	-.412	-.045	-.045	-.045	-.045	-.011	.091	.019	.019	.019
	.821	-.117	-.065	-.074	-.213	-.074	-.074	-.076	-.133	-.133	-.133	-.133
	.852	-.065	-.065	-.074	-.074	-.074	-.074	-.106	-.115	-.115	-.115	-.115
	.881	-.324	-.324	-.362	-.362	-.362	-.362	.359	.402	.402	.402	.375
	.910	-.324	-.324	-.362	-.362	-.362	-.362	.359	.402	.402	.402	.375
	.932	-.324	-.324	-.362	-.362	-.362	-.362	.359	.402	.402	.402	.375
Fan cowl	0.000	.230	.511	1.153	.748	.919						
	.003	-.753	-.552	-.153	-.485	-.981						
	.014	-.627	-.522	-.214	-.427	-.656						
	.031	-.476	-.438	-.266	-.414	-.453						
	.055	-.344	-.581	-.243	-.352	-.517						
	.106	-.593	-.344	-.201	-.216	-.177						
	.172	-.087	-.136	-.071	-.174	-.051						
	.261	-.174	.046	-.119	-.216	.088						
	.302	-.037	-.032	.065	.049	.665						
	.326	-.037	-.025	.024	.017	.111						
	.343	-.037	.075	-.051	.088	.004						
	.387	-.134	.118	.086	.150	.092						
	.442	.152	.043	.027	.052	.147						
	.493	.205	.105	.050	.101	.121						
	.537	.147	.055	.055	.053	.053						
Turbine cowl	.581	.054	.055	.014	-.028	-.054						
	.617	.050	.052	.073	-.032	-.375						
	.653	.054	.059	.073	.072	-.103						
	.690	-.014	.046	.108	-.122	-.190						
	.708		.372	.391	-.676							
	.735		.053	-.035	-.035							
	.768		-.057	-.252	-.252							
	.796		-.006	-.116	-.116							
	.821		-.157	-.182	-.182							
	.852		-.355	-.445	-.445							
	.881		-.355	-.445	-.445							
	.910		-.355	-.445	-.445							
	.932		-.355	-.445	-.445							
	.952		-.355	-.445	-.445							
	.972		-.355	-.445	-.445							
Plug	0.000	.230	.511	1.153	.748	.919						
	.003	-.753	-.552	-.153	-.485	-.981						
	.014	-.627	-.522	-.214	-.427	-.656						
	.031	-.476	-.438	-.266	-.414	-.453						
	.055	-.344	-.581	-.243	-.352	-.517						
	.106	-.593	-.344	-.201	-.216	-.177						
	.172	-.087	-.136	-.071	-.174	-.051						
	.261	-.174	.046	-.119	-.216	.088						
	.302	-.037	-.032	.065	.049	.665						
	.326	-.037	-.025	.024	.017	.111						
	.343	-.037	.075	-.051	.088	.004						
	.387	-.134	.118	.086	.150	.092						
	.442	.152	.043	.027	.052	.147						
	.493	.205	.105	.050	.101	.121						
	.537	.147	.055	.055	.053	.053						
	.581	.054	.055	.014	-.028	-.054						
	.617	.050	.052	.073	-.032	-.375						
	.653	.054	.059	.073	.072	-.103						
	.690	-.014	.046	.108	-.122	-.190						
	.708		.372	.391	-.676							
	.735		.053	-.035	-.035							
	.768		-.057	-.252	-.252							
	.796		-.006	-.116	-.116							
	.821		-.157	-.182	-.182							
	.852		-.355	-.445	-.445							
	.881		-.355	-.445	-.445							
	.910		-.355	-.445	-.445							
	.932		-.355	-.445	-.445							
	.952		-.355	-.445	-.445							
	.972		-.355	-.445	-.445							
	.992		-.355	-.445	-.445							

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(c) $M = 0.750$; Inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	0.000	.873	.401	.690	.867	.931	.723	.375	.841	.886	.827	.376
	.003	-.284	-.815	-.422	.062	-.095	-.816	-.815	-.844	.018	-.354	-.887
	.014	-.306	-.556	-.773	.195	-.227	-.519	-.568	-.844	-.145	-.394	-.861
	.031	-.284	-.291	-.589	-.288	-.145	-.346	-.227	-.659	-.227	-.392	-.861
	.055	-.244	-.330	-.410	-.256	-.169	-.288	-.322	-.322	-.201	-.308	-.776
	.106	-.133	-.248	-.284	-.169	-.072	-.138	-.228	-.234	-.148	-.319	-.275
	.172	-.036	-.092	-.125	-.081	-.036	-.032	.086	-.081	-.078	-.081	-.211
	.261	-.107	-.234	.377	-.177	-.002	-.071	-.199	.103	-.110	.007	-.064
	.302	-.053	-.058	-.075	-.055	.024	-.071	-.026	.046	-.081	.027	-.029
	.326	.137	.016	.523	-.025	.036	.053	.031	.200	-.078	.027	-.035
	.343	.137	.145	.228	.141	-.017	-.311	.191	.237	.106	.059	-.120
	.387	.048	.069	.022	-.017	-.132	.128	.109	.045	-.074	.138	-.220
Turbine cowl	.442	.133	-.078	-.128	-.204	.083	.177	.015	.067	.036	.071	-.073
	.493	.107	.013	-.055	-.177	-.169	-.076	.045	.002	.001	.088	-.068
	.537	.141	-.041	-.137	-.288	-.636	-.014	.045	-.002	-.181	.058	-.047
	.581	.465	-.219	-.248	-.497	-.408	.128	-.014	.110	-.201	.059	-.006
	.617	.222	-.119	-.072	-.326	.715	.107	.027	.031	.312	.099	-.047
	.653	.075	-.137	-.172	-.312	-.663	.123	-.681	-.026	.175	.015	-.041
	.690	-.297	-.263	.286	-.344	.127	-.160	-.011	.039	-.186	.103	-.038
	.708		-.131	.221	-.019							
	.735		-.008		-.455							
	.768		-.533		-.408							
	.796		-.829		-.477							
	.821		-.680		-.812							
	.852		-.330		-.355							
Fan cowl	0.000	.571	.355	.985	.932	.637						
	.003	-.1274	-.809	-.231	.071	-.892						
	.014	.956	-.653	-.307	.099	-.458						
	.031	-.420	-.140	-.805	-.175	-.286						
	.055	-.314	-.301	-.213	-.145	-.239						
	.106	-.114	-.280	-.143	-.087	-.087						
	.172	-.025	-.093	-.054	.057	.057						
	.261	-.081	-.112	.050	-.008	.052						
	.302	-.039	.001	-.011	.112	.177						
	.326	-.059	.077	.089	.153	.206						
	.343	-.309	.248	.199	.283	-.035						
	.387	.272	.124	.086	.185	.033						
	.442	.196	.074	-.002	.089	.323						
Turbine cowl	.493	-.030	.136	.036	.077	.191						
	.537	-.012	.055	.042	-.060	-.172						
	.581	.032	.027	.119	.048	-.172						
	.617	.059	.116	.025	-.169	-.128						
	.653	.152	.066	.048	-.183	-.180						
	.690	.063	.095	.019	-.116	-.183						
	.708		.080	.321	.098							
	.735		-.002		-.008							
	.768		.351		-.353							
	.796		-.090		-.128							
	.821		.110		.153							
	.852		.374		.411							
Plug	0.000											
	.003											
	.014											
	.031											
	.055											
	.106											
	.172											
	.261											
	.302											
	.326											
	.343											
	.387											
	.442											

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(d) M = 0.750; outboard station

x/c	C _p at -												
	α = 0°												
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.760	.901	.781	.619	1.156	.506	.859	.944	.640	1.057	.874	1.050
	.003	-.933	-.679	-.649	-.797	-.405	-.721	-.688	.400	-.700	-.750	-.752	-.662
	.014	-.332	-.572	-.595	-.629	-.464	-.562	-.584	.511	-.573	-.620	-.611	-.410
	.031	-.332	-.666	-.590	-.451	-.465	-.434	-.511	-.451	-.402	-.447	-.637	-.535
	.055	-.274	-.850	-.481	-.443	-.393	-.337	-.110	-.383	-.394	-.376	-.280	-.363
	.106	-.256	-.176	-.339	-.405	-.279	-.275	.299	-.286	-.441	-.199	-.020	-.402
	.172	-.084	1.625	-.164	-.131	-.072	-.093	.466	-.126	-.190	-.100	-.175	-.255
	.261	.406	1.021	-.215	-.326	-.048	-.105	.622	-.174	-.196	-.066	-.346	-.085
	.302	-.102	-.085	.046	-.217	-.158	-.085	-.049	.057	-.073	.092	-.044	-.156
	.326	-.208	.575	.045	.510	-.143	-.009	.234	-.020	.204	.066	-.133	-.009
	.343	-.093	.105	-.085	.128	-.055	-.076	.125	-.064	.148	-.032	.133	-.020
	.387	-.177	.040	-.014	-.078	-.161	-.169	.022	.051	.039	-.023	.166	-.061
Turbine cowl	.442	-.022	-.046	-.090	-.111	-.035	.039	.030	-.079	-.173	-.014	.030	-.023
	.493	.031	-.037	-.096	-.355	-.393	.097	.033	.030	.095	.121	.101	-.033
	.537	-.058	-.632	-.138	-.305	-.582	-.088	-.023	-.020	-.282	-.235	.095	-.051
	.581	-.230	-.073	-.181	-.440	-.452	-.062	-.023	-.082	-.188	-.355	.028	-.014
	.617	-.292	-.191	-.002	-.355	-.938	-.058	.087	.007	-.226	-.464	-.007	-.148
	.553	-.257	-.102	-.102	-.287	-.267	-.085	-.067	-.026	-.279	-.335	-.042	-.003
	.690	-.298	-.274	-.025	-.020	-.355	-.151	.004	.051	-.161	-.320	-.086	-.001
	.708		.258	.267	-.591			.308	.323	-.682		.322	.358
	.735		.022		.651			.039		-.020		.060	
	.768		-.463		-.457			-.327		-.482		-.194	-.464
	.796		-.144		-.473			-.029		-.079		-.038	-.020
	.821		.111		.175			.119		-.166		.125	-.168
.852		-.320		.360			.356		.386		.358	.406	

x/c	α = 4°												
	α = 0°												
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	-.272	.866	1.149	.710	.929							
	.003	-.707	-.742	-.575	-.557	-.1115							
	.014	-.733	-.653	-.583	-.494	-.688							
	.031	-.593	-.668	-.527	-.414	-.488							
	.055	-.357	-.392	-.286	-.358	-.382							
	.106	-.241	-.179	-.227	-.205	-.223							
	.172	-.037	.007	-.079	-.117	-.017							
	.261	-.070	.176	-.138	-.146	-.084							
	.302	-.056	-.028	.063	.031	.160							
	.326	.045	.081	-.008	.116	.042							
	.343	-.048	.128	-.035	.160	.089							
	.387	-.158	.105	.084	.031	.089							
Turbine cowl	.442	.121	.013	.022	.025	.113							
	.493	.165	.179	.051	.025	.142							
	.537	.107	.105	.057	-.037	-.020							
	.581	.045	.016	-.008	-.111	-.084							
	.617	.030	.065	.021	-.105	-.370							
	.653	.054	.108	.069	-.052	-.075							
	.690	.006	.001	.090	-.252	-.255							
	.708		.371	.389	-.673								
	.735		.075		.033								
	.768		-.120		-.464								
	.796		-.014		-.022								
	.821		.152		.172								
.852		.380		.434									

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(e) $M = 0.775$; inboard station

		C_p at -														
		$\alpha = -2^\circ$						$\alpha = 0^\circ$								
x/c		Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl		$\alpha = 1^\circ$														
0.000		.870	.387	.701	.858	.571	.726	.355	.826	.889	.845	.650	.380	.914	.908	.757
.003		.303	.874	1.385	.034	.093	.865	.877	.933	.066	.400	.118	.946	.527	.078	.562
.014		.523	.582	1.138	.205	.233	.538	.578	.595	.139	.397	.838	.994	.451	.130	.410
.031		.252	.423	.386	.393	.149	.364	.429	.339	.1240	.232	.396	.199	.569	.205	.239
.055		.159	.343	.405	.238	.168	.292	.225	.331	.108	.204	.298	.310	.262	.160	.211
.106		.159	.350	.387	.163	.068	.135	.230	.235	.047	.066	.124	.212	.198	.062	.152
.172		.031	.052	.121	.062	.011	.079	.086	.080	.047	.068	.009	.052	.046	.008	.095
.261		.103	.236	.362	.171	.047	.067	.199	.206	.001	.040	.056	.172	.199	.023	.098
.302		.091	.057	.073	.048	.053	.067	.024	.044	.001	.113	.039	.001	.015	.100	.165
.326		.283	.019	.597	.026	.067	.150	.052	.305	.091	.138	.203	.078	.255	.142	.184
.343		.303	.171	.169	.173	.016	.275	.201	.185	.214	.032	.264	.236	.183	.243	.055
.387		.078	.053	.034	.023	.166	.024	.100	.071	.091	.066	.067	.134	.081	.134	.022
.442		.138	.107	.085	.112	.134	.167	.035	.061	.066	.150	.195	.056	.037	.001	.184
.493		.037	.149	.085	.272	.140	.041	.060	.038	.032	.057	.056	.092	.028	.011	.182
.537		.231	.138	.211	.163	.580	.086	.026	.109	.215	.459	.191	.087	.030	.079	.379
.581		.501	.121	.191	.439	.303	.226	.016	.162	.246	.232	.191	.008	.065	.127	.104
.617		.134	.242	.225	.513	.684	.109	.052	.043	.293	.414	.022	.106	.033	.301	.259
.653		.033	.169	.186	.460	.849	.120	.058	.097	.148	.420	.144	.020	.044	.074	.334
.690		.307	.191	.447	.370	.229	.173	.117	.136	.327	.016	.064	.047	.146	.146	.071
.708			.068	.143	.033			.007	.251	.026			.064	.326	.019	
.735			.017		.019			.005		.397			.002		.371	
.768			.411		.345			.502		.363			.372		.346	
.796			.793		.751			.629		.734			.414		.634	
.821			.073		.065			.145		.158			.160		.229	
.852			.309		.308			.342		.405			.362		.420	
Plug		$\alpha = 4^\circ$														
0.000		.537	.380	.577	.946	.688										
.003		.1.271	.768	.854	.076	.688										
.014		.400	.166	.734	.106	.277										
.031		.367	.201	.211	.171	.218										
.055		.124	.202	.163	.137	.056										
.106		.001	.053	.028	.162	.104										
.172		.039	.157	.121	.030	.127										
.261		.022	.014	.000	.121	.191										
.302		.158	.050	.172	.166	.216										
.326		.260	.262	.180	.250	.062										
.343		.165	.158	.110	.138	.075										
.387		.212	.054	.020	.031	.286										
.442		.012	.110	.090	.028	.208										
.493		.063	.127	.057	.014	.230										
.537		.101	.057	.059	.003	.117										
.581		.037	.161	.014	.120	.148										
.617		.169	.062	.006	.168	.196										
.653		.098	.141	.082	.171	.126										
.690			.071	.341	.063											
.708			.003		.325											
.735			.346		.364											
.768			.567		.524											
.796			.126		.222											
.821			.352		.424											
.852																

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(f) $M = 0.775$; outboard station

x/c	C_p at -												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.740	.955	.794	.618	1.146	.502	.868	.943	.638	1.044	.809	.414
	.003	-.712	-.768	-.767	-.877	-.440	-.876	-.781	.478	-.750	-.809	-.789	.875
	.014	-.417	-.621	-.1094	-.663	-.499	-.588	-.648	-.549	-.597	-.634	-.422	.875
	.031	-.333	.323	-.496	-.494	-.373	-.461	-.359	-.498	-.422	-.465	-.412	.875
	.055	-.290	1.005	-.482	-.466	-.344	-.351	.268	-.413	-.405	-.386	-.381	.875
	.106	.451	.053	-.351	-.305	-.285	-.117	.370	-.296	-.326	-.202	-.111	.875
	.172	-.082	.742	-.167	-.220	-.209	-.092	.463	-.126	-.146	-.064	-.217	.875
	.261	.583	1.107	-.221	-.277	.053	-.213	.690	-.186	-.182	-.059	-.289	.875
	.302	-.108	-.082	-.048	-.130	-.127	-.079	.650	.055	-.112	.156	.445	.875
	.326	-.167	.728	-.048	-.130	-.085	.674	.333	.016	-.182	.137	.177	.875
	.343	-.178	.153	-.029	-.169	-.031	-.054	.154	.021	.182	.137	.177	.875
	.387	-.163	.054	.020	-.105	-.088	-.160	.089	.024	.182	.137	.177	.875
Turbine cowl	.442	.037	-.037	-.054	-.023	.045	-.003	.027	-.050	.029	.042	.009	.875
	.493	.100	.085	-.048	-.431	-.285	-.116	.024	.013	.154	.146	.012	.875
	.537	-.027	-.155	-.128	-.237	-.643	-.022	.010	-.033	.177	.298	.053	.875
	.581	-.230	-.029	-.141	-.522	-.601	-.067	-.033	.086	.273	.335	.025	.875
	.617	-.260	-.255	-.135	-.387	-.705	-.113	.030	.047	.202	.589	.026	.875
	.653	-.294	-.102	-.128	-.290	-.485	-.079	.061	.044	.069	.121	.045	.875
	.690	-.268	-.235	-.068	-.336	-.389	-.198	-.135	.032	.211	.332	.005	.875
	.708	.212	-.280	-.280	-.635	-.635	-.833	.313	.041	.833	.351	.362	.875
	.735	.051	-.351	-.351	-.200	-.200	-.041	.052	.041	.041	.052	.052	.875
	.768	-.388	-.388	-.388	-.399	-.399	-.342	-.342	-.342	-.342	-.342	-.342	.875
	.796	-.635	-.635	-.635	-.525	-.525	-.475	-.475	-.475	-.475	-.475	-.475	.875
	.821	-.141	-.141	-.141	-.152	-.152	-.140	-.140	-.140	-.140	-.140	-.140	.875
Plug	.852	.337	.337	.337	.354	.354	.347	.347	.347	.347	.347	.347	.875
	0.000	.299	.675	1.125	.726	.942	.726	.942	.726	.942	.726	.942	.875
	.003	-.634	-.856	-.345	-.607	-.881	-.607	-.881	-.607	-.881	-.607	-.881	.875
	.014	-.934	-.572	-.311	-.511	-.880	-.511	-.880	-.511	-.880	-.511	-.880	.875
	.031	-.493	-.541	-.354	-.361	-.482	-.361	-.482	-.361	-.482	-.361	-.482	.875
	.055	-.361	-.275	-.254	-.344	-.378	-.344	-.378	-.344	-.378	-.344	-.378	.875
	.106	-.323	-.076	-.226	-.251	-.192	-.251	-.192	-.251	-.192	-.251	-.192	.875
	.172	-.077	.103	-.070	-.144	-.070	-.099	.044	.044	-.069	.026	.045	.875
	.261	-.020	.264	-.133	-.276	-.099	.044	.044	.044	.044	.044	.044	.875
	.302	-.039	-.016	-.065	-.054	.243	.243	.243	.243	.243	.243	.243	.875
	.326	-.126	.168	.020	.195	.280	.195	.280	.195	.280	.195	.280	.875
	.343	-.022	.176	-.014	.195	.668	.097	.082	.091	.082	.091	.082	.875
Fan cowl	.387	-.071	.066	-.025	-.005	.091	.091	.091	.091	.091	.091	.091	.875
	.493	.169	.094	.094	.094	.150	.094	.150	.094	.150	.094	.150	.875
	.537	.173	.066	.094	.094	.094	.094	.094	.094	.094	.094	.094	.875
	.581	.057	.077	.006	.045	-.011	.045	-.011	.045	-.011	.045	-.011	.875
	.617	.058	.191	.060	.183	-.344	.183	-.344	.183	-.344	.183	-.344	.875
	.653	.050	.029	.066	.194	-.214	.194	-.214	.194	-.214	.194	-.214	.875
	.690	.041	.100	.066	.122	-.251	.122	-.251	.122	-.251	.122	-.251	.875
	.708	.344	.344	.378	-.055	-.160	-.055	-.160	-.055	-.160	-.055	-.160	.875
	.735	.105	.105	.378	.055	-.160	.055	-.160	.055	-.160	.055	-.160	.875
	.768	-.221	-.221	.378	-.055	-.160	-.055	-.160	-.055	-.160	-.055	-.160	.875
	.796	.015	.015	.378	-.005	-.386	-.005	-.386	-.005	-.386	-.005	-.386	.875
	.821	.162	.162	.378	.215	-.430	.215	-.430	.215	-.430	.215	-.430	.875
Turbine cowl	.852	.391	.391	.378	.430	-.430	.391	.391	.391	.391	.391	.391	.875
	0.000	.299	.675	1.125	.726	.942	.726	.942	.726	.942	.726	.942	.875
	.003	-.634	-.856	-.345	-.607	-.881	-.607	-.881	-.607	-.881	-.607	-.881	.875
	.014	-.934	-.572	-.311	-.511	-.880	-.511	-.880	-.511	-.880	-.511	-.880	.875
	.031	-.493	-.541	-.354	-.361	-.482	-.361	-.482	-.361	-.482	-.361	-.482	.875
	.055	-.361	-.275	-.254	-.344	-.378	-.344	-.378	-.344	-.378	-.344	-.378	.875
	.106	-.323	-.076	-.226	-.251	-.192	-.251	-.192	-.251	-.192	-.251	-.192	.875
	.172	-.077	.103	-.070	-.144	-.070	-.099	.044	.044	-.069	.026	.045	.875
	.261	-.020	.264	-.133	-.276	-.099	.044	.044	.044	.044	.044	.044	.875
	.302	-.039	-.016	-.065	-.054	.243	.243	.243	.243	.243	.243	.243	.875
	.326	-.126	.168	.020	.195	.280	.195	.280	.195	.280	.195	.280	.875
	.343	-.022	.176	-.014	.195	.668	.097	.082	.091	.082	.091	.082	.875
	.387	-.071	.066	-.025	-.005	.091	.091	.091	.091	.091	.091	.091	.875

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(g) $M = 0.800$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.874	.357	.599	.844	.977	.735	.402	.838	.880	.844	.669	.385	.905	.917	.753		
	.003	-.225	-.326	-1.361	.044	-.101	-.898	-.934	-.967	-.075	-.424	-1.094	-.979	-.595	-.073	-.723		
	.014	-.341	-.415	-1.156	-.195	-.233	-.547	-.596	-.572	-.160	-.416	-.813	-.936	-.500	-.135	-.477		
	.031	-.276	-.291	-.024	-.303	-.157	-.371	-.240	-.328	-.238	-.235	-.400	.205	-.376	-.213	-.256		
	.055	-.255	-.347	-.366	-.262	-.165	-.289	-.328	-.331	-.198	-.227	-.310	.322	-.276	-.170	-.235		
	.106	-.129	-.252	-.282	-.157	-.057	-.130	-.223	-.231	-.101	-.052	-.127	.211	.206	-.105	-.030		
	.172	-.022	-.654	-.114	-.076	.072	-.020	.097	.071	.021	.075	-.008	.097	.049	-.048	-.097		
	.241	-.036	-.236	-.792	-.157	.021	-.073	-.150	.335	-.071	.075	-.049	-.173	.311	-.035	-.099		
	.302	-.084	-.652	-.060	-.025	.067	-.061	.012	.030	.051	.134	-.033	.003	.014	.097	.167		
	.326	.434	.056	.912	.005	.018	.088	.076	.432	.110	.156	.335	.081	.365	.137	.197		
	.343	.198	.154	.112	.215	.045	.283	.235	.133	.247	.064	.229	.251	.138	.267	.073		
	.387	-.238	.094	-.030	-.087	-.127	-.130	.168	-.049	-.012	-.047	.188	.022	.011	.145	.027		
Turbine cowl	.442	.031	-.133	-.055	-.017	.172	.003	.213	.094	-.003	.061	-.033	.132	.084	.008	.205		
	.553	.517	.126	-.150	-.276	-.125	-.607	-.205	-.017	-.160	.508	.220	.076	-.035	.118	.180		
	.537	.517	.158	-.193	.146	-.092	.315	.459	.138	.365	-.095	.261	.046	.122	.181	.022		
	.581	.752	.129	-.179	-.459	-.097	.718	.136	.041	.249	-.045	.045	.032	.078	-.256	.302		
	.617	-.137	.158	-.266	-.564	-.777	.123	-.001	.093	-.273	.570	.139	-.011	-.081	-.122	.372		
	.653	.030	-.199	-.195	.510	-.406		-.120	.262	.327	.056	-.698	-.076	.251	.305	.000		
	.590	-.610	-.271	.675	.502			-.025	.262	.075			.078	.276	.062			
	.708		-.039	.153	.072			-.001		-.346			.003		-.329			
	.735		-.022		-.346			-.377		-.254			.422		-.272			
	.768		.371		.518			-.674		-.086			.530		-.679			
	.796		-.658		-.605			-.051		-.069			.354		-.164			
	.821		-.150		-.195			-.356		-.390								
.852		.221		-.250														
Fan cowl	0.000	.615	.358	.577	.947	.724												
	.003	-1.258	-.561	-.277	-.076	-1.004												
	.014	-.793	-.658	-.375	-.123	-.470												
	.031	-.331	-.193	-.575	-.185	-.293												
	.055	-.308	-.210	-.225	-.142	-.217												
	.106	-.116	-.202	-.169	-.110	-.050												
	.172	.002	-.101	-.031	-.071	.111												
	.241	-.034	-.158	-.201	-.004	.125												
	.302	-.018	.017	-.001	.125	.203												
	.326	-.235	.096	-.250	.157	.216												
	.343	-.230	.274	-.148	.265	.076												
	.387	.035	.131	.112	.176	.068												
Turbine cowl	.442	.211	.012	-.023	.020	.224												
	.493	.043	.125	.020	.039	.230												
	.537	.129	.125	.009	-.029	-.309												
	.581	.145	.044	-.112	-.077	-.066												
	.617	.039	.163	.093	-.241	-.180												
	.653	.162	-.015	.047	-.053	-.233												
	.690	.100	.125	.152	-.139	-.131												
	.708		.101	.347	.076													
	.735		.001		-.298													
	.768		-.304		-.306													
	.796		-.491		-.597													
	.821		-.187		-.235													
.852		-.379		-.434														
Plug	0.000																	
	.003																	
	.014																	
	.031																	
	.055																	
	.106																	
	.172																	
	.241																	
	.302																	
	.326																	
	.343																	
	.387																	
	.442																	
	.493																	
	.537																	
	.581																	
	.617																	
	.653																	
	.690																	
	.708																	
	.735																	
	.768																	
	.796																	
	.821																	
	.852																	

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(h) $M = 0.800$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.730	.902	.735	.630	1.177	.545	.890	.955	.669	1.066	.428	.877	1.027	.689	1.014		
	.003	.931	.846	.907	.877	.429	.558	.835	.578	.758	.810	.557	.891	.496	.684	.999		
	.014	.434	.658	.1.066	.671	.476	.595	.680	.557	.614	.639	.582	.692	.461	.575	.722		
	.031	.385	.637	.1.026	.562	.400	.489	.151	.461	.454	.447	.489	.267	.447	.421	.480		
	.055	.312	.1.147	.358	.407	.356	.350	.425	.426	.416	.352	.367	.000	.376	.388	.396		
	.106	.081	.028	.355	.373	.237	.074	.430	.301	.290	.288	.053	.199	.275	.261	.242		
	.172	.084	.956	.167	.231	.069	.077	.526	.123	.172	.066	.078	.357	.104	.158	.076		
	.241	.001	.1.185	.118	.286	.056	.347	.768	.186	.185	.078	.269	.510	.164	.081	.081		
	.302	.100	.082	.046	.158	.020	.069	.042	.059	.017	.244	.053	.030	.060	.016	.372		
	.366	.116	.907	.042	.868	.006	.160	.455	.005	.417	.187	.024	.030	.000	.355	.328		
	.433	.051	.193	.016	.214	.006	.136	.201	.021	.230	.024	.020	.202	.027	.236	.070		
	.500	.024	.087	.008	.120	.008	.042	.095	.059	.024	.016	.135	.125	.057	.081	.132		
Turbine cowl	.567	.008	.104	.058	.121	.056	.062	.021	.010	.039	.173	.057	.005	.014	.082	.125		
	.634	.001	.071	.039	.134	.036	.169	.175	.062	.207	.145	.200	.125	.150	.011	.125		
	.701	.000	.039	.140	.137	.062	.078	.069	.115	.052	.318	.085	.065	.030	.147	.135		
	.768	.000	.267	.104	.147	.042	.065	.005	.147	.446	.352	.012	.027	.071	.247	.323		
	.835	.000	.183	.178	.092	.809	.142	.126	.048	.261	.677	.082	.071	.109	.149	.529		
	.902	.000	.221	.145	.332	.356	.101	.032	.072	.104	.172	.016	.000	.016	.003	.149		
	.969	.000	.232	.093	.172	.500	.158	.189	.065	.031	.318	.135	.000	.090	.166	.269		
	.708	.239	.275	.750	.115	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.735	.387	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.768	.333	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.821	.682	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.852	.682	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
Fan cowl	0.000	.344	.678	1.118	.728	.965	.545	.890	.955	.669	1.066	.428	.877	1.027	.689	1.014		
	.003	.625	.979	.715	.528	.845	.558	.835	.578	.758	.810	.557	.891	.496	.684	.999		
	.014	.445	.845	.364	.528	.425	.585	.860	.587	.761	.829	.567	.902	.507	.722	.722		
	.031	.360	.716	.320	.451	.387	.549	.831	.557	.714	.797	.537	.872	.481	.641	.641		
	.055	.201	.501	.241	.351	.238	.400	.785	.526	.654	.752	.517	.827	.447	.580	.580		
	.106	.075	.373	.173	.213	.001	.274	.630	.436	.416	.616	.407	.700	.376	.396	.396		
	.172	.020	.222	.143	.110	.090	.247	.528	.301	.230	.528	.301	.599	.275	.261	.261		
	.241	.008	.167	.064	.053	.313	.218	.466	.228	.185	.466	.228	.510	.164	.081	.081		
	.302	.038	.120	.021	.256	.318	.160	.416	.201	.158	.416	.201	.481	.104	.181	.181		
	.366	.014	.084	.000	.204	.234	.124	.368	.186	.117	.368	.186	.430	.060	.132	.132		
	.433	.009	.044	.000	.167	.209	.088	.324	.155	.085	.324	.155	.392	.000	.070	.070		
	.500	.000	.000	.000	.132	.172	.068	.283	.121	.059	.283	.121	.352	.000	.032	.032		
Turbine cowl	.567	.000	.000	.000	.100	.142	.056	.242	.100	.039	.242	.100	.312	.000	.013	.013		
	.634	.000	.000	.000	.068	.100	.040	.204	.085	.032	.204	.085	.277	.000	.005	.005		
	.701	.000	.000	.000	.032	.068	.020	.167	.065	.020	.167	.065	.242	.000	.000	.000		
	.768	.000	.000	.000	.015	.048	.015	.132	.048	.015	.132	.048	.217	.000	.000	.000		
	.835	.000	.000	.000	.008	.032	.004	.068	.020	.005	.068	.020	.167	.000	.000	.000		
	.902	.000	.000	.000	.000	.000	.000	.005	.000	.000	.000	.000	.117	.000	.000	.000		
	.969	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.000	.092	.000	.000	.000		
	.708	.353	.386	.1.182	.105	.162	.545	.890	.955	.669	1.066	.428	.877	1.027	.689	1.014		
	.735	.335	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.768	.226	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.821	.682	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		
	.852	.682	.439	.473	.473	.473	.352	.119	.305	.126	.348	.104	.348	.348	.166	.090		

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Continued

(i) $M = 0.825$; inboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.879	.423	.727	.851	.972	.750	.424	.844	.884	.681	.407	.902	.896	.774			
	.003	.340	.949	.1304	.095	.095	.871	.957	.591	.068	.1108	.595	.679	.074	.698			
	.014	.336	.623	.132	.256	.256	.555	.670	.650	.171	.848	.685	.499	.139	.417			
	.031	.277	.296	.644	.306	.144	.390	.249	.002	.255	.391	.223	.041	.212	.277			
	.055	.246	.336	.378	.231	.158	.303	.344	.344	.205	.300	.322	.277	.165	.232			
	.106	.112	.234	.265	.144	.064	.129	.229	.240	.106	.111	.210	.204	.053	.017			
	.172	.009	.101	.095	.014	.079	.015	.092	.070	.021	.054	.067	.098	.040	.113			
	.261	.080	.221	1.080	.144	.040	.055	.013	.033	.065	.048	.169	.456	.024	.116			
	.302	.058	.053	.040	.012	.060	.074	.085	.070	.078	.028	.010	.011	.106	.181			
	.326	.754	.048	1.155	.053	.018	.058	.071	.580	.107	.172	.519	.093	.500	.150			
	.343	.163	.236	.052	.028	.074	.027	.123	.014	.094	.194	.276	.090	.290	.212			
	.442	.155	.048	.012	.025	.075	.217	.264	.029	.086	.162	.130	.062	.160	.043			
Turbine cowl	.492	.048	.048	.012	.025	.075	.217	.264	.029	.086	.162	.130	.062	.160	.043			
	.537	.052	.048	.012	.025	.075	.217	.264	.029	.086	.162	.130	.062	.160	.043			
	.537	.052	.048	.012	.025	.075	.217	.264	.029	.086	.162	.130	.062	.160	.043			
	.581	.070	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.617	.080	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.653	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.690	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.708	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.735	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.768	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.796	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
	.821	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030			
.852	.085	.050	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030	.030				
Plug	0.000	.635	.407	.966	.926	.716	.716	.926	.966	.926	.716	.926	.966	.926	.716			
	.003	.131	.982	.352	.078	.974	.974	.078	.352	.078	.974	.974	.078	.352	.078			
	.014	.579	.450	.355	.112	.517	.517	.112	.355	.112	.517	.517	.112	.355	.112			
	.031	.246	.208	.208	.130	.275	.275	.130	.208	.130	.275	.275	.130	.208	.130			
	.055	.290	.213	.235	.156	.226	.226	.156	.213	.156	.226	.226	.156	.213	.156			
	.106	.105	.195	.175	.055	.039	.039	.055	.195	.175	.039	.039	.056	.212	.360			
	.172	.013	.102	.023	.030	.125	.125	.030	.102	.023	.125	.125	.061	.248	.461			
	.261	.026	.154	.331	.008	.143	.143	.008	.331	.008	.143	.143	.401	.592	.834			
	.302	.006	.023	.003	.138	.190	.190	.138	.003	.138	.190	.190	.011	.592	.834			
	.326	.387	.107	.367	.171	.236	.236	.171	.367	.171	.236	.236	.011	.592	.834			
	.343	.184	.294	.085	.306	.109	.109	.306	.085	.306	.109	.109	.011	.592	.834			
	.387	.105	.138	.089	.044	.078	.078	.044	.138	.089	.078	.078	.011	.592	.834			
Fan cowl	.442	.210	.008	.044	.054	.247	.247	.054	.008	.044	.247	.247	.056	.212	.360			
	.493	.123	.229	.125	.161	.226	.226	.161	.229	.125	.226	.226	.061	.248	.461			
	.537	.249	.073	.034	.112	.304	.304	.112	.073	.034	.304	.304	.061	.248	.461			
	.581	.214	.076	.104	.122	.067	.067	.122	.104	.122	.067	.067	.061	.248	.461			
	.617	.009	.008	.104	.192	.234	.234	.192	.009	.104	.234	.234	.061	.248	.461			
	.653	.155	.068	.085	.177	.283	.283	.177	.068	.177	.283	.283	.061	.248	.461			
	.690	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.708	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.735	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.768	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.796	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.821	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
.852	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461				
Turbine cowl	0.000	.635	.407	.966	.926	.716	.716	.926	.966	.926	.716	.926	.966	.926	.716			
	.003	.131	.982	.352	.078	.974	.974	.078	.352	.078	.974	.974	.078	.352	.078			
	.014	.579	.450	.355	.112	.517	.517	.112	.355	.112	.517	.517	.112	.355	.112			
	.031	.246	.208	.208	.130	.275	.275	.130	.208	.130	.275	.275	.130	.208	.130			
	.055	.290	.213	.235	.156	.226	.226	.156	.213	.156	.226	.226	.156	.213	.156			
	.106	.105	.195	.175	.055	.039	.039	.055	.195	.175	.039	.039	.056	.212	.360			
	.172	.013	.102	.023	.030	.125	.125	.030	.102	.023	.125	.125	.061	.248	.461			
	.261	.026	.154	.331	.008	.143	.143	.008	.331	.008	.143	.143	.401	.592	.834			
	.302	.006	.023	.003	.138	.190	.190	.138	.003	.138	.190	.190	.011	.592	.834			
	.326	.387	.107	.367	.171	.236	.236	.171	.367	.171	.236	.236	.011	.592	.834			
	.343	.184	.294	.085	.306	.109	.109	.306	.085	.306	.109	.109	.011	.592	.834			
	.387	.105	.138	.089	.044	.078	.078	.044	.138	.089	.078	.078	.011	.592	.834			
Plug	.442	.210	.008	.044	.054	.247	.247	.054	.008	.044	.247	.247	.056	.212	.360			
	.493	.123	.229	.125	.161	.226	.226	.161	.229	.125	.226	.226	.061	.248	.461			
	.537	.249	.073	.034	.112	.304	.304	.112	.073	.034	.304	.304	.061	.248	.461			
	.581	.214	.076	.104	.122	.067	.067	.122	.104	.122	.067	.067	.061	.248	.461			
	.617	.009	.008	.104	.192	.234	.234	.192	.009	.104	.234	.234	.061	.248	.461			
	.653	.155	.068	.085	.177	.283	.283	.177	.068	.177	.283	.283	.061	.248	.461			
	.690	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.708	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.735	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.768	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.796	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
	.821	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461			
.852	.088	.034	.274	.270	.041	.041	.270	.034	.274	.270	.270	.061	.248	.461				

TABLE 35.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 4 - Concluded

(1) $M = 0.825$; outboard station

x/c	C _p at															
	α = 0°															
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.748	.899	.817	.660	1.157	.566	.899	.962	.692	1.075	.459	.887	1.031	.711	1.025
	.003	-1.016	-.870	-1.067	-.827	-.415	-.504	-.891	-.687	-.830	-.798	-.669	-.900	-.588	-.713	-.983
	.014	-.426	-.707	-1.020	-.654	-.504	-.621	-.718	-.615	-.623	-.631	-.866	-.737	-.464	-.600	-.684
	.031	-.399	1.012	-1.043	-.607	-.415	-.534	-.660	-.449	-.547	-.573	-.579	-.064	-.356	-.459	-.564
	.055	1.312	1.046	-.507	-.473	-.360	-.365	.572	.474	-.432	-.413	-.359	.094	-.398	-.399	-.404
	.106	1.014	1.191	-.343	-.303	-.158	-.319	.474	.328	-.293	-.261	-.160	.283	-.287	-.278	-.208
	.172	-.072	1.104	-.159	-.166	-.054	-.082	.590	.133	-.154	-.041	-.068	.428	-.103	-.234	-.019
	.261	1.069	1.254	-.230	-.287	.054	.508	.856	.207	-.212	.069	.428	.578	-.174	-.153	.080
	.302	-.096	.678	.046	-.124	-.054	.078	-.049	.048	-.031	.346	.428	.052	-.027	.015	.491
	.326	-.060	1.120	.025	1.104	.054	.272	.598	.040	.556	.228	.228	.412	.473	.473	.368
Turbine cowl	.343	-.029	.222	.057	.253	.020	-.023	.235	.053	.257	.016	-.009	.238	.062	.264	.093
	.387	-.155	.001	.086	-.122	-.085	-.141	.059	.124	-.012	.016	.125	.094	.125	.057	.046
	.442	1.125	.080	-.101	.185	.069	.126	.057	.007	.074	.181	.192	.202	.080	.181	.211
	.493	.050	.012	-.112	-.308	-.169	.436	.086	.020	.109	.154	.086	.059	.077	.024	.200
	.537	.007	.073	-.201	-.407	-.386	.024	.069	.047	.345	.479	.009	.099	.006	.383	.328
	.581	-.253	.146	-.179	-.436	-.685	-.058	.123	.123	.434	.830	.079	.085	.067	.280	.029
	.617	-.296	.154	-.175	-.473	-.882	-.141	.176	.036	.403	.484	.079	.057	.023	.208	.323
	.653	-.296	.138	-.175	-.400	-.402	-.176	.094	.060	.091	.345	.115	.124	.022	.025	.305
	.690	-.450	.335	-.180	-.148	-.436	-.165	.149	.060	.091	.345	.115	.124	.022	.025	.305
	.708	.259	.259	.257	-.737	-.436	-.165	.272	.337	.921	.273	.273	.273	.357	-.105	.053
Plug	.735	.015	.015	.015	-.009	.098	.098	.098	.005	.005	.152	.152	.152	.022	.022	.022
	.768	-.557	-.557	-.572	-.572	-.062	-.062	-.549	.429	.429	.429	.429	.429	.422	.365	.365
	.796	-.093	-.093	-.062	-.062	.137	.137	.148	.337	.337	.337	.337	.337	.322	.265	.265
	.821	.180	.180	-.211	-.211	.390	.390	.390	.388	.388	.388	.388	.388	.357	.182	.182
	.852	.391	.874	1.088	.724	.975	.724	.724	.337	.921	.273	.273	.273	.357	.025	.025
	0.000	.391	.874	1.088	.724	.975	.724	.724	.337	.921	.273	.273	.273	.357	-.105	.053
	.003	-.591	-.895	.481	-.664	-1.082	-.664	-.664	.005	.005	.152	.152	.152	.022	.022	.022
	.014	-.976	-.817	-.389	-.559	-.878	-.559	-.559	.445	.445	.293	.293	.293	.325	.325	.325
	.031	-.792	-.244	-.423	-.429	-.593	-.429	-.429	.429	.429	.422	.422	.422	.365	.365	.365
	.055	-.305	-.076	-.357	-.374	-.390	-.374	-.374	.388	.388	.357	.357	.357	.418	.418	.418
Fan cowl	.106	-.026	.084	-.262	-.259	-.230	-.259	-.259	.388	.388	.357	.357	.357	.418	.418	.418
	.172	-.061	.250	-.087	-.089	-.063	-.089	-.089	.388	.388	.357	.357	.357	.418	.418	.418
	.261	-.268	.391	-.160	-.120	.086	-.120	-.120	.388	.388	.357	.357	.357	.418	.418	.418
	.302	-.042	-.018	.058	.050	.418	.050	.418	.388	.388	.357	.357	.357	.418	.418	.418
	.326	.319	.339	.029	.363	.371	.363	.371	.388	.388	.357	.357	.357	.418	.418	.418
	.343	-.003	.242	.068	.277	.099	.277	.099	.388	.388	.357	.357	.357	.418	.418	.418
	.387	-.124	.127	.032	.079	.084	.102	.076	.388	.388	.357	.357	.357	.418	.418	.418
	.442	.442	.442	.442	.442	.442	.442	.442	.388	.388	.357	.357	.357	.418	.418	.418
	.493	.493	.493	.493	.493	.493	.493	.493	.388	.388	.357	.357	.357	.418	.418	.418
	.537	.537	.537	.537	.537	.537	.537	.537	.388	.388	.357	.357	.357	.418	.418	.418
Turbine cowl	.581	.581	.581	.581	.581	.581	.581	.581	.388	.388	.357	.357	.357	.418	.418	.418
	.617	.617	.617	.617	.617	.617	.617	.617	.388	.388	.357	.357	.357	.418	.418	.418
	.653	.653	.653	.653	.653	.653	.653	.653	.388	.388	.357	.357	.357	.418	.418	.418
	.690	.690	.690	.690	.690	.690	.690	.690	.388	.388	.357	.357	.357	.418	.418	.418
	.708	.708	.708	.708	.708	.708	.708	.708	.388	.388	.357	.357	.357	.418	.418	.418
	.735	.735	.735	.735	.735	.735	.735	.735	.388	.388	.357	.357	.357	.418	.418	.418
	.768	.768	.768	.768	.768	.768	.768	.768	.388	.388	.357	.357	.357	.418	.418	.418
	.796	.796	.796	.796	.796	.796	.796	.796	.388	.388	.357	.357	.357	.418	.418	.418
	.821	.821	.821	.821	.821	.821	.821	.821	.388	.388	.357	.357	.357	.418	.418	.418
	.852	.852	.852	.852	.852	.852	.852	.852	.388	.388	.357	.357	.357	.418	.418	.418
Plug	0.000	.391	.874	1.088	.724	.975	.724	.724	.337	.921	.273	.273	.273	.357	.025	.025
	.003	-.591	-.895	.481	-.664	-1.082	-.664	-.664	.005	.005	.152	.152	.152	.022	.022	.022
	.014	-.976	-.817	-.389	-.559	-.878	-.559	-.559	.445	.445	.293	.293	.293	.325	.325	.325
	.031	-.792	-.244	-.423	-.429	-.593	-.429	-.429	.429	.429	.422	.422	.422	.365	.365	.365
	.055	-.305	-.076	-.357	-.374	-.390	-.374	-.374	.388	.388	.357	.357	.357	.418	.418	.418
	.106	-.026	.084	-.262	-.259	-.230	-.259	-.259	.388	.388	.357	.357	.357	.418	.418	.418
	.172	-.061	.250	-.087	-.089	-.063	-.089	-.089	.388	.388	.357	.357	.357	.418	.418	.418
	.261	-.268	.391	-.160	-.120	.086	-.120	-.120	.388	.388	.357	.357	.357	.418	.418	.418
	.302	-.042	-.018	.058	.050	.418	.050	.418	.388	.388	.357	.357	.357	.418	.418	.418
	.326	.319	.339	.029	.363	.371	.363	.371	.388	.388	.357	.357	.357	.418	.418	.418

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5

(a) $M = 0.700$; inboard station

x/c	C _p at -														
	α = -2°				α = 0°				α = 1°						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.891	.410	.697	1.014	.770	.410	.895	.884	.863	.649	.411	1.014	.913	.784
	.003	-.177	-.713	1.092	-.038	-.538	-.700	-.777	-.393	-.281	-.870	-.716	-.373	-.320	-.455
	.014	-.235	-.503	-.725	-.211	-.407	-.488	-.552	-.329	-.281	-.486	-.482	-.414	-.288	-.400
	.031	-.216	-.386	-.907	-.140	-.290	-.372	-.841	-.243	-.211	-.331	-.360	-.883	-.208	-.253
	.055	-.226	-.333	-.412	-.159	-.266	-.316	-.321	-.220	-.185	-.283	-.305	-.267	-.192	-.262
	.106	-.172	-.262	-.293	-.111	-.179	-.237	-.237	-.140	-.102	-.176	-.225	-.202	-.112	-.096
	.172	-.094	-.160	-.139	-.083	-.091	-.131	-.092	-.028	-.086	-.084	-.119	-.067	.000	.007
	.261	-.182	-.283	-.206	-.108	-.108	-.164	-.173	-.111	-.054	-.152	-.222	-.154	-.074	-.022
	.302	-.162	-.099	-.099	-.067	-.041	-.120	-.060	-.060	.001	-.108	-.152	-.048	.029	.055
	.326	-.109	.026	.108	.004	-.025	-.101	.014	.062	.068	-.103	.026	-.022	.096	.083
	.343	-.082	.094	.124	.357	.382	.326	.178	.062	.251	-.559	.196	.026	.128	.544
	.387	.323	-.019	.011	.655	.357	.111	.076	.035	.507	.319	.048	.064	.288	.487
	.442	-.133	-.080	-.099	-.127	-.179	-.149	.007	-.012	-.028	.079	.038	.019	.026	.192
Turbine cowl	.453	-.031	-.017	-.080	-.671	-.671	-.1043	.007	-.012	-.182	.008	.019	.026	-.109	.333
	.537	-.153	-.031	-.080	-.390	-.743	-.149	.001	-.015	.017	.127	.032	.016	-.045	.192
	.581	.007	-.082	-.192	-.365	-.650	-.048	.049	-.057	.039	.127	.038	.058	-.093	.192
	.617	-.143	.004	-.122	-.365	-.355	.195	.048	-.015	-.050	.127	.038	.058	.016	.067
	.653	.192	-.082	-.080	-.127	-.323	.195	.027	-.012	-.127	.207	.103	.032	-.042	-.035
Plug	.690	-.162	-.029	-.054	-.272	-.083	-.038	.078	.303	.110	.042	.145	.068	-.064	-.163
	.768	-.057	.223	.223	-.012	-.454	.012	-.012	.060	.060	.097	.145	.318	.128	.432
	.735	-.022	-.022	-.022	-.454	.012	-.012	-.012	-.060	.060	-.006	-.006	.318	.128	.432
	.766	-.057	-.057	-.057	-.454	.012	-.012	-.012	-.060	.060	-.006	-.006	.318	.128	.432
	.821	-.176	-.176	-.176	-.131	-.131	-.150	-.240	-.451	-.108	-.087	-.087	.119	.147	.432
.852	.292	.318	.351	.401							.386				
Fan cowl	0.000	.583	.403	1.064	.662	.662	.403	.913	.884	.863	.649	.411	1.014	.913	.784
	.003	-.154	-.713	1.092	-.038	-.538	-.700	-.777	-.393	-.281	-.870	-.716	-.373	-.320	-.455
	.014	-.235	-.488	-.318	-.269	-.416									
	.031	-.383	-.363	-.957	-.198	-.288									
	.055	-.305	-.301	-.216	-.182	-.217									
	.106	-.184	-.221	-.175	-.102	-.099									
	.172	-.082	-.115	-.048	.004	-.022									
	.261	-.145	-.218	-.134	.054	.000									
	.302	-.101	-.041	-.032	.058	.077									
	.326	-.101	.029	-.032	.116	.106									
	.343	-.426	.187	.042	.097	-.550									
	.387	-.297	.058	.084	.173	.503									
	.442	-.072	.032	.007	.109	.161									
Turbine cowl	.493	.015	.046	.046	.225										
	.537	-.111	.062	.036	-.134										
	.581	-.111	.074	-.041	-.166										
	.617	.209	.071	.039	-.122	-.638									
	.653	.083	.091	.058	-.083	.004									
	.690	.083	.155	.097	.116	-.172									
	.768		.142	.328	.109										
	.735		-.003		-.307										
	.768				-.185										
	.796		-.048		-.122										
	.821		.148		-.073										
	.852		.399		.449										

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(b) $M = 0.700$; outboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	.881	1.082	.765	.633	1.161	.587	.509	.982	.672	1.067	.440	.440
	-.148	-.573	.328	-.777	-.325	-.585	-.585	.302	-.660	-.612	-.713	-.583
	.014	-.303	-.647	-.580	-.396	-.488	-.537	-.582	-.544	-.515	-.563	-.531
	.031	.685	-.618	-.448	-.322	-.784	-.942	-.462	-.393	-.405	-.771	-.819
	.055	-.264	.210	.459	.302	.304	.476	.372	.386	.402	.335	.702
	.106	.031	-.291	.326	-.228	.028	.281	-.281	-.276	-.222	-.035	.364
	.172	.133	.041	.164	.160	.125	.125	.129	.238	.167	.127	.256
	.261	-.254	.315	.206	-.218	.236	.195	.177	.199	.212	-.224	-.061
	.302	.167	.106	.365	.160	.101	.080	.106	.038	-.086	-.137	-.068
	.326	-.133	.038	.151	-.093	.140	.009	.007	.001	-.028	-.098	.007
	.343	.354	.017	.161	.036	.238	.095	.109	.059	.388	.226	.091
	.387	.380	.028	.005	.259	.311	.001	.037	-.083	.272	.342	.013
Turbine cowl	.442	.177	.074	.080	.277	.154	.041	.035	.035	.073	.142	.003
	.493	.060	.028	.048	.046	.048	.017	.006	-.006	.059	.028	.033
	.537	.041	.041	.048	.344	.004	.002	.006	.112	.054	.008	.023
	.581	.094	.080	.024	.422	.014	.015	.060	.215	.187	.003	.004
	.617	.119	.024	.054	.470	.009	.069	.017	.057	.341	.042	.075
Plug	.653	.114	.106	.048	.373	.006	.020	.004	.101	.193	.066	.107
	.690	-.201	.028	.015	.357	-.081	.035	.940	.286	-.241	-.016	.081
	.708	.218	.296	.528	.528	.528	.038	.951	-.618		.353	.369
	.735	.028	.028	.077	.554	.554	.030		-.138		.062	.081
	.768	-.116	.069	.088	.588	.588	.135		.315		-.074	.075
Fan cowl	.821	.319		.365			.124		-.051		.156	.181
	.852						.360		.423		.379	.445
	0.000	.316	.456	.734	.935							
	.003	-.629	-.550	-.262	-.573							
	.014	-.774	-.796	-.317	-.673							
	.031	-.358	-.595	-.271	-.338							
	.055	.033	.265	.219	.383							
	.106	.130	.294	.183	.212							
	.172	.212	.142	.142	.074							
	.261	.130	.064	.004	.090							
	.302	.091	.010	.040	.001							
	.326	.228	.085	.054	.036							
	.343	.339	.020	.069	.381							
Turbine cowl	.387	.033	.004	.007	.433							
	.442	.014	.033	.043	.014							
	.493	.020	.046	.004	.094							
	.537	.030	.082	.072	.133							
	.581	.068	.091	.062	.007							
Plug	.617	.107	.078	.041	.125							
	.653	.035	.380	.101	.016							
	.690		.095	-.019	-.096							
	.735		-.041	-.209								
	.768		.036	.010								
Fan cowl	.821		.182	.207								
	.852		.399	.462								
	0.000	.316	.456	.734	.935							
	.003	-.629	-.550	-.262	-.573							
	.014	-.774	-.796	-.317	-.673							
Turbine cowl	.031	-.358	-.595	-.271	-.338							
	.055	.033	.265	.219	.383							
	.106	.130	.294	.183	.212							
	.172	.212	.142	.142	.074							
	.261	.130	.064	.004	.090							
Plug	.302	.091	.010	.040	.001							
	.326	.228	.085	.054	.036							
	.343	.339	.020	.069	.381							
	.387	.033	.004	.007	.433							
	.442	.014	.033	.043	.014							
Fan cowl	.493	.020	.046	.004	.094							
	.537	.030	.082	.072	.133							
	.581	.068	.091	.062	.007							
	.617	.107	.078	.041	.125							
	.653	.035	.380	.101	.016							
Turbine cowl	.690		.095	-.019	-.096							
	.735		-.041	-.209								
	.768		.036	.010								
	.821		.182	.207								
	.852		.399	.462								

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(c) $M = 0.750$; inboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.000	.887	.395	.710	.828	.985	.749	.396	.880	.854	.657	.376	.952	.854	.729			
	.003	.223	.842	.997	.557	.083	.681	.858	.972	.432	.841	.835	.521	.471	.596			
	.014	.262	.552	-1.025	.431	.226	.437	.536	.921	.365	.353	.524	.489	.344	.385			
	.031	.223	.418	.778	.332	.159	.318	.410	.688	.268	.230	.359	.806	.233	.271			
	.055	.231	.351	.428	.288	.162	.278	.334	.337	.230	.186	.289	.322	.204	.204			
	.106	.169	.266	.296	.191	.104	.181	.240	.240	.140	.099	.174	.208	.113	.087			
	.172	.090	.151	.132	.074	.022	.083	.123	.090	.020	.003	.067	.003	.009	.012			
	.261	.183	.279	.207	.180	.095	.110	.240	.172	.105	.040	.138	.111	.064	.005			
	.302	.143	.091	.089	.048	.025	.069	.052	.055	.012	.036	.094	.038	.035	.062			
	.326	.043	.013	.259	.004	.004	.049	.021	.235	.079	.005	.039	.100	.114	.103			
	.343	.515	.101	.586	.513	.005	.068	.153	.226	.412	.024	.191	.153	.263	.164			
	.387	.196	.019	.023	.174	.165	.081	.012	.065	.643	.398	.427	.083	.442	.468			
Turbine cowl	.442	.196	.157	.130	.174	.304	.088	.093	.038	.064	.427	.061	.002	.038	.307			
	.493	.052	.061	.094	.133	.794	.098	.042	.044	.070	.669	.062	.039	.003	.879			
	.537	.096	.096	.157	.247	.680	.018	.032	.021	.202	.420	.005	.056	.009	.242			
	.581	.207	.030	.191	.308	.513	.173	.038	.116	.020	.265	.030	.014	.078	.131			
	.617	.169	.132	.034	.276	.381	.004	.127	.079	.268	.181	.039	.118	.009	.058			
	.653	.185	.049	.102	.232	.396	.191	.026	.055	.061	.186	.198	.036	.073	.073			
	.690	.249	.153	.057	.364	.110	.021	.147	.003	.198	.082	.039	.059	.119	.032			
	.708	.011	.026	.031	.031	.036	.036	.036	.294	.018	.082	.118	.311	.146	.002			
	.735	.015	.016	.016	.016	.016	.016	.016	.016	.391	.000	.000	.000	.359	.000			
	.768	.567	.492	.492	.492	.492	.492	.492	.492	.420	.240	.240	.350	.350	.350			
	.796	.813	.811	.811	.811	.811	.811	.811	.811	.108	.079	.079	.035	.035	.035			
	.821	.078	.136	.136	.136	.136	.136	.136	.136	.182	.112	.112	.152	.152	.152			
.852	.278	.317	.317	.317	.317	.317	.317	.317	.380	.358	.358	.398	.398	.398				
Fan cowl	.000	.592	.389	1.031	.887	.671												
	.003	.1228	.856	.288	.345	.845												
	.014	.577	.525	.349	.301	.439												
	.031	.367	.396	.847	.211	.319												
	.055	.303	.317	.229	.179	.225												
	.106	.179	.220	.176	.091	.085												
	.172	.068	.103	.042	.039	.003												
	.261	.135	.206	.132	.039	.002												
	.302	.082	.027	.021	.087	.108												
	.326	.011	.049	.084	.137	.140												
	.343	.338	.219	.149	.201	.383												
	.387	.087	.046	.102	.286	.499												
Turbine cowl	.000	.592	.389	1.031	.887	.671												
	.003	.1228	.856	.288	.345	.845												
	.014	.577	.525	.349	.301	.439												
	.031	.367	.396	.847	.211	.319												
	.055	.303	.317	.229	.179	.225												
	.106	.179	.220	.176	.091	.085												
	.172	.068	.103	.042	.039	.003												
	.261	.135	.206	.132	.039	.002												
	.302	.082	.027	.021	.087	.108												
	.326	.011	.049	.084	.137	.140												
	.343	.338	.219	.149	.201	.383												
	.387	.087	.046	.102	.286	.499												
Plug	.000	.592	.389	1.031	.887	.671												
	.003	.1228	.856	.288	.345	.845												
	.014	.577	.525	.349	.301	.439												
	.031	.367	.396	.847	.211	.319												
	.055	.303	.317	.229	.179	.225												
	.106	.179	.220	.176	.091	.085												
	.172	.068	.103	.042	.039	.003												
	.261	.135	.206	.132	.039	.002												
	.302	.082	.027	.021	.087	.108												
	.326	.011	.049	.084	.137	.140												
	.343	.338	.219	.149	.201	.383												
	.387	.087	.046	.102	.286	.499												

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued
(d) $M = 0.750$; outboard station

x/c	C _p at -														
	α = -2°						α = 0°						α = 1°		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.803	1.014	.753	.591	1.142	.556	.683	.930	.526	1.043	.810	1.043	.648	.972
	.003	-.231	-.734	.448	-.910	-.398	-.648	-.753	.453	-.770	.693	-.767	.337	-.729	-.876
	.014	-.359	-.622	-.448	-.643	-.431	-.525	-.625	.543	-.579	.558	-.526	-.407	-.564	-.661
	.031	-.439	-.1030	-.657	-.493	-.363	-.569	-.612	.540	-.435	.440	-.641	-.440	-.599	-.676
	.055	-.284	-.119	-.492	-.457	-.322	-.335	-.428	.407	-.408	.358	-.350	-.348	-.388	-.370
	.106	-.041	-.309	-.347	-.331	-.275	-.030	-.292	-.292	-.255	.232	-.036	-.262	-.279	-.264
	.172	-.134	-.132	-.170	-.198	-.119	-.127	-.080	-.132	-.146	.094	-.120	-.129	-.164	-.079
	.261	-.266	-.466	-.220	-.275	-.210	-.233	.408	-.185	-.202	.149	-.217	-.165	-.170	-.108
	.302	-.174	-.111	-.252	-.193	-.131	-.141	-.079	-.094	.035	.085	-.125	-.188	-.002	-.014
	.326	-.125	-.031	-.015	-.078	-.087	-.026	-.002	.015	.023	.005	.012	.030	.003	.030
	.343	-.386	-.093	-.087	.128	-.355	-.226	.113	-.070	.148	.448	.326	.184	.059	.468
	.387	-.054	-.017	-.013	-.066	-.290	-.039	-.053	.036	.001	.020	-.041	.026	.071	.001
Turbine cowl	.442	-.297	-.176	-.090	-.190	-.004	-.189	-.126	-.085	-.120	-.108	-.173	-.100	-.073	-.064
	.493	-.087	-.076	-.031	-.169	-.187	-.102	-.024	.015	.162	.065	.127	.060	.153	.148
	.537	-.160	-.131	-.090	-.151	-.184	-.061	-.032	-.008	-.193	-.064	.042	.036	.164	-.017
	.581	-.050	-.072	-.151	-.401	-.560	-.021	-.085	-.061	-.158	-.191	.008	-.049	-.073	-.079
	.617	-.169	-.054	-.002	-.360	-.831	-.009	.148	.015	.141	.320	.065	.024	-.179	-.344
	.653	-.081	-.072	-.069	-.266	-.301	-.030	-.053	-.005	-.308	.255	.052	.048	.061	-.129
	.690	-.253	-.226	-.008	-.013	-.319	-.066	.033	.054	-.141	-.238	-.010	.066	.005	-.179
	.708		-.238	.268	-.584			.284	.334	.687		.370	.379	.505	.767
	.735		.002	.002	-.087			.069		-.008		.089		.062	
	.768		-.409		-.413			-.280		-.402		-.100		-.449	
	.796		-.102		-.263			-.229		.004		-.100		.062	
	.821		.082		.161			.122		.183		.154		.186	
.852		.306		.346			.352		.406		.379		.439		
Fan cowl	0.000	-.330	-.519	1.127	-.697	-.930									
	.003	-1.115	-.765	-.296	-.647	-.1.012									
	.014	-.602	-.641	-.236	-.527	-.636									
	.031	-.620	-.629	-.364	-.371	-.494									
	.055	-.384	-.405	-.293	-.365	-.371									
	.106	-.033	-.275	-.228	-.209	-.253									
	.172	-.121	-.157	-.086	-.197	-.083									
	.261	-.214	-.008	-.145	-.250	-.074									
	.302	-.117	-.054	-.098	-.023	-.026									
	.326	-.073	-.026	-.050	-.061	-.061									
	.343	-.320	-.180	-.051	-.164	-.467									
	.387	-.011	-.003	-.085	-.005	-.241									
Turbine cowl	.442	-.161	-.068	-.008	-.024	-.026									
	.493	-.122	-.079	-.068	-.111	-.194									
	.537	-.055	-.082	-.035	-.006	.070									
	.581	-.025	-.026	-.006	-.109	-.003									
	.617	-.091	-.118	-.068	-.006	-.165									
	.653	-.095	-.147	-.068	-.211	-.021									
	.690	-.025	-.044	-.100	-.277	-.121									
	.708		-.392	-.401	-.659										
	.735		-.112		-.047										
	.768		-.059		-.377										
	.796		-.032		-.047										
	.821		-.186		-.202										
.852		-.395		-.455											
Plug	0.000														
	.003														
	.014														
	.031														
	.055														
	.106														
	.172														
	.261														
	.302														
	.326														
	.343														
	.387														
	.442														
	.493														
	.537														
	.581														
	.617														
	.653														
	.690														
	.708														
	.735														
	.768														
	.796														
	.821														
	.852														

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued
(e) $M = 0.775$; inboard station

x/c	C _p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																														
	α = -2°						α = 0°						α = 4°																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(f) $M = 0.775$; outboard station

x/c	C_p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 4^\circ$																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued
(E) M = 0.800; Inboard station

x/c	C _p at -																															
	α = -2°						α = 0°						α = 4°																			
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L		
Fan cowl	0.000	.898	.390	.723	.814	.977	.759	.409	.868	.856	.687	.398	.959	.863	.748																	
	.003	-.268	-.894	.923	-.606	-.107	-.784	-.944	-.995	-.531	-.442	-.936	.867	-.448	-.707																	
	.014	-.280	-.597	.923	-.463	-.255	-.469	-.571	-.584	-.394	-.503	-.558	.341	-.365	-.462																	
	.031	-.240	-.455	.200	-.341	-.155	-.331	-.443	-.303	-.286	-.240	-.431	.579	-.281	-.481																	
	.055	-.236	-.359	.281	-.287	-.155	-.277	-.341	-.346	-.237	-.189	-.333	.301	-.211	-.198																	
	.106	-.166	-.263	.286	-.182	-.098	-.171	-.238	-.241	-.135	-.089	-.223	.212	-.106	-.077																	
	.172	-.076	-.137	.286	-.055	-.001	-.065	-.108	-.076	-.005	.006	-.058	.055	-.055	.045																	
	.261	-.174	-.276	.158	-.166	-.074	-.155	-.232	-.165	-.094	-.024	-.132	.215	-.147	.004																	
	.302	-.129	-.080	.072	-.031	.001	-.098	-.043	-.043	.038	.060	-.079	.025	-.028	.077																	
	.326	.394	.005	.575	.036	.028	.348	.206	.504	.095	.095	.191	.056	.283	.133																	
	.343	.379	.168	.568	.743	.028	.348	.206	.504	.095	.095	.191	.056	.283	.133																	
	.387	.791	.088	.019	.897	.028	.776	.095	.455	.602	.259	.307	.231	.296	.252																	
	.442	.010	.014	.074	-.055	.730	.041	-.021	-.008	.071	.761	.081	-.034	.002	.112																	
	.493	-.056	-.027	.090	-.158	.921	.033	.182	.052	.033	.596	.072	.250	.064	.408																	
.537	.283	.139	.186	-.163	.562	.254	-.078	-.059	.372	.785	.072	.250	.064	.408																		
.581	.520	-.101	.131	-.457	.465	.429	.052	-.113	.100	.113	.174	.112	.122	.303																		
.617	.014	-.027	.186	-.352	-.479	-.020	-.065	.125	-.191	.159	.009	.012	.083	.023																		
.653	.181	-.160	.068	-.306	-.592	.192	.054	-.135	-.078	.246	.199	.107	.096	-.217																		
.690	-.219	-.127	-.247	-.357	.368	-.069	.016	-.011	-.305	.254	.030	.048	.029	-.160																		
.735		-.033	.187	-.042			.049	.287	.054			.083	.353	.112																		
.768		-.011		-.481			.008		.111			.007		.270																		
.796		-.444		-.430			-.381		.307			.309		-.284																		
.821		-.770		-.344			-.571		.709			.250		-.329																		
.852		-.027		-.320			.157		.173			.166		-.241																		
		.309		.320			.344		.407			.350		.430																		
Fan cowl	0.000	.633	.403	1.031	.687	.681																										
	.003	-.127	-.915	-.407	-.406	-.963																										
	.014	-.660	-.550	-.415	-.320	-.484																										
	.031	-.378	-.415	-.580	-.228	-.295																										
	.055	-.293	-.318	-.248	-.180	-.209																										
	.106	-.158	-.210	-.183	-.085	-.064																										
	.172	-.048	-.080	-.031	.047	.058																										
	.261	-.117	-.194	-.124	-.024	.048																										
	.302	-.056	-.010	-.010	.103	.133																										
	.326	.197	.071	.250	.152	.160																										
	.343	-.268	.250	.288	.327	.247																										
	.387	-.599	.088	.131	.400	.354																										
	.442	-.040	-.037	.007	.049	.370																										
	.493	.132	.247	.071	.173	.440																										
.537	.152	.033	.039	.192	.155																											
.581	.156	.112	-.056	.090	-.031																											
.617	.046	.163	.023	-.188	-.012																											
.653	.205	.060	.047	.023	-.048																											
.690	.087	.209	.125	-.120	.084																											
.735		.120	.344	.074																												
.768		.012		-.233																												
.796		-.267		-.298																												
.821		-.037		-.204																												
.852		-.179		.265																												
Turbine cowl	0.000	.633	.403	1.031	.687	.681																										
	.003	-.127	-.915	-.407	-.406	-.963																										
	.014	-.660	-.550	-.415	-.320	-.484																										
	.031	-.378	-.415	-.580	-.228	-.295																										
	.055	-.293	-.318	-.248	-.180	-.209																										
	.106	-.158	-.210	-.183	-.085	-.064																										
	.172	-.048	-.080	-.031	.047	.058																										
	.261	-.117	-.194	-.124	-.024	.048																										
	.302	-.056	-.010	-.010	.103	.133																										
	.326	.197	.071	.250	.152	.160																										
	.343	-.268	.250	.288	.327	.247																										
	.387	-.599	.088	.131	.400	.354																										
	.442	-.040	-.037	.007	.049	.370																										
	.493	.132	.247	.071	.173	.440																										
.537	.152	.033	.039	.192	.155																											
.581	.156	.112	-.056	.090	-.031																											
.617	.046	.163	.023	-.188	-.012																											
.653	.205	.060	.047	.023	-.048																											
.690	.087	.209	.125	-.120	.084																											
.735		.120	.344	.074																												
.768		.012		-.233																												
.796		-.267		-.298																												
.821		-.037		-.204																												

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued

(h) $M = 0.800$; outboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.740	1.040	.792	.623	1.157	.551	.708	.929	.689	1.053	.845
	.003	.411	.847	.463	.290	.630	.384	.589	.855	.585	.782	.485
	.014	.382	.880	.437	.261	.597	.357	.562	.828	.558	.750	.459
	.031	.083	1.087	.108	.012	.392	.163	.357	.711	.513	.486	.385
	.055	.292	.923	.327	.379	.341	.358	.117	.444	.439	.377	.359
	.106	.043	.920	.352	.516	.248	.043	.302	.313	.285	.285	.291
	.172	.135	.479	.165	.232	.140	.126	.396	.133	.141	.092	.094
	.261	.272	.689	.225	.281	.216	.244	.611	.193	.209	.149	.296
	.302	.162	.105	.155	.132	.156	.134	.073	.090	.027	.032	.115
	.326	.105	.020	.007	.134	.042	.081	.009	.022	.002	.011	.061
	.345	.162	.176	.001	.219	.441	.298	.189	.009	.024	.040	.129
	.367	.830	.018	.072	.104	.175	.737	.030	.060	.033	.535	.018
	.442	.066	.042	.050	.140	.175	.738	.030	.060	.033	.535	.018
	.493	.064	.075	.181	.314	.191	.037	.180	.069	.137	.046	.051
	.537	.015	.151	.075	.069	.175	.000	.122	.103	.027	.013	.025
	.581	.092	.034	.091	.249	.409	.033	.184	.114	.085	.080	.085
Turbine cowl	0.000	.206	.068	.064	.140	.468	.628	.103	.096	.176	.578	.026
	.003	.206	.157	.102	.430	.810	.013	.093	.031	.078	.228	.073
	.014	.243	.113	.135	.425	.718	.126	.101	.066	.130	.320	.046
	.031	.220	.220	.258	.748	.718	.126	.295	.328	.860	.377	.394
	.055	.045	.045	.113	.113	.113	.126	.126	.328	.860	.377	.394
	.106	.337	.337	.352	.352	.352	.352	.352	.352	.352	.352	.352
	.172	.541	.541	.517	.517	.517	.517	.517	.517	.517	.517	.517
	.261	.168	.168	.140	.140	.140	.140	.140	.140	.140	.140	.140
	.302	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.326	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.345	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.367	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.442	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.493	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
	.537	.353	.353	.360	.360	.360	.360	.360	.360	.360	.360	.360
Plug	0.000	.389	.812	1.105	.715	.969						
	.003	.389	.812	1.105	.715	.969						
	.014	.389	.812	1.105	.715	.969						
	.031	.389	.812	1.105	.715	.969						
	.055	.389	.812	1.105	.715	.969						
	.106	.389	.812	1.105	.715	.969						
	.172	.389	.812	1.105	.715	.969						
	.261	.389	.812	1.105	.715	.969						
	.302	.389	.812	1.105	.715	.969						
	.326	.389	.812	1.105	.715	.969						
	.345	.389	.812	1.105	.715	.969						
	.367	.389	.812	1.105	.715	.969						
	.442	.389	.812	1.105	.715	.969						
	.493	.389	.812	1.105	.715	.969						
	.537	.389	.812	1.105	.715	.969						
	.581	.389	.812	1.105	.715	.969						

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Continued
(i) $M = 0.825$; inboard station

x/\bar{c}	C_p at											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I
Fan cowl	.887	.431	.734	.821	.578	.769	.408	.862	.850	.845	.710	.419
	.287	.947	.883	.873	.134	.801	.979	.1013	.576	.519	.1047	.962
	.003	.314	.883	.484	.251	.498	.846	.766	.429	.384	.524	.616
	.031	.255	.313	.370	.178	.361	.485	.348	.309	.250	.375	.449
	.035	.372	.247	.292	.160	.290	.360	.252	.247	.195	.284	.334
	.106	.165	.262	.180	.092	.180	.243	.256	.138	.091	.158	.220
	.172	.067	.127	.104	.009	.066	.105	.076	.006	.028	.048	.082
	.261	.173	.194	.162	.064	.156	.240	.172	.097	.019	.048	.082
	.302	.118	.069	.017	.014	.097	.042	.042	.044	.072	.068	.017
	.326	.061	.061	.048	.061	.740	.044	.775	.098	.108	.341	.067
	.343	.299	.208	.873	.201	.258	.247	.684	.736	.195	.268	.262
	.387	.747	.074	.079	.097	.757	.117	.046	.809	.069	.744	.148
Turbine cowl	.442	.063	.175	.054	.865	.054	.150	.036	.090	.824	.054	.051
	.493	.102	.082	.084	.879	.054	.115	.093	.472	.390	.267	.040
	.537	.523	.115	.095	.500	.575	.024	.084	.281	.121	.239	.080
	.581	.661	.241	.372	.398	.543	.105	.084	.154	.121	.239	.080
	.617	.020	.025	.393	.401	.103	.132	.084	.154	.242	.050	.020
	.653	.173	.052	.391	.541	.182	.094	.032	.115	.265	.192	.046
	.690	.239	.220	.344	.538	.070	.016	.079	.312	.440	.011	.124
	.728	.025	.115	.066	.066	.114	.114	.288	.103	.054	.054	.054
	.735	.014	.014	.417	.417	.003	.003	.006	.006	.015	.015	.015
	.766	.309	.309	.448	.448	.370	.370	.190	.190	.290	.290	.290
	.796	.699	.699	.642	.642	.620	.620	.455	.455	.519	.519	.519
	.821	.328	.328	.347	.347	.033	.033	.143	.143	.174	.174	.174
Plug	.852	.025	.025	.105	.105	.348	.348	.370	.370	.366	.366	.366
Fan cowl	.658	.414	1.031	.885	.705							
	.175	.973	.463	.428	.950							
	.872	.604	.447	.332	.472							
	.031	.342	.205	.236	.296							
	.055	.283	.317	.250	.184							
	.106	.149	.208	.185	.078							
	.172	.039	.067	.026	.073							
	.261	.106	.187	.127	.055							
	.302	.047	.002	.005	.117							
	.326	.346	.081	.362	.161							
	.343	.224	.279	.378	.410							
	.387	.731	.141	.120	.467							
Turbine cowl	.442	.040	.052	.039	.439							
	.493	.063	.185	.055	.499							
	.537	.284	.031	.252	.189							
	.581	.225	.133	.083	.140							
	.617	.024	.013	.175	.070							
	.653	.197	.151	.155	.010							
	.690	.028	.073	.119	.185							
	.708	.172	.362	.109								
	.735	.021	.021	.163								
	.748	.264	.264	.184								
	.796	.398	.398	.216								
	.821	.193	.193	.249								
Plug	.852	.362	.362	.449								

TABLE 36.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5 - Concluded
(1) M = 0.825; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.781	1.057	.612	1.163	.580	1.075	.938	.668	1.063	.498	.699	1.025	.499	1.019			
	.003	-.435	-.885	.810	-.947	-.811	-.905	.850	-.886	-.818	-.919	-.909	.553	-.806	-.971			
	.014	-.435	-.712	-.107	-.683	-.591	-.732	.706	-.664	-.609	-.774	-.731	-.473	-.828	-.981			
	.031	-.443	-.107	-.107	-.427	-.363	-.255	.758	-.609	-.559	-.201	-.232	.668	-.955	-.990			
	.055	-.306	.025	-.801	-.495	-.359	.071	.430	-.463	-.400	-.354	-.232	.405	-.824	-.968			
	.106	-.063	.334	-.323	-.273	-.058	.320	.339	-.288	-.251	-.048	.295	.295	-.299	-.268			
	.172	-.137	.665	-.164	-.111	-.132	.663	.142	-.149	-.097	-.115	.230	.108	.139	-.074			
	.261	-.286	.752	-.237	-.291	-.262	.711	.218	-.228	-.157	-.225	.377	.179	.181	-.100			
	.302	-.169	-.106	-.093	-.132	-.087	.144	.088	.062	.018	-.056	.033	.041	.007	.007			
	.326	-.106	-.020	-.004	-.048	-.025	.074	.055	.263	.558	.317	.241	.060	.279	.574			
	.343	-.067	.227	.049	.263	.511	.219	.225	.055	.024	.809	.041	.120	.030	-.550			
	.387	-.820	-.025	.080	-.121	-.777	.811	.013	.123	-.024	.198	.031	.057	.039	.072			
Turbine cowl	.442	.082	-.112	-.101	.278	.315	.021	-.074	.167	.198	.060	.043	-.095	.126	.020			
	.493	-.125	.182	.007	-.302	-.262	.074	.124	-.280	.003	.054	-.069	.090	.046	.057			
	.537	.063	.143	-.106	-.291	.044	.008	.039	.177	.016	.042	.141	.017	.251	.200			
	.581	-.157	.208	-.150	-.453	.058	.037	.105	.319	.277	.042	.032	.028	.187	-.487			
	.617	-.110	.007	-.161	-.270	.618	.126	.042	.369	.533	.042	.032	.028	.155	-.213			
	.653	-.118	-.054	-.080	-.507	-.688	.077	.124	-.047	.298	.015	.067	.003	.069	-.051			
	.690	-.384	-.285	-.201	-.503	-.829	-.136	-.055	-.322	-.444	-.032	.036	.094	.017	-.252			
	.708		.214	.214	-.741		.343	.335	-.907		.293	.343		-.104				
	.735		.028		.422		.128		.138		.151			.067				
	.768		-.235		-.566		-.270		-.319		.232			-.268				
	.796		-.552		-.132		-.457		-.523		.216			-.440				
	.821		.054		.061		.181		.175		.204			.245				
.852		.316		.294		.370		.407		.361			.425					
Fan cowl	0.000	.417	.599	1.094	.712	.981												
	.003	-.969	-.915	.499	-.739	-.1071												
	.014	-.988	-.766	-.393	-.603	-.815												
	.031	-.145	-.204	-.522	-.499	-.389												
	.055	-.337	-.110	-.364	-.405	-.378												
	.106	-.039	-.283	-.270	-.224	-.222												
	.172	-.105	.176	-.089	-.086	-.041												
	.261	-.211	.237	-.167	-.151	-.057												
	.302	-.090	.042	.205	.032	.040												
	.326	-.039	.045	.058	.110	.084												
	.343	.326	.245	.066	.283	.570												
	.387	-.772	.058	.108	.074	-.344												
Turbine cowl	.442	.016	.056	.069	.029	.160												
	.493	.044	.048	.061	.005	.102												
	.537	.051	.068	-.055	-.049	.110												
	.581	.083	.252	-.005	-.172	-.162												
	.617	.036	-.034	.145	-.005	-.277												
	.653	.095	.129	-.021	-.086	-.182												
	.690	-.027	-.023	.105	-.193	-.258												
	.708		.363	.373	-.736													
	.735		.174		.128													
	.768		-.241		-.313													
	.796		-.073		-.248													
	.821		.189		.264													
.852		.363		.424														
Plug	0.000	.417	.599	1.094	.712	.981												
	.003	-.969	-.915	.499	-.739	-.1071												
	.014	-.988	-.766	-.393	-.603	-.815												
	.031	-.145	-.204	-.522	-.499	-.389												
	.055	-.337	-.110	-.364	-.405	-.378												
	.106	-.039	-.283	-.270	-.224	-.222												
	.172	-.105	.176	-.089	-.086	-.041												
	.261	-.211	.237	-.167	-.151	-.057												
	.302	-.090	.042	.205	.032	.040												
	.326	-.039	.045	.058	.110	.084												
	.343	.326	.245	.066	.283	.570												
	.387	-.772	.058	.108	.074	-.344												

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a

(a) $M = 0.700$; inboard station

x/c	C _p at -																
	α = -2°						α = 0°										
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row K	Row L
Fan cowl	0.000					.692	.426	-.082	.304	1.021	.827	.425	-.164	.296	.949		
	.003					-.177	-.710	-1.436	-.513	-.014	-.349	-.685	-.189	.237	.889		
	.014					-.269	-.495	-.668	-.283	-.190	-.354	-.469	-.305	.423	.788		
	.031					-.245	1.105	-.819	1.124	.286	-.277	1.105	-.842	1.124	.288		
	.055					-.264	-.316	-.386	.231	.187	-.277	-.296	-.199	.199	.199		
	.106					-.167	-.251	.281	.152	.104	-.165	-.228	-.119	.119	.119		
	.172					-.056	1.08	.165	.036	.293	-.044	.122	.173	.000	.291		
	.261					-.158	-.232	.158	.116	.036	-.146	-.206	-.138	.064	.067		
	.302					-.129	-.069	.069	.031	.031	-.102	-.039	.031	.310	.086		
	.326					-.119	.055	.095	.258	.047	-.102	.029	.051	.265	.086		
	.343					-.177	.159	-.216	.117	-.430	-.461	.205	-.157	.157	.413		
	.387					.057	.076	.037	.136	.532	.534	.131	.077	.150	.560		
Turbine cowl	.442					.017	.024	-.037	.190	.165	.063	-.026	.003	.116	.275		
	.495					-.216	.024	-.004	-.260	.052	-.180	.077	.032	.240	.093		
	.537					-.254	.034	-.011	-.129	.196	-.165	.090	.025	.035	.064		
	.581					-.129	-.056	-.107	-.180	.021	-.092	.048	-.052	.228	.006		
	.617					.123	.028	-.020	.113	.021	.063	.096	.025	.096	.041		
	.653					.244	.060	.015	-.084	.005	.257	.077	.038	.138	.025		
	.690					.143	.149	-.024	-.292	-.123	.145	.128	-.020	.025	.096		
	.708						.005	.297	.091			.090	.321	.061	.419		
	.735						-.024		-.557			-.013		-.468			
	.768						.418		-.593			-.225		-.381			
	.796						-.210		-.180			-.161		-.128			
	.821						.018		.056			.064		.115			
.852						.307		.376			.337		.419				
Fan cowl	0.000					.870	.410	.101	.317	.690							
	.003					-.333	-.688	-.110	-.186	-.622							
	.014					-.376	-.441	-.267	-.109	-.423							
	.031					.286	1.103	.563	1.128	.317							
	.055					-.211	-.261	-.184	-.122	-.228							
	.106					-.099	-.193	.312	-.048	-.090							
	.172					.032	.151	.206	.073	.320							
	.261					.292	.016	-.074	.035	.073							
	.302					.096	-.000	.025	.333	.154							
	.326					.263	.067	.321	.288	.176							
	.343					.183	-.445	.189	.224	.375							
	.387					.170	.577	.516	.230	.596							
Turbine cowl	.442					.375	.051	.138	.625	.566							
	.495					.135	.120	.154	.115	.778							
	.537					.106	.164	.109	.208	.702							
	.581					-.057	.215	.154	.105	.269							
	.617					.042	.218	.122	.000	.272							
	.653					-.045	.312	.154	.073	.202							
	.690					-.089	.259	.344	-.010	.192							
	.708					.100	.206	.424	.179								
	.735					.357	.022		.353								
	.768					-.003	-.016		-.353								
	.796					-.144	-.007		-.119								
	.821					-.103	.193		-.039								
.852					.116	.421		.253									
Plug	0.000																
	.003																
	.014																
	.031																
	.055																
	.106																
	.172																
	.261																
	.302																
	.326																
	.343																
	.387																

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued
(b) $M = 0.700$; outboard station

x/c	C_p at -														
	$\alpha = -2^\circ$					$\alpha = 0^\circ$					$\alpha = 1^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
0.000															
0.003															
0.014															
0.031															
0.055															
0.106															
0.172															
0.261															
0.302															
0.326															
0.343															
0.387															
0.442															
0.493															
0.537															
0.581															
0.617															
0.653															
0.690															
0.708															
0.735															
0.768															
0.796															
0.821															
0.852															
Fan cowl															
Turbine cowl															
Turbine cowl															
Plug															
Plug															
0.000															
0.003															
0.014															
0.031															
0.055															
0.106															
0.172															
0.261															
0.302															
0.326															
0.343															
0.387															
0.442															
0.493															
0.537															
0.581															
0.617															
0.653															
0.690															
0.708															
0.735															
0.768															
0.796															
0.821															
0.852															
Fan cowl															
Turbine cowl															
Turbine cowl															
Plug															
Plug															

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(c) $M = 0.750$; inboard station

x/c	C _p at -															
	α = -2°					α = 0°					α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	.000					.900	.402	.023	.291	.992	.835	.390	-.003	.298	.926	
	.003					.237	.623	-.1433	.520	.079	.652	.849	-.159	.458	.233	
	.014					.286	.536	-.856	.304	.207	.390	.533	.003	.274	.309	
	.031					.259	1.128	-.680	1.147	.277	.328	1.124	-.653	1.138	.274	
	.055					.264	.322	-.393	.149	.187	.284	.310	-.345	.213	.213	
	.106					.162	.249	.278	.149	.097	.165	.237	.275	.125	.102	
	.172					.034	.108	.161	.102	.286	.037	.117	.164	.000	.280	
	.261					.144	.226	.149	.102	.021	.138	.208	.137	.070	.003	
	.302					.109	.056	-.035	.298	.052	.098	.038	.026	.292	.079	
	.343					.096	.018	.249	.260	.073	.090	.035	.170	.234	.099	
Turbine cowl	.387					.321	.138	.226	.172	.283	.333	.161	.225	.187	.274	
	.442					.276	.020	.056	.064	.493	.326	.038	.076	.161	.537	
	.493					.038	.067	.026	-.006	.105	.023	-.009	.000	.052	.201	
	.537					.162	.053	.038	.111	.155	.125	.050	.056	.131	.187	
	.581					.038	.097	-.003	.029	.193	.023	.129	.027	.087	.120	
	.617					.134	.009	.129	.044	.207	.047	.094	.006	.157	.029	
	.653					.086	.108	.065	.248	.088	.087	.094	.006	.146	.000	
	.690					.258	.067	.044	.161	.117	.246	.106	.027	.201	.023	
	.708					.094	.091	.167	.283	.102	.154	.123	.094	.201	.023	
	.735						.029	.290	.012			.097	.334	.347		
Plug	.735						.018	.290	.383			.014		.376		
	.768						.475		.494			.413		.441		
	.796						.220		.581			.135		.079		
	.821						.070		.166			.067		.155		
	.852						.287		.368			.319		.400		
	Fan cowl	.000					.598	.349	.355	.316	.670					
		.003					-.171	.668	.205	.203	.740					
		.014					.559	.484	.302	.134	.497					
		.031					.226	1.130	.293	1.144	.510					
		.055					.320	.276	.196	.123	.317					
.106						.092	.147	.311	.038	.076						
.172						.285	.153	.203	.088	.316						
.261						.034	.072	.073	.041	.094						
.302						.116	.032	.032	.328	.179						
.326						.265	.142	.496	.287	.208						
Turbine cowl	.343					.221	.270	.097	.263	.234						
	.387					.258	.429	.144	.272	.643						
	.442					.119	.056	.082	.760	.424						
	.493					.279	.126	.120	.863	.831						
	.537					.238	.144	.118	.228	.260						
	.581					.040	.402	.050	.059	.404						
	.617					.136	.132	.129	.123	.246						
	.653					.078	.247	.129	.123	.179						
	.690					.051	.300	.220	.032	.331						
	.708					.037	.242	.255	.070	.331						
Plug	.708					.154	.220	.417	.170	.331						
	.735					.365										
	.768					.288										
	.796					.416										
	.821					.288										
	.852					.001										
						.198										
						.098										
						.107										
						.351										

(d) $M = 0.750$; outboard station

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TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(e) $M = 0.775$; inboard station

x/c	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.902	.424	.260	.298	1.002
	.003					-.234	-.862	1.395	-.461	-.091
	.014					-.268	-.541	-.309	-.242	-.242
	.031					-.264	1.136	-.431	1.157	.280
	.055					-.264	-.264	-.264	-.264	-.264
	.066					-.158	-.263	.278	-.144	-.093
	.106					-.026	.112	.162	.312	.285
	.172					-.181	-.220	.164	-.102	-.009
	.261					-.103	-.031	-.029	.296	.084
	.302					-.086	.027	.413	.260	.086
	.326					-.264	.152	.181	.207	-.217
	.343					-.018	.039	.075	.075	.075
	.387					-.111	-.110	-.063	-.057	.055
	.442					.135	.131	-.015	-.113	.344
	.493					.271	.020	-.079	-.034	-.183
Turbine cowl	.537					.084	.216	.068	-.301	.327
	.581					.246	-.029	.039	-.074	.110
	.617					.084	.146	.356	-.225	-.163
	.653						-.004	.297	.047	.249
	.690						-.012		-.326	
	.708						-.431		-.393	
	.735						-.594		-.677	
	.768						.117		.156	
	.796						.289		.395	
	.821									
	.852									
Plug	0.020					.625	.393	.638	.330	.701
	.003					-.1.161	-.892	-.310	-.234	-.839
	.014					-.334	-.421	-.330	-.438	-.438
	.031					-.286	1.161	.027	1.159	.310
	.055					-.202	-.276	-.181	-.124	-.219
	.106					-.081	-.184	.311	-.035	-.068
	.172					.289	-.011	.202	.094	.313
	.261					-.017	.045	-.068	.047	.106
	.302					.129	.027	.036	.321	.190
	.326					.267	.103	.646	.282	.215
	.343					-.193	.275	.328	.279	-.172
	.387					.116	.117	.157	.313	.641
	.442					.081	.072	.086	.854	.389
	.493					.127	.306	.126	.890	.907
	.537					.056	.196	.134	.184	.226
Fan cowl	.581					.499	.145	.038	-.032	.457
	.617					.084	.233	.134	.249	.229
	.653					.265	.241	.143	-.015	.164
	.690					.042	.272	.685	.083	.403
	.708					.121	.216	.424	.162	
	.735					.095	-.022		-.193	
	.768					-.227	-.046		-.245	
	.796					-.101	-.365		-.006	
	.821					-.208	-.171		.229	
	.852					.440	.393		.504	
Turbine cowl	0.020					.864	.393	.638	.330	.701
	.003					-.334	-.421	-.330	-.438	-.438
	.014					-.286	1.161	.027	1.159	.310
	.031					-.202	-.276	-.181	-.124	-.219
	.055					-.081	-.184	.311	-.035	-.068
	.106					.289	-.011	.202	.094	.313
	.172					-.017	.045	-.068	.047	.106
	.261					.129	.027	.036	.321	.190
	.302					.267	.103	.646	.282	.215
	.326					-.193	.275	.328	.279	-.172
	.343					.116	.117	.157	.313	.641
	.387					.081	.072	.086	.854	.389
	.442					.127	.306	.126	.890	.907
	.493					.056	.196	.134	.184	.226
	.537					.499	.145	.038	-.032	.457
Plug	.581					.084	.233	.134	.249	.229
	.617					.265	.241	.143	-.015	.164
	.653					.042	.272	.685	.083	.403
	.690					.121	.216	.424	.162	
	.708					.095	-.022		-.193	
	.735					-.227	-.046		-.245	
	.768					-.101	-.365		-.006	
	.796					-.208	-.171		.229	
	.821					.440	.393		.504	
	.852									

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(f) $M = 0.775$; outboard station

C _p at -															
x/c	α = -2°					α = 0°					α = 1°				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.814	.868	.766	.658	1.166	.685	.861	.861	.961	1.112
	.003					-.251	-.982	.544	-.754	-.421	-.439	-.854	-.854	.418	-.687
	.014					-.378	-.641	-.132	-.605	-.441	-.467	-.636	-.636	-.704	-.546
	.031					-.344	-.528	-.480	-.579	-.342	-.438	-.519	-.519	-.447	-.399
	.055					-.323	-.145	-.451	-.430	-.351	-.353	-.426	-.426	-.396	-.368
	.106					-.052	-.304	-.315	-.297	-.286	-.066	-.288	-.288	-.255	-.253
	.172					-.081	.065	-.137	-.173	-.066	-.078	-.140	-.140	-.182	-.069
	.261					.262	.331	.182	-.198	-.133	-.116	-.228	-.228	-.196	-.100
	.302					-.128	-.080	-.093	-.026	-.032	-.116	-.058	-.052	.013	-.001
	.326					-.094	.056	.031	.008	-.001	-.086	.081	.050	.120	.035
	.343					-.467	.133	-.026	.186	-.040	-.472	.140	-.018	.193	.027
	.387					.135	.055	.028	.047	.101	.244	.055	.075	.103	.162
	.442					-.204	.125	-.035	-.040	.056	.251	.002	.061	.052	.059
	.493					-.262	.141	.018	-.191	.163	.210	.115	.041	.159	.252
	Turbine cowl	.537					-.064	-.032	-.022	-.035	.376	.041	-.030	.004	.058
.581						.071	-.057	-.100	-.035	.376	.041	-.030	.004	.058	.013
.617						.080	.223	.033	.193	.368	.172	.152	.067	.196	-.027
.653						.046	.057	.039	.116	-.291	.164	.118	.086	.165	-.160
.690						.012	.009	.113	.170	-.416	.147	.004	.135	.238	-.182
.708							.314	.340	.788			.390	.384	.859	
.735							.084		.008			.106		.029	
.768							.341		.478			.273		.416	
.796							-.210		.396			.055		.128	
Fan cowl	0.000														
	.003														
	.014														
	.031														
	.055														
	.106														
	.172														
	.261														
	.302														
	.326														
	.343														
	.387														
	.442														
	.493														
Turbine cowl	0.000														
	.003														
	.014														
	.031														
	.055														
	.106														
	.172														
	.261														
	.302														
	.326														
	.343														
	.387														
	.442														
	.493														
Plug	0.000														
	.003														
	.014														
	.031														
	.055														
	.106														
	.172														
	.261														
	.302														
	.326														
	.343														
	.387														
	.442														
	.493														

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Continued

(g) $M = 0.800$; inboard station

x/c	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					0.000				
	.003					.003				
	.014					.014				
	.031					.031				
	.055					.055				
	.106					.106				
	.172					.172				
	.261					.261				
	.302					.302				
	.326					.326				
	.343					.343				
	.357					.357				
	.363					.363				
	.368					.368				
	.373					.373				
Turbine cowl	0.000					0.000				
	.003					.003				
	.014					.014				
	.031					.031				
	.055					.055				
	.106					.106				
	.172					.172				
	.261					.261				
	.302					.302				
	.326					.326				
	.343					.343				
	.357					.357				
	.363					.363				
	.368					.368				
	.373					.373				
Plug	0.000					0.000				
	.003					.003				
	.014					.014				
	.031					.031				
	.055					.055				
	.106					.106				
	.172					.172				
	.261					.261				
	.302					.302				
	.326					.326				
	.343					.343				
	.357					.357				
	.363					.363				
	.368					.368				
	.373					.373				

TABLE 37.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5a - Concluded

(h) $M = 0.800$; outboard station

$x/2$	C_p at									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.806	.851	.767	.669	1.197
	.003	.591	.851	.916	1.084	.459	.844	1.086	.792	.988
	.014	.750	.932	.974	.670	-1.008	.957	.876	.496	.983
	.031	.558	.843	.873	.650	.910	.954	.837	.450	.853
	.055	.375	.640	.676	.495	.445	.805	.730	.406	.458
	.106	.163	.273	.304	.393	.372	.675	.590	.346	.376
	.172	.057	.220	.270	.283	.024	.244	.208	.194	.227
	.261	.277	.259	.193	.132	.008	.084	.047	.031	.015
	.302	.086	.022	.022	.043	.605	.761	.113	.086	.029
	.343	.057	.084	.071	.081	.049	.004	.009	.097	.100
	.387	.395	.180	.327	.239	.028	.117	.097	.040	.132
Turbine cowl	0.000					.459	.844	1.086	.792	.988
	.003	.591	.851	.916	1.084	.459	.844	1.086	.792	.988
	.014	.750	.932	.974	.670	-1.008	.957	.876	.496	.983
	.031	.558	.843	.873	.650	.910	.954	.837	.450	.853
	.055	.375	.640	.676	.495	.445	.805	.730	.406	.458
	.106	.163	.273	.304	.393	.372	.675	.590	.346	.376
	.172	.057	.220	.270	.283	.024	.244	.208	.194	.227
	.261	.277	.259	.193	.132	.008	.084	.047	.031	.015
	.302	.086	.022	.022	.043	.605	.761	.113	.086	.029
	.343	.057	.084	.071	.081	.049	.004	.009	.097	.100
	.387	.395	.180	.327	.239	.028	.117	.097	.040	.132
	.442	.442	.076	.006	.021	.195	.248	.018	.241	.024
Plug	0.000					.459	.844	1.086	.792	.988
	.003	.591	.851	.916	1.084	.459	.844	1.086	.792	.988
	.014	.750	.932	.974	.670	-1.008	.957	.876	.496	.983
	.031	.558	.843	.873	.650	.910	.954	.837	.450	.853
	.055	.375	.640	.676	.495	.445	.805	.730	.406	.458
	.106	.163	.273	.304	.393	.372	.675	.590	.346	.376
	.172	.057	.220	.270	.283	.024	.244	.208	.194	.227
	.261	.277	.259	.193	.132	.008	.084	.047	.031	.015
	.302	.086	.022	.022	.043	.605	.761	.113	.086	.029
	.343	.057	.084	.071	.081	.049	.004	.009	.097	.100
	.387	.395	.180	.327	.239	.028	.117	.097	.040	.132
	.442	.442	.076	.006	.021	.195	.248	.018	.241	.024

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b

(a) $M = 0.700$; Inboard station

x/c	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.732	.451	.900	.878	.878	-1.137	.604	.317	1.042	.913	.676	.511	.001	1.100	.920	.590	
	.003	-.557	-.588	-.938	.013	-.361	-.693	-.256	.020	-.742	-.1360	-.668	-.003	-.668	-.003	.045	-1.076	
	.014	-.446	-.358	-.505	-.274	-.359	-.603	-.365	-.289	-.482	-.651	-.822	-.224	-.822	-.224	.188	-.615	
	.031	-.325	-.503	-.569	-.297	-.274	-.414	-.391	-.901	-.360	-.452	-.581	-.906	-.581	-.906	.045	-.355	
	.055	-.180	-.331	-.116	-.362	-.268	-.356	-.166	-.223	-.284	-.370	-.356	-.212	-.356	-.212	.195	-.317	
	.106	-.180	-.240	-.272	-.182	-.153	-.196	-.230	-.211	-.149	-.190	-.231	-.173	-.190	-.231	.134	-.144	
	.172	.030	-.096	-.128	-.335	-.041	-.026	-.130	-.082	.024	-.020	-.138	-.054	-.124	-.138	.124	-.083	
	.261	.202	-.253	-.208	-.159	-.044	-.036	-.230	-.172	.017	-.108	-.221	-.150	-.067	-.067	.067	-.003	
	.302	.410	-.073	-.095	-.038	.016	.139	-.050	-.063	.069	-.006	-.041	-.041	-.041	-.041	.029	.029	
	.326	.574	-.003	-.202	-.057	-.089	.318	.021	.056	.033	.125	.033	.003	.071	.122	.122	.001	
	.343	.572	.161	-.272	.096	-.060	-.531	.197	.236	.034	-.438	.174	-.192	.174	-.192	.167	.001	
	.387	.033	.004	.004	-.028	.055	.037	.037	.040	.046	.181	.096	.068	.071	.109	.109	.209	
	.442	.076	-.064	-.099	-.086	.013	.134	.015	.031	.027	.145	.164	.001	-.006	.116	.116	.222	
	.493	-.232	.023	.061	-.137	.003	-.215	.011	.002	.069	.107	.166	.075	.033	.145	.145	.209	
Turbine cowl	.537	-.138	-.067	-.073	-.156	-.460	-.147	.012	.027	.069	.031	.065	.023	.001	.001	.147	.105	
	.581	.134	-.080	-.188	-.138	-.421	-.069	-.111	.124	.194	.151	.074	.007	-.067	-.067	.166	-.147	
	.617	-.039	.026	-.099	-.332	-.399	-.016	.027	.012	.133	.156	.019	.052	.023	.023	.035	-.028	
	.653	.057	-.128	-.080	-.249	-.465	.110	.066	-.012	.114	.216	.145	.045	.033	.105	.105	-.102	
	.690	-.135	.036	-.048	-.278	.019	-.050	.062	.033	.268	-.032	.048	.132	.081	.028	.028	-.116	
	.708		-.052	-.234	-.093			.037	.307		.104		.328		.328	.116	.116	
	.735		-.028	.134	-.134			-.012		.124		-.003		.128		.128	.128	
	.768		-.487	-.606	-.134			-.342		.124		-.186		.355		.355	.355	
	.796		-.221	-.175	-.606			-.155		.133		-.144		.321		.321	.321	
	.821		.010	.055	-.175			-.050		.085		.051		.135		.135	.135	
	.852		.289	.313	.055			-.348		.405		.373		.449		.449	.449	
	Plug	0.000	-.439	-.107	1.121	.935	.511											
		.003	-.1551	-.673	.157	.083	-.1324											
		.014	-.632	-.827	-.113	.164	-.638											
.031		-.478	-.660	.718	-.299	.372												
.055		-.381	-.485	-.158	.130	.308												
.106		-.192	-.232	.145	.195	.158												
.172		-.017	-.133	.036	.135	.036												
.261		-.153	-.219	.036	.042	.034												
.302		-.030	-.062	.015	.089	.124												
.326		.017	.028	.017	.131	.147												
.343		-.386	.167	.132	.133	.007												
.387		.116	.067	.089	.185	.281												
.442		.152	.003	.015	.095	.256												
.493		-.133	.086	.051	.195	.291												
.537	-.128	.085	.041	.105	.036													
Fan cowl	.581	.075	.034	.039	.135	.036												
	.617	.042	.085	.041	.047	.054												
	.653	.158	.099	.060	.046	.033												
	.690	.090	.131	.102	.076	.103												
	.708		.118	.118	.118													
	.735		-.004	.334	.118													
	.768		-.057		.324													
	.796		.057		.116													
	.821		.118		.172													
	.852		.382		.457													

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(b) $M = 0.700$; outboard station

x/c	C _p at															
	α = -2°					α = 0°					α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.573	-.851	-.882	-.990	.254	.642	1.043	.595	-.850	.192	.931	1.195	-.680	-.827	
	.003	-.841	-.820	-.204	-.687	-.195	-.196	-.202	-.647	-1.027	-.049	-.074	.195	-.577	-1.322	
	.014	-.643	-.601	-.535	-.617	-.771	-.665	-.306	-.550	-.740	-.640	-.301	-.171	-.502	-.735	
	.031	-.1391	-.1604	-.442	-.475	-.805	-1.035	-.355	-.505	-.518	-.631	-.258	-.258	-.444	-.541	
	.055	-.349	-.355	-.400	-.424	-.399	-.322	-.309	-.434	-.460	-.417	-.237	-.237	-.464	-.438	
	.106	-.436	-.290	-.313	-.379	-.379	-.481	-.277	-.258	-.431	-.539	-.268	-.216	-.286	-.306	
	.172	-.088	.310	-.158	-.212	-.090	-.094	-.319	-.115	-.237	-.070	-.093	-.308	-.087	-.070	
	.261	-.139	.304	-.209	-.237	-.154	-.041	-.248	-.177	-.247	-.092	-.137	-.185	-.152	-.067	
	.302	-.078	-.093	-.012	-.125	-.090	-.084	-.070	-.041	-.124	-.021	-.103	-.061	-.048	-.017	
	.326	.356	.123	-.022	-.096	-.089	100	.131	-.008	-.095	.001	.004	.121	.033	.039	
Turbine cowl	.343	.482	.030	-.154	.039	.186	.511	.689	-.134	.056	.189	.528	.082	.072	-.170	
	.387	.320	-.012	-.009	.128	.193	.229	-.005	.034	.005	.379	.010	.062	.104	.510	
	.442	.103	.041	-.077	.170	.196	.128	-.018	-.037	.053	.063	.122	.004	.059	.030	
	.493	.016	.009	-.061	.000	-.073	.011	.047	-.011	.021	.018	.004	.027	.039	.046	
	.537	-.020	-.022	.029	.042	-.283	.047	.018	-.021	-.040	.067	.059	.056	.032	.104	
	.581	-.088	.077	.135	.353	-.440	-.002	-.028	-.066	.340	-.203	.033	.004	.219	.035	
	.617	-.156	.043	.061	.302	-.482	-.084	.053	.014	-.070	-.231	.066	.049	.035	.106	
	.653	-.151	-.090	-.051	.324	-.430	-.027	-.066	.002	.040	-.211	.031	.082	.123	-.054	
	.690	-.267	-.067	-.032	.363	-.427	-.132	-.005	.040	-.298	-.215	.069	.088	.222	-.228	
	.708	-.247	-.087	-.298	.517	.517	.361	.364		-.624		.386	.396	-.670		
Plug	.735	.285	-.045	.298	.186	.517	.361		-.118		.056		-.248			
	.766	-.481	-.062	-.186	-.662	-.187	-.008		-.579		.119		-.444			
	.796	-.138	-.163	-.062	-.163	-.173	-.096		-.086		-.032		-.077			
	.821	.030	-.062	-.062	-.062	-.095	-.095		-.111		-.127		-.159			
	.852	-.327	-.351	-.351	-.351	.348			-.424		.370		.449			
	Fan cowl	0.000	-.076	.920	1.201	.749	.811									
		.003	.007	-.030	.191	-.523	-.514									
		.014	-.525	-.127	-.069	-.462	-.714									
		.031	-.516	-.347	-.185	-.423	-.559									
		.055	-.433	-.286	-.198	-.443	-.449									
.106		-.535	-.270	-.185	-.314	-.281										
.172		-.099	.294	-.065	-.172	-.175										
.261		-.196	.152	-.133	-.185	-.036										
.302		-.152	-.062	-.062	-.019	-.038										
.326		-.041	.119	-.048	-.036	-.061										
Turbine cowl	.343	.497	.074	-.075	.074	-.165										
	.387	-.177	.016	.074	.157	.393										
	.442	-.059	-.007	.019	.028	.096										
	.493	.061	.035	.045	.128	.167										
	.537	.037	.077	.074	.035	.170										
	.581	.037	.045	-.003	-.049	.048										
	.617	-.007	.090	.067	.048	-.007										
	.653	.129	.093	.106	.033	.016										
	.690	.017	.055	.411	-.004	-.078										
	.708	.398	.398	.691	-.691											
Plug	.735	.067	.067	.146	-.146											
	.766	-.069	-.069	-.304	-.304											
	.796	-.004	-.004	-.068	-.068											
	.821	-.145	-.145	-.183	-.183											
	.852	-.372	-.372	.470	.470											

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(c) $M = 0.750$; inboard station

x/c	C _p at -																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																																				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued.

(d) $M = 0.750$; outboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.493	.803	.832	.571	.972	.351	.880	1.099	.647	.916	.894	1.178	.718	.873			
	.003	-.867	-.818	-.217	-.814	-.794	-.102	-.086	.192	-.777	-.127	-.054	.192	-.674	-.412			
	.014	-.606	-.1490	-.594	-.662	-.650	-.734	-.565	.366	-.607	-.845	-.201	.233	-.362	-.121			
	.031	-.1316	-.1361	-.523	-.480	-.506	-.615	-.890	.399	-.430	-.557	-.523	.310	-.403	-.950			
	.055	-.330	-.341	-.445	-.463	-.436	-.423	-.343	.340	-.418	-.457	-.445	.284	.400	.450			
	.106	-.245	-.308	-.331	-.332	-.324	-.266	-.293	.275	.312	-.292	-.339	.281	.285	.403			
	.172	-.082	.318	-.166	-.207	-.074	-.076	.310	.122	.242	-.133	-.074	.304	.095	.147			
	.261	.302	.439	-.222	-.255	-.130	.115	.375	.184	.277	-.080	.019	.301	.163	.185			
	.302	-.069	-.092	-.010	-.072	-.101	-.032	-.066	.014	.018	-.004	-.038	.035	.012	.035			
	.326	.538	.126	-.022	-.110	-.062	.264	.127	.014	.038	.023	.156	.121	.076	.056			
	.343	.553	.111	-.092	-.131	-.083	.534	.112	.080	.149	-.057	.526	.118	.068	.053			
	.367	-.368	-.027	-.016	-.007	.072	-.305	.051	.044	.032	.182	-.087	.041	.074	.285			
Turbine cowl	.442	-.113	-.154	-.073	-.151	.084	-.089	-.116	-.089	-.171	-.068	-.098	-.044	.038	.041			
	.493	.084	.052	-.060	-.039	-.118	.110	.032	.020	.129	.135	.130	.050	.129	.162			
	.537	-.077	-.104	-.080	-.256	-.204	.044	.001	.029	.242	-.101	.085	.098	.185	.071			
	.581	-.047	-.055	-.177	-.377	-.524	.013	.074	.065	.080	-.157	.046	.050	.224	.065			
	.617	-.179	-.032	-.004	-.336	-.811	-.045	.141	.029	.201	.348	.109	.021	.224	.206			
	.653	-.117	-.116	-.095	-.324	-.480	-.014	-.027	.002	.360	-.310	.072	.021	.230	.118			
	.690	-.320	-.166	-.033	.014	-.347	-.107	.002	.064	.136	-.289	-.056	.399	.224	.165			
	.708	.088	.341	.315	-.594	.322	.322	.345	.695	.396	.399	.767	.399	.767	.461			
	.735	-.019	-.019	.315	-.160	.150	.150	.350	.212	.212	.032	.071	.032	.500	.032			
	.768	-.458	-.458	.535	-.535	.370	.370	.535	.454	.454	-.218	.071	.032	.500	.032			
	.796	-.390	-.390	.538	-.538	.060	.060	.538	.068	.068	-.032	.071	.032	.500	.032			
	.821	.114	.114	.156	.156	.091	.091	.156	.185	.185	-.032	.071	.032	.500	.032			
.852	.318	.318	.357	.357	.448	.448	.357	.402	.402	.438	.438	.402	.438	.438				
Fan cowl	0.000	.152	.566	1.200	.739	.802												
	.003	.118	.098	.181	-.624	1.449												
	.014	-.443	.008	-.133	-.532	1.310												
	.031	-.341	-.233	-.224	-.382	-.512												
	.055	-.425	-.301	-.224	-.397	-.453												
	.106	-.341	-.277	-.203	-.267	-.456												
	.172	-.076	-.299	-.067	-.209	-.026												
	.261	-.045	.276	-.138	-.027	-.065												
	.302	-.058	-.053	-.020	-.074	.089												
	.326	.114	.125	.057	.074	.044												
	.343	.521	.128	-.059	.162	-.044												
	.387	-.231	.035	.085	.027	.368												
Turbine cowl	.442	.085	-.085	.030	.065	.018												
	.493	.105	.063	.072	.115	.195												
	.537	.110	.125	.057	.089	.171												
	.581	.057	.045	-.002	-.088	.030												
	.617	-.012	.085	.057	.039	-.061												
	.653	.122	.122	.060	.159	-.002												
	.690	.017	.066	.116	-.270	-.031												
	.708	.391	.391	.406	.003													
	.735	-.082	-.082	-.003	-.465													
	.768	-.068	-.068	.013	-.465													
	.796	-.821	-.821	.180	-.180													
	.852	.379	.379	.448	.448													
Plug	0.000	.152	.566	1.200	.739	.802												
	.003	.118	.098	.181	-.624	1.449												
	.014	-.443	.008	-.133	-.532	1.310												
	.031	-.341	-.233	-.224	-.382	-.512												
	.055	-.425	-.301	-.224	-.397	-.453												
	.106	-.341	-.277	-.203	-.267	-.456												
	.172	-.076	-.299	-.067	-.209	-.026												
	.261	-.045	.276	-.138	-.027	-.065												
	.302	-.058	-.053	-.020	-.074	.089												
	.326	.114	.125	.057	.074	.044												
	.343	.521	.128	-.059	.162	-.044												
	.387	-.231	.035	.085	.027	.368												

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(e) $M = 0.775$; inboard station

x/c	C_p at -															
	$\alpha = 1^\circ$															
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	.770	.456	.864	.839	.845	.625	.386	1.015	.864	.556	.179	1.056	.895	.606	
	.003	-.756	-.903	-.105	.025	-.376	-.1208	-.880	-.495	.048	-.1.011	-.919	-.146	.076	-.1.275	
	.014	.501	.436	-.432	.300	.443	-.614	-.182	-.495	.283	-.554	-.939	-.303	-.221	-.549	
	.031	-.374	.459	.115	.320	.286	-.448	-.053	-.559	.280	-.347	-.216	-.590	-.246	-.367	
	.055	-.323	.438	.371	.372	.258	-.363	.113	-.295	.277	-.366	-.053	-.227	-.207	-.597	
	.106	-.183	.234	.274	.171	.143	-.189	-.233	-.227	.134	-.183	.222	-.185	-.103	-.134	
	.172	-.089	.093	.113	.177	.065	-.002	.070	-.075	.048	.072	.004	-.084	-.042	.045	
	.261	.633	.258	-.205	.140	.047	.261	-.222	-.177	.078	.033	.157	-.025	-.041	.057	
	.302	.858	.056	-.082	.003	.047	.448	-.039	-.056	.054	.056	.267	-.025	.090	.135	
	.326	.802	.022	.189	.051	.033	.567	.040	.327	.107	.356	.054	.234	.138	.158	
Turbine cowl	.343	.446	.180	.185	.190	.017	.410	.220	.199	.216	.370	.243	.199	.236	.068	
	.387	.021	.631	.036	.053	.003	.006	.051	.060	.110	.018	.071	.085	.127	.191	
	.442	.085	-.132	-.051	.053	.134	-.053	.105	-.073	.039	.082	-.070	.011	.040	.152	
	.493	.157	.180	.064	.073	.047	.053	.105	-.019	.048	.222	-.018	.046	.045	.250	
	.537	.239	.086	.118	.233	.421	.172	.047	-.070	.070	.254	.101	.046	.040	.154	
	.581	.446	.008	-.222	.317	.160	.261	.009	-.134	-.002	.113	.152	.059	.012	.138	
	.617	-.060	.144	.023	.348	.479	.015	.144	.020	-.297	.182	.059	.184	.003	.047	
	.653	.034	.003	.197	.325	.479	.113	.619	.043	.058	.319	.144	.005	-.069	.156	
	.690	-.234	.175	.056	.463	.305	-.053	.657	.009	.204	.046	.169	.085	-.187	.099	
	.708	.087	.174	.028	.028	.028	.028	.065	.304	.043	.046	.169	.085	-.187	.099	
Plug	.735	.009	.174	.028	.028	.028	.028	.065	.304	.043	.046	.169	.085	-.187	.099	
	.768	.380	.009	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	
	.796	.627	.009	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	
	.821	.665	.009	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	
	.852	.278	.009	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	.093	
	Fan cowl	0.000	.499	.035	1.134	.908	.543									
		.003	-.895	.058	.058	.086	1.376									
		.014	-.1.219	.457	-.172	-.137	-.682									
		.031	-.356	.219	-.462	.222	-.368									
		.055	-.356	.186	-.170	.186	-.292									
.106		-.165	.217	-.147	.088	-.119										
.172		.018	.043	.020	.117	.069										
.261		.124	.152	.125	.015	.055										
.302		.136	.012	.012	.117	.167										
.326		.251	.064	.219	.159	.193										
Turbine cowl	.343	.348	.252	.189	.246	.092										
	.387	.003	.081	.120	.148	.266										
	.442	.188	.046	.033	.117	.238										
	.493	.030	.061	.109	.097	.280										
	.537	.128	.055	.075	.067	-.020										
	.581	.149	.011	-.060	.016	.196										
	.617	.034	.143	.022	.076	.038										
	.653	.175	.117	.039	.074	-.062										
	.690	.030	.120	.106	.096	.114										
	.708	.157	.157	.354	.139											
Plug	.735	.008	.008	.008	.183											
	.768	.251	.008	.008	.350											
	.796	.077	.008	.008	.029											
	.821	.124	.008	.008	.210											
	.852	.371	.008	.008	.448											

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(f) $M = 0.775$; outboard station

x/c	C _p at -																
	α = -2°						α = 0°						α = 1°				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L		
Fan cowl	0.000	.492	.900	.834	.570	.983	.286	.900	.997	.563	.855	.299	.875	1.173	.685	.868	
	.003	-.793	-.737	.161	-.675	-.799	-.074	-.032	.188	-.798	-1.221	.118	.117	.186	-.708	-1.368	
	.014	-.627	1.390	-.398	-.674	-.657	.722	.586	-.396	-.637	-.514	-.543	-.175	-.260	-.595	-1.095	
	.031	-1.313	-1.233	-.584	-.457	-.525	-.511	-.872	-.393	-.459	-.535	-.344	-.498	-.357	-.417	-.437	
	.055	-.394	-.366	-.453	-.463	-.443	-.430	-.362	-.362	-.431	-.471	-.416	-.334	-.294	-.409	-.451	
	.106	-.039	-.309	-.337	-.327	-.308	-.300	-.244	-.244	-.332	-.349	-.196	-.288	-.246	-.299	-.324	
	.172	-.088	.314	-.162	-.169	-.076	-.066	.307	.307	-.160	-.109	-.060	.307	-.093	-.174	-.030	
	.261	.457	.541	-.224	-.260	.121	.244	.465	.465	-.197	-.076	.131	.386	-.164	-.180	-.039	
	.302	-.031	-.682	-.309	-.243	-.102	-.015	-.064	-.064	-.009	-.002	-.005	-.053	.001	.015	.037	
	.326	.767	.127	-.014	-.034	.020	.388	.125	.125	.018	.116	.029	.258	.131	.040	.176	
Turbine cowl	.343	.576	.150	.003	.180	.026	.544	.148	-.039	.198	.011	.542	.157	.033	.198	.002	
	.387	.424	.014	-.003	-.026	.042	-.544	-.026	.026	.066	.142	-.391	.018	.057	.108	.215	
	.442	-.131	-.679	.065	-.184	.107	-.146	.113	-.045	-.039	.074	.136	-.090	-.013	-.050	.001	
	.493	.035	.184	.031	-.158	.155	.112	.134	.021	.085	.119	.152	.120	-.002	.204	.218	
	.537	-.030	-.145	-.133	.175	.121	-.083	-.036	.012	-.389	-.028	-.063	.004	-.004	-.239	.077	
	.581	-.000	.042	-.240	-.477	-.437	.061	-.070	.191	.083	-.183	-.366	.004	-.053	.004	-.039	
	.617	-.225	-.156	-.023	-.384	-.790	-.070	.002	.064	-.005	-.234	-.346	.054	.018	-.191	-.273	
	.653	-.115	.051	-.133	-.364	-.872	.002	.095	-.086	.018	-.197	-.273	-.039	.046	.117	-.183	
	.690	-.311	-.264	-.065	-.494	-.389	-.095		.310	.366	-.837	-.273	-.039	.046	.117	-.191	
	.708		.278	.303	-.638												
Plug	.735		.679		.000			.049		-.019			.358	.381	-.877	-.191	
	.768		-.439		.516			-.291		-.471			-.300		-.005	-.005	
	.796		-.564		-.669			-.288		-.406			-.016		-.078	-.078	
	.821		.147		.144			.174		-.226			.126		.232	.232	
	.852		.340		.372			.361		.413			.358		.433	.433	
	Fan cowl	0.000	.177	.854	1.201	.705	.921										
		.003	-.191	.178	.201	-.653	1.371										
		.014	-.474	.036	-.125	-.548	-1.203										
		.031	-.237	-.305	-.236	.407	.478										
		.055	-.339	-.307	-.233	.339	.435										
.106		-.203	-.279	.205	.283	.265											
.172		-.054	.068	.063	.133	.015											
.241		.077	.357	.137	.195	-.004											
.302		-.008	-.440	.022	.039	.073											
.326		.217	.135	.665	.155	.101											
Turbine cowl	.353	.555	.165	-.026	.203	.068											
	.387	-.239	.022	.093	.092	.290											
	.442	-.057	-.077	-.015	.008	.011											
	.493	.170	.113	.084	.155	.242											
	.537	.089	.033	.124	.125	.129											
	.581	.073	.035	.104	-.056	.025											
	.517	.035	.164	.062	-.057	-.046											
	.653	.115	.127	.056	.028	-.012											
	.690	-.004	.022	.084	-.238	-.133											
	.708		.610	.408	-1.139												
Plug	.735		.110		.053												
	.768		-.114		-.435												
	.796		.002		.033												
	.821		.158		.205												
	.852		.376		.445												

TABLE 38.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Continued

(g) $M = 0.800$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.759	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		
Turbine cowl	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.749	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		
Plug	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.749	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		
Fan cowl	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.749	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		
Turbine cowl	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.749	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		
Plug	0.000	.779	.465	-.865	-.819	-.851	-.645	-.415	-.995	-.862	-.693	-.585	-.198	1.086	-.893	-.627		
	-.003	-.749	-.934	-.1126	-.018	-.450	-.1216	-.551	-.514	-.049	-.1045	-.1322	-.316	-.200	-.585	-.1279		
	-.014	-.525	-.876	-.805	-.313	-.401	-.849	-.128	-.478	-.280	-.573	-.1081	-.216	-.311	-.585	-.619		
	-.031	-.334	-.330	-.356	-.334	-.286	-.420	-.428	-.156	-.285	-.355	-.302	-.111	-.251	-.531	-.556		
	-.055	-.233	-.643	-.381	-.283	-.280	-.353	-.209	-.278	-.253	-.293	-.355	-.040	-.227	-.242	-.594		
	-.106	-.186	-.237	-.278	-.173	-.140	-.179	-.221	-.221	-.129	-.134	-.171	-.216	-.186	-.102	-.121		
	-.172	-.081	-.054	-.108	-.076	-.021	-.013	-.032	-.062	-.135	-.055	-.016	-.032	-.041	-.175	-.067		
	-.261	-.258	-.241	-.203	-.132	-.085	-.070	-.210	-.069	-.044	-.044	-.331	-.200	-.149	-.035	-.075		
	-.302	-.685	-.052	-.071	-.076	-.085	-.572	-.026	-.043	-.071	-.114	-.347	-.019	-.027	-.100	-.145		
	-.326	-.858	-.032	-.658	-.053	-.083	-.634	-.435	-.479	-.122	-.135	-.433	-.062	-.351	-.148	-.170		
	-.387	-.390	-.251	-.123	-.228	-.042	-.354	-.244	-.137	-.254	-.339	-.339	-.262	-.149	-.258	-.091		
	-.442	-.078	-.052	-.072	-.036	-.066	-.062	-.062	-.084	-.135	-.108	-.016	-.078	-.108	-.167	-.172		

TABLE 38. - FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 5b - Concluded

(h) $M = 0.800$; outboard station

x/c	C_p at											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	.432	.787	.839	.592	.982	.311	.767	1.001	.591	.864	.891
	.003	-.784	-.838	.092	-.940	-.877	-.032	.020	.192	-.813	-.152	-.318
	.014	-.613	1.335	-.688	-.704	-.677	-.757	-.618	-.411	-.667	-.911	-.1087
	.031	1.235	1.137	-.677	-.596	-.593	-.378	-.841	-.435	-.504	-.585	-.789
	.055	-.418	-.386	-.492	-.488	-.463	-.411	-.364	-.378	-.444	-.469	-.439
	.106	.107	-.324	-.357	-.379	-.309	-.119	-.302	-.296	-.290	-.401	-.344
	.172	-.052	.323	-.164	-.231	-.028	-.044	-.311	-.122	-.178	-.401	-.295
	.261	.647	.660	-.234	-.247	-.111	-.416	.576	-.193	-.186	-.064	-.034
	.302	-.011	-.085	.024	-.065	-.057	.005	-.056	.039	-.005	.012	-.035
	.326	1.017	.125	-.009	.021	-.011	.551	.129	.028	.267	.041	-.049
	.343	.556	.201	.008	.232	.027	.547	.194	.009	.239	.039	.073
	.387	-.568	.000	.087	-.049	-.068	-.610	.050	.050	.055	.120	.038
	.442	.033	-.000	.036	-.000	.111	-.101	.004	.009	-.132	.204	.138
	.493	-.044	-.071	.223	.195	-.187	.058	.222	.113	-.078	.033	.184
Turbine cowl	.537	.005	-.098	.055	.043	-.190	.050	-.089	.078	-.097	.041	.088
	.581	-.044	.065	-.084	-.358	.388	.078	.110	.081	-.406	-.404	-.184
	.617	-.109	-.066	-.147	-.490	-.661	.113	-.056	.069	-.214	-.480	-.127
	.653	-.198	-.115	-.109	-.469	-.834	.025	.080	.067	-.132	-.306	-.171
	.690	-.341	-.180	-.131	-.504	-.1045	-.134	.138	.085	-.178	-.292	-.155
	.708		.261	.228	-.750			.377	.380	-.840		
	.735		.002	-.214	-.214			.107		.009		
	.768		-.381	-.233	-.233			-.348		-.357		
	.796		-.639	-.515	-.515			-.307		-.580		
	.821		.081	-.000	-.000			.159		.208		
	.852		.356	.357	.357			.341		.448		
Plug	0.000	.213	.858	1.191	.702	.838						
	.003	.274	.244	.190	-.687	1.315						
	.014	-.488	.057	-.167	-.592	-.1024						
	.031	-.113	-.191	-.279	-.435	-.395						
	.055	-.301	-.528	-.285	-.518	-.324						
	.106	-.044	-.294	-.074	-.299	-.020						
	.172	-.044	.294	-.074	-.117	-.011						
	.261	.025	-.041	.040	.037	.078						
	.302	.323	.117	.057	.352	.100						
	.343	.518	.153	.010	.241	.037						
	.387	-.472	.073	.065	.135	.233						
	.442	.146	-.055	.005	.026	.097						
	.493	.147	.247	.089	.249	.260						
	.537	-.020	-.033	.068	-.155	.132						
	.581	.131	.027	-.052	.102	.007						
Turbine cowl	.617	.005	.239	.130	-.005	-.098						
	.653	.082	-.001	.057	-.180	-.093						
	.690	-.011	.160	.138	-.074	-.090						
	.708		.340	.376	-.184							
	.735		.119		-.028							
	.768		-.251		-.370							
	.796		.008		-.315							
	.821		.145		-.252							
	.852		.356		.436							
Plug	0.000	.213	.858	1.191	.702	.838						
	.003	.274	.244	.190	-.687	1.315						
	.014	-.488	.057	-.167	-.592	-.1024						
	.031	-.113	-.191	-.279	-.435	-.395						
	.055	-.301	-.528	-.285	-.518	-.324						
	.106	-.044	-.294	-.074	-.299	-.020						
	.172	-.044	.294	-.074	-.117	-.011						
	.261	.025	-.041	.040	.037	.078						
	.302	.323	.117	.057	.352	.100						
	.343	.518	.153	.010	.241	.037						
	.387	-.472	.073	.065	.135	.233						
	.442	.146	-.055	.005	.026	.097						
	.493	.147	.247	.089	.249	.260						
	.537	-.020	-.033	.068	-.155	.132						
	.581	.131	.027	-.052	.102	.007						
	.617	.005	.239	.130	-.005	-.098						
	.653	.082	-.001	.057	-.180	-.093						
	.690	-.011	.160	.138	-.074	-.090						
	.708		.340	.376	-.184							
	.735		.119		-.028							
	.768		-.251		-.370							
	.796		.008		-.315							
	.821		.145		-.252							
	.852		.356		.436							

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6

(a) $M = 0.700$; inboard station

C_p at -											
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L		Row H	Row I	Row J	Row K	Row L
$\alpha = 10^\circ$											
Fan cowl	0.000						.741	.433	.899	.103	.827
	.003						.575	.681	.726	.387	.407
	.014						.559	.488	.549	.282	.407
	.031						.343	.607	.835	.123	.419
	.055						.309	.321	.322	.227	.566
	.106						.178	.234	.160	.144	.256
	.172						.027	.058	.087	.131	.118
	.261						.005	.218	.177	.109	.107
	.302						.086	.048	.141	.186	.004
	.326						.196	.022	.046	.130	.081
Turbine cowl	.343						.196	.209	.222	.131	.148
	.387						.191	.022	.032	.137	.151
	.442						.173	.003	.010	.038	.184
	.493						.153	.007	.016	.052	.180
	.537						.037	.061	.115	.112	.147
	.581						.042	.048	.019	.163	.071
	.617						.196	.013	.013	.087	.049
	.653						.076	.003	.022	.253	.057
	.708							.022	.279	.089	.121
	.735							.019		.096	
Plug	.768							.238		.525	
	.786							.190		.138	
	.821							.054		.086	
	.852							.334		.393	
$\alpha = 20^\circ$											
Fan cowl	0.000						.647	.403	1.114	.173	.475
	.003						.832	.691	.245	.070	.1346
	.014						.479	.453	.064	.144	.621
	.031						.553	.472	.411	1.116	.394
	.055						.267	.355	.282	.134	.281
	.106						.120	.166	.205	.060	.112
	.172						.190	.004	.113	.058	.278
	.261						.047	.040	.163	.004	.100
	.302						.111	.020	.006	.141	.177
	.326						.163	.011	.065	.313	.202
Turbine cowl	.343						.185	.179	.139	.225	.200
	.387						.172	.355	.129	.196	.424
	.442						.009	.310	.030	.055	.392
	.493						.076	.233	.139	.100	.427
	.537						.041	.006	.100	.126	.388
	.581						.057	.193	.033	.090	.315
	.617						.044	.140	.044	.104	.196
	.653						.063	.181	.129	.113	.151
	.690						.134	.108	.168	.006	.193
	.708						.089	.318	.145	.209	
Plug	.735						.001	.078	.396	.349	
	.768						.127	.210	.013	.089	
	.796						.078	.104	.004	.051	
	.821						.137	.210	.051	.251	
	.852						.395	.448	.536		

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(b) $M = 0.700$; outboard station

C_p at -											
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L		Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000						.549	.897	.982	.657	1.044
	.003						.526	.628	.320	.651	.883
	.014						.501	.527	.288	.312	.609
	.031						.448	.437	.272	.470	.599
	.055						.342	.272	.281	.390	.593
	.104						.144	.072	.135	.277	.445
	.172						.071	.081	.135	.380	.164
	.261						.168	.032	.197	.399	.209
	.302						.105	.074	.026	.122	.180
	.329						.085	.068	.010	.073	.090
	.343						.022	.084	.071	.048	.187
Turbine cowl	.387						.043	.091	.032	.048	.171
	.442						.095	.003	.039	.006	.013
	.493						.013	.023	.016	.077	.122
	.537						.003	.045	.061	.077	.193
	.581						.052	.002	.010	.077	.519
	.617						.002	.049	.000	.068	.264
	.653						.002	.026	.032	.316	.325
	.690						.144	.026	.340	.631	
	.708							.026		.110	
	.735							.181		.454	
	.768							.074		.242	
	.796							.107		.126	
Plug	.821							.353		.412	
	.852										
Fan cowl	0.000						.038	.912	1.161	.741	.935
	.003						.1023	.565	.236	.508	.705
	.014						.640	.491	.228	.463	.476
	.031						.543	.410	.248	.424	.511
	.055						.383	.119	.264	.392	.318
	.106						.266	.254	.218	.276	.253
	.172						.067	.079	.082	.253	.104
	.261						.135	.072	.153	.211	.028
	.302						.077	.050	.038	.022	.047
	.326						.058	.083	.044	.004	.511
	.343						.554	.825	.083	.083	.264
	.387						.554	.129	.070	.151	.060
Turbine cowl	.442						.009	.012	.012	.041	.086
	.493						.049	.070	.044	.102	.088
	.537						.033	.083	.064	.005	.157
	.581						.038	.051	.005	.046	.028
	.617						.059	.070	.070	.028	.379
	.653						.098	.083	.064	.014	.169
	.690						.024	.670	.603	.035	.072
	.708							.608	.401	.883	
	.735							.090	.069	.082	
	.768							.649		.030	
	.796							.009		.135	
	.821							.821		.017	
Plug	.852							.353		.422	

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(c) $M = 0.750$; inboard station

x/c	C _p at -																					
	α = -2°						α = 0°						α = 4°									
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L		
Fan cowl	0.000					.742	.408	.880	.107	.821	.657	.407	.978	.119	.703							
	.003					-.748	-.794	-.932	-.415	-.447	-.1052	-.802	-.467	-.337	-.703							
	.014					-.495	-.522	-.589	-.321	-.447	-.369	-.520	-.464	-.287	-.527							
	.021					-.367	-.375	-.427	-.143	-.318	-.409	-.576	-.775	1.149	-.384							
	.055					-.314	-.328	-.334	-.233	-.254	-.330	-.317	-.276	1.149	-.264							
	.106					-.177	-.234	-.165	-.143	-.125	-.175	-.226	-.179	-.114	-.120							
	.172					-.013	-.062	-.088	-.014	-.067	-.002	-.079	.105	.014	-.047							
	.261					-.101	-.211	-.170	-.102	-.000	-.081	-.188	.105	.014	-.047							
	.302					-.084	-.035	-.023	-.167	.062	-.059	-.021	-.024	.187	.102							
	.326					-.061	-.036	-.203	-.147	.126	-.046	.055	.108	.166	.143							
	.343					.262	.238	-.246	.179	-.286	.269	.270	.232	.201	-.278							
	Turbine cowl	.387					.111	.000	.056	.033	.322	.176	.038	.085	.087	.084						
.442						.182	-.088	-.044	.041	-.046	.220	-.044	.018	-.044	-.084							
.493						-.106	.036	.050	.021	.152	.012	.076	.058	-.006	.169							
.537						-.022	-.002	-.005	-.026	-.079	.043	.079	.008	.035	.017							
.581						-.102	-.046	-.134	-.038	.237	.069	-.009	-.076	.032	.181							
.617						-.008	.115	-.085	-.239	-.181	.034	.132	.003	-.138	-.050							
.653						.187	-.026	-.064	-.096	-.222	.220	.055	.014	-.164	-.085							
.690						-.075	.141	.012	-.233	.062	-.002	.114	.073	-.117	-.008							
.708							-.011	.276	-.014	.067	.000	.067	.323	-.125	.340							
.735							-.011		-.391	.087	-.000	.067	.323	-.125	.340							
.768							-.378		-.458	-.376	-.376	-.000	.067	.323	-.125	.340						
Plug		.796					-.140	-.088		-.312	-.085	-.085	-.376	-.401	-.401	-.401						
	.821					.088	.326		.179	.102	.102	-.085	-.401	-.401	-.401							
	.852								.380	.367	.367	-.085	-.401	-.401	-.401							
	Fan cowl	0.000					.454	.382	1.129	.165	.460											
		.003					-.157	-.803	.182	-.079	-.406											
		.014					-.207	-.487	-.088	.163	-.564											
		.031					-.382	.303	-.208	1.149	-.210											
		.055					-.119	-.278	-.108	-.134	-.266											
		.106					-.144	-.196	-.247	-.049	-.093											
		.172					-.033	.121	.156	.076	-.128											
		.261					-.057	.146	.156	.024	.129											
		.302					.002	.012	-.005	.249	.196											
.326						.015	.088	.514	.223	.202												
Turbine cowl		.343					.281	.253	.152	.263	.245											
		.387					.387	.088	.150	.269	.410											
	.442					.312	.033	.071	.228	.334												
	.493					-.051	.124	.112	.293	.389												
	.537					.127	.184	.106	.161	.334												
	.581					-.005	.088	.077	-.044	.436												
	.617					-.022	.150	.112	.258	.190												
	.653					-.034	.021	.176	.033	.135												
	.690					-.025	.206	.135	.033	.135												
	.708					-.148	.235	.173	.217	.331												
	Plug	.735					-.144	.168	.399	.210												
		.768					-.274	.021		.216												
.796						-.373	-.017		-.207													
.821						-.019	.206		.006													
.852						.179	.249		.249													
						.451	.437		.515													

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(d) M = 0.750; outboard station

$x/2$	C_p at -																	
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000																	
	0.003																	
	0.014																	
	0.031																	
	0.055																	
	0.106																	
	0.172																	
	0.261																	
	0.302																	
	0.326																	
	0.343																	
	0.387																	
	0.442																	
Turbine cowl	0.000																	
	0.003																	
	0.014																	
	0.031																	
	0.055																	
	0.106																	
	0.172																	
	0.261																	
	0.302																	
	0.326																	
	0.343																	
	0.387																	
	0.442																	
Plug	0.000																	
	0.003																	
	0.014																	
	0.031																	
	0.055																	
	0.106																	
	0.172																	
	0.261																	
	0.302																	
	0.326																	
	0.343																	
	0.387																	
	0.442																	

Fan cowl	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														
Turbine cowl	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														
Plug	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														

Fan cowl	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														
Turbine cowl	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														
Plug	0.000														
	0.003														
	0.014														
	0.031														
	0.055														
	0.106														
	0.172														
	0.261														
	0.302														
	0.326														
	0.343														
	0.387														
	0.442														

(e) $M = 0.775$; inboard station

C at - p															
x/c	$\alpha = -9^\circ$					$\alpha = 0^\circ$					$\alpha = 1^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.744	.411	.876	.103	.788	.669	.411	.971	.121	.732
	.003					-.878	-.871	-.927	-.439	-.464	-.1080	-.855	-.576	-.339	-.847
	.014					-.521	-.536	-.590	-.338	-.462	-.558	-.467	-.300	-.468	
	.031					-.377	-.154	-.427	1.154	-.198	-.413	-.430	.652	1.157	-.274
	.055					-.322	-.328	-.334	1.240	-.249	-.332	-.323	.287	.204	-.258
	.106					-.173	-.232	.167	-.139	-.166	-.166	-.219	.184	-.109	-.114
	.172					-.003	.069	.091	-.007	-.005	.012	.082	.105	.020	.034
	.261					-.092	-.204	-.168	-.094	-.007	-.073	-.182	.143	.053	.043
	.302					-.046	.049	.299	.147	.124	-.047	.013	.074	.186	.116
	.343					-.203	.254	.201	.211	.226	.310	.279	.196	.225	.213
Turbine cowl	.387					-.184	-.086	-.083	.018	.004	.220	.064	.082	.699	.362
	.442					-.046	.102	-.030	.102	.228	.084	.071	.065	.093	.015
	.493					-.099	-.027	-.041	-.134	-.117	.127	.006	.065	.034	-.013
	.537					-.224	.032	.145	-.005	.357	.140	.006	.064	.023	.259
	.581					-.014	.026	.007	.313	.170	.038	.178	.022	.176	.073
	.617					-.192	.064	.049	-.080	-.271	.220	.034	.019	-.050	-.126
	.653					-.103	-.080	.302	-.178	.197	-.005	.206	.060	-.171	-.096
	.690						-.005		-.305			.001	.310	.034	
	.708						-.384		-.369			.315		.280	
	.725						-.108		-.683			-.084		.375	
Plug	.768						.316		-.214		.352		.225	.288	
	.796														
	.821														
	.852														
Fan cowl	0.000					.481	.393	1.134	.154	.531					
	.003	.398	1.034	.134	.631	-.842	-.867	.153	-.108	1.312					
	.014	-.588	-.521	-.261	-.578	-.1262	-.519	-.122	-.192	-.643					
	.031	-.431	-.425	1.158	-.278	-.356	-.620	.032	1.141	-.102					
	.055	-.342	-.313	1.228	-.267	-.339	-.285	-.120	.150	-.271					
	.106	-.163	-.209	.200	-.087	-.104	.198	.229	-.060	-.088					
	.172	-.020	.098	.118	-.042	.036	.117	.042	.074	-.049					
	.261	-.053	.169	-.127	-.028	.073	.064	.150	.089	.021	.127				
	.302	-.031	-.000	-.084	.146	.047	.017	.094	.072	.228	.197				
	.326	-.010	-.076	.149	.188	.160	.017	.094	.662	.209	.192				
Turbine cowl	.343	.317	.298	.189	.244	-.202	.314	.299	.165	.267	-.192				
	.387	.079	.048	.112	.132	.216	.089	.069	.142	.279	.405				
	.442	-.249	-.042	-.019	.106	.120	-.002	.043	.102	.281	.281				
	.493	.028	.081	.107	.061	.232	-.025	.117	.228	.304	.307				
	.537	.113	.081	.076	.092	.109	.348	.150	.094	.074	.307				
	.581	-.126	-.006	-.042	.047	.218	.484	.069	.004	.040	.497				
	.617	-.083	.180	.028	-.084	.031	.170	.150	.083	.046	.152				
	.653	.249	.056	.033	-.101	-.009	.289	.164	.128	.124	.119				
	.690	.054	.154	.107	-.104	.075	.127	.190	.150	.063	.413				
	.708		.143	.346			.164	.164	.209	.209					
Plug	.735	.008	.008	.219		.013	.013	.161	.161						
	.768	.735	.735	.348		-.091	-.091	-.296	-.296						
	.796	-.293	-.293	-.348		-.016	-.016	.007	.007						
	.821	-.014	-.014	-.184		-.184	-.184	.223	.223						
	.852	.135	.135	.210		.210	.210	.223	.223						
		.382	.382	.443		.443	.443	.417	.417						

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued
(f) $M = 0.775$; outboard station

x/\bar{c}		C_p at $\alpha =$																	
		-2°				0°				4°				8°					
Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.563	.861	.949	.644	1.051	.441	.861	1.051	.441	.861	1.051	.432	.671	.983
	.003					.775	.841	.520	.772	.809	.972	.813	.432	.972	.813	.432	.432	.660	-1.013
	.014					.580	.625	.538	.609	.657	.722	.621	.430	.722	.621	.430	.430	.564	-1.722
	.031					.520	.509	.603	.544	.499	.586	.493	.428	.586	.493	.428	.428	.496	-1.528
	.055					.385	.112	.425	.431	.431	.399	.123	.357	.399	.123	.357	.357	.386	-1.420
	.106					.096	.291	.302	.270	.309	.217	.277	.269	.309	.277	.269	.269	.310	-1.231
	.172					.062	.007	.132	.230	.052	.032	.104	.032	.104	.032	.104	.104	.166	-1.081
	.261					.173	.276	.200	.355	.098	.149	.205	.178	.205	.178	.205	.178	.237	-1.070
	.302					.105	.064	.027	.058	.081	.081	.048	.026	.058	.048	.026	.058	.006	.020
	.326					.071	.075	.021	.038	.001	.052	.083	.043	.052	.083	.043	.052	.164	.046
Turbine cowl	.343					.406	.140	.027	.190	.315	.408	.151	.025	.408	.151	.025	.025	.193	.287
	.387					.137	.052	.018	.029	.024	.207	.040	.066	.207	.040	.066	.066	.080	.057
	.442					.194	.129	.044	.058	.089	.242	.087	.025	.242	.087	.025	.025	.053	.029
	.493					.222	.120	.007	.041	.015	.190	.117	.004	.117	.004	.004	.011	.210	.156
	.537					.130	.061	.021	.340	.052	.035	.006	.011	.035	.006	.011	.011	.186	.026
	.581					.039	.033	.075	.239	.504	.008	.062	.050	.008	.062	.050	.050	.043	.073
	.617					.075	.109	.083	.287	.513	.063	.205	.057	.063	.205	.057	.057	.158	.412
	.653					.018	.027	.001	.230	.335	.022	.019	.038	.022	.019	.038	.038	.395	.302
	.690					.126	.018	.027	.281	.363	.052	.143	.100	.052	.143	.100	.100	.022	.242
	.708						.339	.373	.798	.402								.816	
Plug	.735						.075	.055	.055									.026	
	.768						.277		.513									.026	
	.796						.107		.326									.194	
	.821						.166		.230									.224	
	.852						.361		.402									.421	
Fan cowl	0.000	.360	.868	1.125	.705	.949	.185	.848	1.215	.753	.862		.848	1.215	.753	.862			
	.003	-1.053	.800	.400	.633	.912	.156	.845	.845	.729	.862		.845	.729	.862				
	.014	.879	.482	.326	.540	.912	.175	.619	.619	.430	.862		.619	.430	.862				
	.031	.574	.482	.355	.469	.589	.377	.474	.474	.408	.441		.474	.408	.441				
	.055	.404	.119	.309	.370	.421	.387	.421	.421	.202	.416		.421	.202	.416				
	.106	.239	.264	.236	.266	.387	.330	.262	.262	.188	.230		.262	.188	.230				
	.172	.048	.029	.085	.141	.096	.030	.038	.038	.052	.010		.038	.052	.010				
	.261	.129	.139	.159	.172	.028	.028	.525	.525	.126	.148		.525	.126	.148				
	.302	.065	.034	.031	.031	.048	.034	.030	.030	.078	.097		.030	.078	.097				
	.326	.036	.088	.054	.056	.076	.008	.095	.078	.086	.125		.086	.125	.125				
Turbine cowl	.343	.383	.218	.037	.200	.288	.338	.157	.004	.210	.275		.157	.004	.210				
	.387	.253	.025	.085	.079	.124	.330	.021	.109	.072	.283		.021	.109	.072				
	.442	.239	.065	.037	.009	.003	.220	.035	.007	.066	.238		.035	.007	.066				
	.493	.202	.127	.088	.220	.206	.131	.067	.123	.179	.034		.067	.123	.179				
	.537	.083	.076	.116	.158	.059	.131	.144	.124	.114	.189		.144	.124	.114				
	.581	.032	.064	.009	.082	.008	.042	.061	.001	.063	.094		.061	.001	.063				
	.617	.126	.201	.048	.113	.356	.131	.126	.075	.081	.081		.126	.075	.081				
	.653	.053	.068	.045	.065	.082	.123	.129	.084	.196	.063		.129	.084	.196				
	.690	.010	.397	.088	.266	.181	.034	.061	.129	.269	.080		.061	.129	.269				
	.708		.397	.411	.1220			.429	.438					.429					
Plug	.735		.136	.090			.140	.069					.140						
	.768		.133				.013						.013						
	.796		.014				.038						.038						
	.821		.167				.186						.186						
	.852		.391				.395						.395						

TABLE 39.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 6 - Continued

(g) $M = 0.800$; Inboard station.

x/\bar{c}	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					.753	.433	.874	.101	.784
	.003	-.627	-.415	1.001	.125	-.902	-.936	-.979	-.537	-.496
	.014	-1.248	-.938	-.324	-.296	-.535	-.557	-.592	-.345	-.464
	.031	-.872	-.554	-.386	-.280	-.384	-.384	-.417	1.172	-.070
	.055	-.407	-.243	-.286	1.173	-.331	-.333	-.341	-.243	-.254
	.106	-.341	-.316	-.246	-.191	-.168	-.233	.162	-.135	-.116
	.172	-.157	-.213	-.192	-.089	-.088	.070	.086	.002	.075
	.261	.026	-.055	.108	-.138	-.086	-.200	-.165	-.087	.027
	.302	-.047	-.165	.132	-.024	-.037	.057	.192	.164	.086
	.326	-.023	.006	.200	.154	-.037	.284	.152	.248	.126
	.343	.361	.317	.146	.178	-.796	.067	-.152	.148	.071
	.387	-.300	.071	.119	.375	-.184	-.060	-.006	.110	.250
	.442	-.239	-.040	-.013	.124	-.012	.224	.070	.129	.234
Turbine cowl	.493	.096	.203	.073	.154	.208	-.065	-.062	-.140	.217
	.537	.214	.035	.038	.073	.302	.051	-.135	-.084	.431
	.581	.190	.095	-.067	.078	-.000	-.054	.113	-.235	.201
	.617	.063	.144	.025	-.129	.188	.043	.113	-.175	.017
	.653	.239	.068	.044	-.003	-.086	-.027	-.014	-.313	.213
	.690	.022	.106	.122	-.135		-.011	.257	.062	.291
	.708		.106	.319	.042		-.006	-.264	.125	
	.735		.000		-.156		-.371	-.294	.003	
	.768		-.235		-.302		-.611	-.717	-.356	
	.796		-.225		-.515		-.124	-.134	-.294	
	.821		.163		.249		.335		.144	
	.852		.365		.437				.339	
Plug	0.000									
	.003									
	.014									
	.031									
	.055									
	.106									
	.172									
	.261									
	.302									
	.326									
	.343									
	.387									
	.442									
Fan cowl	.493	-.627	-.415	1.001	.125	-.902	-.936	-.979	-.537	-.496
	.537	-1.248	-.938	-.324	-.296	-.535	-.557	-.592	-.345	-.464
	.581	-.872	-.554	-.386	-.280	-.384	-.384	-.417	1.172	-.070
	.617	-.407	-.243	-.286	1.173	-.331	-.333	-.341	-.243	-.254
	.653	-.341	-.316	-.246	-.191	-.168	-.233	.162	-.135	-.116
	.690	-.157	-.213	-.192	-.089	-.088	.070	.086	.002	.075
	.708	.026	-.055	.108	-.138	-.086	-.200	-.165	-.087	.027
	.735	-.047	-.165	.132	-.024	-.037	.057	.192	.164	.086
	.768	-.023	.006	.200	.154	-.037	.284	.152	.248	.126
	.796	.361	.317	.146	.178	-.796	.067	-.152	.148	.071
	.821	-.300	.071	.119	.375	-.184	-.060	-.006	.110	.250
	.852	-.239	-.040	-.013	.124	-.012	.224	.070	.129	.234
		.096	.203	.073	.154	.208	-.065	-.062	-.140	.217
Turbine cowl	.537	.214	.035	.038	.073	.302	.051	-.135	-.084	.431
	.581	.190	.095	-.067	.078	-.000	-.054	.113	-.235	.201
	.617	.063	.144	.025	-.129	.188	.043	.113	-.175	.017
	.653	.239	.068	.044	-.003	-.086	-.027	-.014	-.313	.213
	.690	.022	.106	.122	-.135		-.011	.257	.062	.291
	.708		.106	.319	.042		-.006	-.264	.125	
	.735		.000		-.156		-.371	-.294	.003	
	.768		-.235		-.302		-.611	-.717	-.356	
	.796		-.225		-.515		-.124	-.134	-.294	
	.821		.163		.249		.335		.144	
	.852		.365		.437				.339	
Plug	0.000									
	.003									
	.014									
	.031									
	.055									
	.106									
	.172									
	.261									
	.302									
	.326									
	.343									
	.387									
	.442									

(h) $M = 0.800$; outboard station

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TABLE 40.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7

(a) $M = 0.700$; inboard station

x/\bar{z}	C_p at -																				
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$								
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	1.305	.495	.800	.946	1.074	1.202	.495	1.001	.988	.964	1.110	.482	1.092	.842						
	.003	-.217	-.022	-1.426	-.444	-.044	-.572	-.026	-.769	-.319	-.338	-.904	-.018	-.370	-.515						
	.014	-.260	.511	-.733	-.387	-.216	-.439	.515	-.596	-.335	-.370	-.520	-.495	-.451	-.287						
	.031	-.217	.417	-.843	-.339	-.153	-.316	.412	-.951	-.251	-.229	-.364	-.395	-.378	-.215						
	.055	-.246	.342	-.740	-.272	-.169	-.293	.333	-.321	-.223	-.201	-.307	-.316	-.257	-.199						
	.106	-.199	.266	-.285	-.194	-.122	-.252	-.239	-.149	-.120	-.208	-.231	-.203	-.121	-.115						
	.172	-.094	.132	-.094	-.031	-.103	.139	-.098	-.029	-.013	-.089	-.122	-.068	.001	-.024						
	.261	-.189	.269	-.197	-.181	-.170	.242	-.173	-.110	-.051	-.146	-.034	-.153	-.074	-.024						
	.302	-.151	.208	-.156	-.100	-.132	.204	-.173	-.064	-.020	-.108	-.034	-.153	-.037	-.060						
	.326	-.073	.266	-.085	-.028	-.165	.187	-.001	.012	-.049	-.137	-.038	-.046	.095	.082						
	.343	-.279	.132	-.022	-.131	-.259	.187	-.192	.115	-.101	-.411	-.098	.082	.046	.082						
	Turbine cowl	.387	-.445	.016	.022	-.228	-.439	.084	.049	.052	-.101	-.137	.038	.046	.095	.082					
.442		.085	-.141	.078	-.159	-.239	.062	.039	.087	.136	.138	.101	.038	.026	-.185						
.493		.090	.063	-.353	.062	-.077	.062	.005	.087	.067	.101	.038	.026	-.227	-.234						
.537		.047	.022	-.766	-.228	-.056	-.094	.046	-.010	-.079	-.079	.101	.038	.026	-.193						
.581		.180	-.144	-.159	-.075	-.094	.013	-.007	-.079	-.079	.101	.038	.026	-.045	-.046						
.617		-.109	.009	-.069	-.362	-.028	.040	-.007	-.079	-.079	.101	.038	.026	-.037	-.037						
.653		.130	-.041	-.059	-.191	-.331	.209	.071	.012	-.092	.233	.095	.054	-.087	-.037						
.690		-.319	-.059	-.041	-.244	.141	.095	.065	.018	-.132	.153	.108	.067	.082	-.165						
.708		.035	.260	.031	.141	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068						
.735		-.016	.235	.141	.141	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068						
.768		-.448	.237	.150	.150	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068						
.796		-.207	.066	.062	.062	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068						
.821	.066	.304	.337	.337	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068							
.852	.304	.304	.337	.337	.095	.099	.306	.127	.098	.148	.120	.333	.020	-.068							
Fan cowl	0.000	1.031	.513	1.153	1.037	.751	.879	.476	1.214	1.086	.598										
	.003	-.274	-.020	-.142	-.282	-.688	-1.553	-.018	-.729	-.178	-1.254										
	.014	-.682	.496	.133	.379	-.094	.492	.492	.718	.200	-.578										
	.031	-.488	.493	-.133	.379	-.094	.492	.492	.718	.200	-.578										
	.055	-.417	.314	.188	-.235	-.341	.382	.718	.162	-.340	-.364										
	.106	-.239	.233	.176	-.107	-.116	.189	.219	.109	.147	-.247										
	.172	-.176	.120	.051	.078	-.008	-.066	.113	.000	.047	-.041										
	.241	-.228	.176	.054	.008	-.085	.194	.097	.047	-.041	-.041										
	.322	.130	.029	.029	.083	-.052	.015	.010	.057	.057	.150										
	.329	.214	.046	.115	.105	-.070	.037	.010	.157	.172	.172										
	.343	.323	.175	.138	-.073	-.194	.132	.043	.197	.172	.172										
	.387	.507	.099	.133	.005	-.369	.126	.043	.197	.172	.172										
Turbine cowl	.442	.056	.018	.149	.221	.171	.054	.066	.125	.116	.116										
	.493	.080	.093	.158	.233	.218	.120	.101	.166	.376	.376										
	.537	.147	.096	.058	-.014	.123	.095	.141	.107	.107	.107										
	.581	.162	.068	-.023	-.170	-.037	.101	.035	.088	-.009	-.009										
	.617	.043	.096	.056	.037	.133	.164	.110	.066	.194	.166										
	.653	.146	.103	.078	.005	.285	.192	.135	.119	.166	.166										
	.690	.099	.093	.008	-.198	.294	.204	.145	.015	-.153	-.153										
	.708	.078	.175	.143	.143	.294	.204	.145	.015	-.153	-.153										
	.735	.092	.353	.357	.357	.294	.204	.145	.015	-.153	-.153										
	.768	.101	.092	.179	.179	.294	.204	.145	.015	-.153	-.153										
	.796	.045	.156	.131	.131	.294	.204	.145	.015	-.153	-.153										
	.821	.156	.400	.196	.196	.294	.204	.145	.015	-.153	-.153										
.852	.400	.400	.449	.449	.294	.204	.145	.015	-.153	-.153											
Plug	0.000	1.031	.513	1.153	1.037	.751	.879	.476	1.214	1.086	.598										
	.003	-.274	-.020	-.142	-.282	-.688	-1.553	-.018	-.729	-.178	-1.254										
	.014	-.682	.496	.133	.379	-.094	.492	.492	.718	.200	-.578										
	.031	-.488	.493	-.133	.379	-.094	.492	.492	.718	.200	-.578										
	.055	-.417	.314	.188	-.235	-.341	.382	.718	.162	-.340	-.364										
	.106	-.239	.233	.176	-.107	-.116	.189	.219	.109	.147	-.247										
	.172	-.176	.120	.051	.078	-.008	-.066	.113	.000	.047	-.041										
	.241	-.228	.176	.054	.008	-.085	.194	.097	.047	-.041	-.041										
	.322	.130	.029	.029	.083	-.052	.015	.010	.057	.057	.150										
	.329	.214	.046	.115	.105	-.070	.037	.010	.157	.172	.172										
	.343	.323	.175	.138	-.073	-.194	.132	.043	.197	.172	.172										
	.387	.507	.099	.133	.005	-.369	.126	.043	.197	.172	.172										

TABLE 40.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(b) $M = 0.750$; inboard station

x/c	C_p at															
	$\alpha = -2^\circ$					$\alpha = 0^\circ$					$\alpha = 1^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.000	1.293	.471	.799	1.060	1.188	.499	.988	.971	.926	1.104	.483	1.082	.993	.816	
	.003	.259	.024	-1.452	.532	.780	-.619	-.942	-.454	-.429	-1.068	-.016	-.361	-.411	-.616	
	.014	.336	.579	-1.144	.255	.473	-.550	-.633	-.363	-.400	-.350	-.544	-.301	-.346	-.465	
	.031	.355	.661	-.661	.340	.326	-.230	-.894	-.265	-.232	-.374	-.427	-.867	-.283	-.289	
	.055	.343	.376	.390	.172	.200	-.347	.339	-.232	-.204	-.317	-.333	-.275	-.209	-.226	
	.106	.237	.282	.289	.128	.208	-.253	-.239	-.141	-.113	-.205	-.235	-.204	-.121	-.112	
	.172	.199	.182	.130	.027	.085	-.127	-.090	-.098	.001	-.080	-.116	-.061	.004	.010	
	.261	.186	.182	.094	-.078	-.138	-.236	-.170	-.030	-.038	-.141	-.027	-.187	-.073	.013	
	.302	.147	.087	.082	.061	-.115	-.041	-.053	.030	.050	-.093	-.027	-.030	.056	.076	
	.326	.151	.002	.441	-.002	-.024	.164	.159	.087	.076	-.024	.053	.076	.110	.104	
Turbine cowl	.343	.042	.107	.833	.015	-.223	.164	.230	.167	-.056	-.210	.201	-.213	.181	.041	
	.387	.367	.024	.024	.015	-.361	.044	.073	.070	-.098	-.356	.079	.090	.104	.044	
	.442	.021	.030	-.104	.172	.071	.027	-.033	-.058	.030	.092	.027	.004	.016	.110	
	.493	.069	.004	-.079	.087	.135	.046	.079	-.061	.215	.157	.079	.059	.013	.192	
	.537	.378	.044	-.073	.393	.036	.010	.016	.010	.403	.040	.090	.033	.027	.203	
	.581	.371	.010	-.158	.258	.105	.030	-.104	.016	.135	.058	.030	-.038	-.050	.087	
	.617	.104	-.013	-.044	-.303	.006	.124	-.044	-.269	-.135	.032	.110	.024	-.070	.053	
	.653	.160	-.070	-.076	.212	.213	.036	-.027	-.061	-.150	.226	.050	.053	-.138	.055	
	.690	-.056	-.076	-.073	.294	.105	.102	.027	-.192	.050	.157	.059	.059	-.155	-.047	
	.708	.073	.073	.267	.035	.105	.102	.319	.073	.073	.173	.173	.361	.175	.047	
Plug	.735	.019	.019	.691	.491	.001	-.001	.320	.320	.004	.004	-.004	-.351	-.360	.041	
	.768	.487	.487	.488	.488	.396	.396	.434	.434	.147	.147	-.087	-.087	-.087	.041	
	.796	.619	.619	.622	.622	.101	.101	.201	.201	.127	.127	.041	.172	.172	.041	
	.821	.678	.678	.684	.684	.339	.339	.400	.400	.376	.376	.420	.420	.420	.420	
	.852	.290	.290	.334	.334	.430	.430	.492	.492	.376	.376	.420	.420	.420	.420	
	Fan cowl	0.000	1.049	.477	1.143	1.010	.743	.454	1.226	1.060	.615	1.104	.483	1.082	.993	.816
		.003	-.1254	-.014	-.222	-.347	-.890	-.016	.165	-.226	-.1340	-.016	-.226	-.1340	-.016	-.226
		.014	.555	.542	-.382	.295	-.483	-.533	-.144	-.215	-.326	-.533	-.144	-.215	-.326	-.465
		.031	.420	.410	-.924	.207	.301	-.404	.653	-.167	-.326	-.404	-.167	-.326	-.404	-.465
		.055	.331	.331	-.225	.190	.233	-.334	.130	-.141	-.235	-.334	-.141	-.235	-.334	-.465
.106		.153	.231	.182	.102	.102	-.178	.116	.058	-.078	-.178	.116	.058	-.078	-.465	
.172		.068	.108	.039	.023	.023	-.044	.004	.064	.059	-.044	.004	.064	.059	-.465	
.261		.124	.205	.134	.039	.018	-.079	-.093	.016	.078	-.079	.016	.078	.078	-.465	
.302		.072	.014	-.016	.089	.109	.023	.002	.016	.170	.023	.002	.016	.170	-.465	
.326		.003	.064	.055	.134	.134	.025	.025	.025	.152	.025	.025	.025	.152	-.465	
Turbine cowl	.343	.193	.232	.193	.203	-.174	.247	.130	.172	.007	.247	.130	.172	.007	-.465	
	.387	.348	.092	.101	.146	.012	.087	.139	.192	.124	.087	.139	.192	.124	-.465	
	.442	.117	.004	.035	.083	.203	-.041	.070	.238	.332	-.041	.070	.238	.332	-.465	
	.493	.173	.155	.266	.080	.193	.139	.110	.238	.332	.139	.110	.238	.332	-.465	
	.537	.053	.598	.672	.075	.099	.099	.039	.018	.070	.099	.039	.018	.070	-.465	
	.581	.057	.635	-.034	.018	.088	.099	.039	.018	.070	.099	.039	.018	.070	-.465	
	.617	.083	.141	.064	.018	.023	.156	.119	.021	.144	.156	.119	.021	.144	-.465	
	.653	.247	.158	.081	.003	.015	.176	.142	.021	.144	.176	.142	.021	.144	-.465	
	.690	.191	.129	.081	.082	.068	.185	.150	.021	.144	.185	.150	.021	.144	-.465	
	.708	.158	.158	.372	.183	.288	.242	.399	.233	.030	.242	.399	.233	.030	-.465	
Plug	.735	.609	.609	.327	.327	.016	.016	.016	.016	.016	.016	.016	.016	.016	-.465	
	.768	.325	.325	.321	.321	-.024	-.024	-.024	-.024	-.024	-.024	-.024	-.024	-.024	-.465	
	.796	.854	.854	.828	.828	.033	.033	.033	.033	.033	.033	.033	.033	.033	-.465	
	.821	.161	.161	.134	.134	.216	.216	.216	.216	.216	.216	.216	.216	.216	-.465	
	.852	.400	.400	.450	.450	.430	.430	.492	.492	.430	.430	.492	.492	.430	-.465	

TABLE 40.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued

(c) $M = 0.775$; inboard station

x/\bar{c}	C_p at -														
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$		
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	1.303	.488	.808	.925	1.053	1.191	.499	.967	.957	.898	1.111	.498	1.058	.973	.818
	-.286	-.020	1.283	-.586	-.124	-.824	-.024	-.995	-.502	-.442	-.1052	-.017	-.634	-.435	-.750
	.003	-.337	-.590	-.457	-.250	-.497	-.575	-.666	-.385	-.512	-.565	-.445	-.546	-.359	-.497
	.014	-.253	-.475	-.420	-.181	-.307	-.364	-.350	-.245	-.256	-.377	-.445	-.796	-.246	-.276
	.031	-.261	-.376	-.299	-.124	-.181	-.263	-.252	-.144	-.218	-.227	-.340	-.291	-.214	-.222
	.055	-.208	-.280	-.254	-.064	-.017	-.088	-.131	-.093	-.117	-.203	-.239	-.212	-.118	-.104
	.106	-.092	-.152	-.124	-.075	-.086	-.176	-.263	-.103	-.038	-.075	-.113	-.058	.013	.019
	.172	-.092	-.124	-.064	-.075	-.086	-.176	-.263	-.103	-.038	-.075	-.113	-.058	.013	.019
	.261	-.187	-.280	-.201	-.176	-.001	-.116	-.046	-.020	-.050	-.087	-.214	-.146	-.060	-.000
	.302	-.141	-.075	-.039	-.024	-.053	-.036	-.236	.083	.077	-.054	-.060	-.060	.090	.090
	.326	.331	.603	.024	.024	.053	.036	.236	.083	.077	-.054	-.060	-.060	.090	.090
	.343	-.216	.142	-.201	.155	-.072	.176	-.202	.186	.077	-.182	.214	-.184	.120	.117
	.387	-.315	.040	.038	.021	-.189	-.319	.066	.666	.099	-.101	.082	.090	.123	.033
Turbine cowl	.442	.049	-.064	-.061	.056	.117	.041	-.001	-.062	.072	.058	.046	-.003	.019	.079
	.493	.074	.161	.057	.017	.128	.080	-.024	.156	.197	.153	.082	.079	.046	.276
	.537	.190	-.056	-.091	-.233	.034	.039	.014	.128	.439	.103	.038	.071	.408	.008
	.581	.529	.021	-.230	-.326	.173	.025	-.092	.061	-.062	.128	.046	-.069	.043	-.000
	.617	-.083	.116	-.303	-.296	-.441	.138	-.021	.264	.139	.050	.186	-.009	.178	-.060
	.653	.157	.067	-.105	-.228	.198	.061	.057	-.098	-.202	.224	.077	.019	-.014	-.069
	.690	-.025	-.135	.062	-.424	.423	.699	.091	.661	.124	.161	.145	.071	-.014	-.038
	.708	.029	.216	-.028	.423		.091	.310	.050			.153	.351	.104	.038
	.735	-.006	.441	-.441			-.008		.038			.011	.351	.104	.038
	.768	-.492		-.386			-.350		.336			-.332		-.380	
	.796	-.662		-.791			-.114		.592			-.661		-.052	
	.821	.090		.089			.116		.203			.129		.216	
	.852	.295		.352			.330		.383			.367		.413	
Fan cowl	1.074	.504	1.138	1.035	.781	.562	.471	1.234	1.047	.610					
	-.1247	-.015	-.325	-.370	-.943	-.512	-.013	.116	.257	-.129					
	.014	-.523	-.396	-.313	-.556	-.1247	.079	.175	.532	.467					
	.031	-.330	-.435	-.220	-.302	-.352	.009	.435	.149	.320					
	.055	-.328	-.334	-.190	-.226	-.319	.315	.351	.139	.257					
	.106	-.152	-.236	-.182	-.057	-.170	.314	.112	.052	.074					
	.172	-.059	-.100	-.037	-.031	-.033	.058	.006	.071	.071					
	.261	-.121	-.109	.037	.029	-.036	.175	.006	.071	.085					
	.302	-.083	-.133	.037	.029	-.036	.175	.006	.071	.085					
	.326	.003	.012	.034	.119	-.020	.068	.022	.143	.183					
	.343	.083	.117	.144	.144	.108	.068	.135	.183	.211					
	.387	-.338	.086	.117	.007	-.149	.264	.149	.249	.229					
	Turbine cowl	.442	.036	.052	.062	.146	.150	.033	.077	.173	.317				
.493		.185	.130	.056	.250	.216	.076	.077	.266	.315					
.537		.117	.089	.079	.176	.154	.146	.113	.080	.025					
.581		.157	.128	.045	-.037	.158	.077	.025	.049	.131					
.617		.086	.163	.045	-.037	.158	.077	.025	.049	.131					
.653		.243	.136	.059	.018	.162	.179	.124	.173	.159					
.690		.198	.103	.092	.010	.274	.126	.126	.137	.131					
.708		.215	.388	-.116	-.067	.274	.203	.157	.036	.028					
.735		.012		.294			.234	.409	.230						
.768		-.133		-.286			.022		.186						
.796		-.048		-.337			-.016		-.191						
.821		.158		-.015			.203		.006						
.852		.388		.212			.417		.241						

TABLE 40.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Continued
(d) $M = 0.800$; inboard station

x/\bar{c}	C_p at									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	1.298	.504	.822	.924	1.053	1.201	.435	.571	.941	.909
	-.341	-.024	1.239	-.625	-.134	-.883	-.033	-.038	-.537	-.498
	-.333	-.627	1.202	-.489	-.260	-.501	.628	-.641	-.431	-.414
	-.265	-.527	-.016	-.357	-.184	-.346	.499	-.360	-.288	-.259
	-.277	-.395	-.382	-.299	-.181	-.318	.439	-.373	-.240	-.209
	-.209	-.285	-.255	-.197	-.118	-.211	.237	-.257	-.138	-.109
	-.090	-.150	-.118	-.060	-.008	-.080	.123	-.086	-.007	-.006
	-.185	-.279	-.200	-.158	-.081	-.159	.036	-.170	-.101	-.025
	-.134	-.068	-.008	-.031	-.010	-.104	.036	-.044	.040	-.081
	.326	.575	.016	.794	.042	.045	.179	.058	.098	.098
	.343	.185	.174	.150	.052	.171	.204	.159	.221	.025
	.387	-.273	.079	.026	.018	-.278	.093	.069	.116	-.078
	.442	.109	-.034	-.063	.005	.095	.031	.004	-.054	.095
Turbine cowl	.593	.069	.077	.137	.011	.055	.217	.114	.213	.156
	.537	.336	-.097	-.134	-.066	.171	.002	-.023	-.170	.380
	.581	.698	-.042	-.100	-.011	.258	.104	-.137	.041	.182
	.617	.026	-.068	.105	-.286	-.402	-.004	.088	-.198	.043
	.553	.149	-.063	-.121	-.248	-.531	.159	.054	-.086	.225
Plug	.690	-.087	-.121	-.389	.570	.063	-.017	.009	-.269	.216
	.708	-.034	.219	.029			.172	.330	.074	.141
	.735	-.010	.391	.331			.604		.390	.009
	.768	-.432	.278	-.278			.397		.345	.281
	.796	-.704	.457	-.457			.433		.345	.281
Fan cowl	.821	.003	-.239	-.239			.140		.195	.165
	.952	.322	.336	.336			.333		.405	.360
Turbine cowl	0.000	1.078	.499	1.114	.936	.776	.931	1.242	1.037	.468
	.003	-.1237	-.015	-.461	-.378	-.945	-.1427	.013	-.257	-.1313
	.014	-.847	-.608	-.650	-.331	-.507	-.1203	.076	-.255	-.582
	.031	-.374	.450	-.651	-.223	-.256	.281	.435	-.187	-.303
	.055	-.326	-.340	-.245	-.191	-.235	.281	.355	-.137	-.218
	.106	-.184	-.229	-.184	-.091	-.041	.153	.121	.050	-.600
	.172	-.047	-.082	-.031	.030	.045	.054	.011	.076	.082
	.261	-.107	-.192	-.139	-.054	.037	-.039	.095	.024	.097
	.302	-.355	.001	-.005	.005	.129	.012	.021	.193	.193
	.329	.154	.082	.265	.150	.136	.294	.219	.153	.221
	.387	-.238	.109	.127	.169	.016	.127	.127	.276	.045
	.442	.056	.045	.002	.032	.124	.116	.074	.205	.126
	.493	.192	.153	.059	.106	.321	.219	.082	.147	.255
Plug	.537	.192	.066	.045	.038	-.218	.223	.159	.126	.221
	.581	.198	.082	.045	.032	.124	.223	.095	.147	.063
	.617	.084	.235	.032	.162	.301	.211	.095	.000	.218
	.653	.236	.173	.048	-.162	.301	.164	.193	.095	.158
	.690	.239	.217	.111	-.136	.364	.287	.133	.008	.113
Plug	.708	.172	.367	.100	.027	.027	.254	.145	-.021	.052
	.735	.022	.022	.231	.027	.027	.254	.145	-.021	.052
	.768	-.289	.310	.310	.047	.047	.047	.137	.137	.137
	.796	-.028	.107	.107	.016	.016	.016	.026	.026	.026
	.821	.184	.253	.253	.195	.195	.195	.250	.250	.250
Plug	.852	.383	.431	.431	.431	.431	.412	.412	.476	.476

TABLE 40.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 7 - Concluded

(e) M = 0.825; inboard station

x/c	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	1.301	.534	.636	.910	1.033	1.221	.531	.980	.955
	.703	-.332	-.025	1.231	-.633	1.132	-.827	-.020	-.056	-.569
	.014	-.340	-.646	1.152	-.491	-.258	-.497	-.437	-.746	-.428
	.031	-.267	-.638	.606	-.385	-.182	-.347	-.336	-.269	-.301
	.055	-.275	-.391	-.257	-.304	-.182	-.317	-.371	-.363	-.245
	.066	-.209	-.284	-.279	-.185	-.111	-.202	-.256	-.249	-.134
	.172	-.079	-.142	-.106	-.053	.005	-.064	-.112	-.076	.000
	.261	-.179	-.279	-.193	-.157	-.061	-.152	-.236	-.167	.018
	.302	-.125	-.041	-.058	-.015	.033	-.094	-.028	-.035	.081
	.326	.916	.026	1.028	.053	.066	.350	.661	.498	.139
	.343	-.156	.211	-.104	.226	-.028	-.144	.241	-.094	.253
	.387	-.240	.077	.046	.008	-.126	-.229	.119	.031	.114
	.442	.151	.018	.023	.026	.231	.159	.015	.010	.033
	.493	.036	-.089	-.061	-.154	.021	.105	.119	.066	.116
Turbine cowl	.537	.532	.010	-.155	.046	.453	.301	-.025	-.061	.039
	.581	.862	-.089	-.139	.188	.181	.354	.076	-.035	.180
	.617	.036	-.036	-.140	-.370	.319	.059	.090	.086	.023
	.653	.151	-.668	-.010	-.337	-.484	.186	.686	-.134	-.091
	.690	-.060	-.147	-.155	-.334	.730	.090	.048	.137	-.086
	.708		-.002	.173	.056			.664	.226	.334
	.735		-.002		.327			.010		
	.768		-.401		-.446			.333		
	.796		-.740		-.646			-.564		
	.821		-.109		-.142			.059		
	.852		.145		.236			.320		
Fan cowl	0.000	1.128	.525	1.163	.989	.801	1.019	.534	1.236	1.038
	.703	-.124	-.013	-.433	-.434	-.968	-.139	-.007	-.044	-.274
	.014	-.573	-.633	-.498	-.325	-.486	-.123	-.008	-.216	-.274
	.031	-.359	-.361	-.285	-.246	-.287	-.403	.432	-.236	-.182
	.055	-.259	-.341	-.252	-.168	-.213	-.253	.432	-.137	-.197
	.066	-.239	-.222	-.184	-.082	-.076	-.153	.196	-.112	-.137
	.172	-.113	-.082	-.058	.055	.003	-.008	-.066	.021	.094
	.261	-.174	-.189	-.128	-.018	.073	-.043	.157	-.686	.036
	.302	-.117	.012	.005	.119	.177	.003	.036	.036	.165
	.326	.229	.046	.320	.157	.115	.337	.115	.324	.203
	.343	-.194	.279	-.079	.283	.038	-.101	.308	-.081	.302
	.387	-.286	.132	.127	.192	.030	-.204	.140	.171	.229
	.442	.079	.025	.081	.040	.174	.138	.082	.062	.109
	.493	.079	.284	.127	.301	.337	.238	.207	.135	.148
Turbine cowl	.537	.229	.066	.076	-.023	-.254	.337	.112	.687	.186
	.581	.171	.185	-.079	.235	.139	.264	.097	.026	.059
	.617	-.009	.152	.119	-.158	.012	.168	.273	.138	.041
	.653	.156	.106	.155	.012	-.061	.212	.143	.140	.023
	.690	.091	.119	.081	-.087	.152	.276	.209	.179	-.071
	.708		.233	.389	.108			.250	.418	.226
	.735		.022		-.160			.036	-.073	
	.768		-.222		-.216			-.193	-.266	
	.796		-.280		-.495			.021	.021	
	.821		-.190		-.261			.209	.277	
	.852		.371		.451			.413	.472	
Plug	0.000	1.301	.534	.636	.910	1.033	1.221	.531	.980	.955
	.703	-.332	-.025	1.231	-.633	1.132	-.827	-.020	-.056	-.569
	.014	-.340	-.646	1.152	-.491	-.258	-.497	-.437	-.746	-.428
	.031	-.267	-.638	.606	-.385	-.182	-.347	-.336	-.269	-.301
	.055	-.275	-.391	-.257	-.304	-.182	-.317	-.371	-.363	-.245
	.066	-.209	-.284	-.279	-.185	-.111	-.202	-.256	-.249	-.134
	.172	-.079	-.142	-.106	-.053	.005	-.064	-.112	-.076	.000
	.261	-.179	-.279	-.193	-.157	-.061	-.152	-.236	-.167	.018
	.302	-.125	-.041	-.058	-.015	.033	-.094	-.028	-.035	.081
	.326	.916	.026	1.028	.053	.066	.350	.661	.498	.139
	.343	-.156	.211	-.104	.226	-.028	-.144	.241	-.094	.253
	.387	-.240	.077	.046	.008	-.126	-.229	.119	.031	.114
	.442	.151	.018	.023	.026	.231	.159	.015	.010	.033
	.493	.036	-.089	-.061	-.154	.021	.105	.119	.066	.116
	.537	.532	.010	-.155	.046	.453	.301	-.025	-.061	.039

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8

(a) $M = 0.700$; outboard station

C _p at -															
x/c	α = 0°														
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.958	.935	.903	1.153	.579	.896	.989	.734	1.074	.454	.504	1.082	.749	1.012
	.003	-.147	-.535	.418	-.274	-.433	-.575	.261	-.537	-.563	-.697	-.608	.209	-.464	-.741
	.014	-.273	-.470	-.599	-.361	-.418	-.491	-.433	-.421	-.486	-.513	-.479	-.324	-.431	-.567
	.031	-1.218	-.418	-.758	-.293	-.688	-.433	-.543	-.421	-.434	-.789	-.453	-.423	-.564	-.380
	.055	-.258	-.006	-.421	-.406	-.292	.011	-.335	-.379	-.453	-.333	-.308	-.290	-.373	-.422
	.106	-.287	-.266	-.308	-.371	-.346	-.238	-.258	-.279	-.257	-.435	-.239	-.232	-.223	-.334
	.172	-.118	-.065	-.152	-.281	-.103	-.018	-.173	-.289	-.198	-.115	-.041	-.096	-.160	-.257
	.261	-.224	.208	.210	-.242	-.195	.011	.173	.260	.208	-.193	.044	-.164	-.205	-.279
	.302	-.142	-.054	-.035	-.150	-.113	-.060	.021	.121	.121	.110	.060	.034	.001	-.011
	.326	-.123	-.004	-.013	-.334	-.089	.021	.121	.121	.121	.097	.083	-.034	-.044	-.021
	.343	-.090	.072	-.062	-.623	-.685	.086	.118	.066	-.605	-.697	.083	-.106	-.063	-.603
	.367	-.054	.114	.000	.252	.483	.118	.053	-.021	.373	.163	.122	.063	.105	.363
Turbine cowl	.442	.178	.032	.070	.312	.129	.005	.053	-.021	.240	.163	.021	-.002	.107	-.008
	.493	.234	.004	.042	.535	.695	.075	.028	-.003	.027	.089	.070	.027	.024	-.011
	.537	-.016	-.074	-.029	-.222	-.621	.011	.028	-.007	-.008	-.091	.063	.047	-.015	.124
	.581	-.089	-.081	-.245	-.342	-.684	-.030	.021	-.047	-.116	-.071	.011	-.028	-.218	-.034
	.617	-.079	.075	-.035	-.310	-.608	.037	.021	-.082	-.118	-.021	.034	.040	-.108	-.057
	.653	-.113	-.084	-.061	-.394	-.684	.063	.015	-.005	-.124	-.066	.066	.040	.098	-.053
	.690	-.130	-.081	-.026	-.445	-.689	.053	.047	.366	-.431	-.062	.063	.073	.131	-.280
	.708		.295	.368			.365	.365	-.618			.378	.378	-.664	
	.735		.009	.093			.028	.082	-.089			.053		-.134	
	.766		-.428	-.510			-.141	.337	-.337			-.102		-.351	
	.821		-.123	.049			-.063	.118	-.150			-.024		-.199	
	.852		.337	.362			.374	.418				.138		.166	
Plug	0.000	.338	.904	1.129	.787	.558	.881	1.217	.904	.850					
	.003	-.949	-.564	-.108	-.439	-.901	-.625	-.241	-.361	-.1236					
	.014	-.546	-.467	-.226	-.374	-.629	-.486	-.028	-.306	-.651					
	.031	-.822	-.434	-.402	-.620	-.445	-.593	-.268	-.848	-.448					
	.055	-.937	.627	-.243	-.351	-.513	.056	-.138	-.328	-.387					
	.106	-.415	-.230	-.200	-.261	-.274	-.359	-.138	-.251	-.232					
	.172	-.109	-.012	-.071	-.177	-.158	-.002	-.028	-.222	-.102					
	.261	-.172	-.129	-.142	-.200	-.083	.001	.103	-.093	.014					
	.302	-.100	-.048	-.012	.033	.026	-.038	.011	.027	.014					
	.326	-.075	.033	.049	-.025	-.073	.063	.082	.033	.121					
	.343	-.701	.085	-.077	.078	-.591	.082	.082	.031	.101					
	Fan cowl	.387	.449	.140	.079	.130	.392	.441	.111	.240	.424				
.442		.060	-.006	.020	.017	.056	.164	.004	.059	.111					
.493		-.027	.085	.046	.120	.078	.124	.089	.234	.263					
.537		-.071	.092	.066	.023	.201	.130	.108	.121	.250					
.581		-.007	.043	.004	-.032	.052	.121	.056	-.038	.179					
.617		-.090	.069	.072	.075	.068	.157	.115	.040	.208					
.653		-.041	.101	.014	.014	.081	.163	.121	.085	.150					
.690		-.090	.085	.156	.007	-.009	.092	.134	.082	.069					
.708			.423	.403	-.687		.465	.439	-.616						
.735			.055		-.064		.154		.004						
.766			-.058		-.337		.037		-.157						
.821			.010		-.096		.082		.020						
.852		.166		-.201		.222		.256							
Turbine cowl			.357		.466		.429		.515						

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued

(b) $M = 0.750$; outboard station

x/\bar{c}	C_p at											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000	.310	.490	.795	.647	1.156	.491	.873	1.057	.725	1.000	
	.003	-.221	-.703	.544	-.715	-.385	-.910	-.744	.306	-.584	-.826	
	.014	-.314	-.555	-.765	-.535	-.406	-.529	-.537	-.380	-.455	-.617	
	.031	-.184	-.326	-.878	-.600	-.309	-.618	-.555	-.499	-.585	-.423	
	.055	-.278	-.020	-.461	-.397	-.300	-.344	.001	-.318	-.387	-.414	
	.106	-.225	-.286	-.325	-.356	-.276	-.322	-.253	-.247	-.279	-.364	
	.172	-.124	-.065	-.155	-.206	-.276	-.114	-.022	-.093	-.217	-.158	
	.261	-.221	.379	.218	-.253	-.203	-.189	.081	-.170	-.167	-.226	
	.302	-.137	.057	-.041	-.179	-.159	-.100	-.055	-.022	-.128	-.004	
	.326	-.110	.000	.009	-.168	-.141	-.078	.022	.037	.042	.040	
	.343	-.543	.652	.094	.130	-.435	-.529	.167	.084	.154	-.435	
	.387	.435	.006	.009	-.041	-.150	.488	.001	.072	-.002	.305	
	.442	.261	-.136	.077	-.197	.103	.183	-.005	.011	-.019	-.081	
	.483	.535	.000	.050	-.250	-.185	.209	.049	.046	.140	.134	
	.537	-.031	-.050	.071	-.197	-.120	.037	.090	.052	.066	.048	
	.581	-.119	-.106	.150	-.377	-.491	.001	.031	-.078	.111	-.046	
	.617	-.146	.035	.014	-.371	-.638	.023	.125	.031	-.093	-.287	
Turbine cowl	.653	.031	-.136	.077	-.221	-.385	.019	.063	.049	.016	-.264	
	.690	.411	-.145	.012	-.239	-.444	.072	.055	.055	.382	-.390	
	.708	.708	.308	.256	-.591			.377	.383	-.741		
	.735	.735	-.003		.009			.090		.045		
	.768	.768	.402		-.441			-.093		-.505		
Plug	.796	.796	.207		-.403			-.028		-.117		
	.821	.821	.698		-.156			.143		.164		
	.852		.311		.333			.271		.419		
Fan cowl	0.000	-.320	.865	1.127	.773	.565	.160	.838	1.222	.838	.859	
	.003	-.057	-.699	.205	-.478	-.979	-.1232	-.740	.342	-.377	-.1260	
	.014	-.556	-.551	-.260	-.416	-.616	-.1130	-.545	-.063	-.380	-.592	
	.031	-.663	-.506	-.447	-.755	-.419	-.574	-.450	-.290	-.725	-.439	
	.055	-.748	.015	-.260	-.407	-.369	-.353	.035	.160	-.319	-.266	
	.106	-.273	-.243	-.207	-.227	-.268	-.286	-.290	.148	-.260	-.242	
	.172	-.100	.009	.071	-.139	-.259	-.083	.002	.027	-.077	-.057	
	.261	-.167	-.036	.142	-.183	.153	-.127	.126	.101	.066	.026	
	.302	.087	-.035	.009	.039	.044	-.052	-.024	.017	.085	.105	
	.326	-.056	.036	.057	.050	.033	-.026	.055	.088	.114	.135	
	.343	-.525	.172	.068	.163	.431	-.525	.165	-.042	.179	.416	
	.387	.507	.030	.086	.049	.419	.540	.082	.118	.102	.476	
	.442	.112	.012	.033	.047	-.035	.275	.026	.061	.070	.123	
	.483	.055	.065	.068	.130	.160	.328	.082	.088	.123	.153	
	.537	.050	.083	.068	.074	.157	.124	.064	.118	.120	.279	
	.581	.015	.051	-.009	-.103	.044	-.120	.082	.043	.015	.164	
	.617	.108	.154	.077	.050	.033	.142	.165	.118	.079	.164	
Turbine cowl	.653	.024	.107	.095	.130	.071	.155	.194	.103	.143	.123	
	.690	.015	.089	.172	-.371	-.044	.253	.171	.147	.101	.626	
	.708		.421	.409	-.631			.443	.443	-.751		
	.735		.113		.033			.144		.035		
	.768		-.071		-.022			.011		-.454		
Plug	.796		.015		-.059			.070		.023		
	.821		.166		.195			.209		.235		
	.852		.591		.452			.413		.485		

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued
(c) $M = 0.775$; outboard station

x/\bar{c}	C_p at																			
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 4^\circ$							
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.814	.815	.807	.651	1.133	.583	.868	.963	.692	1.071	.488	.868	1.051	.319	1.003				
	.003	-.251	-.768	.655	-.722	-.364	-.711	-.778	.442	-.664	-.670	-.923	-.800	.374	-.827	-.862				
	.014	-.328	-.587	-.1017	-.554	-.412	-.499	-.579	.505	-.520	-.540	-.533	-.583	.395	-.569	-.616				
	.031	-.154	-.581	-.666	-.626	-.322	-.767	-.573	.715	-.506	-.384	-.521	-.596	.308	-.577	-.621				
	.055	-.285	-.023	-.468	-.401	-.299	-.334	-.066	.380	-.370	-.384	-.343	-.402	.350	-.356	-.365				
	.106	-.192	-.292	-.332	-.398	-.293	-.165	-.261	.278	-.370	-.322	-.245	-.250	.244	-.320	-.311				
	.172	-.120	-.059	-.156	-.251	-.150	-.114	-.046	.117	-.212	-.156	-.105	-.029	.051	-.150	-.150				
	.261	-.222	.507	-.224	-.257	-.175	-.207	.244	.188	-.209	-.173	-.181	-.158	.165	-.187	-.218				
	.302	-.137	-.091	-.034	-.144	-.169	-.114	-.057	.029	.051	.043	-.092	.043	.045	.107	-.053				
	.326	-.107	.003	-.006	-.203	-.028	-.080	.017	.028	.112	.016	-.063	.031	.046	.191	-.053				
	.343	-.463	.130	-.042	.183	.344	.436	.215	.040	.192	.373	.453	.221	-.046	.191	-.053				
	.387	.355	.026	.011	.059	.138	.404	.036	.034	.056	.019	.442	.034	.068	.078	.135				
Turbine cowl	.442	.376	-.119	-.062	-.127	.135	.217	-.111	-.023	.060	.152	.238	.086	-.049	-.046	.090				
	.493	.749	.111	.006	-.291	.336	.310	.113	.008	.081	.014	.285	.110	.028	.208	.186				
	.537	-.018	-.167	-.136	-.138	-.223	-.084	-.046	.031	-.235	.019	.035	.028	.016	-.184	-.101				
	.581	-.041	-.023	-.288	-.440	-.308	-.603	.113	.070	-.240	-.342	-.008	-.023	-.034	-.022	-.029				
	.617	-.179	-.170	.003	-.364	-.714	.075	.113	.070	-.195	.410	.052	.215	.068	.159	-.029				
	.653	.206	.024	.150	-.387	-.593	.035	.030	.017	.418	.342	.090	.008	.028	.339	-.441				
	.690	.672	-.268	.020	-.392	-.460	.158	.051	.031	.317	.353	.162	.158	.102	.167	-.280				
	.708	.705	.077	.286	-.629			.365	.368	.817			.360	.368						
	.735	.735	.077	.053	-.053			.070	.059	.059			.121		.050					
	.768	.768	-.428	-.031	-.381			.253	.382	.382			.267		.050					
	.796	.796	-.507	-.657	-.152			-.040	.314	.314			.008		.050					
	.821	.821	.130	.152				.147	.231	.231			.144		.225					
.852	.328	.328	.372				.348	.403	.403			.368		.420						
Fan cowl	0.000	-.399	-.874	1.125	.772	.969	.209	.847	1.223	.827	.895									
	.003	-.1013	-.834	.251	-.491	-.978	-.181	-.812	.389	.365	-.1.244									
	.014	-.733	-.570	-.289	-.424	-.636	-.1.210	-.560	.082	.385	.871									
	.031	-.580	-.539	-.471	-.740	-.429	-.531	-.499	.276	.687	.430									
	.055	-.342	-.018	-.275	-.382	-.316	-.331	.031	.170	.379	.283									
	.106	-.154	-.249	-.215	-.234	-.274	-.258	.276	.151	.170	.102									
	.172	-.091	.019	-.067	-.189	-.155	-.084	.014	.105	.139	.025									
	.261	-.164	.052	-.141	-.214	-.025	-.119	.202	.105	.240	.074									
	.302	-.079	-.031	.006	.052	.060	-.034	.017	.014	.090	.074									
	.326	-.049	.035	.063	.105	.088	-.013	.057	.091	.158	.071									
	.343	-.469	.146	-.022	.204	.326	.433	.020	.020	.223	.399									
	Turbine cowl	.387	.486	.029	.092	.066	.004	.344	.028	.037	.088	.503								
.442		.125	-.036	.028	.224	.196	.420	.028	.037	.088	.503									
.493		.108	.063	.092	.224	.196	.420	.028	.037	.088	.503									
.537		.104	.109	.129	-.228	.111	.153	.094	.131	.170	.238									
.581		.023	.049	-.002	-.177	.049	.102	.094	.020	.158	.260									
.617		.104	.171	.055	-.135	-.005	.140	.176	.091	.122	.147									
.653		.062	.089	.075	.043	.032	.234	.122	.102	.079	.113									
.690		.074	.086	.129	-.211	-.189	.332	.122	.102	.079	.113									
.708		.384	.384	.407	-.211	-.189	.332	.122	.102	.079	.113									
.735		.123	.123	.150	.015			.162	.446	.311	.000									
.768		-.042	-.042	-.015	-.015			.162	.446	.311	.000									
Plug		.796	.005	.005	.037	.037			.054	.446	.311	.000								
	.821	.148	.148	.213	.213			.054	.446	.311	.000									
	.852	.387	.387	.407	-.211	-.189	.332	.122	.102	.079	.113									
	0.000	-.399	-.874	1.125	.772	.909	.209	.847	1.223	.827	.895									
	.003	-.1013	-.834	.251	-.491	-.978	-.181	-.812	.389	.365	-.1.244									
	.014	-.733	-.570	-.289	-.424	-.636	-.1.210	-.560	.082	.385	.871									
	.031	-.580	-.539	-.471	-.740	-.429	-.531	-.499	.276	.687	.430									
	.055	-.342	-.018	-.275	-.382	-.316	-.331	.031	.170	.379	.283									
	.106	-.154	-.249	-.215	-.234	-.274	-.258	.276	.151	.170	.102									
	.172	-.091	.019	-.067	-.189	-.155	-.084	.014	.105	.139	.025									
	.261	-.164	.052	-.141	-.214	-.025	-.119	.202	.105	.240	.074									

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Continued
(d) $M = 0.800$; outboard station

C_p at -											
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H
Fan cowl	0.000	.786	.864	.812	.669	1.164	.584	.864	.962	.695	1.073
	.003	.345	.848	.783	.730	.407	.760	.880	.519	.665	.700
	.014	.337	.640	.783	.591	.445	.508	.613	.515	.529	.553
	.031	.288	.657	.759	.689	.355	.422	.624	.798	.548	.418
	.055	.238	.630	.733	.610	.336	.336	.619	.400	.366	.393
	.106	.153	.368	.340	.301	.255	.010	.275	.291	.328	.293
	.172	.116	.062	.163	.239	.187	.112	.032	.122	.184	.149
	.261	.227	.660	.237	.263	.171	.210	.350	.198	.176	.176
	.302	.133	.092	.049	.206	.138	.120	.062	.040	.111	.019
	.326	.100	.002	.005	.217	.119	.080	.009	.025	.242	.022
	.343	.394	.161	.002	.220	.076	.398	.172	.003	.234	.054
	.387	.185	.002	.082	.076	.260	.259	.050	.061	.025	.323
Turbine cowl	.442	.581	.055	.166	.008	.120	.295	.068	.006	.078	.209
	.493	.996	.038	.135	.255	.336	.409	.156	.137	.097	.370
	.537	.039	.185	.128	.079	.135	.031	.122	.062	.024	.071
	.581	.068	.145	.059	.328	.358	.018	.052	.119	.295	.317
	.617	.092	.166	.174	.338	.605	.125	.079	.101	.203	.521
	.653	.487	.095	.090	.331	.681	.145	.074	.076	.152	.276
	.690	.960	.114	.117	.434	.787	.267	.163	.036	.187	.382
	.708	.200	.028	.268	.749	.047	.110	.363	.352	.095	.871
	.735	.028	.028	.047	.047	.110	.760	.110	.329	.095	.112
	.768	.319	.319	.317	.317	.329	.329	.329	.329	.320	.320
	.796	.613	.613	.608	.608	.613	.613	.613	.613	.613	.613
	.821	.052	.052	.076	.076	.178	.178	.178	.228	.532	.228
	.852	.347	.347	.356	.356	.374	.374	.374	.408	.608	.374
Plug											
Fan cowl	0.000	.421	.858	1.105	.779	1.001	.282	.832	1.211	.819	.910
	.003	.993	.900	.355	.535	.101	.352	.628	.745	.965	.910
	.014	.993	.630	.327	.445	.249	.189	.628	.745	.965	.910
	.031	.465	.606	.267	.371	.459	.370	.568	.622	.835	.835
	.055	.331	.093	.232	.372	.326	.313	.017	.193	.338	.376
	.106	.302	.258	.229	.245	.231	.069	.277	.174	.246	.219
	.172	.087	.086	.076	.171	.022	.065	.002	.035	.078	.040
	.261	.157	.135	.156	.149	.022	.065	.002	.035	.078	.040
	.302	.078	.035	.016	.043	.057	.044	.016	.002	.082	.058
	.326	.045	.027	.008	.095	.087	.011	.039	.085	.248	.139
	.343	.383	.165	.008	.236	.318	.403	.194	.017	.250	.273
	.387	.289	.071	.000	.119	.149	.482	.110	.113	.142	.367
Turbine cowl	.442	.200	.180	.030	.013	.149	.482	.110	.113	.142	.367
	.493	.289	.071	.000	.013	.149	.482	.110	.113	.142	.367
	.537	.000	.003	.063	.182	.133	.518	.137	.063	.256	.288
	.581	.081	.008	.038	.090	.008	.054	.110	.102	.010	.159
	.617	.045	.224	.125	.201	.235	.020	.020	.014	.126	.126
	.653	.118	.016	.055	.185	.114	.380	.072	.099	.051	.079
	.690	.196	.188	.139	.120	.153	.457	.115	.126	.104	.074
	.708	.330	.330	.382	.1184	.120	.457	.115	.126	.104	.074
	.735	.139	.139	.038	.038	.120	.457	.115	.126	.104	.074
	.768	.235	.235	.372	.372	.120	.457	.115	.126	.104	.074
	.796	.016	.016	.174	.174	.120	.457	.115	.126	.104	.074
	.821	.166	.166	.263	.263	.120	.457	.115	.126	.104	.074
	.852	.371	.371	.440	.440	.120	.457	.115	.126	.104	.074

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 8 - Concluded
(e) $M = 0.825$; outboard station

x/c	C_p at -												
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	
Fan cowl	0.000	.774	.868	.830	.673	1.157	.604	.874	.975	.717	1.081	.522	.868
	.003	-.339	-.888	.934	.785	-.408	-.767	-.901	.619	.737	-.737	-.871	1.044
	.014	-.355	-.662	-.1001	-.589	-.439	-.531	-.635	.554	.541	-.541	-.559	.834
	.031	-.1019	-.654	-.1072	-.633	-.343	-.440	-.609	.754	.531	-.531	-.502	.818
	.055	-.300	-.549	-.305	-.408	-.311	-.334	-.022	.353	.343	-.329	-.005	.818
	.106	-.068	-.313	-.328	-.269	-.233	.263	.278	.291	.276	-.334	.127	.835
	.172	-.107	-.018	-.147	-.196	-.154	.698	.009	.117	.156	.172	.085	.860
	.261	-.225	.831	-.231	-.259	-.178	-.201	.447	.187	.157	.156	.180	.816
	.302	-.131	-.084	-.026	-.201	-.117	-.106	.054	.022	.072	-.004	.353	.816
	.326	-.327	.006	.008	.009	-.013	-.057	.014	.035	.436	.024	.055	.816
	.343	-.327	.219	.045	.259	.303	.319	.214	.036	.436	.024	.055	.816
	.387	-.046	-.029	.113	.068	.442	.447	.041	.134	.281	.218	.053	.816
	.442	.946	-.092	.068	.184	.210	.490	.037	.006	.059	.113	.066	.816
	.493	1.229	-.023	.155	.184	.223	.531	.106	.185	.216	.450	.142	.816
	.537	-.082	-.171	.014	-.136	-.139	-.000	-.117	.034	.106	.046	-.026	.816
Turbine cowl	.581	-.131	-.171	-.126	-.426	-.285	.008	.185	.020	.303	.066	.157	.021
	.617	-.084	.032	-.126	-.424	-.355	.034	.059	.075	.287	.546	.063	.018
	.653	1.911	-.135	-.128	-.445	-.607	.358	.017	.012	.203	.367	.150	.042
	.690	1.217	-.249	-.171	-.471	-.756	.385	.038	.054	.326	.320	.100	.021
	.708	-.100	.771	.213	-.746	.256	.385	.038	.054	.326	.320	.100	.021
	.735	-.302	.100	.246	-.246	.145	.385	.038	.054	.326	.320	.100	.021
	.768	-.670	-.302	.248	-.248	.145	.385	.038	.054	.326	.320	.100	.021
	.821	-.029	-.670	.319	-.319	.203	.385	.038	.054	.326	.320	.100	.021
	.852	-.234	-.029	.398	-.398	.156	.385	.038	.054	.326	.320	.100	.021
				.283		.412				.423			.350
Plug	0.000	.455	.874	1.106	.773	1.012	.289	.843	1.201	.805	.931	.522	.868
	.003	-.915	-.526	.501	-.581	1.034	-.1061	-.932	.568	.443	-.222	.871	1.044
	.014	-.946	-.628	-.347	-.468	-.728	-.1175	-.632	.568	.443	-.222	.871	1.044
	.031	-.195	-.576	-.562	-.686	-.439	-.648	.561	.264	.099	-.367	.818	.834
	.055	-.221	-.006	-.320	-.335	-.316	-.243	.015	.208	.286	-.238	.818	.834
	.106	-.010	-.254	-.241	-.222	-.230	.201	.281	.177	.147	-.239	.818	.834
	.172	-.077	.006	-.073	-.143	-.015	-.050	.021	.032	.152	.026	.818	.834
	.261	-.163	.314	-.160	-.151	-.023	-.117	.426	.127	.077	.036	.818	.834
	.302	-.065	-.023	.001	.053	.066	-.031	.006	.005	.091	.102	.818	.834
	.326	-.029	.032	.069	.234	.103	.005	.044	.094	.447	.023	.818	.834
	.343	-.317	.232	.064	.297	.337	.337	.231	.052	.285	.312	.818	.834
	.387	.203	.080	.151	.108	.072	.390	.110	.094	.172	.267	.818	.834
	.442	.474	-.015	.164	.077	.090	.657	.019	.044	.078	.175	.818	.834
	.493	.006	-.094	-.047	.002	.006	.579	.215	.110	.327	.329	.818	.834
	.537	-.006	-.094	.206	.014	.002	.171	.001	.068	.037	.217	.818	.834
Fan cowl	.581	-.093	.206	.106	-.253	-.146	.131	.036	.186	.107	.120	.818	.834
	.617	-.025	-.028	.106	.045	-.211	.099	.215	.016	.179	.052	.818	.834
	.653	.372	.167	.031	-.227	-.138	.626	.023	.081	.126	.018	.818	.834
	.690	.329	-.091	.127	-.177	-.311	.540	.226	.155	.048	-.029	.818	.834
	.708		.403	.403	-.552			.363	.384	-.592		.818	.834
	.735		.177		.158			.170	.020	.020		.818	.834
	.768		-.254		.335			.164		.336		.818	.834
	.821		-.194		.518			.021		.302		.818	.834
	.852		-.182		.260			.184		.267		.818	.834
			.359		.457			.357		.434		.818	.834

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12

(a) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.5$; Inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	.462	.354	.679	.765	1.012	.462	.376	.834	.795	.865								
	-.006	-.809	-1.557	-.748	-.006	-.256	-.784	-1.107	-.771	-.242								
	-.149	-.569	-.726	-.551	-.164	-.304	-.573	-.710	-.545	-.277								
	-.031	-.180	-.550	-.441	-.193	-.247	-.553	-.521	-.432	-.280								
	-.055	-.269	-.405	-.409	-.238	-.304	-.400	-.438	-.396	-.274								
	-.106	-.313	-.342	-.361	-.290	-.336	-.323	-.345	-.313	-.306								
	-.172	-.244	-.224	-.201	-.203	-.254	-.244	-.208	-.188	-.193	-.254							
	-.261	-.370	-.342	-.325	-.374	-.355	-.320	-.256	-.316	-.361								
	-.302	-.307	-.185	-.153	-.177	-.291	-.169	-.156	-.171	-.290								
	-.326	-.263	-.059	-.099	-.090	-.274	-.083	-.204	-.080	-.261								
	-.343	-.174	.029	.332	.048	-.451	.068	.339	.057	-.435								
Turbine cowl	-.387	-.459	-.054	-.025	-.087	-.451	-.462	-.048	-.080	-.442								
	-.442	-.019	-.262	-.124	-.251	-.087	-.124	-.252	-.238	-.100								
	-.493	-.237	-.022	-.057	-.154	-.237	-.006	-.048	-.154	-.261								
	-.537	-.310	-.131	-.086	-.293	-.187	-.304	-.092	-.283	-.193								
	-.581	-.386	-.183	-.216	-.319	-.234	-.099	-.196	-.231	-.487								
	-.617	-.275	-.102	-.134	-.235	-.257	-.105	-.140	-.219	-.267								
	-.653	-.019	-.112	-.057	-.167	-.016	-.124	-.044	-.248	-.174								
	-.690	-.069	-.061	-.003	-.090	-.079	-.055	-.004	-.041	-.145								
	-.708		.042	-.275	-.118	-.079	-.055	.285	-.102									
	-.735		.020	-.032	-.016	-.032	-.016		-.032									
	-.768		-.137	-.164	-.137	-.137	-.137		-.219									
Plug	-.796		-.080	-.125	-.073	-.073	-.157		-.166									
	-.821		-.157	-.172	-.172	-.157	-.032		-.439									
	-.852		-.028	.435														
Fan cowl	.453	.375	.973	.771	.709	.453	.400	1.082	.787	.578								
	-.591	-.183	-.543	-.618	-.531	-.802	-.531	-.197	-.457	-.900								
	-.407	-.525	-.552	-.573	-.366	-.518	-.543	-.380	-.547	-.505								
	-.321	-.582	-.421	-.417	-.310	-.517	-.517	-.316	-.444	-.468								
	-.340	-.592	-.383	-.385	-.317	-.376	-.376	-.342	-.382	-.374								
	-.106	-.321	-.324	-.304	-.304	-.370	-.349	-.300	-.305	-.312								
	-.172	-.242	-.205	-.183	-.243	-.227	-.204	-.172	-.192	-.250								
	-.261	-.337	-.321	-.263	-.346	-.319	-.316	-.281	-.315	-.331								
	-.302	-.277	-.164	-.167	-.272	-.259	-.185	-.168	-.173	-.257								
	-.326	-.239	-.078	-.222	-.246	-.224	-.082	-.242	-.086	-.237								
	-.343	-.169	-.102	-.344	-.057	-.417	-.164	.129	-.364	-.058	-.399							
Turbine cowl	-.442	-.014	-.116	-.246	-.107	-.015	-.255	-.130	-.241	-.086								
	-.493	-.226	-.052	-.149	-.253	-.227	-.021	-.050	-.150	-.254								
	-.537	-.280	-.135	-.097	-.204	-.278	-.140	-.095	-.305	-.160								
	-.581	-.248	-.119	-.198	-.233	-.252	-.104	-.234	-.283	-.155								
	-.617	-.267	-.122	-.148	-.256	-.262	-.130	-.152	-.192	-.247								
	-.653	-.014	-.142	-.058	-.246	-.015	-.140	-.053	-.237	-.183								
	-.690	-.090	-.041	-.007	-.130	-.107	-.021	-.027	-.021	-.131								
	-.708		.035	.269														
	-.735		.009	.028														
	-.768		-.142	-.217														
	Plug	-.796		-.090	-.133													
-.821			-.147	-.158														
-.852			-.046	-.426														

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued
(b) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

x/c	C_p at -																															
	$\alpha = -2^\circ$						$\alpha = 0^\circ$						$\alpha = 1^\circ$																			
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L		
Fan cowl	0.000	.536	.344	.683	.739	.993	.540	.371	.830	.760	.852																					
	.003	.101	.929	1.392	-.855	-.088	-.422	-.535	-1.229	-.925	-.318																					
	.014	.182	-.435	-1.259	-.617	-.188	-.321	-.625	-.719	-.418	-.218																					
	.031	.202	-.673	-.501	-.438	-.217	-.263	-.646	-.616	-.480	-.298																					
	.055	.283	-.443	-.507	-.452	-.253	-.313	-.627	-.479	-.424	-.258																					
	.106	.323	-.364	-.385	-.344	-.270	-.341	-.358	-.327	-.318	-.262																					
	.172	.280	-.233	-.207	-.214	-.270	-.249	-.216	-.199	-.200	-.262																					
	.261	.392	-.361	-.392	-.274	-.350	-.373	-.339	-.275	-.333	-.377																					
	.302	.318	.192	-.160	-.188	-.308	-.295	-.375	-.275	-.333	-.377																					
	.326	.260	-.056	-.035	-.085	-.264	-.246	-.074	-.105	-.074	-.247																					
Turbine cowl	.343	.133	.056	-.039	.112	.300	.119	.116	.240	.118	.255																					
	.387	.577	-.056	-.047	.126	.585	-.550	-.047	.056	-.115	.574																					
	.442	.021	-.018	-.128	.127	.072	-.018	-.038	-.123	.145	.087																					
	.493	.339	.178	-.158	.347	.329	-.327	.149	-.152	-.351	.351																					
	.537	.237	.181	-.216	.097	.361	-.237	.184	-.222	-.080	.300																					
	.581	.381	.321	-.241	-.358	.144	-.373	.237	.256	.342	.377																					
	.617	.243	.011	-.172	-.297	.361	-.235	.023	.181	-.348	.359																					
	.653	.021	.213	-.050	.141	-.150	-.018	-.216	-.053	.098	-.144																					
	.690	.087	.051	-.012	-.229	-.162	-.056	-.056	-.041	-.224	-.133																					
	.708		.052	.282	.159	-.159			.041	.224	-.133																					
Plug	.735	.064	.098	.098	.098			.067	.295	.153																						
	.768	-.006	-.306	-.388	-.327	-.327	-.327	-.327	-.327	.110																						
	.796	-.018	-.018	-.018	-.012	-.012	-.012	-.012	-.009	.009																						
	.821	.166	.166	.162	.169	.169	.169	.169	.165	.165																						
	.852	.076	.076	.433	.433	.433	.433	.433	.437	.437																						
	Fan cowl	0.000	.535	.345	.591	.768	.711	.533	.385	1.071	.760	.562																				
		.003	.805	-1.005	-.746	-.696	-.820	-1.212	-.550	-.279	-.799	-1.241																				
		.014	.465	-.641	-.641	-.638	-.429	-.507	-.632	-.413	-.610	-.487																				
		.031	.361	-.682	-.493	-.500	-.365	-.367	-.617	-.360	-.499	-.428																				
		.055	.334	-.438	-.429	-.435	-.344	-.268	-.419	-.360	-.425	-.372																				
.106		.379	-.356	-.362	-.330	-.327	-.360	-.337	-.325	-.325	-.328																					
.172		.252	.215	-.199	-.203	-.271	-.230	-.208	-.176	-.195	-.251																					
.261		.361	.347	-.292	-.341	-.368	-.331	-.337	-.281	-.331	-.340																					
.302		.299	.179	-.179	-.184	-.280	-.256	-.256	-.168	-.175	-.257																					
.326		.217	.088	-.120	-.080	-.239	-.215	-.074	-.141	-.078	-.225																					
Turbine cowl	.343	.125	.133	-.056	.120	.274	.114	.159	.241	.132	.257																					
	.387	.534	.056	-.056	-.112	.588	.536	.051	.045	-.110	.578																					
	.442	.024	-.024	-.126	.123	.083	-.013	-.045	-.112	.126	.069																					
	.493	.321	.173	-.161	.156	-.362	.305	.165	-.153	-.357	-.251																					
	.537	.253	.202	-.237	-.277	-.305	-.215	-.208	-.226	-.057	-.284																					
	.581	.379	.257	-.262	-.350	.338	.363	.255	-.257	.345	-.367																					
	.617	.243	.058	-.196	-.359	-.362	-.235	.005	-.194	.375	-.363																					
	.653	.054	-.237	-.077	.077	-.156	-.235	-.235	-.074	.074	-.160																					
	.690	-.119	.046	-.071	-.224	-.165	-.017	-.017	-.074	-.234	-.145																					
	.708		.037	.270	.135	-.158		.034	.291	.144	-.145																					
Plug	.735	.034	.086	.086	.086			.031	.089	.089																						
	.768	-.034	-.034	-.034	-.034			-.031	-.089	-.089																						
	.796	-.030	-.030	-.030	-.030			-.021	-.089	-.089																						
	.821	.151	.151	.146	.146			.154	.154	.154																						
	.852	.046	.046	.415	.415			.037	.416	.416																						

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(c) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

C _p at -																		
x/c	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000	.570	.374	.713	.746	.970	.847	-1.226	-1.023	-1.219	-.036	-.029						
	.003	-.100	-.989	1.336	-.935	-.138	-1.226	-1.023	-1.219	1.308	-1.090							
	.014	-.183	-.682	-.659	-.188	-1.176	-1.023	-1.219	1.308	-1.003								
	.031	-.199	-.765	-.955	-.231	-1.176	-1.023	-1.219	1.308	-1.003								
	.055	-.282	-.453	-.455	-.267	-1.176	-1.023	-1.219	1.308	-1.003								
	.106	-.313	-.371	-.391	-.355	-.474	-.473	-.585	-.353	-.344								
	.172	-.404	-.235	-.227	-.217	-.270	-.283	-.244	-.229	-.294								
	.261	-.404	-.235	-.227	-.217	-.270	-.283	-.244	-.229	-.294								
	.302	-.315	-.153	-.162	-.191	-.312	-.305	-.185	-.186	-.305								
	.326	-.252	-.095	-.127	-.084	-.256	-.244	-.120	-.064	-.118	-.254							
	.343	-.108	-.134	-.176	-.150	-.234	-.294	-.134	-.137	-.116	-.311							
	.387	-.589	-.053	-.061	-.157	-.087	-.211	-.081	-.064	-.082	-.206							
	.442	-.017	-.182	-.084	-.125	-.087	-.011	-.120	-.092	-.132	-.192							
	.493	-.318	-.330	-.008	-.310	-.290	-.205	-.131	-.098	-.147	-.231							
Turbine cowl	.537	-.351	-.003	-.059	-.135	-.563	-.241	-.143	-.106	-.152	-.222							
	.581	-.253	-.254	-.121	-.174	-.047	-.233	-.154	-.124	-.164	-.141							
	.617	-.288	-.114	-.095	-.290	-.377	-.225	-.143	-.126	-.161	-.260							
	.653	-.017	-.028	-.134	-.180	-.011	-.134	-.092	-.132	-.203								
	.690	-.141	-.123	-.017	-.155	-.126	-.106	-.087	-.118	-.152								
	.708		-.129	.285	.044		-.112	-.106	-.113									
	.735		-.081	.081	.081		-.115	-.110	-.110									
	.768		-.218	.262	.262		-.078	-.028	-.028									
	.796		-.033	.003	.003		-.014	-.036	-.036									
	.821		-.165	.219	.219		-.065	-.036	-.036									
Plug	.852		-.207	.433	.433		-.171	.064	.064									
Fan cowl	0.000	-.570	-.380	.949	.773	-.712	-.576	-.372	1.071	-.766	-.576							
	.003	-.832	-.1055	-.755	-.897	-.911	-1.187	-.989	-.306	-.893	-1.331							
	.014	-.665	-.640	-.660	-.634	-.429	-.464	-.659	-.446	-.625	-.467							
	.031	-.360	-.741	-.543	-.516	-.378	-.414	-.704	-.379	-.500	-.424							
	.055	-.366	-.450	-.442	-.448	-.338	-.381	-.435	-.368	-.433	-.379							
	.106	-.388	-.358	-.369	-.333	-.327	-.389	-.345	-.340	-.328	-.323							
	.172	-.249	-.218	-.195	-.197	-.265	-.231	-.211	-.177	-.195	-.246							
	.241	-.366	-.352	-.296	-.344	-.369	-.333	-.345	-.289	-.342	-.242							
	.302	-.283	-.176	-.179	-.175	-.279	-.253	-.169	-.180	-.252	-.242							
	.326	-.230	-.078	-.031	-.068	-.228	-.206	-.071	-.057	-.210	-.210							
	.343	-.105	-.171	-.180	-.155	-.203	-.171	-.073	-.071	-.163	-.193							
	.387	-.568	-.031	-.081	-.135	-.071	-.056	-.073	-.073	-.133	-.088							
	.442	-.017	-.182	-.092	-.128	-.031	-.012	-.139	-.082	-.122	-.012							
	.493	-.319	-.308	-.017	-.327	-.338	-.314	-.303	-.017	-.328	-.320							
Turbine cowl	.537	-.307	-.226	-.084	-.127	-.593	-.284	-.009	-.019	-.139	-.498							
	.581	-.324	-.242	-.118	-.223	-.259	-.242	-.125	-.125	-.224	-.334							
	.617	-.259	-.117	-.126	-.324	-.383	-.264	-.132	-.121	-.328	-.376							
	.653	-.017	-.054	-.145	-.290	-.194	-.012	-.045	-.121	-.212	-.155							
	.690	-.158	-.126	-.070	-.133	-.124	-.153	-.103	-.094	-.103	-.122							
	.708		.129	.263	.025			.134	.263	.035								
	.735		.067	.072	.072			.047	.066	.066								
	.768		-.341	-.276	-.276			-.351	-.266	-.266								
	.796		-.025	-.025	-.025			-.043	-.094	-.094								
	.821		-.157	.211	.211			-.153	.216	.216								
Plug	.852		.105	.425	.425			.097	.425	.425								

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued
(d) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
Fan cowl	0.000	.649	.375	.714	.740	.955	.635	.401	.831	.759	.838	
	.003	-.122	-1.060	-1.285	-1.001	-.138	-.512	-1.058	-1.230	-1.037	-.442	
	.014	-.213	-.691	-1.233	-.651	-.219	-.362	-.892	-.980	-.621	-.342	
	.031	-.208	-.823	-1.111	-.694	-.244	-.274	-.918	-.748	-.646	-.301	
	.055	-.298	-.463	-.341	-.483	-.274	-.344	-.449	-.487	-.467	-.317	
	.106	-.341	-.385	-.387	-.363	-.331	-.357	-.369	-.390	-.347	-.325	
	.172	-.269	-.234	-.207	-.219	-.276	-.253	-.220	-.202	-.200	-.274	
	.261	-.418	-.387	-.296	-.385	-.423	-.292	-.366	-.301	-.366	-.399	
	.302	-.322	-.194	-.166	-.195	-.317	-.258	-.180	-.169	-.176	-.293	
	.326	-.250	-.088	-.477	-.084	-.249	-.229	-.072	-.122	-.062	-.220	
	.343	-.098	.175	-.131	.182	-.176	-.077	.194	-.129	.193	-.165	
	.387	-.642	-.083	-.062	-.185	-.738	-.621	.056	.070	.152	-.741	
	.442	-.018	-.091	-.053	-.180	-.293	-.013	.075	-.051	.154	-.263	
	.493	-.306	-.277	-.142	-.184	-.143	-.306	.272	.153	.181	-.154	
Turbine cowl	.537	-.421	-.164	-.234	-.309	-.689	-.397	.134	.231	.320	-.670	
	.581	-.245	-.226	-.217	-.124	-.308	-.226	.239	.214	.097	-.184	
	.617	-.325	-.140	-.145	-.055	-.241	-.309	.140	.161	-.067	-.247	
	.653	-.018	-.166	-.045	-.089	-.233	-.013	.153	.037	-.114	-.228	
	.690	-.106	-.052	-.026	-.170	-.140	-.112	.043	-.067	-.190	-.100	
	.708		-.033	.276	.214	.052	.212	.294				
	.735		-.124	.117	.135	.129						
	.768		-.333	.309	.350	.296						
	.796		-.066	.189	.201	.254						
	.821		-.202	.241	.213	.234						
Plug	.852		-.380	.443	.425							
	0.000	.622	.375	.555	.759	.714	.639	.400	1.073	.766	.583	
	.003	-.941	-1.053	-.808	-.958	-.937	-1.201	-1.031	-.347	-.947	1.509	
	.014	-.859	-.573	-.976	-.627	-.423	-.882	-.681	-.476	-.632	-.798	
	.031	-.371	-.194	-.595	-.632	-.383	-.394	-.781	-.414	-.657	-.418	
	.055	-.368	-.532	-.463	-.456	-.356	-.351	-.452	-.395	-.450	-.377	
	.106	-.384	-.364	-.380	-.342	-.337	-.378	-.358	-.360	-.344	-.333	
	.172	-.231	-.218	-.197	-.198	-.286	-.237	-.212	-.188	-.197	-.263	
	.261	-.373	-.366	-.307	-.358	-.317	-.346	-.358	-.309	-.358	-.355	
	.302	-.283	-.172	-.183	-.174	-.274	-.202	-.169	-.188	-.173	-.255	
Fan cowl	.326	-.219	-.070	-.110	-.082	-.223	-.202	.067	.068	.206	-.206	
	.343	-.080	.202	.132	.184	.152	.071	.230	.142	.201	-.138	
	.387	-.616	-.046	.055	.149	.733	-.621	.045	.055	.146	-.725	
	.442	-.019	-.003	-.051	-.146	.158	-.013	-.024	.050	.168	-.219	
	.493	-.305	-.294	-.162	-.198	.195	-.303	.285	.164	-.192	-.185	
	.537	-.381	-.062	-.229	-.320	.043	-.355	.107	.244	-.339	-.627	
	.581	-.237	-.275	-.212	-.084	-.133	-.245	.263	.197	-.081	-.230	
	.617	-.312	-.167	-.167	-.068	-.261	-.314	.156	.174	-.083	-.271	
	.653	-.019	-.140	-.032	-.141	-.228	-.013	.126	.010	-.159	-.246	
	.690	-.131	-.011	-.140	-.184	-.149	-.133	-.024	-.204	-.249	-.121	
Turbine cowl	.708		.027	.296	.192			.055	.324	.191		
	.735		.110	.101	.101			.122		.105		
	.768		-.358	.282	.282			-.371		-.279		
	.796		-.005	-.353	-.353			-.077		-.445		
	.821		.193	.246	.246			.203		.247		
	.852		.221	.425	.425			.205		.431		
Plug	0.000	.622	.375	.555	.759	.714	.639	.400	1.073	.766	.583	
	.003	-.941	-1.053	-.808	-.958	-.937	-1.201	-1.031	-.347	-.947	1.509	
	.014	-.859	-.573	-.976	-.627	-.423	-.882	-.681	-.476	-.632	-.798	
	.031	-.371	-.194	-.595	-.632	-.383	-.394	-.781	-.414	-.657	-.418	
	.055	-.368	-.532	-.463	-.456	-.356	-.351	-.452	-.395	-.450	-.377	
	.106	-.384	-.364	-.380	-.342	-.337	-.378	-.358	-.360	-.344	-.333	
	.172	-.231	-.218	-.197	-.198	-.286	-.237	-.212	-.188	-.197	-.263	
	.261	-.373	-.366	-.307	-.358	-.317	-.346	-.358	-.309	-.358	-.355	
	.302	-.283	-.172	-.183	-.174	-.274	-.202	-.169	-.188	-.173	-.255	
	.326	-.219	-.070	-.110	-.082	-.223	-.202	.067	.068	.206	-.206	

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(e) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.5$; inboard station

x/c	C _p at -																		
	α = -2°						α = 0°												
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row K	Row L		
Fan cowl	0.000	.843	.404	.743	.768	.962	.773	.415	.868	.780	.849								
	.003	-.106	-1.019	-1.256	-1.006	-.152	-.515	-1.003	-1.156	-1.051	-.483								
	.014	-.213	-.754	-1.159	-.798	-.212	-.372	-.728	-.959	-.684	-.339								
	.031	-.216	-.798	-1.076	-.703	-.267	-.305	-.793	-.829	-.698	-.331								
	.055	-.321	-.906	-.915	-.594	-.275	-.331	-.417	-.588	-.436	-.313								
	.106	-.339	-.392	-.335	-.363	-.350	-.344	-.378	-.417	-.357	-.347								
	.172	-.260	-.236	-.197	-.217	-.280	-.251	-.217	-.199	-.200	-.266								
	.261	-.437	-.425	-.311	-.418	-.476	-.410	-.359	-.321	-.399	-.423								
	.302	-.326	-.195	-.144	-.186	-.319	-.257	-.176	-.168	-.174	-.287								
	.326	-.239	-.083	-.070	-.088	-.241	-.181	-.061	-.041	-.051	-.216								
	.343	-.078	.720	.070	.218	-.115	.053	.258	.081	.321	.192								
	.387	-.793	.319	.152	.732	.708	.502	.303	.376	.577	.192								
	.442	-.013	.319	.152	.732	.708	.502	.303	.376	.577	.192								
	.493	-.242	-.122	-.132	-.107	-.253	-.151	-.123	-.117	-.117	-.093								
Turbine cowl	.537	-.303	-.288	-.275	-.311	-.087	-.350	-.212	-.223	-.295	-.052								
	.561	-.344	-.164	-.071	-.282	.702	-.351	-.155	-.289	-.300	-.378								
	.617	-.280	-.117	.165	-.331	-.382	-.253	.107	-.098	-.025	-.200								
	.653	-.021	-.164	-.052	-.199	-.012	-.163	-.108	-.022	-.195	-.135								
	.690	-.134	-.112	-.088	-.246	-.162	-.135	-.146	.276	.194									
	.708		.114	.287	.133			.172		.237									
	.735		.140		.174			.331		.343									
	.768		-.322		-.259			.362		.302									
	.796		-.174		-.497			.268		.477									
	.821		.230		.280			.442											
	.852		.720		.439														
	Fan cowl	0.000	.754	.401	.955	.786	.703	.787	.427	1.081	.779	.557							
		.003	-.552	-1.016	-.834	-.996	-.980	-.122	-.567	-.959	-.925	1.244							
		.014	-.448	-.756	-.660	-.626	-.422	-.911	-.676	-.471	-.640	-.912							
.031		-.370	-.795	-.663	-.671	-.406	-.363	-.816	-.582	-.731	-.386								
.055		-.383	-.418	-.431	-.443	-.351	-.389	-.393	-.416	-.412	-.367								
.106		-.401	-.271	-.392	-.346	-.343	-.397	-.364	-.380	-.344	-.338								
.172		-.252	-.213	-.195	-.156	-.267	-.235	-.211	-.185	-.194	-.252								
.261		-.386	-.387	-.319	-.377	-.388	-.350	-.380	-.331	-.383	-.365								
.302		-.280	-.171	-.184	-.173	-.267	-.250	-.167	-.188	-.168	-.247								
.326		-.211	-.062	-.349	-.055	-.209	-.189	-.055	-.081	-.048	-.194								
.343		-.067	.229	-.088	.220	-.102	-.050	.269	.081	.234	.079								
.387		-.692	-.057	.084	-.144	.718	-.687	.055	.085	.155	.726								
.442		-.015	-.247	-.119	-.225	-.330	-.011	-.252	.141	.236	.349								
.493		-.278	-.208	-.137	-.112	-.133	-.276	-.154	-.141	.074	.126								
Turbine cowl	.537	-.478	-.257	-.283	-.314	-.658	-.456	-.271	-.255	-.299	-.650								
	.581	-.319	-.153	-.104	-.220	.091	-.302	-.138	-.111	-.260	.030								
	.617	-.255	-.105	.076	-.309	-.291	-.260	-.133	.080	-.294	-.331								
	.653	-.018	-.176	-.096	-.125	-.154	-.011	-.198	-.110	-.108	-.187								
	.690	-.149	-.080	-.044	-.285	-.141	-.147	-.105	-.011	-.273	-.134								
	.708		.085	.258	.132			.126	.259	.141									
	.735		.057		.150			.121		.172									
	.768		-.251		-.267			-.318		-.257									
	.796		-.171		-.511			-.238		-.548									
	.821		-.219		-.269			-.236		-.278									
	.852		.425		.431			.389		.445									

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(f) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C_p at											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row M	Row H	Row I	Row J	Row K	Row L	Row M
0.030							.663	.027	.554	.446	.570	
.003							.966	-1.526	1.538	-1.484	.812	
.014							.420	.652	.694	.441	.422	
.031							.351	.636	.594	.515	.360	
.055							.338	.467	.479	.444	.318	
.136							.357	.368	.364	.331	.321	
.172							.249	.223	.200	.202	.267	
.251							.360	.325	.255	.315	.367	
.332							.300	.178	.162	.176	.299	
.376							.259	.095	.233	.086	.280	
.383							.329	.082	.156	.379	.567	
.387							.344	.067	.031	.057	.350	
.462							.015	.127	.095	.128	.115	
.493							.230	.114	.088	.137	.502	
.537							.205	.117	.092	.150	.496	
.581							.225	.152	.128	.180	.353	
.617							.243	.160	.114	.134	.280	
.633							.013	.092	.053	.086	.150	
.650							.104	.031	.021	.066	.092	
.708								.011	.035	.323		
.735								.014	.011	.011		
.768								.005	.011	.011		
.796								.014	.011	.011		
.821								.091	.093	.093		
.852								.018	.166	.166		
$\alpha = 1^\circ$												
0.030							.584	.028	.873	.447	.237	
.003							.1432	-1.441	.743	-1.504	.812	
.014							.752	.650	.548	.649	.724	
.031							.448	.603	.417	.524	.443	
.055							.403	.443	.382	.440	.388	
.106							.384	.350	.331	.333	.330	
.172							.259	.222	.190	.204	.262	
.251							.327	.327	.279	.317	.330	
.326							.264	.177	.183	.185	.272	
.387							.229	.097	.283	.095	.253	
.442							.337	.081	.158	.082	.249	
.473							.343	.049	.036	.059	.249	
.537							.020	.135	.103	.136	.214	
.581							.229	.132	.100	.146	.198	
.617							.204	.162	.167	.162	.198	
.633							.283	.157	.146	.175	.256	
.653							.251	.135	.132	.172	.285	
.708							.092	.110	.075	.101	.182	
.735							.117	.049	.039	.088	.130	
.768								.011	.015	.001		
.796								.008	.014	.014		
.821								.030	.030	.030		
.852								.004	.004	.004		
								.059	.072	.072		
								.036	.148	.148		
$\alpha = 2^\circ$												
0.030							.679	.005	.710	.447	.400	
.003							-1.479	-1.322	-1.482	-1.346	-1.432	
.014							.533	.654	.583	.640	.508	
.031							.410	.632	.521	.401	.480	
.055							.375	.440	.437	.362	.382	
.106							.378	.347	.350	.327	.327	
.172							.255	.222	.200	.204	.259	
.251							.340	.324	.273	.311	.350	
.326							.283	.177	.174	.179	.288	
.387							.249	.094	.088	.088	.241	
.442							.337	.085	.152	.079	.566	
.473							.347	.049	.037	.059	.253	
.537							.021	.129	.101	.134	.221	
.581							.230	.126	.097	.140	.201	
.617							.201	.139	.101	.156	.198	
.633							.280	.161	.137	.166	.340	
.653							.249	.126	.113	.163	.282	
.708							.021	.101	.065	.092	.172	
.735							.119	.040	.033	.075	.117	
.768								.001	.011	.011		
.796								.002	.002	.002		
.821								.017	.024	.024		
.852								.005	.005	.005		
								.079	.081	.081		
								.005	.157	.157		

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(g) $M = 0.750$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C _p at -																
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row L	
	α = -2°						α = 0°						α = 1°				
Fan cowl	0.000						.717	.073	.568	.484	.583						
	.003						.954	-1.477	1.393	-1.484	.989						
	.014						.924	.902	-1.372	-1.036	.438						
	.031						.932	.674	.4303	.527	.370						
	.055						.933	.484	.4930	.468	.341						
	.106						.981	.365	.382	.347	.335						
	.172						.921	.233	.213	.214	.273						
	.251						.973	.347	.285	.341	.385						
	.302						.939	.186	.169	.185	.312						
	.326						.927	.096	.169	.091	.285						
	.343						.924	.052	.119	.058	.500						
	.387						.939	.043	.102	.032	.361						
Turbine cowl	0.000						.914	.143	.102	.144	.226						
	.003						.924	.137	.095	.144	.229						
	.014						.9254	.143	.102	.173	.250						
	.031						.9280	.181	.144	.203	.247						
	.055						.9257	.131	.116	.173	.294						
	.106						.9017	.111	.067	.100	.176						
	.172						.9112	.040	.032	.082	.114						
	.251							.009	.030	.021							
	.302							.012	.063								
	.326							.008	.012	.012							
	.343							.018	.012	.087							
	.387							.085	.036	.160							
Plug	0.000																
	.003																
	.014																
	.031																
	.055																
	.106																
	.172																
	.251																
	.302																
	.326																
	.343																
	.387																

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(h) $M = 0.775$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

C _p at -												
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row L
Fan cowl	0.000					-1.074	-1.352	-1.343	-1.308	-1.128		
	.003					-1.074	-1.352	-1.343	-1.308	-1.128		
	.014					-1.074	-1.352	-1.343	-1.308	-1.128		
	.031					-1.074	-1.352	-1.343	-1.308	-1.128		
	.055					-1.074	-1.352	-1.343	-1.308	-1.128		
	.106					-1.074	-1.352	-1.343	-1.308	-1.128		
	.172					-1.074	-1.352	-1.343	-1.308	-1.128		
	.261					-1.074	-1.352	-1.343	-1.308	-1.128		
	.332					-1.074	-1.352	-1.343	-1.308	-1.128		
	.387					-1.074	-1.352	-1.343	-1.308	-1.128		
	.442					-1.074	-1.352	-1.343	-1.308	-1.128		
	.493					-1.074	-1.352	-1.343	-1.308	-1.128		
Turbine cowl	0.000					-1.074	-1.352	-1.343	-1.308	-1.128		
	.003					-1.074	-1.352	-1.343	-1.308	-1.128		
	.014					-1.074	-1.352	-1.343	-1.308	-1.128		
	.031					-1.074	-1.352	-1.343	-1.308	-1.128		
	.055					-1.074	-1.352	-1.343	-1.308	-1.128		
	.106					-1.074	-1.352	-1.343	-1.308	-1.128		
	.172					-1.074	-1.352	-1.343	-1.308	-1.128		
	.261					-1.074	-1.352	-1.343	-1.308	-1.128		
	.332					-1.074	-1.352	-1.343	-1.308	-1.128		
	.387					-1.074	-1.352	-1.343	-1.308	-1.128		
	.442					-1.074	-1.352	-1.343	-1.308	-1.128		
	.493					-1.074	-1.352	-1.343	-1.308	-1.128		
Plug	0.000					-1.074	-1.352	-1.343	-1.308	-1.128		
	.003					-1.074	-1.352	-1.343	-1.308	-1.128		
	.014					-1.074	-1.352	-1.343	-1.308	-1.128		
	.031					-1.074	-1.352	-1.343	-1.308	-1.128		
	.055					-1.074	-1.352	-1.343	-1.308	-1.128		
	.106					-1.074	-1.352	-1.343	-1.308	-1.128		
	.172					-1.074	-1.352	-1.343	-1.308	-1.128		
	.261					-1.074	-1.352	-1.343	-1.308	-1.128		
	.332					-1.074	-1.352	-1.343	-1.308	-1.128		
	.387					-1.074	-1.352	-1.343	-1.308	-1.128		
	.442					-1.074	-1.352	-1.343	-1.308	-1.128		
	.493					-1.074	-1.352	-1.343	-1.308	-1.128		

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(i) $M = 0.800$; $P_{t,e}/P_{t,\infty} = 1.3$; Inboard station

x/c	C _p at -														
	α = -2°					α = 0°					α = 1°				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000					-.772	-.119	-.629	-.518	-.570					
	.003					-1.049	1.306	1.285	1.251	-1.182					
	.014					-.454	-1.122	-1.267	-1.120	-.424					
	.031					-.388	-1.058	-1.122	-.962	-.394					
	.055					-.372	-.416	-.329	-.438	-.373					
	.106					-.398	-.383	-.386	-.362	-.365					
	.172					-.276	-.238	-.211	-.218	-.288					
	.261					-.404	-.386	-.310	-.381	-.419					
	.322					-.318	-.197	-.178	-.199	-.332					
	.326					-.265	-.098	-.009	-.090	-.286					
	.343					-.182	.005	-.038	-.028	-.427					
	.387					-.286	-.063	-.030	-.055	-.234					
Turbine cowl	.442					-.019	-.119	-.119	-.182	-.256					
	.493					-.265	-.146	-.114	-.166	-.223					
	.537					-.316	-.189	-.119	-.212	-.283					
	.581					-.281	-.155	-.166	-.191	-.068					
	.617					-.276	-.141	-.119	-.177	-.337					
	.653					-.022	-.152	-.081	-.144	-.223					
	.690					-.131	-.052	-.038	-.087	-.131					
	.708						.010	.029	.018						
	.735						.010	.002	.002						
	.768						-.011	-.010	-.017						
	.796						-.016	-.010	-.082						
	.821						-.083	.010	.082						
.852						.156		.154							
Fan cowl	0.000					.779	.132	.877	.531	.309					
	.003					1.314	1.303	.980	1.280	1.256					
	.014					-1.282	-1.104	-.597	-1.126	-1.262					
	.031					-.415	-.915	-.683	-.904	-.490					
	.055					-.383	-.411	-.449	-.433	-.365					
	.106					-.351	-.358	-.379	-.352	-.355					
	.172					-.272	-.224	-.220	-.193	-.208					
	.261					-.386	-.368	-.317	-.368	-.357					
	.322					-.299	-.268	-.193	-.189	-.276					
	.326					-.258	-.085	-.028	-.083	-.238					
	.343					-.413	-.015	-.042	-.020	-.406					
	.387					-.271	-.050	-.026	-.048	-.216					
Turbine cowl	.442					-.231	-.133	-.115	-.181	-.224					
	.493					-.223	-.136	-.109	-.167	-.221					
	.537					-.275	-.284	-.193	-.123	-.270					
	.581					-.052	-.179	-.175	-.133	-.056					
	.617					-.329	-.266	-.123	-.186	-.325					
	.653					-.218	-.160	-.060	-.153	-.216					
	.690					-.134	-.053	-.050	-.050	-.129					
	.708						-.004	.017	.007						
	.735						-.004	.007	.007						
	.768						-.020	.002	.002						
	.796						.013	.007	.007						
	.821						.080	.082	.071						
.852						.155	.152	.143							
Plug	0.000					.773	.140	.753	.531	.309					
	.003					1.312	1.288	1.279	1.279	1.256					
	.014					-1.085	-1.026	-.842	-1.140	-1.262					
	.031					-.308	-.803	-.482	-.942	-.490					
	.055					-.398	-.386	-.379	-.433	-.365					
	.106					-.411	-.375	-.356	-.355	-.355					
	.172					-.292	-.224	-.204	-.208	-.270					
	.261					-.376	-.372	-.313	-.368	-.357					
	.322					-.296	-.189	-.193	-.189	-.276					
	.326					-.243	-.087	-.022	-.082	-.238					
	.343					-.177	-.011	-.041	-.020	-.406					
	.387					-.275	-.057	-.025	-.047	-.216					

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(i) $M = 0.825$; $P_{t,e}/P_{t,\infty} = 1.3$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 2^\circ$			
	Row H	Row I	Row J	Row L	Row H	Row I	Row J	Row L	Row H	Row I	Row J	Row L
Fan cowl	0.030											
	.003				.842	.188	.547	.534				
	.014				-1.036	1.282	1.239	1.234				
	.031				-.509	-1.064	-1.202	-1.202				
	.055				-.393	-1.072	-1.064	-1.064				
	.106				-.390	-.399	-.395	-.371				
	.172				-.413	-.371	-.345	-.350				
	.251				-.262	-.225	-.199	-.203				
	.302				-.419	-.402	-.319	-.397				
	.326				-.316	-.194	-.176	-.193				
	.343				-.254	-.088	-.198	-.080				
	.387				-.146	-.050	-.067	-.002				
Turbine cowl	.442				-.293	-.059	-.023	-.049				
	.493				-.015	-.166	-.124	-.180				
	.537				-.254	-.101	-.115	-.151				
	.581				-.334	-.176	-.124	-.219				
	.617				-.277	-.176	-.190	-.151				
	.653				-.295	-.202	-.019	-.209				
	.690				-.015	-.129	-.082	-.125				
	.708				-.125	-.043	-.038	-.112				
	.735					.014	.032	.024				
	.758					-.037		-.014				
	.796					-.022		.016				
	.821					-.086		.091				
	.852					.359		.158				
Fan cowl	0.000											
	.033											
	.014											
	.031											
	.055											
	.106											
	.172											
	.251											
	.302											
	.326											
	.343											
	.387											
	.442											
Turbine cowl	.493											
	.537											
	.581											
	.617											
	.653											
	.690											
	.708											
	.735											
	.758											
	.796											
	.821											
	.852											
Plug	0.000											
	.033											
	.014											
	.031											
	.055											
	.106											
	.172											
	.251											
	.302											
	.326											
	.343											
	.387											
	.442											

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(k) $M = 0.700$; $P_{t,e}/P_{t,\infty} = 1.0$; inboard station

x/\bar{c}	C_p at -															
	$\alpha = -2^\circ$				$\alpha = 0^\circ$				$\alpha = 1^\circ$							
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
0.030						.803	-.517	.081	-.028	-.004						
.003						-1.264	-1.335	-1.146	-1.502	-1.097						
.014						-1.077	-1.309	-1.159	-1.330	-.939						
.021						-.589	-.992	-.969	-.969	-.616						
.025						-.373	-.598	-.877	-.509	-.357						
.136						-.387	-.339	-.358	-.345	-.341						
.172						-.269	-.233	-.198	-.219	-.280						
.251						-.354	-.316	-.242	-.303	-.361						
.302						-.288	-.192	-.166	-.186	-.286						
.326						-.237	-.118	-.127	-.106	-.238						
.343						-.326	-.127	-.140	-.128	-.345						
.387						-.224	-.066	-.057	-.073	-.209						
.442						-.015	-.118	-.089	-.122	-.183						
.493						-.202	-.121	-.089	-.138	-.219						
.537						-.218	-.130	-.092	-.138	-.196						
.581						-.215	-.137	-.109	-.141	-.325						
.617						-.205	-.124	-.134	-.135	-.241						
.653						-.012	-.102	-.066	-.096	-.154						
.690						-.107	-.063	-.044	-.073	-.131						
.708							-.028	-.025	-.028							
.735							-.021	-.015	-.025							
.758									-.012							
.796									-.005							
.821									-.002							
.852									-.037							
	$\alpha = 2^\circ$												$\alpha = 4^\circ$			
0.030																
.003																
.014																
.021																
.025																
.106																
.172																
.251																
.302																
.326																
.343																
.387																
.442																
.493																
.537																
.581																
.617																
.653																
.690																
.708																
.735																
.758																
.796																
.821																
.852																
	$\alpha = 2^\circ$												$\alpha = 4^\circ$			
0.030																
.003																
.014																
.021																
.025																
.106																
.172																
.251																
.302																
.326																
.343																
.387																
.442																
.493																
.537																
.581																
.617																
.653																
.690																
.708																
.735																
.758																
.796																
.821																
.852																

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(1) $M = 0.750$; $P_{t,c}/P_{t,\infty} = 1.0$; inboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	0.000						.839	-.439	.101	.016	.073	
	.003	-.400	.137	.052	-.061	-.595	-.1265	1.402	1.402	1.362	-1.079	
	.014	1.395	1.395	-1.305	-.984	-1.060	-1.065	1.402	1.402	1.362	-.932	
	.031	-.877	-.637	-.928	-.913	-.974	-.693	1.402	1.402	1.362	-.699	
	.055	-.570	-.418	-.586	-.551	-.812	-.383	-.848	-.425	-.381	-.351	
	.106	-.400	-.389	-.339	-.324	-.274	-.409	-.354	-.363	-.351	-.351	
	.172	-.258	-.237	-.220	-.221	-.227	-.276	-.240	-.220	-.225	-.287	
	.251	-.185	-.135	-.189	-.351	-.308	-.371	-.343	-.287	-.340	-.378	
	.302	-.117	-.020	-.109	-.277	-.245	-.250	-.191	-.179	-.198	-.298	
	.326	-.120	-.118	-.118	-.324	-.207	-.250	-.118	-.182	-.110	-.251	
	.343	-.232	-.053	-.068	-.210	-.224	-.319	-.112	-.115	-.110	-.325	
	.387	-.015	-.123	-.097	-.183	-.013	-.227	-.042	-.048	-.069	-.210	
	.442	-.206	-.132	-.103	-.145	-.013	-.209	-.121	-.089	-.125	-.133	
	.493	-.229	-.141	-.112	-.151	-.207	-.223	-.155	-.097	-.145	-.228	
Turbine cowl	0.000						-.227	-.116	-.116	-.145	-.210	
	.003	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.014	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.031	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.055	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.106	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.172	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.251	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.302	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.326	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.343	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.387	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.442	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.493	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
	.581	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	
Plug	0.000						-.111	-.057	-.027	-.027	-.125	
	.003	-.400	.137	.052	-.061	-.595	-.1265	1.402	1.402	1.362	-1.079	
	.014	1.395	1.395	-1.305	-.984	-1.060	-1.065	1.402	1.402	1.362	-.932	
	.031	-.877	-.637	-.928	-.913	-.974	-.693	1.402	1.402	1.362	-.699	
	.055	-.570	-.418	-.586	-.551	-.812	-.383	-.848	-.425	-.381	-.351	
	.106	-.400	-.389	-.339	-.324	-.274	-.409	-.354	-.363	-.351	-.351	
	.172	-.258	-.237	-.220	-.221	-.227	-.276	-.240	-.220	-.225	-.287	
	.251	-.185	-.135	-.189	-.351	-.308	-.371	-.343	-.287	-.340	-.378	
	.302	-.117	-.020	-.109	-.277	-.245	-.250	-.191	-.179	-.198	-.298	
	.326	-.120	-.118	-.118	-.324	-.207	-.250	-.118	-.182	-.110	-.251	
	.343	-.232	-.053	-.068	-.210	-.224	-.319	-.112	-.115	-.110	-.325	
	.387	-.015	-.123	-.097	-.183	-.013	-.227	-.042	-.048	-.069	-.210	
	.442	-.206	-.132	-.103	-.145	-.013	-.209	-.121	-.089	-.125	-.133	
	.493	-.229	-.141	-.112	-.151	-.207	-.223	-.155	-.097	-.145	-.228	
	.581	-.281	-.258	-.232	-.242	-.207	-.227	-.116	-.116	-.145	-.210	

TABLE 41.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(m) $M = 0.775$; $p_{t,e}/p_{t,\infty} = 1.0$; inboard station

x/c	C _p at -																				
	α = -2°						α = 0°						α = 4°								
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	
Fan cowl	0.030										-1.275	-1.341	-1.344	-1.48	-1.066	-1.114					
	0.033									-1.237	-1.341	-1.341	-1.303	-1.303	-1.151						
	0.04									-1.592	-1.341	-1.341	-1.303	-1.303	-1.062						
	0.031									-1.698	-1.363	-1.363	-1.344	-1.344	-1.069						
	0.055									-1.404	-1.441	-1.441	-1.596	-1.596	-1.069						
	0.106									-1.423	-1.535	-1.535	-1.538	-1.538	-1.346						
	0.172									-1.279	-1.242	-1.242	-1.214	-1.225	-1.292						
	0.251									-1.384	-1.363	-1.363	-1.301	-1.380	-1.394						
	0.302									-1.309	-1.200	-1.200	-1.186	-1.202	-1.309						
	0.326									-1.254	-1.119	-1.119	-1.128	-1.112	-1.261						
	0.343									-1.312	-1.105	-1.105	-1.114	-1.106	-1.312						
	0.387									-1.229	-1.063	-1.063	-1.046	-1.069	-1.213						
	0.442									-1.015	-1.125	-1.125	-1.125	-1.128	-1.191						
	0.493									-1.215	-1.136	-1.136	-1.100	-1.151	-1.236						
	0.537									-1.240	-1.144	-1.144	-1.108	-1.154	-1.222						
Turbine cowl	0.591									-1.240	-1.158	-1.158	-1.123	-1.162	-1.123						
	0.617									-1.221	-1.142	-1.142	-1.125	-1.151	-1.261						
	0.653									-1.015	-1.119	-1.119	-1.077	-1.112	-1.182						
	0.690									-1.118	-1.069	-1.069	-1.052	-1.086	-1.128						
	0.708										-1.029	-1.029	-1.029	-1.032							
	0.735										-1.027	-1.027	-1.027	-1.027							
	0.768										-1.021	-1.021	-1.021	-1.018							
	0.821										-1.010	-1.010	-1.010	-1.010							
	0.852										-1.004	-1.004	-1.004	-1.004							
											-0.958	-0.958	-0.958	-1.004							
	Fan cowl	0.030									-1.170	-1.237	-1.237	-1.343	-1.305	-1.130					
		0.033									-1.124	-1.237	-1.237	-1.343	-1.305	-1.132					
		0.04									-1.020	-1.170	-1.170	-1.343	-1.305	-1.087					
		0.031									-1.299	-1.307	-1.307	-1.493	-1.493	-1.014					
		0.055									-1.104	-1.299	-1.299	-1.343	-1.343	-1.014					
0.106										-1.334	-1.365	-1.365	-1.493	-1.493	-1.014						
0.172										-1.379	-1.345	-1.345	-1.397	-1.397	-1.014						
0.251										-1.222	-1.224	-1.224	-1.235	-1.235	-1.014						
0.302										-1.334	-1.354	-1.354	-1.365	-1.365	-1.014						
0.326										-1.195	-1.206	-1.206	-1.204	-1.204	-1.014						
0.343										-1.122	-1.153	-1.153	-1.111	-1.111	-1.014						
0.387										-1.116	-1.119	-1.119	-1.079	-1.079	-1.014						
0.442										-1.069	-1.052	-1.052	-1.013	-1.013	-1.014						
0.493										-1.127	-1.102	-1.102	-1.131	-1.131	-1.014						
0.537										-1.141	-1.111	-1.111	-1.151	-1.151	-1.014						
Turbine cowl	0.591									-1.150	-1.119	-1.119	-1.159	-1.159	-1.014						
	0.617									-1.164	-1.134	-1.134	-1.168	-1.168	-1.014						
	0.653									-1.127	-1.127	-1.127	-1.159	-1.159	-1.014						
	0.690									-1.147	-1.091	-1.091	-1.120	-1.120	-1.014						
	0.708									-1.080	-1.063	-1.063	-1.092	-1.092	-1.014						
	0.735									-1.041	-1.041	-1.041	-1.036	-1.036	-1.014						
	0.768									-1.035	-1.035	-1.035	-1.038	-1.038	-1.014						
	0.821									-1.029	-1.029	-1.029	-1.027	-1.027	-1.014						
	0.852									-1.021	-1.021	-1.021	-1.019	-1.019	-1.014						
										-1.016	-1.016	-1.016	-1.016	-1.016	-1.014						
										-1.013	-1.013	-1.013	-1.013	-1.013	-1.014						
										-1.013	-1.013	-1.013	-1.013	-1.013	-1.014						
										-1.013	-1.013	-1.013	-1.013	-1.013	-1.014						
										-1.013	-1.013	-1.013	-1.013	-1.013	-1.014						

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(n) $M = 0.800$; $P_{te}/P_{t,\infty} = 1.0$; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000					.831	-.253	.217	.139	.139								
	.003					1.309	1.291	1.255	1.255	1.255								
	.014					-1.115	1.291	1.291	1.255	-1.060								
	.031					-.657	1.296	1.293	1.324	-1.324								
	.055					-.443	1.296	1.293	1.324	-1.324								
	.106					-.435	1.296	1.293	1.324	-1.324								
	.172					-.275	1.296	1.293	1.324	-1.324								
	.261					-.398	1.296	1.293	1.324	-1.324								
	.302					-.313	1.296	1.293	1.324	-1.324								
	.326					-.254	1.296	1.293	1.324	-1.324								
	.343					-.297	1.296	1.293	1.324	-1.324								
	.387					-.232	1.296	1.293	1.324	-1.324								
	.442					-.014	1.296	1.293	1.324	-1.324								
	.493					-.248	1.296	1.293	1.324	-1.324								
	.537					-.266	1.296	1.293	1.324	-1.324								
Turbine cowl	.591					-.230	1.296	1.293	1.324	-1.324								
	.617					-.014	1.296	1.293	1.324	-1.324								
	.653					-.123	1.296	1.293	1.324	-1.324								
	.690						1.296	1.293	1.324	-1.324								
	.708						1.296	1.293	1.324	-1.324								
	.735						1.296	1.293	1.324	-1.324								
	.758						1.296	1.293	1.324	-1.324								
	.786						1.296	1.293	1.324	-1.324								
	.821						1.296	1.293	1.324	-1.324								
	.852						1.296	1.293	1.324	-1.324								
Fan cowl	0.000					.883	-.279	.485	.126	.123								
	.003					1.240	1.289	1.289	1.256	-1.094								
	.014					-1.178	1.289	1.296	1.256	-1.100								
	.031					-1.016	1.096	-1.034	-1.314	-1.037								
	.055					-.797	-.624	-.374	-.507	-.812								
	.106					-.469	-.428	-.390	-.393	-.314								
	.172					-.277	-.234	-.212	-.216	-.254								
	.261					-.317	-.344	-.339	-.366	-.339								
	.302					-.207	-.193	-.217	-.203	-.265								
	.326					-.250	-.112	-.026	-.108	-.230								
	.343					-.277	-.107	-.112	-.094	-.282								
	.387					-.231	-.061	-.112	-.067	-.214								
	.442					-.214	-.128	-.104	-.132	-.181								
	.493					-.239	-.139	-.112	-.157	-.227								
	.537					-.228	-.153	-.120	-.165	-.216								
Turbine cowl	.581					-.237	-.160	-.143	-.176	-.206								
	.617					-.269	-.221	-.153	-.167	-.285								
	.653					-.193	-.134	-.099	-.129	-.197								
	.690					-.152	-.085	-.072	-.102	-.148								
	.708					-.095	-.045	-.045	-.045									
	.735					-.035	-.040											
	.758					-.032	-.034											
	.786					-.024	-.026											
	.821					-.016	-.021											
	.852					-.013	.168											

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(p) M = 0.700; windmilling; inboard station

[illegible]

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(q) $M = 0.750$; windmilling; inboard station

x/c	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.030									
	.003									
	.014									
	.031									
	.055									
	.106									
	.172									
	.251									
	.302									
	.326									
Turbine cowl	.363									
	.387									
	.442									
	.493									
	.537									
	.581									
	.617									
	.653									
	.690									
	.708									
Plug	.735									
	.768									
	.796									
	.821									
	.852									
Fan cowl	0.030									
	.003									
	.014									
	.031									
	.055									
	.106									
	.172									
	.251									
	.302									
	.326									
Turbine cowl	.363									
	.387									
	.442									
	.493									
	.537									
	.581									
	.617									
	.653									
	.690									
	.708									
Plug	.735									
	.768									
	.796									
	.821									
	.852									

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(r) $M = 0.775$; windmilling; inboard station

C _p at												
x/c	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row L
Fan cowl	0.000											
	.003											
	.014											
	.031											
	.055											
	.126											
	.172											
	.261											
	.302											
	.326											
Turbine cowl	.343											
	.387											
	.442											
	.493											
	.537											
	.591											
	.617											
	.653											
	.690											
	.708											
Plug	.735											
	.758											
	.796											
	.821											
	.852											
Fan cowl	0.000											
	.003											
	.014											
	.031											
	.055											
	.126											
	.172											
	.251											
	.302											
	.343											
Turbine cowl	.387											
	.442											
	.493											
	.537											
	.581											
	.617											
	.653											
	.690											
	.708											
	.735											
Plug	.758											
	.796											
	.821											
	.852											

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Continued

(s) M = 0.800; windmilling; inboard station

x/c	C _p at -																	
	α = -2°						α = 0°						α = 1°					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L			
Fan cowl	0.000					.870	-.383	.074	.015	.021								
	.003	-.389	1.286	.201	.022	-.142	1.288	1.288	1.255	-1.230								
	.014	1.289	1.286	1.286	1.250	-1.137	1.313	1.288	1.255	-1.230								
	.031	-1.114	-1.329	-1.063	-1.058	-.931	1.291	1.288	1.285	-1.230								
	.055	-.810	-.847	-.301	-.938	-.549	-.875	-1.234	1.285	-1.230								
	.106	-.479	-.451	-.279	-.314	-.472	-.406	-.242	-.969	-.516								
	.172	-.260	-.239	-.209	-.273	-.272	-.234	-.188	-.274	-.350								
	.251	-.354	-.317	-.333	-.375	-.397	-.353	-.307	-.214	-.286								
	.302	-.287	-.196	-.214	-.216	-.314	-.212	-.195	-.374	-.497								
	.326	-.236	-.131	-.012	-.118	-.263	.030	.030	-.214	-.317								
	.343	-.300	-.128	-.038	-.078	-.210	.058	.058	-.119	-.258								
	Fan cowl	0.000					.870	-.396	.348	.015	.0230							
.003		-.389	1.286	.201	.022	-.142	1.290	1.290	1.259	-1.132								
.014		1.289	1.286	1.286	1.250	-1.137	1.291	1.290	1.259	-1.132								
.031		-1.114	-1.329	-1.063	-1.058	-.931	1.291	1.290	1.259	-1.132								
.055		-.810	-.847	-.301	-.938	-.549	-.820	-.384	-.425	-1.081								
.106		-.479	-.451	-.279	-.314	-.472	-.516	-.384	-.425	-1.081								
.172		-.260	-.239	-.209	-.273	-.272	-.249	-.219	-.237	-1.081								
.251		-.354	-.317	-.333	-.375	-.397	-.375	-.319	-.338	-1.081								
.302		-.287	-.196	-.214	-.216	-.314	-.289	-.227	-.199	-1.081								
.326		-.236	-.131	-.012	-.118	-.263	-.201	-.122	-.004	-1.081								
.343		-.300	-.128	-.038	-.078	-.210	-.298	-.120	-.112	-1.081								
Turbine cowl		0.000					.870	-.396	.348	.015	.0230							
	.003	-.389	1.286	.201	.022	-.142	1.290	1.290	1.259	-1.132								
	.014	1.289	1.286	1.286	1.250	-1.137	1.291	1.290	1.259	-1.132								
	.031	-1.114	-1.329	-1.063	-1.058	-.931	1.291	1.290	1.259	-1.132								
	.055	-.810	-.847	-.301	-.938	-.549	-.820	-.384	-.425	-1.081								
	.106	-.479	-.451	-.279	-.314	-.472	-.516	-.384	-.425	-1.081								
	.172	-.260	-.239	-.209	-.273	-.272	-.249	-.219	-.237	-1.081								
	.251	-.354	-.317	-.333	-.375	-.397	-.375	-.319	-.338	-1.081								
	.302	-.287	-.196	-.214	-.216	-.314	-.289	-.227	-.199	-1.081								
	.326	-.236	-.131	-.012	-.118	-.263	-.201	-.122	-.004	-1.081								
	.343	-.300	-.128	-.038	-.078	-.210	-.298	-.120	-.112	-1.081								
	Plug	0.000					.870	-.396	.348	.015	.0230							
.003		-.389	1.286	.201	.022	-.142	1.290	1.290	1.259	-1.132								
.014		1.289	1.286	1.286	1.250	-1.137	1.291	1.290	1.259	-1.132								
.031		-1.114	-1.329	-1.063	-1.058	-.931	1.291	1.290	1.259	-1.132								
.055		-.810	-.847	-.301	-.938	-.549	-.820	-.384	-.425	-1.081								
.106		-.479	-.451	-.279	-.314	-.472	-.516	-.384	-.425	-1.081								
.172		-.260	-.239	-.209	-.273	-.272	-.249	-.219	-.237	-1.081								
.251		-.354	-.317	-.333	-.375	-.397	-.375	-.319	-.338	-1.081								
.302		-.287	-.196	-.214	-.216	-.314	-.289	-.227	-.199	-1.081								
.326		-.236	-.131	-.012	-.118	-.263	-.201	-.122	-.004	-1.081								
.343		-.300	-.128	-.038	-.078	-.210	-.298	-.120	-.112	-1.081								
.387		-.215	-.080	-.099	-.104	-.132	.033	.033	-.103	-.238								

TABLE 42.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 12 - Concluded

(t) M = 0.825; windmilling; inboard station

x/2	C_p at -									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	0.037	0.336	0.253	0.072	0.084	0.831	0.315	0.995	0.068
	0.003	1.266	1.245	1.245	1.213	-1.311	-1.101	1.243	1.243	1.211
	0.014	1.266	1.245	1.245	1.213	-1.311	-1.089	1.243	1.243	1.211
	0.031	1.266	1.245	1.245	1.213	-1.311	-1.078	1.243	1.243	1.211
	0.055	1.266	1.245	1.245	1.213	-1.311	-1.067	1.243	1.243	1.211
	0.136	1.266	1.245	1.245	1.213	-1.311	-1.056	1.243	1.243	1.211
	0.172	1.266	1.245	1.245	1.213	-1.311	-1.045	1.243	1.243	1.211
	0.251	1.266	1.245	1.245	1.213	-1.311	-1.034	1.243	1.243	1.211
	0.302	1.266	1.245	1.245	1.213	-1.311	-1.023	1.243	1.243	1.211
	0.326	1.266	1.245	1.245	1.213	-1.311	-1.012	1.243	1.243	1.211
Turbine cowl	0.000	0.037	0.336	0.253	0.072	0.084	0.831	0.315	0.995	0.068
	0.003	1.266	1.245	1.245	1.213	-1.311	-1.101	1.243	1.243	1.211
	0.014	1.266	1.245	1.245	1.213	-1.311	-1.089	1.243	1.243	1.211
	0.031	1.266	1.245	1.245	1.213	-1.311	-1.078	1.243	1.243	1.211
	0.055	1.266	1.245	1.245	1.213	-1.311	-1.067	1.243	1.243	1.211
	0.136	1.266	1.245	1.245	1.213	-1.311	-1.056	1.243	1.243	1.211
	0.172	1.266	1.245	1.245	1.213	-1.311	-1.045	1.243	1.243	1.211
	0.251	1.266	1.245	1.245	1.213	-1.311	-1.034	1.243	1.243	1.211
	0.302	1.266	1.245	1.245	1.213	-1.311	-1.023	1.243	1.243	1.211
	0.326	1.266	1.245	1.245	1.213	-1.311	-1.012	1.243	1.243	1.211
Plug	0.000	0.037	0.336	0.253	0.072	0.084	0.831	0.315	0.995	0.068
	0.003	1.266	1.245	1.245	1.213	-1.311	-1.101	1.243	1.243	1.211
	0.014	1.266	1.245	1.245	1.213	-1.311	-1.089	1.243	1.243	1.211
	0.031	1.266	1.245	1.245	1.213	-1.311	-1.078	1.243	1.243	1.211
	0.055	1.266	1.245	1.245	1.213	-1.311	-1.067	1.243	1.243	1.211
	0.136	1.266	1.245	1.245	1.213	-1.311	-1.056	1.243	1.243	1.211
	0.172	1.266	1.245	1.245	1.213	-1.311	-1.045	1.243	1.243	1.211
	0.251	1.266	1.245	1.245	1.213	-1.311	-1.034	1.243	1.243	1.211
	0.302	1.266	1.245	1.245	1.213	-1.311	-1.023	1.243	1.243	1.211
	0.326	1.266	1.245	1.245	1.213	-1.311	-1.012	1.243	1.243	1.211

TABLE 43.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 13

(a) $M = 0.700$; outboard station

x/\bar{c}	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row L
Fan cowl	.0000	.888	.747	.662	.561	1.105	.685	.778	.786	.552	1.074	
	.003	-.047	-.802	.329	-.757	-.018	-.301	-.734	.178	-.320	-.320	
	.014	-.179	-.594	-.645	-.573	-.215	-.291	-.600	-.600	-.598	-.340	
	.031	-.693	-.454	-.620	-.509	-.170	-.253	-.422	-.527	-.524	-.240	
	.055	-.263	-.429	-.508	-.412	-.279	-.317	-.412	-.457	-.411	-.330	
	.106	-.328	-.352	-.372	-.331	-.315	-.430	-.342	-.355	-.314	-.327	
	.172	-.247	-.215	-.206	-.221	-.263	-.253	-.195	-.195	-.211	-.259	
	.261	-.376	-.311	-.244	-.073	-.389	-.366	-.294	-.243	-.065	-.379	
	.302	-.312	-.193	-.149	-.325	-.296	-.295	-.189	-.151	-.320	-.282	
	.326	-.267	-.107	-.056	-.170	-.257	-.259	-.093	-.055	-.162	-.243	
	.343	-.612	.055	-.193	.018	-.434	.624	.069	-.186	.025	-.427	
	.387	-.221	.149	-.034	-.125	-.063	-.220	.132	-.029	-.120	-.091	
	.442	-.350	.238	.117	-.279	-.189	-.269	.211	-.122	-.272	-.185	
Turbine cowl	.493	-.150	.008	.072	-.105	-.241	.156	.018	.065	.110	-.307	
	.537	-.312	.177	.072	-.202	-.296	.311	.071	-.065	.181	-.288	
	.581	-.276	.129	.147	-.163	-.263	.285	.138	-.152	.191	-.259	
	.617	-.267	.091	.056	-.131	-.705	.272	.090	-.055	.091	-.698	
	.653	-.166	.120	.069	-.183	-.060	.194	.128	-.068	-.211	-.136	
Plug	.690	-.070	.020	.015	-.112	-.160	-.097	.015	-.010	-.059	-.143	
	.708		.310	.345	-.015			.339	.344	-.017		
	.735		.036		-.205			.034		-.064		
	.768		-.072		-.218			-.125		-.233		
	.796		-.315		-.008			-.020		-.013		
Fan cowl	.821		.154		.192			.152		.190		
	.852		.371		.437			.379		.432		
	.0000	.459	.778	.965	.553	1.004	.241	.778	1.090	.529	.840	
	.003	-.523	-.711	.152	-.695	-.601	-.918	-.683	1.136	-.658	-.921	
	.014	-.420	-.577	.453	-.611	-.556	-.508	-.555	.291	-.611	-.605	
	.031	-.323	-.418	.415	-.514	-.288	-.385	-.437	-.291	-.511	-.366	
	.055	-.346	-.405	.405	-.401	.359	-.354	-.386	.335	-.398	-.382	
	.106	-.369	.329	.335	.310	.330	.247	.329	.306	.317	-.329	
	.172	-.252	.192	.185	.207	.255	.240	.259	.170	.217	-.269	
	.261	-.352	.287	.246	.065	.365	.517	.284	.240	.059	-.344	
	.302	-.284	.169	.153	.313	.265	.253	.186	.154	.367	-.243	
	.326	-.249	.083	.055	.162	.230	.223	.084	.052	.136	-.213	
	.343	-.249	.073	.055	.135	.094	.023	.069	.045	.045	-.382	
Turbine cowl	.387	-.213	.118	.025	.113	.084	.021	.069	.026	.107	-.108	
	.442	-.375	.192	.052	.235	.191	.372	.182	.119	.259	-.175	
	.493	-.350	.016	.035	.110	.249	.149	.022	.032	.098	-.223	
	.537	-.320	.150	.090	.207	.249	.311	.147	.087	.211	-.220	
	.581	-.291	.076	.162	.194	.235	.282	.151	.178	.204	-.253	
	.617	-.278	.076	.067	.078	.082	.282	.096	.080	.049	-.682	
	.653	-.188	.128	.064	.239	.155	.185	.141	.061	.265	-.169	
	.690	-.164	.009	.023	.036	-.133	-.110	.002	.029	.030	-.152	
	.708		.305	.337	.016			.302	.334	.610		
	.735		.031		.007			.025		.067		
	.768		-.128		.275			-.096		.346		
	.796		-.023		.013			-.026		-.001		
	.821		.149		.181			.140		.177		
	.852		.376		.429			.369		.426		

TABLE 43.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 13 - Continued

(b) $M = 0.750$; outboard station

x/\bar{c}	C_p at									
	$\alpha = -2^\circ$					$\alpha = 0^\circ$				
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	.829	.727	.666	.523	1.120	.602	.751	.780	.517	1.049
	-.113	-.938	.459	-.870	-.115	-.334	-.898	.286	-.832	-.408
	-.165	-.684	-.161	-.667	-.213	-.347	-.667	.618	-.685	-.358
	-.690	-.504	-.590	-.560	-.174	-.272	-.485	.606	-.567	-.272
	-.283	-.451	-.524	-.437	-.292	-.322	-.429	.496	-.443	-.349
	-.176	-.336	-.385	-.345	-.328	-.408	-.374	.337	-.340	-.340
	.172	-.260	-.219	-.204	-.272	-.235	.199	.219	-.278	-.278
	.261	-.390	-.326	-.251	-.071	-.375	.216	.255	.066	-.399
	.302	-.319	-.198	-.146	-.345	-.302	.182	.337	-.281	-.281
	.326	-.263	-.163	-.053	-.282	-.246	.083	.153	.169	.231
	.343	-.497	-.142	-.111	-.248	-.476	.152	.112	.125	.231
	.387	-.463	-.129	-.133	-.289	-.384	.054	.048	.114	.623
Turbine cowl	.442	-.058	-.123	-.103	-.123	-.116	.153	.124	.090	.116
	.493	-.294	-.123	-.342	-.278	-.234	.089	.089	.328	-.496
	.537	-.177	-.143	-.189	-.110	-.184	.159	.153	.122	.122
	.581	-.416	-.178	-.145	-.419	-.411	.173	.214	.225	-.408
	.617	-.222	-.020	-.354	-.643	-.246	.013	.013	.325	-.699
	.653	-.163	-.184	-.015	-.046	-.202	.194	.031	.028	-.049
	.690	-.028	-.052	-.219	-.387	-.105	.013	.013	.219	-.399
	.708	-.328	.369	-.012			.312	.341	.013	
	.735	-.095		.082			.085		.066	
	.768	-.056	-.434				.138		.434	
	.796	.037	.058				.016		.045	
	.821	.165	.194				.161		.178	
	.852	.366	.444				.376		.425	
Fan cowl	.356	.751	.815	.510	.914	.226	.744	1.056	.510	.836
	-.723	-.914	.236	-.882	-.762	-.104	-.899	.211	-.867	-.836
	-.452	-.667	-.502	-.682	-.490	-.505	-.658	.347	-.684	-.513
	-.343	-.467	-.470	-.564	-.326	-.402	-.481	.347	-.566	-.434
	-.055	-.361	-.432	-.439	-.384	-.393	.425	.376	-.434	-.419
	.106	-.570	-.353	-.331	-.340	-.619	.353	.333	.334	-.346
	.172	-.252	-.199	-.214	-.273	-.248	.190	.181	.222	-.260
	.261	-.358	-.377	-.246	-.379	-.337	.306	.266	.057	.360
	.302	-.284	-.176	-.159	-.244	-.257	.179	.154	.337	.248
	.326	-.237	-.051	-.134	-.217	-.222	.093	.164	.169	.284
	.343	-.467	.161	-.107	-.241	-.432	.158	.118	.164	.251
	.387	-.387	.051	-.048	-.114	-.349	.058	.118	.143	-.512
	.442	-.120	.118	-.113	.125	-.176	.052	.118	.113	.114
Turbine cowl	.493	-.237	.082	-.124	.152	-.258	.100	.129	.337	.575
	.537	-.184	.156	-.167	.105	-.193	.199	.152	.154	-.689
	.581	-.414	.173	-.224	.376	-.464	.193	.283	.283	-.346
	.617	-.246	.064	-.465	-.735	-.254	.004	.004	.402	-.772
	.653	-.248	-.253	-.057	-.037	-.219	-.083	.034	.034	-.042
	.690	-.114	.043	.027	-.370	-.137	.116	.067	.222	-.301
	.735		.366	-.017			.334	.327	.016	
	.768		.076	.072			.160		.067	
	.796		-.188	-.449			-.225		.457	
	.821		.030	.357			.016		.058	
	.852		.161	.178			.150		.170	
			.384	.419			.365		.411	

TABLE 43.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 13 - Continued

(c) $M = 0.775$; outboard station

x/c	C _p at -																			
	α = -2°						α = 0°						α = 4°							
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L
Fan cowl	0.000	.805	.751	.649	.533	1.119	.639	.751	.785	.519	1.050									
	.003	-.126	-.1030	.622	-.908	.112	-.427	-.985	.355	-.958	-.444									
	.014	-.225	-.713	-.1172	-.681	-.239	-.348	-.681	-.745	-.689	-.379									
	.031	-.290	-.543	-.1953	-.586	.183	-.289	-.506	-.723	-.597	-.275									
	.055	-.358	-.378	-.270	-.485	.367	-.331	-.453	-.520	-.467	-.363									
Turbine cowl	.106	-.425	-.217	-.362	-.352	.538	-.419	-.365	-.391	-.348	-.346									
	.172	-.459	-.133	-.259	-.252	.270	-.252	-.210	-.207	-.218	-.278									
	.261	-.406	-.339	-.259	-.367	.617	-.388	-.330	-.272	-.058	-.411									
	.382	-.327	-.197	-.144	-.174	.293	-.306	-.194	-.150	-.354	-.281									
	.537	-.283	-.097	-.044	-.170	.239	-.247	-.085	-.057	-.168	-.224									
Plug	.735	-.623	.132	.061	.176	.182	.019	.052	.302	.137	.888									
	.881	-.423	.092	.116	.324	.307	.295	.096	.323	.468										
	.961	-.231	.037	.116	.050	.357	.244	.010	.115	.049	.382									
	.981	-.167	.167	.106	.143	.200	.377	.233	.106	.142	.193									
	.991	-.082	.157	.065	.284	.813	.331	.190	.029	.249	.826									
Fan cowl	.996	-.157	.125	.177	.177	.072	.204	.012	.146	.210	.089									
	.999	-.166	.247	.069	.142	.163	.179	.007	.143	.122										
	.708	.385	.385	.344	.013	.069	.081	.375	.339	.015										
	.735	.096	.096	.378	.378	.135	.135	.688	.081	.360										
	.768	.006	.006	.594	.594	.135	.135	.602	.081	.360										
Turbine cowl	.821	.165	.165	.277	.277	.155	.155	.602	.081	.360										
	.852	.389	.389	.433	.433	.155	.155	.572	.372	.421										
	.000	.383	.751	.887	.533	.839	.234	.758	1.057	.526	.839									
	.003	-.742	-.933	.301	-.937	.818	-.118	-.545	.256	-.935	-.121									
	.014	-.448	-.702	-.529	-.697	.513	-.556	.703	-.357	-.706	-.703									
Plug	.031	-.352	-.504	-.534	-.592	.338	-.412	.502	-.374	-.599	-.407									
	.055	-.361	-.445	-.459	-.443	.338	-.395	.446	-.396	-.452	-.421									
	.106	-.570	-.365	-.370	-.338	.338	-.363	.363	-.351	-.342	-.353									
	.172	-.251	-.209	-.195	-.214	.270	-.254	.212	-.190	-.220	-.271									
	.261	-.369	-.317	-.270	-.047	.386	-.339	.321	-.276	-.054	-.373									
Fan cowl	.382	-.275	-.181	-.161	.344	.259	-.263	.187	-.173	.347	-.251									
	.537	-.231	-.075	-.050	.160	.208	-.218	.081	.059	.164	-.198									
	.735	-.623	.204	.055	.184	.200	.407	.197	.067	.181	.198									
	.881	-.423	.009	.066	.195	.855	.602	.056	.006	.141	.853									
	.961	-.279	.059	.066	.195	.184	.003	.022	.078	.178	.189									
Turbine cowl	.991	-.364	.006	-.080	.081	.355	.271	.257	.081	.331	.503									
	.996	-.347	-.124	-.124	.121	.194	.260	.014	.101	.091	.288									
	.999	-.263	-.002	-.147	.265	.118	.203	.031	.048	.356	.825									
	.999	-.217	.067	.343	.013	.070	.195	.226	.016	.144	.102									
	.999	-.098	.098	.046	.046	.373	.373	.373	.328	.631	.085									

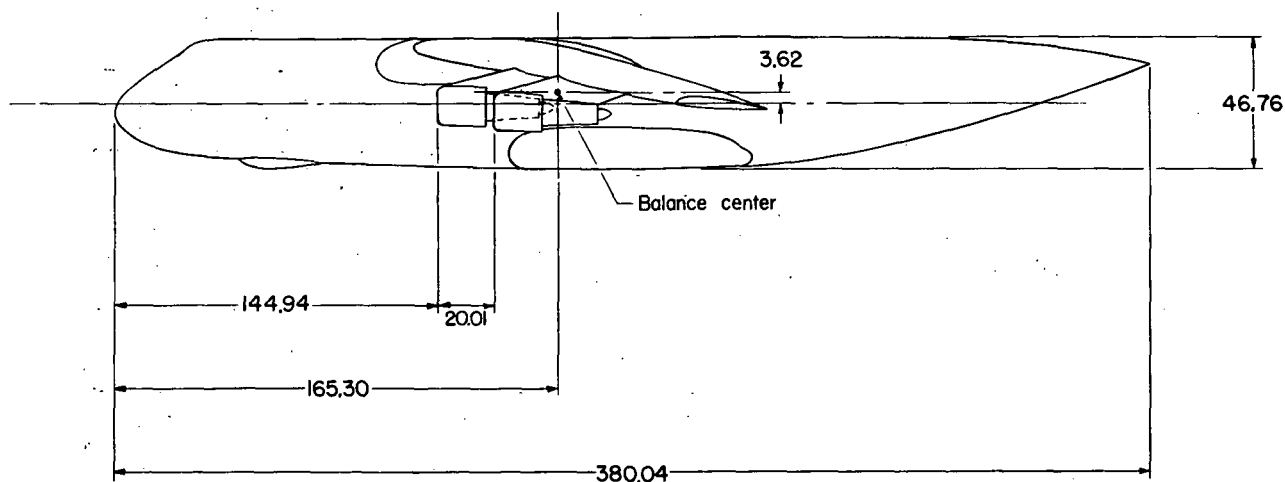
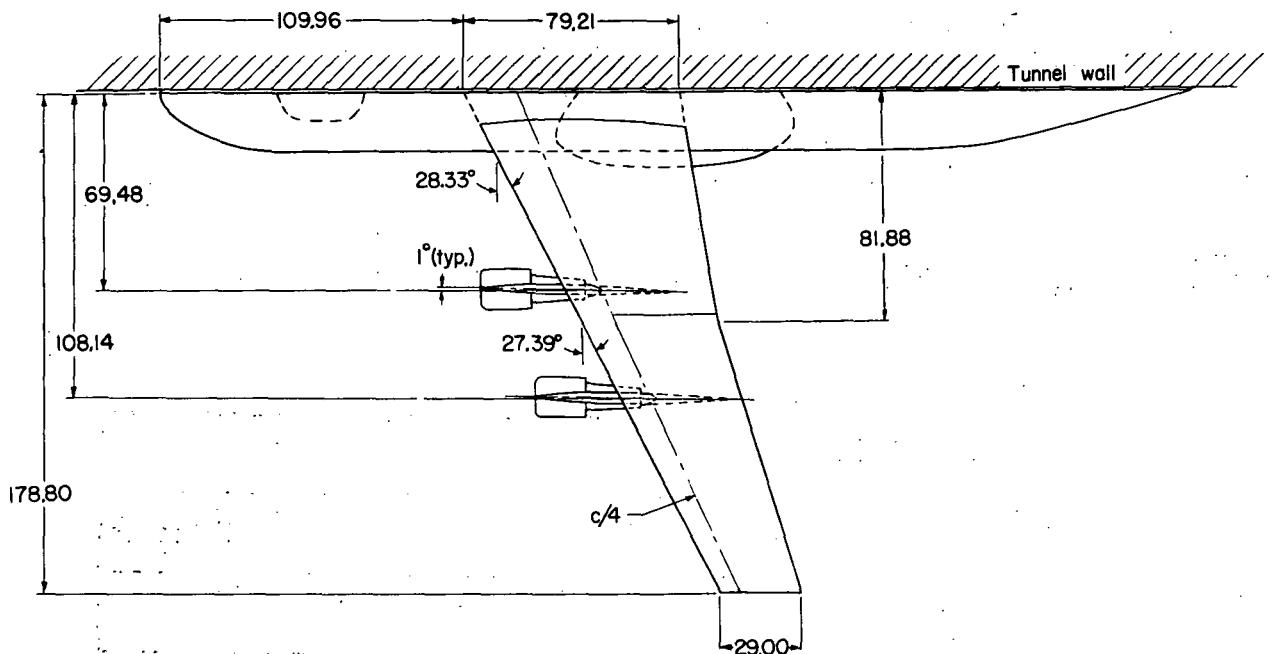
TABLE 43.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 13 - Continued

(d) M = 0.800; outboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row H	Row I	Row J	Row K	Row L	Row H	Row L
Fan cowl	.000	.757	.751	.712	.542	1.125	.581	.771	.817	.548		
	.003	-.157	-.994	.850	-.974	-.168	-.442	-.538	.459	-.1.029		
	.014	-.230	-.786	-.703	-.703	-.238	-.361	-.692	.852	-.483		
	.031	-.768	-.659	-.1144	-.719	-.263	-.293	-.603	-.852	-.399		
	.055	-.361	-.477	-.367	-.475	-.320	-.331	-.461	-.769	-.706		
	.106	-.295	-.591	-.388	-.360	-.347	-.369	-.378	.507	-.472		
	.172	-.271	-.219	-.203	-.233	-.270	-.255	-.293	-.399	-.355		
	.261	-.428	-.353	-.265	-.054	-.434	-.394	-.346	-.203	-.220		
	.302	-.322	-.192	-.147	-.374	-.293	-.394	-.276	-.051	-.423		
	.326	-.265	-.088	-.040	-.176	-.225	-.244	-.075	.158	-.372		
	.343	-.360	.175	.003	.215	-.157	-.244	.175	.003	-.279		
	.387	-.893	-.128	.073	.168	-.841	-.766	.107	.073	-.214		
	.442	.085	.097	-.104	-.002	-.287	.101	.086	.157	-.154		
	.493	-.347	-.297	-.048	-.227	-.260	-.350	-.281	.107	-.853		
Turbine cowl	.537	-.325	-.136	-.160	-.263	-.540	-.293	-.128	.053	-.258		
	.581	-.295	-.168	-.165	-.130	-.255	-.293	-.201	.163	-.274		
	.617	-.377	-.292	-.109	-.143	-.513	-.380	-.142	.122	-.540		
	.653	-.238	.003	-.066	-.135	-.333	-.233	.003	.142	-.236		
	.690	-.112	-.131	.001	-.265	-.130	-.122	.034	.176	-.551		
	.738		.279	.368	-.013			-.051	.301	-.347		
	.735		.161		.115			.387	.166	-.124		
	.768		.061		.365				.016			
	.796		.035		-.121				.112			
	.821		.204		.264				.380			
	.852		.387		.438				.269			
									.196			
									.275			
									.441			
Fan cowl	.000	.411	.745	.935	.516	.922	.268	.745	1.059	.534		
	.003	-.852	-.1.034	.384	-.977	-.816	-.1.177	.591	.332	-.1.020		
	.014	-.444	-.758	-.551	-.708	-.501	-.917	.710	.344	-.1.188		
	.031	-.363	-.592	.723	-.694	-.352	-.458	.605	.415	-.395		
	.055	-.360	-.468	-.484	-.445	-.396	-.394	.461	.426	-.399		
	.106	-.515	-.380	-.391	-.345	-.340	-.378	.434	.468	-.448		
	.172	-.254	-.206	-.158	-.218	-.273	-.243	.378	.372	-.353		
	.261	-.379	-.337	-.286	-.043	-.464	-.353	.204	.190	-.220		
	.302	-.292	-.179	-.166	-.042	-.260	-.266	.337	.329	-.946		
	.326	-.230	-.074	-.053	-.159	-.203	-.214	.179	.154	-.377		
	.343	-.368	.177	.003	.223	-.113	.075	.051	.051	-.160		
	.387	-.751	-.096	.065	.154	-.854	.341	.185	.000	-.187		
	.442	.098	.076	.107	-.013	-.164	.701	.067	.067	-.136		
	.493	-.287	-.283	.066	.219	-.252	.103	.084	.110	-.117		
Turbine cowl	.537	-.306	-.133	.169	.278	-.237	.321	.284	.075	-.225		
	.581	-.304	-.198	.132	.112	-.263	.317	.113	.212	-.320		
	.617	-.390	-.273	.145	-.140	-.380	.342	.204	.111	-.469		
	.653	-.246	-.024	-.032	-.216	-.366	-.399	.285	.195	-.117		
	.690	-.138	-.179	-.063	-.306	-.162	-.249	.038	.000	-.277		
	.738		.276	.378	-.016		-.144	.129	.315	-.198		
	.735		.156		.090			.270	.087			
	.768		-.214		-.371			.155	.087			
	.796		.033		-.176			-.260	.150			
	.821		.199		.256			-.027	.380			
	.852		.384		.427			.380	.418			

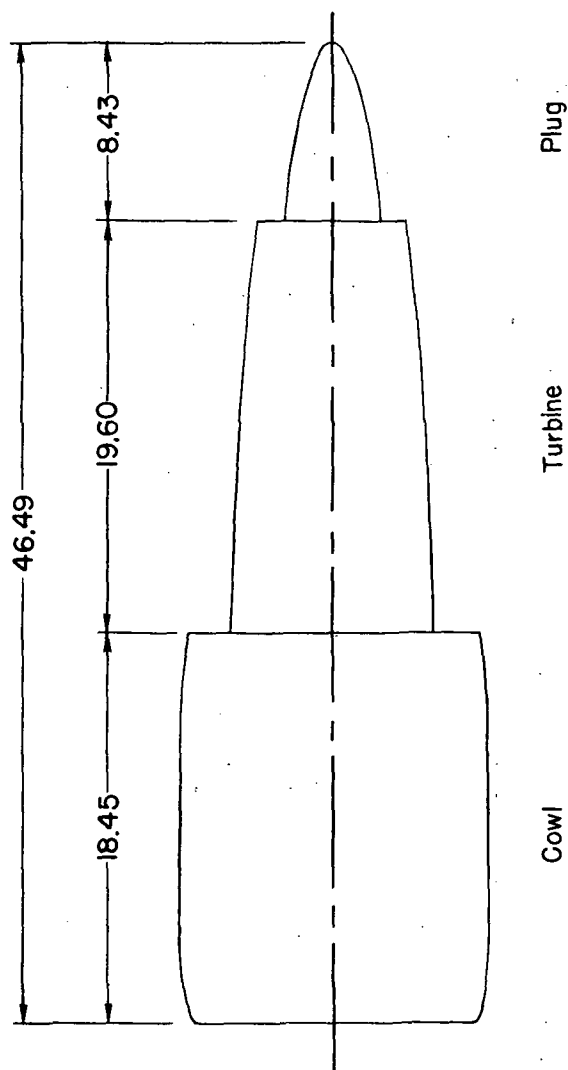
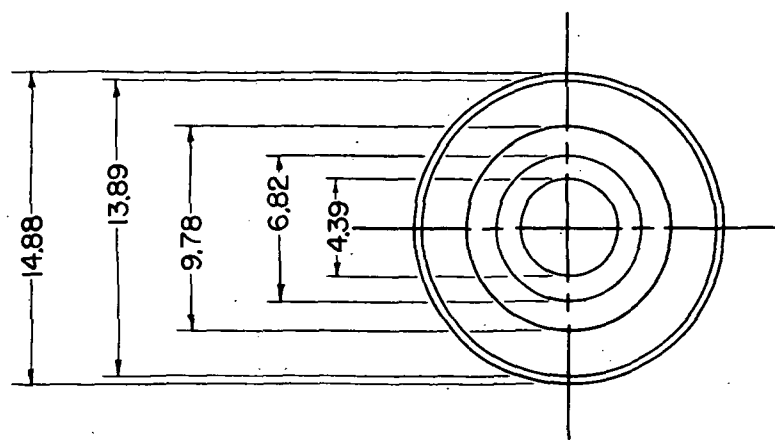
TABLE 43.- FAN-JET ENGINE SURFACE PRESSURE COEFFICIENTS FOR CONFIGURATION 13 - Concluded
(e) $M = 0.825$; outboard station

x/c	C_p at -											
	$\alpha = -2^\circ$						$\alpha = 0^\circ$					
	Row H	Row I	Row J	Row K	Row L	Row L	Row H	Row I	Row J	Row K	Row L	Row L
Fan cowl	.791	.759	.715	.538	1.119	.576	.576	.752	.828	.550	1.043	
	-.167	-.966	1.171	-.970	1.198	-.198	-.537	-.933	.618	-.987	-.518	
	-.235	-.761	-1.056	-.701	-.568	-.374	-.374	-.655	-.855	-.764	-.411	
	-.654	-.681	-1.079	-.740	-.504	-.306	-.306	-.662	-.759	-.305	-.305	
	-.055	-.506	-.914	-.304	-.321	-.395	-.395	-.623	-.823	-.447	-.382	
	1.066	-.400	-.330	-.366	-.339	-.323	-.323	-.393	-.218	-.266	-.266	
	1.172	-.219	-.156	-.227	-.274	-.259	-.259	-.206	-.203	-.227	-.282	
	.261	-.371	-.271	-.651	-.583	-.421	-.421	-.369	-.289	-.644	-.571	
	.502	-.190	-.035	-.160	-.587	-.314	-.314	-.193	-.160	-.395	-.275	
	.326	-.079	-.144	-.665	-.571	-.423	-.423	-.072	-.044	-.165	-.204	
	.343	-.258	.052	-.255	-.106	-.309	-.309	.044	.252	-.104	-.104	
Turbine cowl	.387	-.017	.103	-.174	-.792	-.621	-.621	.039	.134	-.167	-.801	
	.442	-.277	.252	-.216	-.653	-.552	-.552	.217	.106	-.248	-.547	
	.537	-.352	.323	-.275	-.470	-.377	-.377	.217	.106	-.248	-.547	
	.581	-.295	.030	-.342	-.586	-.335	-.335	.662	.118	-.335	-.573	
	.617	-.287	-.115	-.201	-.476	-.301	-.301	.173	.152	-.199	-.502	
	.653	-.289	-.045	-.330	-.070	-.272	-.272	.083	.075	-.065	-.107	
	.690	-.148	-.038	-.167	-.133	-.136	-.136	-.041	-.041	-.188	-.149	
	.738	.337	.337	-.012	.163	.211	.211	.331	.331	.160	-.018	
	.735	.151	.151	.163	.163	.145	.145	.145	.145	.160	.160	
	.768	-.074	-.074	-.318	.163	-.144	-.144	.144	.144	-.332	-.332	
	.821	.241	.241	.292	.292	-.134	-.134	.230	.230	-.286	-.286	
	.852	.383	.383	.438	.438	.396	.396	.433	.433	.433	.433	
Fan cowl	.430	.771	.910	.550	.885	.273	.273	.784	1.056	.551	.854	
	-.901	-.964	.512	-.985	-.867	-.135	-.135	-.637	.422	-.978	-.1219	
	-.568	-.755	.520	-.735	-.542	-.967	-.967	-.722	.376	-.769	-.899	
	-.031	-.350	.656	-.754	-.364	-.454	-.454	.717	.562	-.784	-.360	
	.055	-.380	.442	-.434	-.403	-.392	-.392	.428	.446	.413	-.433	
	1.066	-.270	.385	-.411	-.355	-.367	-.367	.381	.402	-.358	-.363	
	1.172	-.245	.202	-.215	-.275	-.248	-.248	.201	.190	-.219	-.269	
	.261	-.358	.305	-.034	.416	.358	.358	.356	.309	-.041	-.399	
	.302	-.286	.173	-.168	.393	.257	.257	.175	.177	.392	-.235	
	.326	-.220	.065	-.047	.186	.261	.261	.064	.051	.156	-.177	
	.343	-.280	.039	.267	.102	.287	.287	.280	.032	.268	-.096	
	.387	-.757	.083	.279	.160	.206	.206	.064	.047	.164	-.810	
Turbine cowl	.442	-.089	.137	-.135	.802	.753	.753	.082	.047	-.164	-.810	
	.493	-.293	.217	-.124	.537	.020	.020	.097	.136	.269	-.502	
	.597	-.375	.292	-.189	.236	.324	.324	.232	.131	.130	-.109	
	.581	-.356	.235	-.292	.471	.358	.358	.232	.278	.131	-.420	
	.617	-.301	.042	-.354	.553	.326	.326	.040	.083	.363	-.507	
	.653	-.293	.132	-.212	.516	.329	.329	.211	.121	.190	-.533	
	.690	-.160	.085	-.079	.102	.318	.318	.183	.115	.062	-.104	
	.708	-.106	-.021	-.185	-.178	-.159	-.159	.105	.061	-.258	-.232	
	.735	.344	.344	.016	.332	.332	.332	.332	.332	.015	.163	
	.768	.158	.158	.162	.140	.140	.140	.263	.263	.163	.163	
	.796	-.191	-.191	-.317	-.263	-.263	-.263	.226	.226	-.407	-.316	
	.821	-.251	-.251	-.382	-.283	-.283	-.283	.278	.278	-.407	-.273	
	.852	.393	.393	.443	.443	.378	.378	.433	.433	.433	.433	



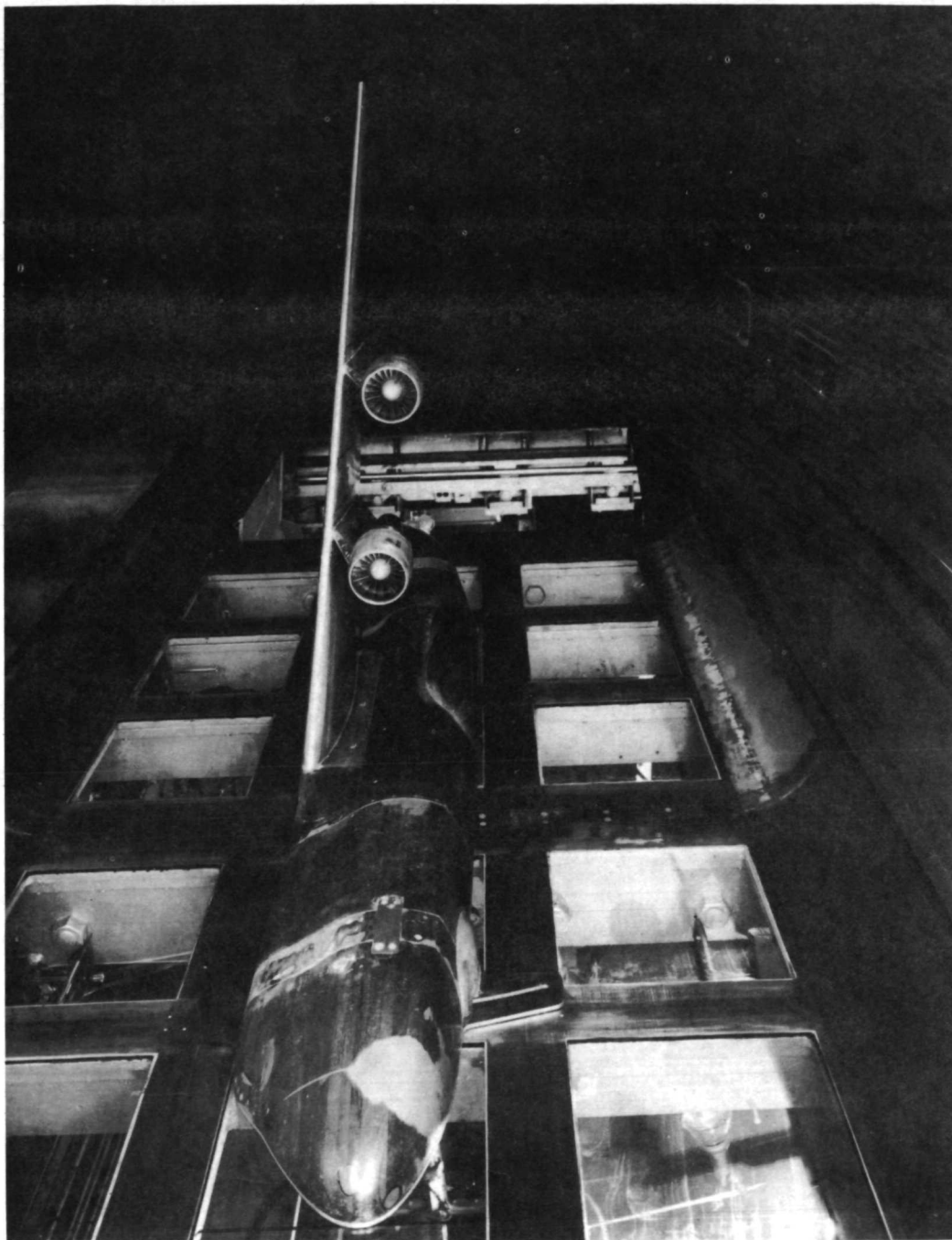
(a) Details of test model.

Figure 1.- Drawing of semispan transport model. All dimensions are in centimeters.



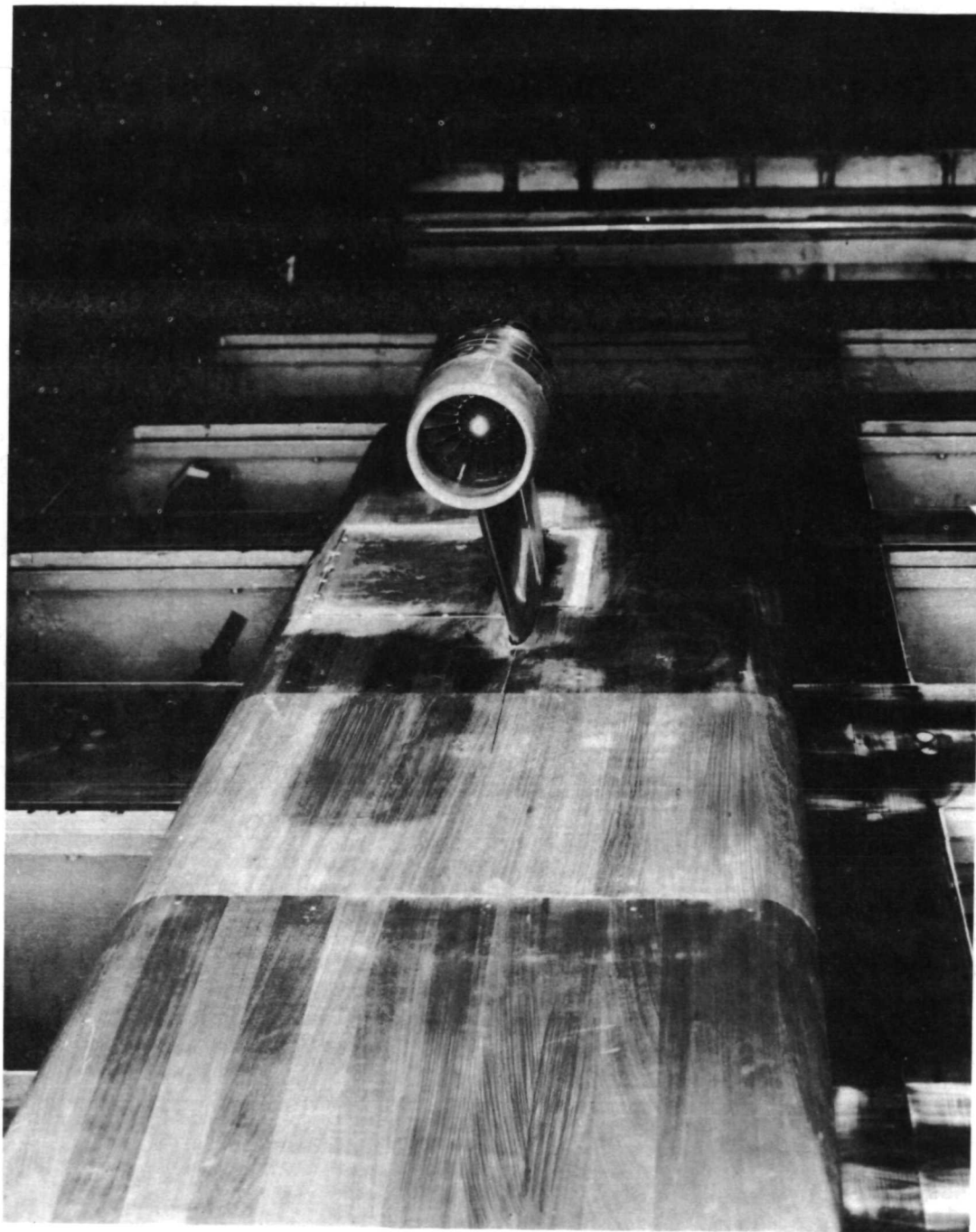
(b) Details of model engine.

Figure 1.- Concluded.

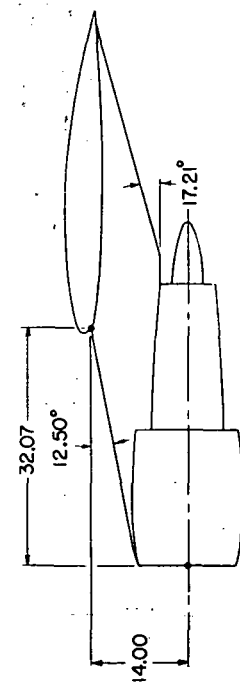


L-66-1766

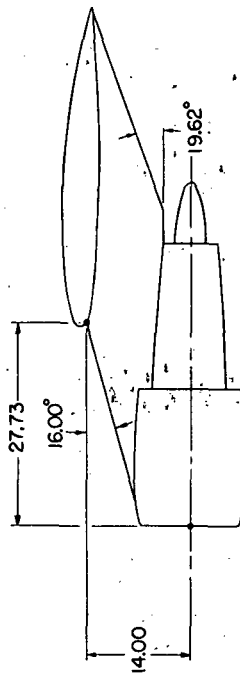
Figure 2.- Semispan model mounted on wall of Langley 8-foot transonic pressure tunnel.



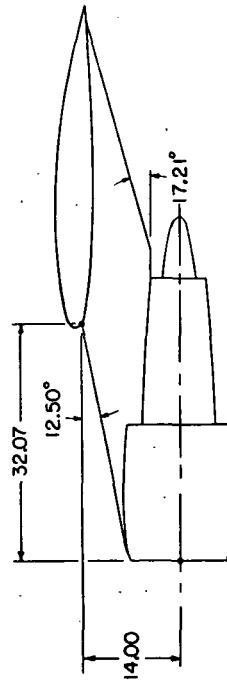
L-65-4643
Figure 3.- Model engine and elongated pylon wall-mounted in Langley 8-foot transonic pressure tunnel



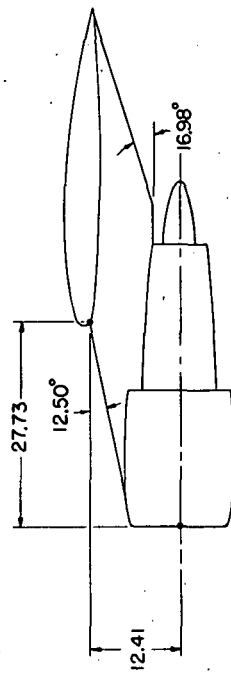
Configurations 1, 7, and 8
Basic



Configuration 3
Longitudinal variation



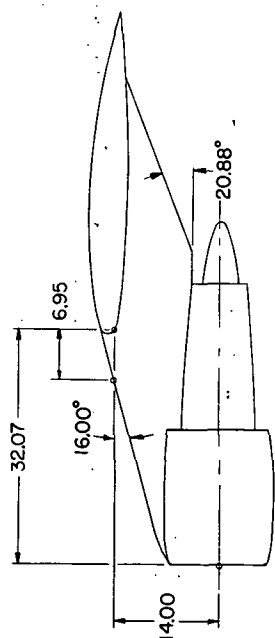
Configuration 2
Thickness variation



Configuration 4
Longitudinal and vertical variation

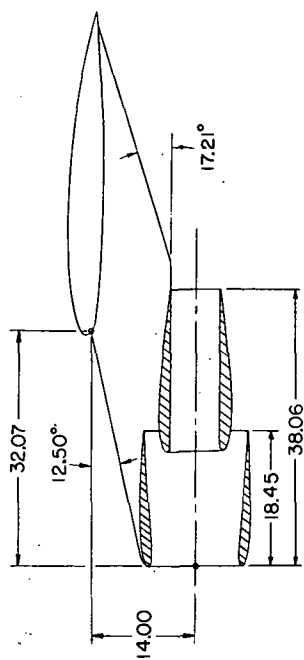
(a) Locations of powered engine.

Figure 4.- Details of engine locations. Juncture of pylon leading edge and wing is at 1.4 percent of the local chord. All dimensions are in centimeters.



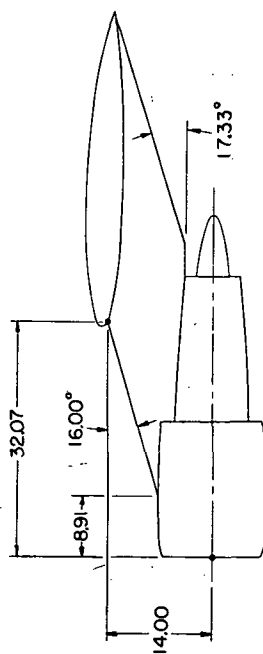
Configuration 5

Leading-edge extension



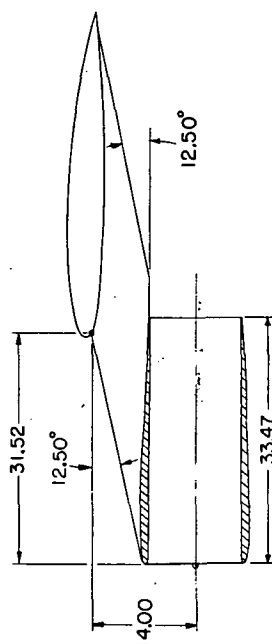
Configuration 9

Flow-through nacelle



Configuration 6

Sweep variation

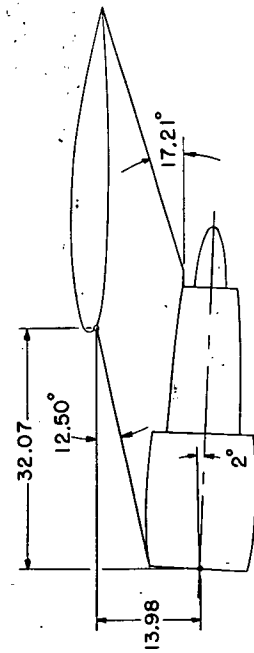


Configuration 10

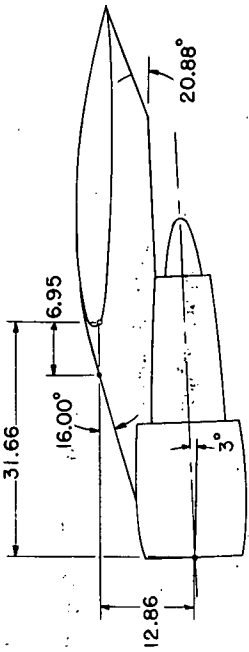
Flow-through nacelle

(b) Locations of powered engine and flow-through nacelle.

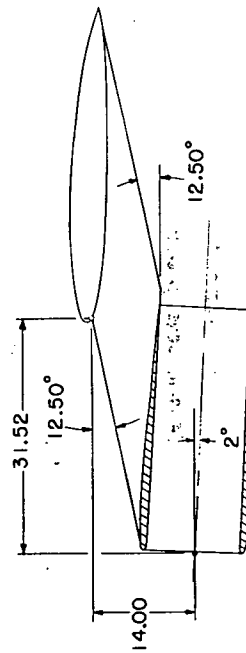
Figure 4.- Continued.



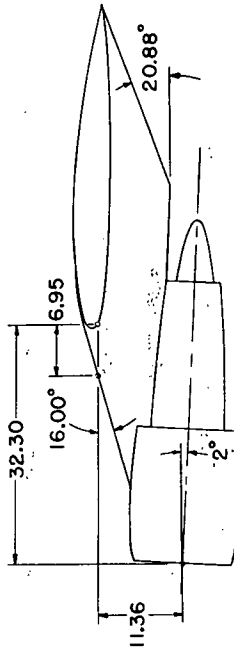
Configuration 1a
 $i = 2^\circ$



Configuration 5a
 $i = -3^\circ$

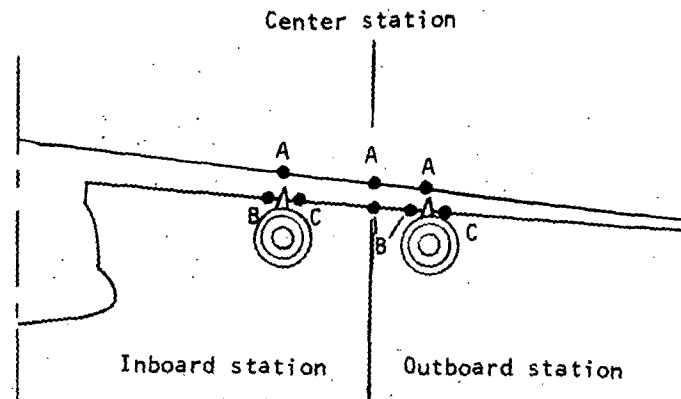


Configuration 10a
 $i = 2^\circ$



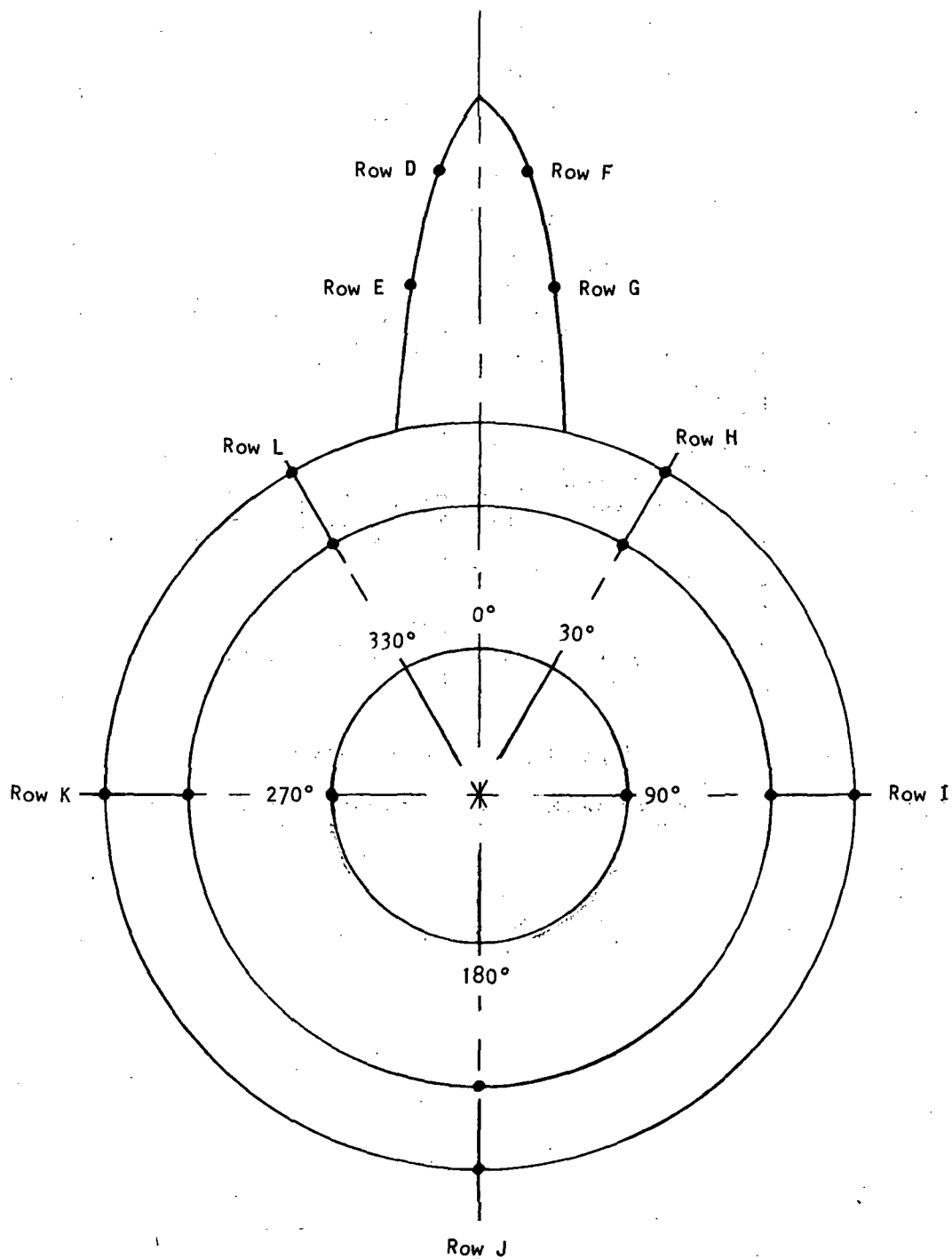
Configuration 5b
 $i = 2^\circ$

(c) Locations of engines and flow-through nacelles with incidence.
Figure 4.- Concluded.



(a) Stations and wing rows.

Figure 5.- Location of instrumentation rows.



(b) Pylon and engine rows.

Figure 5.- Concluded.



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— NATIONAL AERONAUTICS AND SPACE ACT OF 1958

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